WO-42

	flydubai	W/O	Barcode	Registration
	Dubai	7911262 Working Copy	WO7911262	A6FEW 73NG 738

	ATA	Position	Zone	Area
V Replacement Event	32 LANDING GEAR	NLG NOSE LANDING GEAR		

	Description	Serialnumber
162A1100-13	NLG COMPONENT INSTL	MAL01926S

Reference	Description

	Due at TAH	Due at TAC
05.Jan.2029		28307

Description Step 1 — Udit Bansal (FZ04850), 02.Oct.2024

REPLACE PN:162A1100-13 SN:MAL01926S (NLG COMPONENT INSTL)

1.REPLACE PN:162A1100-13 SN:MAL01926S (NLG COMPONENT INSTL)
INSTALLED AT: NLG(NOSE LANDING GEAR)
INFO/TRIGGERED BY: 162A2301-2/E4669 RS(RESTORE) RESTORE / 162A2301-2 RESTORE

RECORD THE FOLLOWING:
OFF: P/N:_____ S/N:_____

ON: P/N:_____ S/N:_____

2. REMOVED NLG TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6-FEY.

Action Step 1-1

REPLACED PN:162A1100-13 SN:MAL01926S (NLG COMPONENT INSTL)

1.REPLACED PN:162A1100-13 SN:MAL01926S (NLG COMPONENT INSTL) AS PER MPD TASK 32-090-00, REV. 41 DTD 17.10.2024

INSTALLED AT: NLG(NOSE LANDING GEAR)

INFO/TRIGGERED BY: 162A2301-2/E4669 RS(RESTORE) RESTORE / 162A2301-2 RESTORE

RECORDED THE FOLLOWING:
OFF: P/N: 162A1100-13
S/N: MAL01926S
ON: P/N:162A1100-14
S/N: GK15668Y5241

2. REMOVED NLG PROVIDED WITH SERVICEABLE TAG AND CREDITED TO STORES FOR ONWARD INSTALLATION ON A6-FEY.

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 2 — Munal Rao (FZ04849), 14.Jan.2025

NLG STEERING VALVE ASSY REPLACEMENT ALONG WITH NLG

1. CARRY OUT REPLACEMENT OF NLG STEERING VALVE ASSY P/N: 383900-1011. REFER AMM 32-51-11.

2. REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6-FEY.

RECORD THE FOLLOWING:
OFF : P/N: _____ S/N:_____

ON: P/N:_____ S/N:_____

Action Step 2-1

NLG STEERING VALVE ASSY REPLACED ALONG WITH NLG LEG ASSY.

1. DISASSEMBLY NOT CARRIED OUT.

2. N/A

RECORDED THE FOLLOWING:
OFF : P/N: 383900-1011 S/N: 0921A
ON: P/N: 383900-1011 S/N: 5860A

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 3 — Munal Rao (FZ04849), 14.Jan.2025

NLG TAXI LIGHT REPLACEMENT ALONG WITH NLG

1.CARRY OUT REPLACEMENT OF NLG TAXI LIGHT ASSY P/N: 50-0199-13. REFER AMM TASK 32-21-00-000-801.

2. REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6-FEY.

RECORD THE FOLLOWING:
OFF : P/N: _____ S/N:_____

ON: P/N:_____ S/N:_____

NOTE P/N: 50-0199-15 CANNOT BE INSTALLED AS A REPLACEMENT.

Action Step 3-1

NLG TAXI LIGHT REPLACEMENT ALONG WITH NLG

1.NLG TAXI LIGHT ASSY FOUND ALREADY INSTALLED ON RECEIVED NLG.

2. REMOVED NLG ROUTED ALONG WITH TAXI LIGHT ASSY AS SERVICEABLE.

RECORDED
~~RECORD~~ THE FOLLOWING:

OFF : P/N: 50-0199-13 S/N:4111
ON: P/N:50-0199-13 S/N:_6880

NOTE : NOTED

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 4 — Munal Rao (FZ04849), 14.Jan.2025

NLG RETRACT ACTUATOR REPLACMENT ALONG WITH NLG

1.CARRY OUT REPLACEMENT OF NLG RETRACT ACTUATOR P/N: 273A1101-2. REFER AMM 32-33-11.

2. REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6-FEY.

RECORD THE FOLLOWING:

OFF : P/N: _____ S/N:_____
ON: P/N:_____ S/N:_____

Action Step 4-1

NLG RETRACT ACTUATOR REPLACED ALONG WITH NLG

1.CARRIED OUT REPLACEMENT OF NLG RETRACT ACTUATOR P/N: 273A1101-2 AS PER AMM 32-33-11-000-801 AND 32-33-11-400-801, REV. 86 DTD 15.02.2025.

2. REMOVED UNIT PROVIDED WITH SERVICEABLE TAG AND CREDITED TO STORES FOR ONWARD INSTALLATION ON A6-FEY.

RECORDED THE FOLLOWING:
OFF : P/N: 273A1101-1 S/N:1101/0292
ON: P/N:273A1101-2 S/N : 1101/5701

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 5 — Munal Rao (FZ04849), 14.Jan.2025

LH STEERING ACTUATOR REPLACEMENT ALONG WITH NLG

1. CARRY OUT REPLACEMENT OF THE FOLLOWING NLG STEERING ACTUATORS AS PER AMM 32-51-51.

1) NLG STEERING ACTUATOR ASSY LH P/N : 275A1101-9.

Action Step 5-1

LH STEERING ACTUATOR REPLACED ALONG WITH NLG LEG ASSY.

DISASSEMBLY NOT CARRIED OUT

1) NLG STEERING ACTUATOR ASSY LH P/N : 275A1101-9.

.T TO BE PROVIDED WITH SERVICEABLE TAG AND
ORES FOR ONWARD INSTALLATION ON A6-FEY.

HE FOLLOWING:

N: _____ S/N:_____

/N:_____ S/N:_____

2. N/A.

RECORDED THE FOLLOWING:
OFF P/N: 275A1101-7 S/N: EFS5818
ON P/N: 275A1101-9 S/N: TASV11093

27/3/25

GAT 374

Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers									
Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 6 — Munal Rao (FZ04849), 14.Jan.2025

RH STEERING ACTUATOR REPLACEMENT ALONG WITH NLG

1. CARRY OUT REPLACEMENT OF THE FOLLOWING NLG STEERING ACTUATORS AS PER AMM 32-51-51.
1) RH NLG STEERING ACTUATOR ASSY RH P/N : 275A1101-10.

2. REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6-FEY.

RECORD THE FOLLOWING:

OFF : P/N: _____ S/N:_____
ON: P/N:_____ S/N:_____

Action Step 6-1

RH STEERING ACTUATOR REPLACED ALONG WITH NLG LEG ASSY

1. DISASSEMBLY NOT CARRIED OUT.
1) RH NLG STEERING ACTUATOR ASSY RH P/N: 275A1101-10

2. N/A.

27/3/25

GAT 374

RECORDED THE FOLLOWING:

OFF P/N: 275A1101-8 S/N: EFS5849
ON P/N: 275A1101-10 S/N: TASV11080

27/3/25

GAT 374

Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers									
Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 7 — Munal Rao (FZ04849), 14.Jan.2025

FIRST INSPECTION OF THE NLG STEERING ACTUATOR INST.

CARRY OUT FIRST INSPECTION FIRST INSPECTION OF THE NLG STEERING ACTUATOR INSTALLATION.

DATE:_____

TIME OF INSPECTION : _____ UTC

Action Step 7-1

N/A.

N/A.

27/3/25

GAT 374

flydubai	flydubai Dubai	W/O 7911262 Working Copy	Barcode WO7911262	Registration A6FEW 73NG 738

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 8 — Munal Rao (FZ04849), 14.Jan.2025

INDEPENDENT INSPECTION OF THE NLG STEERING ACTUATOR INST.

CARRY OUT INDEPENDENT INSPECTION OF THE NLG STEERING ACTUATOR INSTALLATION.

DATE: _____

TIME OF INSPECTION : _____ UTC

Action Step 8-1

N/A.

N/A.

27/3/25

GAT 374

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 9 — Munal Rao (FZ04849), 14.Jan.2025

FIRST INSPECTION OF THE NLG STEERING ACTUATOR TESTING/RIGGING

CARRY OUT FIRST INSPECTION OF THE NLG STEERING ACTUATOR TESTING/ RIGGING .

DATE:_____

TIME OF INSPECTION : _____ UTC

Action Step 9-1

CARRIED OUT FIRST INSPECTION OF THE NLG STEERING ACTUATOR TESTING/RIGGING AS PER AMM 32-51-51-720-801 (REV 86, 15 FEB 2025).

DATE: 27/03/25

TIME OF INSPECTION: 09:45

27/3/25

GAT 374

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

	flydubai	W/O	Barcode	Registration
	Dubai	7911262 Working Copy	WO7911262	A6FEW 73NG 738

on Step 10 — Munal Rao (FZ04849), 14.Jan.2025	Action Step 10-1
DEPENDANT INSPECTION OF THE NLG STEERING ACTUATOR TESTING/RIGGING	
CARRY OUT INDEPENDANT INSPECTION OF THE NLG STEERING ACTUATOR TESTING/ RIGGING . DATE:_____ TIME OF INSPECTION : _____ UTC	CARRIED OUT INDEPENDANT INSPECTION OF THE NLG STEERING ACTUATOR TESTING/RIGGING AS PER AMM 32-51-51-720-801 (REV 86, 15 FEB 2025). DATE: 27/03/25 TIME OF INSPECTION: 1000 UTC 27/3/25 GAT 082

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Workorder Transfer

Flight Hours	Flight Cycles	Days	Work deferred in accordance with	Limitation / Downgrades	Date
Due at TAH	**Duet at TAC**	**Due Date**			**Stamp / Sign**
	28307	**05.Jan.2029**			

Released To Service

Certifies that the work specified except as otherwise specified was carried out in accordance with CAR 145 and in respect to that work the aircraft /aircraft component is considered ready for release to service	Date 27/03/25 Stamp / Sign GAT 082

flydubai	Task Card	Rotable Controlled Taskcard	Page 1 / 2	Tally Seq No: 42	TCREQLI.99631
		Work Package: FEW/H-25		A.M.M. REV:	

Task Card no:	Task Card Title:	A/C Area:
32-090-00 / ITEM-1	**RESTORE THE NOSE LANDING GEAR ASSEMBLY.**	N/A

Part Number:	Serial Number:	Description:	Pos:
162A1100-13	MAL01926S	NLG COMPONENT INSTL	NLG

Operator Revision: 02/31.Dec.2015 Type Codes: RST
Special Codes:

MSG Codes: 8 Internal Check:

Step classificatior CMR: ☐ ETOPS: ☐ CDCCL: ☐ AD: ☐ ALI: ☐

A/C Registration: FEW
A/C Type: 73NG
Related Task Card:

Workorder No: 7911262
Work Package: FEW/H-25
Project No: BASE_CHECKS

ITEM-1 -

Zones: 115,116,713

ACCESS AND OTHER REQUIREMENTS

Panel	Description
113AW	113AW
114BW	114BW
113BW	113BW
114AW	114AW
712AR	712AR
113AC	113AC
711AL	711AL
114AC	114AC

DOCUMENT REFERENCES

Doc Typ	Reference
AMM	32-21-00-000
AMM	32-21-00-400
AMM	32-21-21-000
AMM	32-21-21-400
AMM	32-33-11-000
AMM	32-33-11-400
AMM	32-33-51-000
AMM	32-33-51-400

MATERIAL REQUIREMENTS

Partnumber	Description	Qty	SN/Batch no
No parts assigned to this Task Item	REFER ATTACHMENTS		

TOOLING REQUIREMENTS

Partnumber	Description	S/N used (tools)
No tools assigned to this Task Item	P.T.O	

PERF. TASKCARD 32-090-00, Rev. 02 Step-1 / Job Description	Mech	Insp
Restore the nose landing gear assembly. INTERVAL NOTE: Whichever comes first. Per SL-32-165 Restoration should Include: Component Installation P/N 162A1100 (CMM 32-21-12) Drag Strut Assembly P/N 162A2100 (CMM 32-21-22) Lock Mechanism P/N 162A2120 (CMM 32-21-22) Steering Actuators P/N 275A1101 (CMM 32-51-52)* Steering Metering Valve P/N S275A210 (Parker CMM 32-50-17)* All installation components should also be restored (CMM 32-21-07)	MIR1338	27/3/25 GAT 572

flydubai	Task Card	Rotable Controlled Taskcard	Page 2 / 2	Tally Seq No: 42	TCREQLI.99631
		Work Package: FEW/H-25		A.M.M. REV:	

Task Card no:	Task Card Title:	A/C Area:
32-090-00 / ITEM-1	**RESTORE THE NOSE LANDING GEAR ASSEMBLY.**	N/A

Part Number:	Serial Number:	Description:	Pos:
162A1100-13	MAL01926S	NLG COMPONENT INSTL	NLG

All system components and attachments (CMM 32-21-16)
NOTE: TASK COMPONENT RELATED

MR1338 — 27/3/25 — GAT 574

TASKCARD FINDING(Tick applicable Box): [✓]YES []NO
(Please enter Taskcard Reference also on Workorder)
FINDING WORKORDER NO . HHN24/000323/0225/123

If Job Stopped or interrupted :
Task Card Performed Up to Page : _____ Step: _____
Remarks:

If Job Stopped or interrupted :
Task Card Performed Up to Page : _____ Step: _____
Remarks:

Manhours used to perform taskcard : _____

Station	Date	Accomplished
HYD	27/3/25	GAT 574

CAT C Verification — GAT C-042

TOOLS DETAILS:

	P/N		S/N	CAL DUE DT.
1)	500-197-30	CALIPER	GAT/HTC/003	19/12/2025
2)	184-313S	FEELER GUAGE	2208225573	3/12/2025
3)	C32033-1	R/I TRUNNION PIN	N/A	—
4)	TS-2002-101-00	NLG CABLE TENSIOMETER	71650	22/7/2025
5)	(QJR3200C) QD3R250	TORQUE WRENCH	0511031135	21/5/2025
6)	QD2FR75B	TORQUE WRENCH	1118121366	7/1/2026
7)	QD1R50	TORQUE WRENCH	0420600979	7/1/2026
8)	QD2R200	TORQUE WRENCH	0219600215	3/10/2025
9)	F70200-14	INFLATION TOOL-STRUT	PG-005	17/9/25

flydubai

737-600/700/800/900

Restore Nose Landing Gear

Type:Routine Card	ATA:32--	Flow:-	Work Area:-

EVENT DETAILS

Tail #	Work Order	Base	Start	End
-	-	-	-	-

REQUIREMENT 32-090-00

PERIODICITY

Version	Threshold	Interval	Limit	Threshold Date	Limit Date
1.1	18000 FC	18000 FC			
1.2	10 YR	10 YR			

Whichever comes first

SOURCES

Type	Reference	Title	Version	Version Date	Originator	Mandatory
MPD	32-090-00				BOEING	true
MRB	D626A001-MRBR		5	Mar2003	BOEING	true

INTERVAL NOTES: Whichever comes first.

AIRPLANE NOTES:

ENGINE NOTES:

ACCESS NOTES:

SPECIAL NOTES:

ENGINEERING COMMENTS:

ANY FINDINGS

YES	NO

IRC No. : HMV24/0003?3/0225/123

GAT 374

GAT 499

20/3/25

ZONES: 000, 113, 114, 115, 116, 710, 713, 730, 740
ACCESS: 711AL, 712AR, 113AC, 113AW, 113BW, 114AC, 114AW, 114BW
SKILLS:
CERTIFICATIONS:
CONDITIONS:

PARTS

Operator Part No	Specification	Qty	Unit of Measure	Type	Description	Manufacturer
C00308	MIL-C-11796	1		Consumable	Compound - Corrosion Preventive, Petrolatum Hot Application	81205
C00528	MIL-C-11796 CLASS III	1		Consumable	Compound - Corrosion Preventive, Petroleum Hot Application (Soft Film)	81205

D00013	MIL-PRF-23827 (NATO G-354) (SUPERSEDES MIL-G-23827)	1	Consumable	Grease - Aircraft And Instrument Grease	81205
D00013	MIL-PRF-23827 (NATO G-354) (S	1	Consumable	Grease - Aircraft And Instrument Grease	81205
D00153	BMS3-11 TYPE IV	1	Consumable	Fluid - Hydraulic Fluid, Fire Resistant	81205
D00633	BMS3-33	1	Consumable	Grease - Aircraft General Purpose	81205
G00018	A-A-59503 TYPE I GRADE B, MIL-PRF-27401 TYPE I GRADE A	1	Consumable	Nitrogen - Gaseous, Pressurizing, 99.5 Percent Pure	81205
G00018	A-A-59503 TYPE I GRADE B, MIL	1	Consumable	Nitrogen - Gaseous, Pressurizing, 99.5 Percent Pure	81205
G50136	BMS3-38	1	Consumable	Compound - Corrosion Inhibiting, Non-drying	81205
G50225	NASM20995	1	Consumable	Lockwire - MS20995C20, Corrosion Resistant Steel - 0.020 Inch (0.508 mm) Diameter	81205
G50237	BMS3-38	1	Consumable	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	81205

TOOLS

Operator Part No	Mfr Part No	Qty	Description	Manufacturer
COM-4938	-	1	Insertion/Removal - contact (size 16, backshell side) (Part #: M81969/1-03, Supplier: 11139, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -ALL) (Part #: M81969/14-03, Supplier: 11139, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -ALL) (Part #: MS3447-20, Supplier: 58960, A/P Effectivity: 737-ALL)	81205
SPL-10305	-	1	Outrigger Equipment - Nose Landing Gear Installation/Removal	81205

20/3/25 GAT 499

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

			(Part #: C32049-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	
SPL-14475	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-6 NLG Trunnion Pin Puller Assy is included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14476	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-20 Small Slide Hammer with CG240-8 Side Screw are included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14477	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-2 NLG Alignment Slugs are included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14478	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-5 Drag Strut Trunnion Pin Puller Assy is included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14479	-	1	R/I Equipment - Trunnion Pin, NLG (Note: CG240-9 Slide Hammer and CG240-8 Slide Screw are included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14480	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-4 Drag Strut Alignment Slug is included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14481	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-3 Alignment Pin Assy is included in C32033-1 kit)	81205
SPL-1521	-	1	Tool - Strut Inflation, Landing Gear (Part #: F70200-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1559	-	1	Adjustable Spanner Wrench (2.00 to 4.00 Inch Dia. Retainer) JXB ALL; 737-800 (Part #: F72959-34, Supplier: 81205) (Part #: F72959-35, Supplier: 81205) (Part #: F72959-36, Supplier: 81205) (Part #: F72959-41, Supplier: 81205) (Opt Part #: F72959-5, Supplier: 81205) (Opt Part #: F72959-6, Supplier: 81205) (Opt Part #: F72959-7, Supplier: 81205)	81205

GAT 499

20/3/25

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

			(Opt Part #: F72959-8, Supplier: 81205)	
SPL-1821	-	1	Wrench - Spanner, Nose Landing Gear Trunnion Pin Retainer Nut (Part #: F72959-40, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1866	-	1	Expander Set - Spring, Main Landing Gear (Part #: C32014-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1871	-	1	Strap - Retention, NLG/MLG Inner Cylinder (Part #: C32030-10, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
STD-6545	-	1	Rope - Nylon	81205

REFERENCES

Reference Number	Type	Title	Version Number	Version Date
07-11-01-580-815	AMM:task	Lift the Airplane with the Jacks	-	-
07-11-01-580-816	AMM:task	Lower the Airplane Off the Jacks	-	-
07-11-03-201	AMM:pgblk	JACK AIRPLANE AXLES - MAINTENANCE PRACTICES	-	-
07-11-03-580-802	AMM:task	Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E	-	-
07-11-21-580-801	AMM:task	Lift the Airplane Nose with the Nose Jack at Jack Point D	-	-
07-11-21-580-802	AMM:task	Lower the Airplane Nose Off of the Jack	-	-
10-11-05-201	AMM:pgblk	CHOCK INSTALLATION	-	-
12-12-00-610-801	AMM:task	Hydraulic Reservoir Servicing	-	-
12-15-41-610-801	AMM:task	Nose Landing Gear Shock Strut Fluid Check	-	-
12-15-41-610-802	AMM:task	Nose Landing Gear Shock Strut Servicing	-	-
12-15-41-610-805	AMM:task	Nose Landing Gear Shock Strut Servicing, Airplane on Jacks	-	-
12-21-21-640-801	AMM:task	Nose Landing Gear Upper End Components Servicing	-	-
12-21-21-640-802	AMM:task	Nose Landing Gear Lower End Components Servicing	-	-
29-09-00-860-802	AMM:task	Hydraulic Reservoirs Depressurization	-	-
29-11-00-860-801	AMM:task	Hydraulic System A or B Pressurization	-	-
29-11-00-860-805	AMM:task	Hydraulic System A or B Power Removal	-	-

GAT 499 20/3/25

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Type:Routine Card	ATA:32--	Flow:-	Work Area:-	
32-00-01-080-801	AMM:task	Landing Gear Downlock Pins Removal	-	-
32-00-01-480-801	AMM:task	Landing Gear Downlock Pins Installation	-	-
32-33-00-710-801	AMM:task	Operational Test for the Nose Landing Gear	-	-
32-33-00-710-802	AMM:task	Nose Landing Gear Test - Component Replacement	-	-
32-33-52-000-801	AMM:task	Nose Gear Lock Spring Removal	-	-
32-33-52-400-801	AMM:task	Nose Gear Lock Spring Installation	-	-
32-35-00-730-801	AMM:task	Nose Gear Manual Extension System Test - Airplane on Jacks	-	-
32-45-21-000-801	AMM:task	Nose Landing Gear Wheel and Tire Assembly Removal	-	-
32-45-21-400-801	AMM:task	Nose Landing Gear Wheel and Tire Assembly Installation	-	-
32-51-00-700-801	AMM:task	Nose Wheel Steering System Test	-	-
32-51-00-820-802	AMM:task	Nose Wheel Steering System Adjustment	-	-
32-51-31-000-802	AMM:task	Removal of the Steering System Cables for the Nose Gear	-	-
32-51-31-400-802	AMM:task	Installation of the Steering System Cables for the Nose Gear	-	-
32-61-41-400-801	AMM:task	Nose Landing Gear Lock Sensor Installation	-	-
32-61-51-400-801	AMM:task	Nose Landing Gear Down Position Sensor Clearance Adjustment	-	-
33-45-01-000-801	AMM:task	Taxi Light Housing Assembly - Removal	-	-
33-45-01-400-801	AMM:task	Taxi Light Housing Assembly - Installation	-	-
53-14-01-020-801	AMM:task	Nose Wheel Well Access Panels - Removal	-	-
53-14-01-420-801	AMM:task	Nose Wheel Well Access Panels - Installation	-	-

NOTES: Restore the nose landing gear assembly.

GAT 499

20/3/25

FYR13378

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH | INSP

JXB ALL

TASK 32-21-00-000-801

1. Nose Landing Gear - Removal (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)
07-11-03 P/B 201	JACK AIRPLANE AXLES - MAINTENANCE PRACTICES
07-11-03-580-802	Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E (P/B 201)
07-11-21-580-801	Lift the Airplane Nose with the Nose Jack at Jack Point D (P/B 201)
10-11-05 P/B 201	CHOCK INSTALLATION
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-45-21-000-801	Nose Landing Gear Wheel and Tire Assembly - Removal (P/B 401)
32-51-31-000-802	Nose Gear Steering System Cables Removal (P/B 401)
33-45-01-000-801	Taxi Light Housing Assembly - Removal (P/B 201)
53-14-01-020-801	Nose Wheel Well Access Panels - Removal (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
COM-4938	Insertion/Removal Tool - Contact (size 16, backshell side) **737-800** (Part #: M81969/1-03, Supplier: 11851) (Part #: M81969/14-03, Supplier: 11139)
SPL-1559	Adjustable Spanner Wrench (2.00 to 4.00 Inch Dia. Retainer) **737-800** (Part #: F72959-34, Supplier: 81205) (Part #: F72959-35, Supplier: 81205)

GAT 499 | 2013125 | GAT 499 | 2013125 | GAT 499 | 2013125 | GAT 499 | 2013125 | GAT 499

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	(Part #: F72959-36, Supplier: 81205) (Part #: F72959-41, Supplier: 81205) (Opt Part #: F72959-5, Supplier: 81205) (Opt Part #: F72959-6, Supplier: 81205) (Opt Part #: F72959-7, Supplier: 81205) (Opt Part #: F72959-8, Supplier: 81205)		
SPL-1821	Adjustable Spanner Wrench (0.75 to 2.00 Inch Dia. Retainer, 0.12 x 0.12 Key Arm) **737-800** (Part #: F72959-40, Supplier: 81205) (Opt Part #: F72959-4, Supplier: 81205)		
SPL-1871	Strap - Retention, NLG/MLG Inner Cylinder **737-800** (Part #: C32030 -31, Supplier: 81205) (Opt Part #: C32030-10, Supplier: 81205)	AM	N 20/3/25 GAT 499
SPL-10305	Outrigger Equipment - Nose Landing Gear Installation/ Removal **737-800** (Part #: C32049-1, Supplier: 81205)		
SPL-14475	NLG Trunnion Pin Puller Assy, C32033-6 (included in C32033 Eqpt) **737-800** (Part #: C32033-1, Supplier: 81205)		
SPL-14476	Small Slide Hammer, C32033-20 (included in C32033 Eqpt) **737-800** (Part #: C32033-1, Supplier: 81205)		
STD-6545	Rope - Nylon		

C. Location Zones

Zone	Area	MECH	INSP
115	Nose Landing Gear Wheel Well - Left	AM CT0994	GAT 499 N 20/3/25
116	Nose Landing Gear Wheel Well - Right		
713	Nose Landing Gear		

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH | INSP

D. Access Panels

Number	Name/Location
113BW	Forward Nose Wheel Well Panel

E. Prepare for the Removal

SUBTASK 32-21-00-480-006

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT. THIS CAN CAUSE INJURIES TO PERSONS, AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed on the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-21-00-860-003

(2) For hydraulic system A, make sure that the hydraulic power is removed (TASK 29-11-00-860-805).

SUBTASK 32-21-00-860-025

(3) Open this circuit breaker and install safety tag:
CAPT Electrical System Panel, P18-3

Row	Col	Number	Name
E	4	(C00700)	HEATERS DRAIN MAST AIR

SUBTASK 32-21-00-860-004

(4) Open these circuit breakers and install safety tags:
CAPT Electrical System Panel, P18-3

Row	Col	Number	Name
JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053			
D	14	(C00123)	EXT LIGHTING NOSE GEAR TAXI

JXB ALL

F/O Electrical System Panel, P6-3

Row	Col	Number	Name
B	17	(C00129)	LANDING GEAR LATCH & PRESS WARN
C	15	(C01355)	LANDING GEAR AIR/GND SYS 2
C	16	(C01356)	LANDING GEAR AIR/GND SYS 1
C	18	(C01398)	LANDING GEAR TAKEOFF WARNING CUTOFF
D	1	(C01399)	PSEU PRI
D	2	(C01400)	PSEU ALTN

MECH: GAT 499 — INSP: GAT 499

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

				MECH	INSP
D	15	(C01401)	LANDING GEAR AIR/GND RELAY		GAT 499
D	16	(C01432)	LANDING GEAR ALTN EXTEND SOL	R C10768	C 20/3/25
D	17	(C01027)	LANDING GEAR NOSE GEAR STEER		
D	18	(C00451)	LANDING GEAR AURAL WARN		

SUBTASK 32-21-00-020-002

(5) Disconnect the NWS-A/B cables [49] (TASK 32-51-31-000-802).
- (a) Install the rig pin NS2 in the drum for the Captain's control wheel.
- (b) To get access to the cable turnbuckle [1], do this step:
  - 1) Open this access panel:
    (TASK 53-14-01-020-801)

Number	Name/Location
113BW	Forward Nose Wheel Well Panel

- (c) Remove the locking clips [2] for the cable turnbuckles [1] (View B, Figure 401).
- (d) Loosen the tension at the cable turnbuckles [1].
- (e) Disconnect the NWS-A/B cables [49] at the cable turnbuckles [1].
- (f) Remove the bolts [88], washers [89], and pulleys [87] from the bracket that is outboard of the left trunnion pin [63].
  - 1) Remove the NWS-A/B cables [49] from the guard [90] and grooves of the pulleys [87].
- (g) Attach the identification tags to the NWS-A/B cables [49].

MECH: R MRJ32P — INSP: C 20/3/25 GAT 499

SUBTASK 32-21-00-020-003

(6) Do this task to disconnect the flexible conduits [73] (View C, Figure 401).
- (a) Remove the screws [10] and washers [3] from the junction box cover [9].
- (b) Remove the junction box cover [9] from the junction box [5].
- (c) Disconnect the terminal block [72] from the junction box [5].
  - 1) Pull the terminal block [72] out to get access to the wires during the removal.
- (d) Identify the wires in the terminal block [72] that go to the landing gear and label them or make a drawing to be used during installation.
- (e) Use a contact insertion / removal tool, COM-4938 , tool to remove the wires from the terminal block [72] inside the junction box [5].
- (f) Attach a string to each wire removed from the terminal block [72].
  - 1) Make sure that the length of string is sufficient to go through the rigid conduit [4] to point "A".
- (g) If it is necessary, remove the lockwire from the flexible conduits [73].
- (h) Disconnect the flexible conduits [73] at point "A" and pull the wires through.
  - 1) Do not pull the string all the way through the conduit.

    NOTE: The string is necessary to pull the wires back through the rigid conduit on installation.
- (i) Disconnect the string from the wires.

MECH: R 20/03/2025 — INSP: GAT 300

SUBTASK 32-21-00-020-021

(7) Do this task to disconnect the flexible conduit [80] (View L, Figure 401).
- (a) Remove the screws [85] and washers [84] from the junction box cover [86].

MECH: R 20/03/2025 — INSP: GAT 300

MYRO780 R 20/03/2025 GAT 300

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____ GAT 300

Item: _____ Completed through item: _____ Sign: _____

	MECH	INSP

(b) Remove the junction box cover [86] from the junction box [82].
(c) Disconnect the terminal block [81] from the junction box [82].
1) Pull the terminal block [81] out to get access to the wires during the removal.
(d) Identify the wires in the terminal block [81] that go to the landing gear and label them or make a drawing to be used during installation.
(e) Use a contact insertion / removal tool, COM-4938 , tool to remove the wires from the terminal block [81] inside the junction box [82].
(f) Attach a string to each wire removed from the terminal block [81].
1) Make sure that the length of string is sufficient to go through the rigid conduit [83] to the flexible conduit [80].
(g) If it is necessary, remove the lockwire from the flexible conduit [80].
(h) Disconnect the flexible conduit [80] from the rigid conduit [83] and pull the wires through.
1) Do not pull the string all the way through the conduit.
NOTE: The string is necessary to pull the wires back through the rigid conduit on installation.
(i) Disconnect the string from the wires.

MECH: ✓ 20/03/2025 GAT 300 | ✓ GAT 300 | ✓ 20/03/2025 | Straight IRO180 20/03/2025 GAT 300

JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053

SUBTASK 32-21-00-000-001

(8) If it is necessary, remove the taxi light, do this task: Taxi Light Housing Assembly - Removal, TASK 33-45-01-000-801.

MECH: NOT REMOVED — N/A — 20/03/2025 GAT 300

JXB ALL

F. Nose Landing Gear Removal

SUBTASK 32-21-00-020-015

(1) Disconnect the hydraulic lines [25] (View E, Figure 401).
(a) Loosen the swivel nuts [77] to disconnect the hydraulic lines [25] from the swivel [24].
(b) Put caps on the hydraulic lines [25].
(c) Install plugs on each of the hydraulic ports of the swivel [24].
(d) Put tags to identify the swivel ports and hydraulic lines [25] that will prevent the crossing of the lines during the installation.

MECH: ✓ GAT 499 | INSP: ✓ 20/3/25

SUBTASK 32-21-00-020-006

(2) Disconnect the right upper rod assembly [26] (View F, Figure 401).
(a) Remove the pin [31], nut [32], and washer [30].
(b) Remove the bolt [27], washer [28], and bushing [29] from the clevis of the right trunnion [23].
(c) Use a rope to hold the upper rod assembly [26] away from the work area.

MECH: ✓ GAT 499 | INSP: ✓ 20/3/25

SUBTASK 32-21-00-020-007

(3) Disconnect the left upper rod assembly [41] (View H, Figure 401).
(a) Remove the pin [43], nut [42], and washer [44].
(b) Remove the bolt [47], washer [46], and bushing [45] from the clevis of the left trunnion [40].
(c) Use a rope to hold the upper rod assembly [41] away from the work area.

MECH: ✓ GAT 499 | INSP: ✓ 20/3/25

SUBTASK 32-21-00-020-008

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(4) Remove each pulley [50] from the bracket [51] (View I, Figure 401).
(a) Remove the nut [52] and washer [53] from the bolt [47].
(b) Remove the bolt [47] and washer [48] that hold the pulley [50] and cable guard [54] from the bracket [51].
(c) Remove the washer [48] from the bolt [47].
(d) Remove the pulley [50] and cable guard [54] from the bracket [51].
(e) Remove the NWS-A/B cables [49] from the groove of the pulley [50].

SUBTASK 32-21-00-020-009

(5) Remove the seal [65] (View J, Figure 401).
(a) Remove the lockwire from the screw [58] and lock [57].
(b) Remove the screw [58] that holds the lock [57] to the left trunnion pin [63].
(c) Remove the lock [57] from the left trunnion pin [63].
(d) Use the spanner wrench, SPL-1821 (face lug adapter), or spanner wrench set, SPL-1559, to remove the trunnion pin nut [68] from the left trunnion pin [63].
(e) Pull the seal ring [69] and seal [65] from the left trunnion pin [63].
1) If it is difficult to pull the seal [65], hold the eyebolt [67] and pull the seal [65] out of the left trunnion pin [63].
(f) Remove the nut [64], washer [56], and inboard retainer [66] from the eyebolt [67].
(g) Remove the eyebolt [67] from the left trunnion pin [63].
(h) Remove the outboard retainer [66] from the eyebolt [67].
(i) Isolate the segments of the seal [65] from the NWS-A/B cables [49].
(j) Pull the NWS-A/B cables [49] from the trunnion pin nut [68], seal ring [69], and left trunnion pin [63].
(k) Attach identification tags to the NWS-A/B cables [49].
(l) Wind the NWS-A/B cables [49] into a coil and attach it to the trunnion of the gear with a tape.

SUBTASK 32-21-00-020-010

WARNING: DO NOT REMOVE THE VALVE BODY UNTIL YOU DEFLATE THE SHOCK STRUT FULLY. THE AIR PRESSURE CAN BLOW THE VALVE BODY OUT AND CAUSE INJURIES TO PERSONNEL.

(6) Deflate the shock strut [17] (View D, Figure 401).
(a) Remove the dust cap [14] for the gas charging valve [16].
(b) Loosen the swivel nut [15] a maximum of two turns.
(c) Deflate the shock strut slowly to prevent the leakage of the fluid through the gas valve.
NOTE: Fluid in the shock strut will have bubbles when the pressure is released.
(d) Loosen the swivel nut [15] fully when all of the pressure in the shock strut [17] is released.

SUBTASK 32-21-00-480-007

(7) Install the retention strap, SPL-1871, with the marker facing up to hold the shock strut [17] in its compressed position.

SUBTASK 32-21-00-580-002

(8) Do this task: Lift the Airplane Nose with the Nose Jack at Jack Point D, TASK 07-11-21-580-801.

GAT 499

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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(a) Make sure that the chocks are installed at the main landing gear after the nose landing gear is lifted.

SUBTASK 32-21-00-580-005

(9) If it is necessary, do this task: Lift the Airplane with the Jacks, TASK 07-11-01-580-815.

SUBTASK 32-21-00-020-017

(10) Do this task: Nose Landing Gear Wheel and Tire Assembly - Removal, TASK 32-45-21-000-801.

SUBTASK 32-21-00-020-016

(11) Disconnect the lower drag strut assembly [19] (View D, Figure 401).
(a) Remove the pin [11], nut [22], washer [21], and bolt [13] from the pin [18].
(b) Remove the nut [12] and washer [20] from the pin [18].
(c) Hold the lower drag strut assembly [19] in position and remove the pin [18].
(d) Use a STD-6545 nylon rope, to hold the lower drag strut assembly [19] away from the work area.
NOTE: When the pin is removed the lower drag strut will move forward.

SUBTASK 32-21-00-580-003

(12) If it is necessary to lift the gear with an axle jack to reduce the weight of the gear when the left trunnion pin [63] and right trunnion pin [37] are removed, do this task: Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E, TASK 07-11-03-580-802.

SUBTASK 32-21-00-020-011

(13) Remove the right trunnion pin [37] (View G, Figure 401).
(a) Remove the pin [35], nut [36], washer [34], and bolt [33].
(b) Remove the swivel [24] from the right trunnion pin [37].
(c) Use the NLG Trunnion Pin puller, C32033-6, SPL-14475, and small slide hammer, C32033-20, SPL-14476, to remove the right trunnion pin [37] from the right trunnion [23].
1) Make sure that the spacer [39] that is between the sidewall of the nose wheel well and right trunnion [23] is caught.

SUBTASK 32-21-00-020-012

(14) Remove the left trunnion pin [63] (View J, Figure 401).
(a) Remove the pin [59], nut [60], washer [61], and bolt [62].
(b) Use the trunnion NLG Trunnion Pin puller, C32033-6, SPL-14475, and small slide hammer, C32033-20, SPL-14476, to remove the left trunnion pin [63] from the left trunnion [40].
1) Make sure that the spacer [71] that is between the sidewall of the nose wheel well and left trunnion [40] is caught (View K, Figure 401).
(c) Remove the cable guard [70] from the left trunnion pin [63].

SUBTASK 32-21-00-860-005

(15) Measure the thickness of the spacer [39] and spacer [71] that were between the sidewalls of the wheel well and nose landing gear trunnion.
(a) Record the measurement of the washer stack for use when installing the washer stack.
NOTE: The thickness of the washer stacks are special for each airplane.

LH WASHER - 0.077 IN
RH WASHER - 0.078 IN

20/3/25

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

SUBTASK 32-21-00-020-013

(16) Remove the nose landing gear from the airplane.
  (a) Move the nose landing gear forward in the wheel well.
  (b) Turn the gear approximately 90 degrees such that it is clear of the wheel well sidewalls and nose wheel well doors.
  (c) Lower the nose landing gear at point E until it clears the nose wheel well (PAGEBLOCK 07-11-03/201).
  (d) If it is necessary, use a dolly or the Outrigger R/I Equipment, SPL-10305, to transport the nose landing gear.
    1) When the Outrigger R/I Equipment, SPL-10305, is installed and will be left unattended, chock the nose landing gear tires and lock the caster brake (PAGEBLOCK 10-11-05/201).

Figure 401. Nose Landing Gear Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

Sheet 4 - Effectivity: JXB ALL

Sheet 5 - Effectivity: JXB ALL

Sheet 6 - Effectivity: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

Sheet 7 - Effectivity: JXB ALL

Sheet 8 - Effectivity: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

Sheet 9 - Effectivity: JXB ALL

Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

GAT 499 20/3/25 GAT 499 20/3/25 GAT 499

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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JXB ALL

TASK 32-21-21-000-801

2. **Nose Landing Gear Drag Strut Removal** (Figure 401) (Figure 402)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-03-580-802	Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E (P/B 201)
07-11-21-580-801	Lift the Airplane Nose with the Nose Jack at Jack Point D (P/B 201)
10-11-05 P/B 201	CHOCK INSTALLATION
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-080-801	Landing Gear Downlock Pins Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-33-52-000-801	Nose Gear Lock Spring Removal (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1871	Strap - Retention, NLG/MLG Inner Cylinder **737-800** (Part #: C32030 -31, Supplier: 81205) (Opt Part #: C32030-10, Supplier: 81205)
SPL-14478	Drag Strut Trun Pin Puller Assy, C32033-5 (included in C32033 Eqpt) **737-800** (Part #: C32033-1, Supplier: 81205)
SPL-14479	Slide Hammer, C32033-22, with Slide Screw, C32033-21 (included in C32033 Eqpt) **737-800** (Part #: C32033-1, Supplier: 81205)

JXB ALL

C. Location Zones

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Zone	Area
115	Nose Landing Gear Wheel Well - Left
116	Nose Landing Gear Wheel Well - Right
713	Nose Landing Gear

D. Prepare for the Removal

SUBTASK 32-21-21-480-001

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT. THIS CAN CAUSE INJURIES TO PERSONS, AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed on all the landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-21-21-580-001

(2) Do this task: Lift the Airplane Nose with the Nose Jack at Jack Point D, TASK 07-11-21-580-801.

NOTE: Lower the jack that supports the tail of the airplane while you lift the nose landing gear.

SUBTASK 32-21-21-580-002

(3) If it is necessary to lift the gear with an axle jack to reduce the weight of the gear when you remove the right trunnion pin [1] and left trunnion pin [2], do this task: Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E, TASK 07-11-03-580-802.

SUBTASK 32-21-21-860-001

(4) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-21-21-860-002

(5) Open these circuit breakers and install safety tags:
CAPT Electrical System Panel, P18-3

Row	Col	Number	Name
JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053			
D	14	(C00123)	EXT LIGHTING NOSE GEAR TAXI

JXB ALL

F/O Electrical System Panel, P6-3

Row	Col	Number	Name
B	17	(C00129)	LANDING GEAR LATCH & PRESS WARN
C	15	(C01355)	LANDING GEAR AIR/GND SYS 2
C	16	(C01356)	LANDING GEAR AIR/GND SYS 1

MECH	INSP
GAT 499	
	20/3/25
GAT 499	
	20/3/25
GAT 499	
MR1342	20/3/25
~~N/A~~	GAT 499
	20/3/25
GAT 499	
	20/3/25
GAT 499	
	20/3/25
GAT 499	
MR1342	20/3/25

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

				MECH	INSP
C	18	(C01398)	LANDING GEAR TAKEOFF WARNING CUTOFF		
D	1	(C01399)	PSEU PRI		GAT 499
D	2	(C01400)	PSEU ALTN		
D	15	(C01401)	LANDING GEAR AIR/GND RELAY	[illegible]	[illegible] 20/3/25
D	16	(C01432)	LANDING GEAR ALTN EXTEND SOL		
D	17	(C01027)	LANDING GEAR NOSE GEAR STEER		
D	18	(C00451)	LANDING GEAR AURAL WARN		

SUBTASK 32-21-21-480-002

(6) Install chocks around the tires of all the landing gear (PAGEBLOCK 10-11-05/201). — GAT 499

SUBTASK 32-21-21-020-001

(7) Do this task: Nose Gear Lock Spring Removal, TASK 32-33-52-000-801. — [illegible] [illegible] 20/3/25

SUBTASK 32-21-21-080-001

(8) Remove the downlock pin for the nose landing gear, do this task: Landing Gear Downlock Pins Removal, TASK 32-00-01-080-801. — [illegible] [illegible] 20/3/25 GAT 499

E. Nose Landing Gear Drag Strut Removal

SUBTASK 32-21-21-020-002

(1) If necessary, disconnect the hydraulic hose connection from the rod end of the retract actuator [3]. — [illegible] [illegible] 20/3/25 GAT 499

NOTE: These steps are only necessary if the retract actuator will be removed.

(a) Disconnect the hydraulic UP-hose at the bracket on the retract actuator [3]. — [illegible] [illegible] 20/3/25 GAT 499

(b) Disconnect the hydraulic DOWN-hose at the fitting on the retract actuator [3]. — [illegible] [illegible] 20/3/25 GAT 499

(c) Install caps on the hydraulic hoses and the fittings on the retract actuator [3].

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

SUBTASK 32-21-21-020-003

(2) Disconnect the rod end of the retract actuator [3]. — [illegible] MR1342 [illegible] 20/3/25 GAT 499

(a) Remove the cotter pin [50] from the bolt [16].

(b) Remove the nut [12] and washer [14] from the bolt [16].

(c) Remove the bolt [16] from the pin [9].

(d) Remove the nut [15] and washer [11] from the pin [9].

(e) Remove the pin [9] that holds the rod end of the retract actuator [3] and washers [18] from the upper drag strut [17].

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

SUBTASK 32-21-21-020-014

(3) Disconnect the rod end of the retract actuator [3]. — N/A [illegible] 20/3/25 GAT 499

(a) Remove the nuts [46], washers [45], washers [44], bolts [43], and retainer [47] from the pin [48].

(b) Remove the pin [48] that holds the rod end of the retract actuator [3] and washers [18] from the upper drag strut [17].

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

JXB ALL

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SUBTASK 32-21-21-020-004

(4) Disconnect the forward lock link [20].

NOTE: This step will also disconnect the lower drag strut [10] from the upper drag strut [17].

(a) Remove the cotter pin [13] from the bolt [22].
(b) Remove the nut [25] and washer [24] from the bolt [22].
(c) Remove the bolt [22] from the pin [19].
(d) Remove the nut [23] and washer [21] from the pin [19].
(e) Remove the pin [19] to disconnect the forward lock link [20] from the upper drag strut [17] and lower drag strut [10].

SUBTASK 32-21-21-020-005

WARNING: MAKE SURE THAT YOU FULLY EXTEND THE SHOCK STRUT FOR THE NOSE LANDING GEAR BEFORE YOU DISCONNECT THE DRAG STRUT. IF YOU DO NOT FULLY EXTEND THE SHOCK STRUT, IT CAN MOVE QUICKLY AND CAN CAUSE INJURY TO PERSONNEL AND DAMAGE TO EQUIPMENT.

(5) Disconnect the lower drag strut [10] from the shock strut [29].
(a) Remove the cotter pin [26] from the bolt [28].
(b) Remove the nut [33] and washer [32] from the bolt [28].
(c) Remove the bolt [28] from the pin [30].
(d) Remove the nut [27] and washer [31] from the pin [30].
(e) Remove the pin [30] that holds the lower drag strut [10] from the shock strut [29].

SUBTASK 32-21-21-480-003

(6) Use the retention strap, SPL-1871, to hold the shock strut [29] in the compressed position.

SUBTASK 32-21-21-020-006

(7) Disconnect the upper drag strut [17] from the bracket [51].

NOTE: This step will also remove the bracket, that contains the down position sensor target. Keep the bracket for the installation procedure.

(a) Remove the nut [34] and washer [35] from the bolt [36].
(b) Remove the bolt [36] and washer [35] to disconnect the upper drag strut [17] from the bracket [51].
(c) Remove the washer [35] from the bolt [36].

SUBTASK 32-21-21-020-007

(8) Remove the right trunnion pin [1] and left trunnion pin [2].
(a) Remove the cotter pin [38] from the bolt [40].
(b) Remove the nut [37] and washer [39] from bolt [40].
(c) Remove the bolt [40] that hold the right trunnion pin [1] and left trunnion pin [2] to the sidewall of the nose wheel well.
(d) Remove the right trunnion pin [1] and left trunnion pin [2] to disconnect the upper drag strut [17] from the sidewall of the nose wheel well.

NOTE: Use the Drag Strut Trunnion Pin puller, C32033-5, SPL-14478, and slide hammer, C32033-22, with slide screw, C32033-21,

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	20/3/25 GAT 499
MR1342	20/3/25 GAT 499

SPL-14479, to remove the trunnion pins from inboard side of the upper drag strut.

JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

(e) Remove and discard the packing [42].

JXB ALL

SUBTASK 32-21-21-020-008

(9) Remove the upper drag strut [17] from the airplane.

Figure 401. Nose Landing Gear Drag Strut Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

Sheet 3 - Effectivity: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

Sheet 4 - Effectivity: JXB ALL

Sheet 5 - Effectivity: JXB ALL

Sheet 6 - Effectivity: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment

Sheet 1 - Effectivity: JXB ALL

GAT 499 20/3/25 MR1342

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Item: _____ Completed through item: _____ Sign: _____

Type: Routine Card **ATA:**32-- **Flow:**- **Work Area:**-

MECH	INSP

JXB ALL

TASK 32-33-51-000-801

3. Nose Landing Gear Lock Mechanism Removal (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-21-580-801	Lift the Airplane Nose with the Nose Jack at Jack Point D (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1866	Expander Set - Spring, Main Landing Gear and Nose Landing Gear **737-800** (Part #: C32014-29, Supplier: 81205) (Opt Part #: C32014-1, Supplier: 81205) (Opt Part #: C32014-20, Supplier: 81205)

JXB ALL

C. Location Zones

Zone	Area
113	Area Above and Outboard of Nose Landing Gear Wheel Well - Left
114	Area Above and Outboard of Nose Landing Gear Wheel Well - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors

D. Prepare for the Removal

SUBTASK 32-33-51-480-001

WARNING: MAKE SURE THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR COULD RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

MR1342

GAT 499 — 20/3/25

PARTIAL SIGN OFF STATUS:

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(1) Make sure the downlock pins are installed in the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-33-51-580-001

WARNING: REMOVE PERSONS AND EQUIPMENT FROM THE NOSE GEAR PATH. WHEN THE NOSE GEARS RETRACT, THEY CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(2) Do this task: Lift the Airplane Nose with the Nose Jack at Jack Point D, TASK 07-11-21-580-801.

SUBTASK 32-33-51-860-001

(3) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

E. Nose Landing Gear Lock Mechanism Removal

SUBTASK 32-33-51-020-001

(1) Do these steps to disconnect the spring assembly [2] from the aft lock link assembly [3] and the wheel well fitting:
- (a) Remove the pins [8], nuts [4], and washers [5] from shaft [6] where the spring assembly [2] are attached to the aft lock link assembly [3] and the wheel well fitting (Detail B).
- (b) Use the expander set, SPL-1866 to extend the spring assembly [2].
- (c) Remove the pins [8], nuts [9] and washers [5] from the bolts [10] that hold the spring assembly [2] to the forward lock link assembly [1] (Detail D).
- (d) Remove the spring assembly [2], bolts [10], washers [11], and sleeves [12] from the forward lock link assembly [1].
- (e) Remove the spring assembly [2] from the aft lock link assembly [3].
- (f) Install the bolts [10], washers [11], spring assembly [2], and sleeves [12] to attach the spring assembly [2] to the forward lock link assembly [1].
- (g) Install the washers [5] and nuts [9], but do not tighten it.

SUBTASK 32-33-51-020-002

(2) Remove the pin [18], nut [19], washers [20, 21, 28], and pin [22] to disconnect the rod end of the lock actuator from the aft lock link assembly [3] (Detail C).

SUBTASK 32-33-51-020-003

(3) Remove the pin [29], nut [28], washer [27] and bolt [25] from the pin [23] that connects the forward lock link assembly [1] to the drag strut (Detail F).

SUBTASK 32-33-51-020-004

(4) Remove the nut [26], washer [24], and pin [23] to disconnect the forward lock link assembly [1] from the drag strut.

SUBTASK 32-33-51-020-005

(5) Do the steps that follow to remove the lock sensor [14] from the aft lock link assembly [3] (Detail E):
- (a) Remove the screws [17] from the clamps that attach the wire bundle to the aft lock link assembly [3].
- (b) Remove the nuts [13], washers [15], and screws [16] from the lock sensor [14].
- (c) Remove the lock sensor [14] from the aft lock link assembly [3].

SUBTASK 32-33-51-020-006

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INSP: GAT 499 — 20/3/25 — GAT 499 — 20/3/25 — GAT 499 — 20/3/25 — GAT 499 — 20/3/25

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(6) Remove the bushing [7] and shaft [6] to disconnect the aft lock link assembly [3] from the wheel well fitting (Detail C).

SUBTASK 32-33-51-020-007

(7) Remove the lock mechanism as a complete assembly.

Figure 401. Nose Gear Lock Mechanism Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

Sheet 4 - Effectivity: JXB ALL

GAT 499 20/3/25

MR1342

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	20/3/25	GAT 499
	20/3/25	GAT 499
MKBB	20/3/25	GAT 499

JXB ALL

TASK 32-33-11-000-801

4. Nose Gear Retract Actuator Removal (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
10-11-05 P/B 201	CHOCK INSTALLATION
29-09-00-860-802	Hydraulic Reservoirs Depressurization (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)

B. Location Zones

Zone	Area
113	Area Above and Outboard of Nose Landing Gear Wheel Well - Left
114	Area Above and Outboard of Nose Landing Gear Wheel Well - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors
730	Subzone - Left Main Landing Gear and Landing Gear Doors
740	Subzone - Right Main Landing Gear and Landing Gear Doors

C. Prepare for the Removal

SUBTASK 32-33-11-480-001

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT. THIS CAN CAUSE INJURIES TO PERSONS, AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed in the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-33-11-860-001

(2) For hydraulic system A, do this task: Hydraulic Reservoirs Depressurization, TASK 29-09-00-860-802.

SUBTASK 32-33-11-580-001

(3) Install chocks around the tires of the nose landing gear (CHOCK INSTALLATION, PAGEBLOCK 10-11-05/201).

D. Nose Gear Retract Actuator Removal

NOTE: The pin [8] or pin [16] together with pin [11], nut [10], washer [12], bolt [14], castellated nut [13] and washer [9] can replace or be replaced by pin [23] or pin [24] together with nuts [21], washers [20] and washers [19], bolts [18], and retainer [22].

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

	MECH	INSP

SUBTASK 32-33-11-020-001

(1) Disconnect the hydraulic UP-hose at the bracket on the retract actuator [1].

SUBTASK 32-33-11-020-002

(2) Disconnect the hydraulic DOWN-hose at the fitting on the retract actuator [1].

SUBTASK 32-33-11-480-002

(3) Install caps on the hydraulic hoses and the fittings on the retract actuator [1].

MECH: A INSP: N 20/3/25 GAT 499

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

SUBTASK 32-33-11-020-003

(4) Hold the retract actuator [1] in its position and disconnect the rod end of the retract actuator [1] from the drag brace (Detail B):

(a) Remove the pin [11], the nut [10], the washer [12], and the bolt [14] from the castellated nut [13] and the pin [8].

(b) Remove the castellated nut [13], the washer [9], the pin [8] and the washers [15] from the airplane.

SUBTASK 32-33-11-020-004

(5) Hold the retract actuator [1] in its position and disconnect the head end of the retract actuator [1] from the top fitting (Detail C):

(a) Remove the pin [11], the nut [10], the washer [12], and the bolt [14] from the castellated nut [13] and the pin [16].

(b) Remove the castellated nut [13], the washer [9], the pin [16] and the washers [17] from the airplane.

MECH: A INSP: N 20/3/25 GAT 499

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

SUBTASK 32-33-11-020-008

(6) Hold the retract actuator [1] in its position and disconnect the rod end of the retract actuator [1] from the drag brace (Detail B):

(a) Remove the bolts [18], the washers [19], the washers [20], the nuts [21] and the retainer [22] from the pin [23].

(b) Remove the washers [15] and the pin [23] from the airplane. — N/A

MECH: A INSP: N 20/3/25 GAT 499

SUBTASK 32-33-11-020-009

(7) Hold the retract actuator [1] in its position and disconnect the head end of the retract actuator [1] from the top fitting (Detail C):

(a) Remove the bolts [18], the washers [19], the washers [20], the nuts [21], and the retainer [22] from the pin [24].

(b) Remove the washers [17] and the pin [24] from the airplane.

JXB ALL

SUBTASK 32-33-11-020-005

(8) Remove the retract actuator [1] from the airplane.

MECH: A MRB188 INSP: N 20/3/25 GAT 499

SUBTASK 32-33-11-020-006

(9) If the replacement retract actuator [1] does not have the tube assembly installed, remove these parts from the retract actuator [1] you removed:

(a) The reducer [2].

(b) The screws [4], the washers [5], and the clamp [3].

(c) The tube assembly [6].

(d) The elbow fitting [7].

MECH: A MRB188 INSP: N 20/3/25 GAT 499

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

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SUBTASK 32-33-11-480-003

(10) Install a plug in the extend port of the retract actuator [1].

Figure 401. Nose Gear Retract Actuator Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

Sheet 3 - Effectivity: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

Sheet 4 - Effectivity: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

Sheet 5 - Effectivity: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

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	GAT 499

JXB ALL

TASK 32-33-11-400-801

5. **Nose Gear Retract Actuator Installation** (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-21-580-801	Lift the Airplane Nose with the Nose Jack at Jack Point D (P/B 201)
07-11-21-580-802	Lower the Airplane Nose Off of the Jack (P/B 201)
12-12-00-610-801	Hydraulic Reservoir Servicing (P/B 301)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-080-801	Landing Gear Downlock Pins Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)

B. Consumable Materials

Reference	Description	Specification
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827 (NATO G-354) (Supersedes MIL-G-23827)
D00153	Fluid - Hydraulic Fluid, Fire Resistant (Interchangeable And Intermixable With BMS 3-11 Type V)	BMS3-11 Type IV
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

C. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
1	Retract actuator	32-33-11-01-036	JXB ALL

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

				MECH	INSP
11	Pin	32-33-11-01-017	JXB 004, 005, 007-009		

D. Location Zones

Zone	Area
113	Area Above and Outboard of Nose Landing Gear Wheel Well - Left
114	Area Above and Outboard of Nose Landing Gear Wheel Well - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors
730	Subzone - Left Main Landing Gear and Landing Gear Doors
740	Subzone - Right Main Landing Gear and Landing Gear Doors

E. Nose Gear Retract Actuator Installation

NOTE: The pin [8] or pin [16] together with pin [11], nut [10], washer [12], bolt [14], castellated nut [13] and washer [9] can replace or be replaced by pin [23] or pin [24] together with nuts [21], washers [20] and washers [19], bolts [18], and retainer [22].

SUBTASK 32-33-11-080-001

(1) Remove the plugs from the port and the fitting on the retract actuator [1].

SUBTASK 32-33-11-020-007

(2) If the replacement retract actuator [1] does not have the tube assembly [6] installed, install these parts from the retract actuator [1] you removed:
- (a) The elbow fitting [7].
- (b) The tube assembly [6].
- (c) The screws [4], washers [5], and clamp [3].
- (d) The reducer [2].

SUBTASK 32-33-11-980-001

(3) Make sure that the retract actuator [1] is in the retracted position.

SUBTASK 32-33-11-610-001

(4) Make sure that the retract actuator [1] is full of D00153 hydraulic fluid .
- (a) Put a cap on the hydraulic fitting on the retract actuator [1] where the DOWN-hose will attach.
- (b) Put a cap on the fitting on the retract actuator [1] where the UP-hose will attach.

SUBTASK 32-33-11-600-001

(5) Prepare the retract actuator [1] for installation on the top fitting:

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

(a) Apply D00633 grease , or D00013 grease , to the bushings of the retract actuator [1] and the shaft of the pin [16].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN,

MECH: N/A — INSP: GAT 499 2/13/25

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IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: **DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.**

(b) Apply G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to the threads and thread reliefs of the castellated nut [13] and to the faces of the washer [9]. — MECH: N/A — INSP: 2/13/25 GAT 499

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

(c) Apply D00633 grease , or D00013 grease , to the bushings of the retract actuator [1] and the shaft of the pin [24]. — INSP: 2/13/25 GAT 499

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

SUBTASK 32-33-11-420-014

(6) Hold the retract actuator [1] in its position and do these steps to connect the head end of the retract actuator [1] to the top fitting (View C, Figure 401):

(a) Install the pin [16] and washers [17].

NOTE: **The head end of pin must be on the same side as the hydraulic hoses to prevent physical interference.**

JXB ALL

1) If it is necessary, trim the washers [17] around the entire circumference to prevent part interference with the angle bracket. — MECH: N/A — INSP: 2/13/25

NOTE: **This step is optional.**

INDEPENDENT INSPECTION

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

(b) Install the washer [9] and castellated nut [13] on the pin [16].

(c) Tighten the castellated nut [13] to 50 ft-lb (67.8 N·m) - 58 ft-lb (78.6 N·m). — INSP: GAT 499

(d) Loosen the castellated nut [13] to the nearest castellation and install the bolt [14], washer [12], and nut [10] on the castellated nut [13] and pin [16].

(e) Install the pin [11] on the nut [10] and bolt [14].

(f) Wipe of excess G50237 Cor-Ban 27L Compound , and G50136 corrosion inhibiting material .

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

SUBTASK 32-33-11-420-005

(7) Hold the retract actuator [1] in its position and do these steps to connect the head end of the retract actuator [1] to the top fitting (View C, Figure 401):

(a) Install the pin [24] and washers [17]. — INSP: GAT 499, 2/13/25

NOTE: **The head end of pin must be on the same side as the hydraulic hoses to prevent physical interference.**

INDEPENDENT INSPECTION 2/13/25 0420

JXB ALL — MECH: MR13382 — INSP: 2/13/25 0420 GAT 499

PARTIAL SIGN OFF STATUS:

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INDEPENDENT INSPECTION GAT 082

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1) If it is necessary, trim the washers [17] around the entire circumference to prevent part interference with the angle bracket.
NOTE: This step is optional.

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

(b) Install the retainer [22] with the flat surface touching the retract actuator [1] on the pin [24].
(c) Install the bolts [18], washers [19], washers [20], and nuts [21] on the retainer [22] and pin [24].

JXB ALL

SUBTASK 32-33-11-420-011

(8) Prepare the retract actuator [1] for installation on the drag brace:

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

(a) Apply D00633 grease , or D00013 grease , to the bushings of the retract actuator [1] and the shaft of the pin [8].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: REMOVE UNWANTED CORROSION-INHIBITING COMPOUND FROM SURFACES WHICH WILL BE LUBRICATED. IF YOU APPLY CORROSION-INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT CAN OCCUR.

(b) Apply G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to the threads and thread reliefs of the pin [8] and to the faces of the washers [15].

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

(c) Apply D00633 grease , or D00013 grease , to the bushings of the retract actuator [1] and the shaft of the pin [23].

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

SUBTASK 32-33-11-420-015

(9) Hold the retract actuator [1] in its position and do these steps to connect the rod end of the retract actuator [1] to the drag brace (View B, Figure 401):
(a) Install the pin [8] and washers [15].
(b) Install the washer [9] and castellated nut [13] on the pin [8].
(c) Tighten the castellated nut [13] to 50 ft-lb (67.8 N·m) - 58 ft-lb (78.6 N·m).
(d) Loosen the castellated nut [13] to the nearest castellation and install the bolt [14], washer [12], and nut [10] on the castellated nut [13] and pin [8].
(e) Install the pin [11] on the nut [10] and bolt [14].
(f) Wipe of excess G50237 Cor-Ban 27L Compound , and G50136 corrosion inhibiting material .

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SUBTASK 32-33-11-420-006

(10) Hold the retract actuator [1] in its position and do these steps to connect the rod end of the retract actuator [1] to the drag brace (View B, Figure 401):
(a) Install the pin [23] and washers [15].
(b) Install the retainer [22] with the flat surface touching the retract actuator [1] on the pin [23].
(c) Install the bolts [18], washers [19], washers [20], and nuts [21] on the retainer [22] and pin [23].

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SUBTASK 32-33-11-080-002

(11) Remove the caps from the hydraulic hoses.

SUBTASK 32-33-11-420-003

(12) Connect the hydraulic hoses to the retract actuator [1].
(a) Tighten the B-nut on the hydraulic UP-hose to 133 in-lb (15.0 N·m) - 147 in-lb (16.6 N·m).
(b) Tighten the B-nut on the hydraulic DOWN-hose to 256.5 in-lb (29.0 N·m) - 283.5 in-lb (32.0 N·m).

140 IN.LB 21/3/25
270 IN.LB

SUBTASK 32-33-11-600-002

(13) Apply D00633 grease , to the lubrication fittings to lubricate these items:
(a) The head end bushings of the retract actuator [1].
(b) The bearing of the attach fitting for the head end.
(c) The rod end bearing of the retract actuator [1].
(d) The bushings of the attach fitting for the rod end.

F. Nose Gear Retract Actuator Installation Test

NOTE: This test gives two methods to do a check of the nose gear retract actuator. Method one is with the airplane lifted on jacks and method two is with the airplane on the ground if jacks are not available.

SUBTASK 32-33-11-700-005

(1) Installation test with the airplane on jacks (method one).

WARNING: KEEP PERSONNEL AND EQUIPMENT AWAY FROM THE APPLICABLE LANDING GEAR. THE LANDING GEAR EXTENDS AND RETRACTS QUICKLY. THIS CAN CAUSE INJURIES TO PERSONNEL, AND DAMAGE TO EQUIPMENT.

(a) Lift the airplane for nose gear retraction, do this task: Lift the Airplane Nose with the Nose Jack at Jack Point D, TASK 07-11-21-580-801.
(b) For hydraulic system A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.
(c) Remove the downlock pin for the nose landing gear, do this task: Landing Gear Downlock Pins Removal, TASK 32-00-01-080-801.

WARNING: MAKE SURE THAT YOU INSTALL THE DOWNLOCK PINS IN THE MAIN LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE MAIN LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

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(d) Use the override trigger and move the control lever for the landing gear from the DN position to the UP position and back to DN seven times.
NOTE: It will bleed the air from the retract actuator and the lines. Examine for the correct gear operation.
(e) Examine the hydraulic connections for leaks.
(f) Extend the nose landing gear and install the downlock pin, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.
(g) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.
(h) Do this task: Lower the Airplane Nose Off of the Jack, TASK 07-11-21-580-802.
(i) Examine the hydraulic reservoirs for the correct servicing, do this task: (TASK 12-12-00-610-801).
1) Do a servicing if it is necessary.

SUBTASK 32-33-11-700-006

(2) Installation test with the airplane off jacks (method two).
(a) Make sure that you installed the hydraulic hoses correctly.
WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT. THIS CAN CAUSE INJURIES TO PERSONS, AND DAMAGE TO EQUIPMENT.
(b) For hydraulic system A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.
(c) Use the override trigger to move the control lever for the landing gear from the DN position to the UP position for three times.
NOTE: Hold the lever in each position for three seconds before you move the landing gear lever to the subsequent position.
(d) Make sure that the retract actuator [1] tries to extend when you select the UP position.
(e) Move the control lever for the landing gear to the DN position.
(f) Do a check for leakage with the hydraulic pressure applied.
(g) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.
(h) Make sure that there are no leaks in the hydraulic connections.
(i) Examine the hydraulic reservoirs for the correct servicing, do this task: (TASK 12-12-00-610-801).
1) Do a servicing if it is necessary.

N/A -

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TASK 32-33-51-400-801

6. **Nose Gear Lock Mechanism Installation** (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-61-41-400-801	Nose Landing Gear Lock Sensor Installation (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1866	Expander Set - Spring, Main Landing Gear and Nose Landing Gear **737-800** (Part #: C32014-29, Supplier: 81205) (Opt Part #: C32014-1, Supplier: 81205) (Opt Part #: C32014-20, Supplier: 81205)

JXB ALL

C. Consumable Materials

Reference	Description	Specification
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827 (NATO G-354) (Supersedes MIL-G-23827)
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38

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G50237 Compound - Corrosion Inhibiting, Non-drying - BMS3-38 Cor-Ban 27L

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
1	Link assembly	32-33-51-02-120	JXB ALL
2	Spring assembly	32-33-51-02-030	JXB ALL
3	Link assembly	32-33-51-02-155	JXB ALL
8	Pin	32-21-21-02-045	JXB ALL
18	Pin	32-33-21-01-016	JXB ALL
29	Pin	32-21-21-02-075	JXB ALL

E. Location Zones

Zone	Area
113	Area Above and Outboard of Nose Landing Gear Wheel Well - Left
114	Area Above and Outboard of Nose Landing Gear Wheel Well - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors

F. Procedure

SUBTASK 32-33-51-420-006

(1) If you replace one of the lock links, you must install or remove the shims on the completed lock mechanism assembly to adjust the overcenter dimension.
NOTE: The overcenter dimension is set on the assembly.

SUBTASK 32-33-51-420-001

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: REMOVE UNWANTED CORROSION-INHIBITING COMPOUND FROM SURFACES WHICH WILL BE LUBRICATED. IF YOU APPLY CORROSION-INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT CAN OCCUR.

(2) Do the steps that follow to connect the aft lock link assembly [3] to the fitting on the aft wheel well wall (Figure 401, View B, C):
(a) Apply a thin layer of G50237 Cor-Ban 27L Compound to the threads and thread reliefs of the shaft [6] and the nuts [4].

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NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(b) Apply a thin layer of G50237 Cor-Ban 27L Compound to the faces of the washers [5].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(c) Apply a thin layer of G50237 Cor-Ban 27L Compound to the new pins [8].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(d) Move the aft lock link assembly [3] into the fitting.

(e) Install the shaft [6] and bushing [7] to connect the aft lock link assembly [3] to the fitting.

SUBTASK 32-33-51-420-002

(3) Do the steps that follow to connect the forward lock link assembly [1] to the drag strut (Figure 401, View F):

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(a) Apply a thin layer of G50237 Cor-Ban 27L Compound to the threads and thread reliefs of the pin [23] and the nut [26], and to the faces of the washer [24].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

CAUTION: REMOVE UNWANTED CORROSION-INHIBITING COMPOUND FROM SURFACES WHICH WILL BE LUBRICATED. IF YOU APPLY CORROSION-INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT CAN OCCUR.

(b) Apply D00633 grease , to the chrome plate on the pin [23].

(c) Install the pin [23], washer [24], and nut [26].

(d) Torque the nut [26] to 95 in-lb (10.7 N·m) - 115 in-lb (13.0 N·m) above the run-on torque. 110 IN. LB 21/3/25

1) If it is necessary, loosen the nut [26] to align the nearest castellation with the hole in the pin [23].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(e) Apply a thick layer of G50237 Cor-Ban 27L Compound to the shank and threads of the bolt [25].

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NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(f) Apply a thin layer of G50237 Cor-Ban 27L Compound , to the threads of the nut [28], the faces of the washer [27], and the new pin [29].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(g) Install the bolt [25], washer [27], and nut [28].

(h) Torque the nut [28] to 18 in-lb (2.0 N·m) – 25 in-lb (2.8 N·m) above the run-on torque. 20 IN LB 21/3/25 GAT 499

1) If it is necessary, loosen the nut [28] to align the nearest castellation with the hole in the bolt [25].

(i) Install the new pin [29].

SUBTASK 32-33-51-420-003

(4) Do the steps that follow to connect the aft lock link assembly [3] of the lock mechanism to the rod end of the lock actuator (Figure 401, View C):

(a) Apply D00633 grease , or D00013 grease (Optional), to the bore of the rod end and aft lock link assembly [3], and the pin [22].

(b) Install the pin [22], washer [20], washer [21], two washers [28], and nut [19].

(c) Torque the nut [19] to 18 ft-lb (24.4 N·m) - 24 ft-lb (32.5 N·m). 20 FT·LB 21/3/25

1) If it is necessary, loosen the nut [19] to align the nearest castellation with the hole on the pin [22].

(d) Install the new pin [18].

SUBTASK 32-33-51-420-004

(5) Do these steps to connect the spring assembly [2] to the forward lock link assembly [1] and the wheel well fitting (Figure 401, View B, D):

(a) Remove the washers [5] and nuts [9] that attach the spring assembly [2] to the forward lock link assembly [1].

(b) Remove the spring assembly [2], bolts [10], washers [11], and sleeves [12] from the link assembly [1].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(c) Apply a thick layer of G50237 Cor-Ban 27L Compound to the shank and threads of the bolts [10].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(d) Apply a thin layer of G50237 Cor-Ban 27L Compound , to the threads of the nuts [9], the faces of the washers [5] and sleeves [12], and the new pins [8].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(e) Apply D00633 grease , to the chrome plated surfaces of the sleeves [12] and the bolts [10].

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(f) Apply a thin layer of G50237 Cor-Ban 27L Compound to the threads and thread reliefs of the nuts [4], the faces of the washers [5], and the pins [8].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(g) Apply D00633 grease , to the chrome plated surfaces of the shaft [6].

(h) Use the expander set, SPL-1866 to extend the spring assembly [2].

(i) Connect the spring assembly [2] to the forward lock link assembly [1] (Figure 401, View D):

1) Put the spring assembly [2] in its location at the forward lock link assembly [1].
2) Install the bolts [10], sleeves [12], and the washers [11] to connect the spring assembly [2] to the forward lock link assembly [1].
3) Install the washers [5] and the nuts [9] on the bolts [10].
4) Torque the nuts [9] to 20 in-lb (2.3 N·m) – 24 in-lb (2.7 N·m) above the run-on torque. 22 IN-LB 21/3/25

CAUTION: POINT THE PIN DOWN DURING INSTALLATION TO KEEP RISK OF CHAFING DAMAGE TO CONDUIT TO A MINIMUM.

a) Install the new pins [8].

(j) Connect the spring assembly [2] to the shaft [6] in the wheel well fitting (Figure 401, view B):

1) Install the shaft [6] in the wheel well fitting.
2) Put the spring assembly [2] in its location on the shaft [6].
3) Install the washers [5] and nuts [4] on the shaft [6].
4) Torque the nuts [4] to 20 in-lb (2.3 N·m) – 24 in-lb (2.7 N·m) above the run-on torque. 22 IN-LB 21/3/25

a) Install the new pins [8].

(k) Remove the expander set, SPL-1866 from spring assembly [2].

SUBTASK 32-33-51-420-005

(6) Do the steps that follow to install the lock sensor [14] on the aft lock link assembly [3] (Figure 401, View E):

(a) Put the lock sensors [14] on the aft lock link assembly [3].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(b) Apply a thin layer of G50237 Cor-Ban 27L Compound to the threads and thread reliefs of the screws [16], and the faces of the washers [15].

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NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(c) Install the screws [16], washers [15], and nuts [13] on the lock sensor [14].

(d) Install the screws [17] to attach clamps for the wire bundles to the aft lock link assembly [3].

SUBTASK 32-33-51-700-001

(7) Do a post installation check of the nose gear lock sensors [14] (TASK 32-61-41-400-801).

G. Nose Gear Lock Mechanism Installation Test

SUBTASK 32-33-51-480-003

WARNING: MAKE SURE YOU INSTALL THE DOWNLOCK PINS IN THE MAIN LANDING GEAR TO PREVENT THE ACCIDENTAL OPERATION OF THE GEAR. IF THE GEAR RETRACTS, IT CAN CAUSE INJURY TO PERSONS OR DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed in the main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-33-51-860-002

(2) For hydraulic system A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-33-51-580-002

WARNING: REMOVE PERSONNEL AND EQUIPMENT FROM THE NOSE GEAR PATH WHEN THE NOSE GEAR RETRACTS. IF YOU DO NOT OBEY, DAMAGE TO EQUIPMENT AND INJURY TO PERSONNEL CAN OCCUR.

(3) Extend and retract the nose landing gear several times.

NOTE: Examine the nose gear and lock mechanism for the correct operation.

H. Put the Airplane Back to its Usual Condition

SUBTASK 32-33-51-480-004

(1) Extend the nose landing gear and install the ground lock assembly, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-33-51-860-003

(2) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-33-51-580-003

(3) Do this task: Lower the Airplane Off the Jacks, TASK 07-11-01-580-816.

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TASK 32-21-21-400-801

7. **Nose Landing Gear Drag Strut Installation** (Figure 401) (Figure 402)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-21-580-802	Lower the Airplane Nose Off of the Jack (P/B 201)
12-21-21-640-801	Nose Landing Gear Upper End Components Servicing (P/B 301)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
32-33-00-710-801	Operational Test for the Nose Landing Gear (P/B 501)
32-33-00-710-802	Nose Landing Gear Test - Component Replacement (P/B 501)
32-33-52-400-801	Nose Gear Lock Spring Installation (P/B 401)
32-35-00-730-801	Nose Gear Manual Extension System Test - Airplane on Jacks (P/B 501)
32-51-00-700-801	Nose Wheel Steering System Test (P/B 501)
32-61-51-400-801	Nose Landing Gear Down Position Sensor Clearance Adjustment (P/B 501)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-14480	Drag Strut Align Slug, C32033-4 (included in C32033 Eqpt) **737-800** (Part #: C32033-1, Supplier: 81205)
SPL-14481	R/I Equipment - Trunnion Pin, NLG (Alignment Pin Assy is included in kit) **737-800** (Part #: C32033-1, Supplier: 81205)

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C. Consumable Materials

PARTIAL SIGN OFF STATUS:

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Reference	Description	Specification
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
13	Pin	32-21-21-02-075	JXB ALL
26	Pin	32-21-00-02-005	JXB ALL
38	Pin	32-21-21-02-005	JXB ALL
42	Packing	32-21-21-02-022	JXB ALL
50	Pin	32-33-11-01-017	JXB 004, 005, 007-009

E. Location Zones

Zone	Area
115	Nose Landing Gear Wheel Well - Left
116	Nose Landing Gear Wheel Well - Right
713	Nose Landing Gear

F. Prepare for the Installation

SUBTASK 32-21-21-620-001

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: MAKE SURE THAT YOU REMOVE UNWANTED CORROSION INHIBITING COMPOUND FROM SURFACES THAT YOU WILL LUBRICATE. IF YOU APPLY CORROSION INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT COULD OCCUR.

(1) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:

(a) The shank and the threads of the bolts [40].

NOTE: Apply a thick layer of the compound to the bolts [40].

(b) The threads and thread reliefs of the nuts [37].

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(c) The faces of the washers [39].
(d) The cotter pin [38].

SUBTASK 32-21-21-160-001
(2) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

SUBTASK 32-21-21-640-001
(3) Lubricate the chrome plated surfaces of the right trunnion pin [1] and left trunnion pin [2] with D00633 grease .

SUBTASK 32-21-21-160-002
(4) Remove all unwanted D00633 grease .

G. Nose Landing Gear Drag Strut Installation

SUBTASK 32-21-21-560-001
(1) Put the upper drag strut [17] in its position for installation on the sidewall of the nose wheel well.

SUBTASK 32-21-21-580-003
(2) Align the upper drag strut [17] with the mating hole in the sidewall of the nose wheel well.

SUBTASK 32-21-21-420-001
(3) Install the right trunnion pin [1] and left trunnion pin [2].
(a) Make sure that the trunnion pin seal [41] is installed in the trunnion pin.

JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

(b) Apply D00633 grease , and install new packing [42] inside the nose wheel well drag brace bushing.

JXB ALL

(c) Use the Drag Strut alignment slug, C32033-4, SPL-14480, to align the upper drag strut [17] and sidewall of the nose wheel well.
(d) Put the right trunnion pin [1] and left trunnion pin [2] through the upper drag strut [17] and sidewall of the nose wheel well.
1) Make sure that the hole in the right trunnion pin [1] and left trunnion pin [2] for the bolts [40] are aligned.
a) Use nose landing gear trunnion pin alignment pin assembly, SPL-14481.
(e) Put the bolt [40] through the right trunnion pin [1] and left trunnion pin [2].
(f) Install the washer [39] and nut [37] on the bolt [40].
(g) Torque the nut [37] to 20 in-lb (2.3 N·m) - 24 in-lb (2.7 N·m) more than the run-on torque. 22IN·LB 21/3/25
(h) If it is necessary, loosen the nut [37] to the nearest castellation to align the holes for the cotter pin [38].
(i) Install the cotter pin [38] in the bolt [40].
(j) Rotate seal [41] as required to fully seat seal against crossbolt.

CAUTION: WEAR SAFETY GLASSES DURING APPLICATION, AND IF APPLICATION OF SOLVENT RESULTS IN IRRITATION, USE AN AIR PURIFYING ORGANIC VAPOR CARTRIDGE RESPIRATOR. USE NITRILE GLOVES AND WASH YOUR HANDS WITH MOISTURE CREAM AFTER APPLICATION. STORE DPM 6380-4

PARTIAL SIGN OFF STATUS:

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CLEANER/SOLVENT IN FLAMMABLE RESISTANT CONTAINERS IN QUANTITIES NO LARGER THAN 1 GALLON.

(k) Reapply G50237 Cor-Ban 27L Compound , as required to coat the interior surface of the trunnion pin.

SUBTASK 32-21-21-420-002

(4) Connect the upper drag strut [17] to the bracket [51].
- (a) Put the washer [35] on the bolt [36].
- (b) Align the hole in the upper drag strut [17] to the hole in the bracket [51].
- (c) Put the bolt [36] through the bracket [51] and upper drag strut [17].
- (d) Install the washer [35] and nut [34] on the bolt [36].

SUBTASK 32-21-21-620-002

WARNING: **USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.**

CAUTION: **MAKE SURE THAT YOU REMOVE UNWANTED CORROSION INHIBITING COMPOUND FROM SURFACES THAT YOU WILL LUBRICATE. IF YOU APPLY CORROSION INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT COULD OCCUR.**

(5) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:
- (a) The threads and thread reliefs of the pin [30] and bolt [28].
- (b) The threads of the nut [27] and nut [33].
- (c) The faces of the washers [31] and washer [32].
- (d) To the cotter pin [26].

SUBTASK 32-21-21-160-003

(6) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

SUBTASK 32-21-21-640-002

(7) Lubricate the shank of the pin [30] and bolt [28] with D00633 grease .

SUBTASK 32-21-21-160-004

(8) Remove all unwanted D00633 grease .

SUBTASK 32-21-21-420-003

(9) Connect the lower drag strut [10].
- (a) Make sure that the drag brace lubrication fittings are on the right side.
- (b) Put the lower drag strut [10] in its position on the shock strut [29].
- (c) Put the pin [30] through the shock strut [29] and lower drag strut [10].
- (d) Install the washer [31] and nut [27] on the pin [30].
- (e) Torque the nut [27] to 600 in-lb (67.8 N·m) - 700 in-lb (79.1 N·m).
- (f) Loosen the nut [27] to the nearest castellation to align the holes for the bolt [28].
- (g) Install the bolt [28] in the pin [30].
- (h) Install the washer [32] on the bolt [28].
- (i) Install the nut [33] on the bolt [28].

INDEPENDENT INSPECTION 21/3/25 0740 GAT 082 CR-2

GAT 499 21/3/25 GAT 499 21/3/25 GAT 499 21/3/25 GAT 499 21/3/25 GAT 499 21/3/25 GAT 499 21/3/25 GAT 499 650 IN-LB 21/3/25 CR-2 CR-2 21/3/25 0730 GAT 499 MRB38 21/3/25 0730

INDEPENDENT INSPECTION GAT 082 21/3/25 0740

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Item: _____ Completed through item: _____ Sign: _____

(j) If it is necessary, loosen the nut [33] to the nearest castellation to align the holes for the cotter pin [26].
(k) Install the cotter pin [26] in the bolt [28].

SUBTASK 32-21-21-620-003

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(10) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:
(a) The threads and thread reliefs of the pin [19] and bolt [22].
(b) The threads of the nut [23] and nut [25].
(c) The faces of the washers [21] and washer [24].
(d) The cotter pin [13].

SUBTASK 32-21-21-160-005

(11) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

SUBTASK 32-21-21-420-009

(12) If nut and bolt is installed in the downlock pin hole, remove it and install a downlock pin in that location.

SUBTASK 32-21-21-420-004

(13) Connect the forward lock link [20] and lower drag strut [10].

NOTE: This step will also connect the lower drag strut [10] to the upper drag strut [17].

(a) Put the lower drag strut [10] and forward lock link [20] in their positions on the upper drag strut [17].
(b) Put the pin [19] through the forward lock link [20], upper drag strut [17], and lower drag strut [10].
(c) Install the washer [21] and nut [23] on the pin [19].
(d) Torque the nut [23] to 95 in-lb (10.7 N·m) - 115 in-lb (13.0 N·m) more than the run-on torque. 100 IN.LB 21/3/25
(e) Loosen the nut [23] to the nearest castellation to align the hole for the bolt [22].
(f) Install the bolt [22] in the pin [19].
(g) Install the washer [24] and nut [25] on the bolt [22].
(h) If it is necessary, loosen the nut [25] to the nearest castellation to align the holes for the cotter pin [13].
(i) Install the cotter pin [13] in the bolt [22].

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

SUBTASK 32-21-21-420-011

MECH	INSP
	21/3/25 0845 GAT 499
	21/3/25 GAT 499
	21/3/25 GAT 499
	21/3/25 GAT 499
	21/3/25 GAT 499
	21/3/25 GAT 499
	21/3/25 0755 GAT 499
N/A	21/3/25 0855 GAT 499
	21/3/25 GAT 499

INDEPENDENT INSPECTION GAT 082 21/3/25 0745
INDEPENDENT INSPECTION GAT 082 21/3/25 0805
INDEPENDENT INSPECTION GAT 082 21/3/25 0805
INDEPENDENT INSPECTION

PARTIAL SIGN OFF STATUS:

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(14) If the retract actuator was removed, hold the retract actuator [3] and do these steps to connect the actuator to the upper drag strut [17].
(a) Put the rod end of the retract actuator [3] and washers [18] in their position on the upper drag strut [17].
(b) Put the pin [9] through the upper drag strut [17], rod end of the retract actuator [3], and washers [18].
(c) Install the washer [11] and castellated nut [15] on the pin [9].
(d) Torque the nut [15] to 50 ft-lb (68 N·m) - 58 ft-lb (79 N·m).
(e) If it is necessary, loosen the nut [15] to the nearest castellation to align the hole for the bolt [16].
(f) Install the bolt [16] in the pin [9].
(g) Install the washer [14] and nut [12].
(h) If it is necessary, loosen the nut [12] to the nearest castellation to align the hole for the cotter pin [50].
(i) Install the cotter pin [50] in the bolt [16].

INDEPENDENT INSPECTION

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

SUBTASK 32-21-21-420-008

(15) If the retract actuator was removed, hold the retract actuator [3] and do these steps to connect the actuator to the upper drag strut [17].
(a) Put the rod end of the retract actuator [3] and washers [18] in their position on the upper drag strut [17].
(b) Put the pin [48] through the upper drag strut [17], rod end of the retract actuator [3], and washers [18].
(c) Install the retainer [47] with the flat surface touching the upper drag strut [17] on the pin [48].
(d) Install the bolts [43], washers [44], washers [45], and nuts [46] on the retainer [47].

INDEPENDENT INSPECTION GAT 082 21/3/25 0825

JXB ALL

SUBTASK 32-21-21-080-002

(16) If they were installed, remove the caps from the hydraulic hoses.

SUBTASK 32-21-21-420-006

(17) If the retract actuator was removed, connect the hydraulic hoses to the retract actuator [3].

SUBTASK 32-21-21-720-001

(18) Do a post installation check of the nose landing gear down position sensor (TASK 32-61-51-400-801).

NOTE: This step is necessary because the bracket [51], that contains the down position sensor target, was removed.

H. Put the Airplane Back to Its Usual Condition

SUBTASK 32-21-21-420-007

(1) Do this task: Nose Gear Lock Spring Installation, TASK 32-33-52-400-801.

SUBTASK 32-21-21-610-001

(2) Lubricate the upper components of the nose landing gear (TASK 12-21-21-640-801).

SUBTASK 32-21-21-860-005

(3) Remove the safety tags and close these circuit breakers:

MECH	INSP
S-97	GAT 499
N/A	21/3/25
	GAT 499
	21/3/25 GAT 499
	21/3/25 0820 GAT 499
	GAT 300
S-97 24/03/2025	
S-97 24/03/2025	GAT 300
	MRO38 21/3/25 0840 GAT 499

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CAPT Electrical System Panel, P18-3

Row	Col	Number	Name
JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053			
D	14	(C00123)	EXT LIGHTING NOSE GEAR TAXI

JXB ALL

F/O Electrical System Panel, P6-3

Row	Col	Number	Name
B	17	(C00129)	LANDING GEAR LATCH & PRESS WARN
C	15	(C01355)	LANDING GEAR AIR/GND SYS 2
C	16	(C01356)	LANDING GEAR AIR/GND SYS 1
C	18	(C01398)	LANDING GEAR TAKEOFF WARNING CUTOFF
D	1	(C01399)	PSEU PRI
D	2	(C01400)	PSEU ALTN
D	15	(C01401)	LANDING GEAR AIR/GND RELAY
D	16	(C01432)	LANDING GEAR ALTN EXTEND SOL
D	17	(C01027)	LANDING GEAR NOSE GEAR STEER
D	18	(C00451)	LANDING GEAR AURAL WARN

SUBTASK 32-21-21-860-003

(4) For hydraulic system A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-21-21-710-001

(5) Do this task: Operational Test for the Nose Landing Gear, TASK 32-33-00-710-801.

NOTE: It is optional to perform the Nose Landing Gear Test - Component Replacement, TASK 32-33-00-710-802 instead of the operational test if a hydraulic cart is not available.

SUBTASK 32-21-21-710-002

(6) Do this task: Nose Gear Manual Extension System Test - Airplane on Jacks, TASK 32-35-00-730-801.

SUBTASK 32-21-21-710-003

(7) Do this task: Nose Wheel Steering System Test, TASK 32-51-00-700-801.

SUBTASK 32-21-21-860-004

(8) Do this task: Lower the Airplane Nose Off of the Jack, TASK 07-11-21-580-802.

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27/3/25	
MR1338	27/3/25

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TASK 32-21-00-400-801

8. Nose Landing Gear - Installation (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title	MECH	INSP
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)		
07-11-03-580-802	Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E (P/B 201)		
07-11-21-580-802	Lower the Airplane Nose Off of the Jack (P/B 201)		
10-11-05 P/B 201	CHOCK INSTALLATION		
12-15-41-610-801	Nose Landing Gear Shock Strut Fluid Check (P/B 301)		
12-15-41-610-802	Nose Landing Gear Shock Strut Servicing, Airplane on the Ground (P/B 301)		
12-15-41-610-805	Nose Landing Gear Shock Strut Servicing, Airplane on Jacks (P/B 301)		
12-21-21-640-801	Nose Landing Gear Upper End Components Servicing (P/B 301)		
12-21-21-640-802	Nose Landing Gear Lower End Components Servicing (P/B 301)		
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)		
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)		
32-33-00-710-801	Operational Test for the Nose Landing Gear (P/B 501)		
32-33-00-710-802	Nose Landing Gear Test - Component Replacement (P/B 501)		
32-35-00-730-801	Nose Gear Manual Extension System Test - Airplane on Jacks (P/B 501)		
32-45-21-400-801	Nose Landing Gear Wheel and Tire Assembly - Installation (P/B 401)		
32-51-00-700-801	Nose Wheel Steering System Test (P/B 501)		
32-51-00-820-802	Nose Wheel Steering System Adjustment (P/B 501)		
32-51-31-400-802	Nose Gear Steering System Cables Installation (P/B 401)		
33-45-01-400-801	Taxi Light Housing Assembly - Installation (P/B 201)		
53-14-01-420-801	Nose Wheel Well Access Panels - Installation (P/B 401)		

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B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
COM-4938	Insertion/Removal Tool - Contact (size 16, backshell side) **737-800** (Part #: M81969/1-03, Supplier: 11851) (Part #: M81969/14-03, Supplier: 11139)
SPL-1521	Tool - Strut Inflation, Landing Gear **737-800** (Part #: F70200-35, Supplier: 81205) (Opt Part #: F70200-1, Supplier: 81205) (Opt Part #: F70200-14, Supplier: 81205) (Opt Part #: F70200-17, Supplier: 81205) (Opt Part #: F70200-18, Supplier: 81205)
SPL-1559	Adjustable Spanner Wrench (2.00 to 4.00 Inch Dia. Retainer) **737-800** (Part #: F72959-34, Supplier: 81205) (Part #: F72959-35, Supplier: 81205) (Part #: F72959-36, Supplier: 81205) (Part #: F72959-41, Supplier: 81205) (Opt Part #: F72959-5, Supplier: 81205) (Opt Part #: F72959-6, Supplier: 81205) (Opt Part #: F72959-7, Supplier: 81205) (Opt Part #: F72959-8, Supplier: 81205)
SPL-1821	Adjustable Spanner Wrench (0.75 to 2.00 Inch Dia. Retainer, 0.12 x 0.12 Key Arm) **737-800** (Part #: F72959-40, Supplier: 81205) (Opt Part #: F72959-4, Supplier: 81205)
SPL-1871	Strap - Retention, NLG/MLG Inner Cylinder **737-800**

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	(Part #: C32030 -31, Supplier: 81205) (Opt Part #: C32030-10, Supplier: 81205)		
SPL-10305	Outrigger Equipment - Nose Landing Gear Installation/ Removal **737-800** (Part #: C32049-1, Supplier: 81205)		
SPL-14477	Alignment Pin Assy, C32033-2 (included in C32033 Eqpt) **737-800** (Part #: C32033-1, Supplier: 81205)	2/13/25 GAT 499	
SPL-14481	R/I Equipment - Trunnion Pin, NLG (Alignment Pin Assy is included in kit) **737-800** (Part #: C32033-1, Supplier: 81205)		

JXB ALL

C. Consumable Materials

Reference	Description	Specification	MECH	INSP
C00308	Compound - Corrosion Preventive, Petrolatum Hot Application	MIL-C-11796		
C00528	Compound - Corrosion Preventive, Petroleum Hot Application (Soft Film)	MIL-C-11796 Class III		
D00633	Grease - Aircraft General Purpose	BMS3-33		
G00018	Nitrogen - Gaseous, Pressurizing, 99.5 Percent Pure	A-A-59503 Type I Grade B, MIL-PRF-27401 Type I Grade A	MR1358	2/13/25 GAT 499
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38		
G50225	Lockwire - MS20995C20, Corrosion Resistant Steel - 0.020 Inch (0.508 mm) Diameter	NASM20995		
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38		

D. Expendables/Parts

PARTIAL SIGN OFF STATUS:

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AMM Item	Description	AIPC Reference	AIPC Effectivity
11	Pin	32-21-00-02-005	JXB ALL
19	Strut assembly	32-21-21-02-155	JXB ALL
31	Pin	32-22-11-01A-005	JXB ALL
35	Pin	32-21-00-02-040	JXB ALL
43	Pin	32-22-11-01A-005	JXB ALL
59	Pin	32-21-00-02-040	JXB ALL
74	Packing	32-21-00-02-062	JXB ALL

E. Location Zones

Zone	Area
115	Nose Landing Gear Wheel Well - Left
116	Nose Landing Gear Wheel Well - Right
713	Nose Landing Gear

F. Access Panels

Number	Name/Location
113BW	Forward Nose Wheel Well Panel

G. Prepare for the Installation

SUBTASK 32-21-00-480-011

(1) If it is necessary, use a dolly or the Outrigger R/I Equipment, SPL-10305, to move the nose landing gear.

(a) When the Outrigger R/I Equipment, SPL-10305, is installed and will be left unattended, chock the nose landing gear tires and lock the caster brake (PAGEBLOCK 10-11-05/201).

SUBTASK 32-21-00-480-008

(2) Install the retention strap, SPL-1871, with the marker facing up to hold the shock strut [17] in its compressed position.

SUBTASK 32-21-00-620-001

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO

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	GAT 499 2/13/25

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Type:Routine Card | ATA:32-- | Flow:- | Work Area:-

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THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS. (3) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items: (a) The threads and thread reliefs of the bolt [62] and bolt [33] (b) The threads of the nut [60] and nut [36] (c) The faces of the washer [61] and washer [34]. SUBTASK 32-21-00-160-001 (4) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate).		GAT 499 21/3/25
SUBTASK 32-21-00-640-002 (5) Lubricate the following items with D00633 grease : (a) The shank of the bolt [62] and bolt [33] (b) The shank of left trunnion pin [63] and right trunnion pin [37] (c) The spacer [39] and spacer [71] (d) The cable guard [70]. SUBTASK 32-21-00-160-002 (6) Remove all unwanted D00633 grease .		GAT 499 21/3/25
H. Nose Landing Gear Installation SUBTASK 32-21-00-560-002 (1) You may optionally measure the net freeplay between the wheel well trunnion bushing faces and the strut trunnion bushing faces, before installation, as follows: **NOTE: This measurement will assist in reduction of extra gear removals because the net freeplay requirement of 0.008 in. (0.20 mm) – 0.015 in. (0.38 mm) is not met after installation.** (a) Measure the distance between the wheel well trunnion bushing faces and make a record of it. [Value 1] ____________. (b) Measure the distance over the strut trunnion bushing faces and make a record of it. [Value 2] ___________. (c) Measure the thickness of the spacer [39] and spacer [71] removed as value 3. [Value 3] ________. (d) Subtract the three measurements. Value 1 - Value 2 - Value 3 = Value 4 ________. (e) Make sure that the value 4 equals 0.008 in. (0.20 mm) – 0.015 in. (0.38 mm) total axial freeplay. (f) If it is necessary, use a different spacer [39] and spacer [71] thickness to obtain the required freeplay dimension. **NOTE: It is recommended to make the spacer [39] and spacer [71] of equal thickness.**	N/A	GAT 499 2/3/25
SUBTASK 32-21-00-560-001 (2) Put the nose landing gear in its position for installation on the sidewall of the nose wheel well. SUBTASK 32-21-00-580-004 (3) Lift the nose landing gear with an axle jack until the right trunnion [23] and left trunnion [40] align with the mating hole in the sidewall of the nose wheel well (TASK 07-11-03-580-802).	AR1538	GAT 499 21/3/25

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SUBTASK 32-21-00-420-001

(4) Install the right trunnion pin [37] (View G, Figure 401).

(a) Put the spacer [39] in its position between the right trunnion [23] and the sidewall of the nose wheel well.

JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

(b) Apply D00633 grease , and install new packing [74] inside nose wheel well trunnion bushing.

INDEPENDENT INSPECTION 21/3/25 0520 JXB ALL GAT 082

(c) Use the alignment pin assy, C32033-2, SPL-14477, to put the right trunnion pin [37] through the right trunnion [23], spacer [39], and sidewall of the nose wheel well.

(d) Use the nose landing gear trunnion pin alignment pin assembly, SPL-14481, to make sure that the hole in the right trunnion pin [37] for the bolt [33] aligns with the mating hole in the right trunnion [23].

(e) Put the swivel [24] through the right trunnion pin [37].

1) Make sure that the hole in the swivel [24] for the bolt [33] aligns with the mating hole in the right trunnion pin [37] and right trunnion [23].

(f) Put the bolt [33] through the right trunnion [23], right trunnion pin [37], and swivel [24].

(g) Install the washer [34] and nut [36] on the bolt [33].

(h) Tighten the nut [36] to 20 in-lb (2.3 N·m) - 24 in-lb (2.7 N·m) plus the run-on torque. 22 IN.LB 21/3/25 GAT 499

(i) Install the cotter pin [35] in the bolt [33].

1) If it is necessary, loosen the nut [36] to the nearest castellation to align the holes for the cotter pin [35].

SUBTASK 32-21-00-420-002

(5) Install the left trunnion pin [63] (View K, Figure 401).

(a) Put the spacer [71] in its position between the left trunnion [40] and sidewall of the nose wheel well.

JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

(b) Apply D00633 grease , and install new packing [74] inside nose wheel well trunnion bushing.

INDEPENDENT INSPECTION 21/3/25 0520 JXB ALL GAT 082

(c) Use the alignment pin assy, C32033-2, SPL-14477, to put the left trunnion pin [63] through the left trunnion [40], spacer [71], and left sidewall of the nose wheel well.

(d) Use the nose landing gear trunnion pin alignment pin assembly, SPL-14481, to make sure that the hole in the left trunnion pin [63] for the bolt [62] aligns with the mating hole in the left trunnion [40].

(e) Put the cable guard [70] into the left trunnion pin [63].

1) Make sure that the hole in the cable guard [70] for the bolt [62] aligns with the mating hole in the left trunnion pin [63] and left trunnion [40].

PARTIAL SIGN OFF STATUS:

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(f) Put the bolt [62] through the left trunnion [40], left trunnion pin [63], and cable guard [70].
(g) Install the washer [61] and nut [60] on the bolt [62].
(h) Tighten the nut [60] to 20 in-lb (2.3 N·m) - 24 in-lb (2.7 N·m) plus the run-on torque. GAT 499 22 IN. LB 21/3/25
(i) Install the cotter pin [59] in the bolt [62].
1) If it is necessary, loosen the nut [60] to the nearest castellation to align the holes for the cotter pin [59].

SUBTASK 32-21-00-220-002

(6) Measure the total axial freeplay in the trunnion.

NOTE: If the total axial freeplay is not correct, this will not cause any damage to the landing gear structure, but there may be a loud noise during landing gear extension/retraction.

(a) Use a thickness gage to measure the clearance between the spacer [39], spacer [71], and bushing faces of the sidewall or the trunnion on the two sides of the nose landing gear. GAT 499
(b) Make sure that the combined clearance of the two trunnions is 0.008 in. (0.203 mm) - 0.015 in. (0.381 mm). 0.300mm 21/3/25
(c) If it is necessary to get the correct clearance, adjust the spacer [39] and spacer [71] stack thickness.

SUBTASK 32-21-00-080-003

(7) Remove the axle jack after the correct clearance has been achieved.

SUBTASK 32-21-00-420-003

(8) Install each pulley [50] on the bracket [51] (View I, Figure 401).
(a) Put the NWS-A/B cables [49] in its position on the groove of the pulley [50].
(b) Put the cable guard [54] and pulley [50] in their position on the bracket [51].
(c) Install the washer [48] on the bolt [47].
(d) Put the bolt [47] through the pulley [50], cable guard [54], and bracket [51].
(e) Install the washer [53] and nut [52] on the bolt [47].

SUBTASK 32-21-00-420-004

CAUTION: MAKE SURE THAT THE STEERING CABLES DO NOT CROSS. CROSSED CABLES CAN INCREASE CABLE FRICTION WHICH CAN CAUSE DAMAGE TO THE CABLES.

(9) Connect the NWS-A/B cables [49] (TASK 32-51-31-400-802).
(a) Make sure that the rig pin NS2 is in the drum for the Captain's control wheel.
(b) Put the NWS-A/B cables [49] through the inboard side of the trunnion pin nut [68], seal ring [69], and left trunnion pin [63].
1) Make sure that the one NWS-A/B cable [49] goes over the trunnion pin bolt and one NWS-A/B cable [49] goes under the trunnion pin bolt where the cables go through the trunnion.
(c) For the pulleys [87] on the bracket that is outboard of the left trunnion pin [63], do these steps:
1) Put the NWS-A/B cables [49] in the grooves of the left trunnion pin [63] and through the guard [90].
2) Install the pulleys [87], bolts [88], and washers [89].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR

INSPECTION 22/3/25 0530 GAT 082 CR-2

INSPECTION 22/3/25 0530 CR-2 GAT 082

PARTIAL SIGN OFF STATUS:

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SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

INDEPENDENT INSPECTION 22/3/25 GAT 082 0545

CR-2

a) Install the bolts [88] with G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (optional).

b) Tighten the bolts [88] to 73 in-lb (8.2 N·m) - 77 in-lb (8.7 N·m). 75 IN-LB 22/3/25 GAT 499

(d) Apply a thin layer of C00308 corrosion preventive compound , or C00528 compound , to the threads on the internal and external surfaces of the mating adjustable parts on the cable turnbuckles [1].

(e) Connect the NWS-A/B cables [49] to the cable turnbuckles [1] (View B, Figure 401).

(f) Remove the identification tags attached to the NWS-A/B cables [49].

(g) Install the locking clips [2] on the cable turnbuckles [1].

SUBTASK 32-21-00-410-001

(10) Install the seal [65] (View J, Figure 401).

(a) Assemble the seal [65] around the cables [49] at the inboard end of the left trunnion pin [63].

(b) Install one retainer [66] on the eyebolt [67].

(c) Put the eyebolt [67] through the seal [65].

(d) Install the other retainer [66], washer [56], and nut [64] on the eyebolt [67].

(e) Tighten the nut [64] sufficiently to clamp the seal [65].

(f) Loosen the nut [64] to approximately 1/4 turn until you can turn the eyebolt [67] with your fingers.

(g) Put the seal [65] into the left trunnion pin [63] until it touches the stop.

(h) Install the seal ring [69] into the left trunnion pin [63].

SUBTASK 32-21-00-620-002

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(11) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to the trunnion pin nut [68].

SUBTASK 32-21-00-420-005

CR-2

(12) Install the trunnion pin nut [68] (View J, Figure 401).

(a) Install the trunnion pin nut [68] on the inboard side of the left trunnion pin [63].

INDEPENDENT INSPECTION 22/3/25 0605 GAT 082

GAT 499 22/3/25 0600

PARTIAL SIGN OFF STATUS:

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INDEPENDENT INSPECTION

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(b) Use the spanner wrench, SPL-1821, or spanner wrench set, SPL-1559, to tighten the trunnion pin nut [68] to 100 in-lb (11.3 N·m) - 150 in-lb (16.9 N·m).
(c) If it is necessary, loosen the trunnion pin nut [68] to align with the vernier of the lock holes.
(d) Install the lock [57] on the inboard side of the left trunnion pin [63].
(e) Install the screw [58] that holds the lock [57] to the left trunnion pin [63].
(f) Install the lockwire on the screw [58].

SUBTASK 32-21-00-710-001

(13) Make sure that the seal [65] will turn freely from the extended cable position of the gear to a simulated gear retracted cable position.

NOTE: This will verify that the cables will not become twisted when the gear is retracted.

SUBTASK 32-21-00-620-003

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(14) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:
(a) The threads and thread reliefs of the bolt [27] and the bolt [47]
(b) The threads of the nut [32] and the nut [42]
(c) The faces of the washer [28], washer [30], washer [44], and washer [46].

SUBTASK 32-21-00-160-005

(15) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate).

SUBTASK 32-21-00-640-003

(16) Lubricate the chrome plated surfaces of the bolt [27], bolt [47], bushing [29], and bushing [45] with D00633 grease .

SUBTASK 32-21-00-160-006

(17) Remove all unwanted D00633 grease .

SUBTASK 32-21-00-420-006

(18) Connect the left upper rod assembly [41] (View H, Figure 401).
(a) Put the upper rod assembly [41] in its position on the clevis of the left trunnion [40].
(b) Put the bushing [45] through the clevis of the left trunnion [40] and rod end of the upper rod assembly [41].
(c) Install the washer [46] on the bolt [47].

PARTIAL SIGN OFF STATUS:

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(d) Put the bolt [47] through the clevis of the left trunnion [40] and rod end of the upper rod assembly [41].
(e) Install the washer [44] and nut [42] on the bolt [47].
(f) Tighten the nut [42] to 90 in-lb (10.2 N·m) - 125 in-lb (14.1 N·m). 110 in-lb
(g) Install the pin [43] in the bolt [47].
1) If it is necessary, loosen the nut [42] to the nearest castellation to align the holes for the pin [43].

INDEPENDENT INSPECTION GAT 082 27/3/25 0700

SUBTASK 32-21-00-420-007

(19) Connect the right upper rod assembly [26] (View F, Figure 401).
(a) Put the upper rod assembly [26] in its position on the clevis of the right trunnion [23].
(b) Put the bushing [29] through the clevis of the right trunnion [23] and rod end of the upper rod assembly [26].
(c) Install the washer [28] on the bolt [27].
(d) Put the bolt [27] through the clevis of the right trunnion [23] and rod end of the upper rod assembly [26].
(e) Install the washer [30] and nut [32] on the bolt [27].
(f) Tighten the nut [32] to 90 in-lb (10.2 N·m) - 125 in-lb (14.1 N·m). 110 in-lb
(g) Install the pin [31] in the nut [32].
1) If it is necessary, loosen the nut [32] to the nearest castellation to align the holes for the pin [31].

INDEPENDENT INSPECTION GAT 082 22/3/25 0645

SUBTASK 32-21-00-420-008

(20) Connect the hydraulic lines [25] (View E, Figure 401).
(a) Remove the plugs from the hydraulic ports of the swivel [24].
(b) Remove the caps from the hydraulic lines [25].
(c) Connect the hydraulic lines [25] to the swivel [24].
(d) Remove the tags from the swivel [24] ports and hydraulic lines [25].

SUBTASK 32-21-00-620-004

WARNING: **USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.**

CAUTION: **DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.**

(21) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:
(a) The threads and thread reliefs of the pin [18]
(b) The threads of the nut [12]
(c) The faces of the washer [20] and washer [21].

SUBTASK 32-21-00-160-003

PARTIAL SIGN OFF STATUS:

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(22) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate).

SUBTASK 32-21-00-640-004

(23) Lubricate the shank of the pin [18] and bolt [13] with D00633 grease .

SUBTASK 32-21-00-160-004

(24) Remove all unwanted D00633 grease .

SUBTASK 32-21-00-420-009

(25) Connect the lower drag strut assembly [19] (View D, Figure 401).
- (a) Make sure that the drag brace lubrication fittings are on the right side.
- (b) Put the lower drag strut assembly [19] in its position on the clevis of the shock strut [17].
- (c) Put the pin [18] through the clevis of the shock strut [17] and the lower drag strut assembly [19].
- (d) Install the washer [20] and nut [12] on the pin [18].
- (e) Tighten the nut [12] to 600 in-lb (67.8 N·m) - 700 in-lb (79.1 N·m).
  1) If it is necessary, loosen the nut [12] to the nearest castellation to align the holes for the bolt [13].
- (f) Install the bolt [13] in the pin [18].
- (g) Install the washer [21] on the bolt [13].
- (h) Install the nut [22] on the bolt [13].
- (i) Install the pin [11] in the bolt [13].
  1) If it is necessary, loosen the nut [22] to the nearest castellation to align the holes for pin [11].

INDEPENDENT INSPECTION 22/3/25 0705 GAT 082

SUBTASK 32-21-00-420-012

(26) Do this task: Nose Landing Gear Wheel and Tire Assembly - Installation, TASK 32-45-21-400-801.

SUBTASK 32-21-00-420-011

(27) Do this task to connect the wires to the terminal block [72] (View C, Figure 401).
- (a) For airplanes that have three rigid conduits on the junction box [5], open the tee fitting on the aft side of the junction box [5].
- (b) Pull the wires through the conduit.
  1) Attach the strings that hang out of the end of the rigid conduits [4] to the wires that hang out of the flexible conduits [73].
  2) Pull the wires through the rigid conduits [4] to the junction box [5].
  3) Connect the flexible conduits [73] after the clamps [8] at point "A".
  4) If it is necessary, install G50225 MS20995C20 lockwire , on the flexible conduits [73].
     NOTE: The jam nuts with safety wire holes drilled through the corners must not be lockwired.
  5) Remove the strings attached to the wires.
- (c) Connect the wires to the terminal block [72] with a contact insertion / removal tool, COM-4938 , tool using the drawing you made during the removal.
- (d) Connect the terminal block [72] to the junction box [5].
- (e) Install the screws [10] and washers [3] to close the junction box cover [9] on the junction box [5].

SUBTASK 32-21-00-420-017

PARTIAL SIGN OFF STATUS:

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(28) Do this task to connect the wires to the terminal block [81] (View L, Figure 401).
(a) Pull the wires through the conduit.
1) Attach the strings that hang out of the end of the rigid conduit [83] to the wires that hang out of the flexible conduit [80].
2) Pull the wires through the rigid conduit [83] to the junction box [82].
3) Connect the flexible conduit [80] to the rigid conduit [83].
4) If it is necessary, install G50225 MS20995C20 lockwire , on the flexible conduit [80].
NOTE: The jam nuts with safety wire holes drilled through the corners must not be lockwired.
5) Remove the strings attached to the wires.
(b) Connect the wires to the terminal block [81] with a contact insertion / removal tool, COM-4938 , tool using the drawing you made during the removal.
(c) Connect the terminal block [81] to the junction box [82].
(d) Install the screws [85] and washers [84] to the close the junction box cover [86] on the junction box [82].

I. Put the Airplane Back to Its Usual Condition

SUBTASK 32-21-00-410-002

(1) Close this access panel:
(TASK 53-14-01-420-801)

Number	Name/Location
113BW	Forward Nose Wheel Well Panel

SUBTASK 32-21-00-080-004

(2) Remove the retention strap, SPL-1871, from the nose landing gear.

SUBTASK 32-21-00-610-003

(3) Lubricate the nose landing gear (TASK 12-21-21-640-801, TASK 12-21-21-640-802).

SUBTASK 32-21-00-610-001

(4) To fully service the shock strut with the airplane on jacks, hold the shock strut in a vertical position and do these steps to service the shock strut with fluid:
(a) Install on the shock strut a retention strap, SPL-1871.
(b) Do this task: Nose Landing Gear Shock Strut Servicing, Airplane on Jacks, TASK 12-15-41-610-805.
(c) Remove the retention strap, SPL-1871, from the nose landing gear.

SUBTASK 32-21-00-610-002

(5) To inflate the shock strut with the airplane on jacks, do these steps:
NOTE: To do these steps, you must have accomplished the steps to add fluid to the strut at the start of this task.
(a) Install the tool, SPL-1521, on the gas valve.
(b) Inflate the shock strut with G00018 nitrogen , until you reach the fully extended pressure that is on the servicing placard (TASK 12-15-41-610-805).
(c) Close the gas valve.
(d) Remove the tool, SPL-1521, from the gas valve.
(e) Install the cap on the gas valve.

JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053

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Item: _____ Completed through item: _____ Sign: _____

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SUBTASK 32-21-00-420-014

(6) If it is necessary, install the taxi light, do this task: Taxi Light Housing Assembly - Installation, TASK 33-45-01-400-801.

PRE-INSTALLED N/A R 23/03/2025 GAT 300

JXB ALL

SUBTASK 32-21-00-860-006

(7) For hydraulic system A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-21-00-790-001

(8) Check the swivel [24] for leakage.

27/3/25 GAT 374

SUBTASK 32-21-00-860-007

(9) Remove the safety tags and close these circuit breakers:
CAPT Electrical System Panel, P18-3

Row	Col	Number	Name
JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053			
D	14	(C00123)	EXT LIGHTING NOSE GEAR TAXI

JXB ALL

F/O Electrical System Panel, P6-3

Row	Col	Number	Name
B	17	(C00129)	LANDING GEAR LATCH & PRESS WARN
C	15	(C01355)	LANDING GEAR AIR/GND SYS 2
C	16	(C01356)	LANDING GEAR AIR/GND SYS 1
C	18	(C01398)	LANDING GEAR TAKEOFF WARNING CUTOFF
D	1	(C01399)	PSEU PRI
D	2	(C01400)	PSEU ALTN
D	15	(C01401)	LANDING GEAR AIR/GND RELAY
D	16	(C01432)	LANDING GEAR ALTN EXTEND SOL
D	17	(C01027)	LANDING GEAR NOSE GEAR STEER
D	18	(C00451)	LANDING GEAR AURAL WARN

27/3/25 GAT 374

SUBTASK 32-21-00-860-026

(10) Remove the safety tag and close this circuit breaker:
CAPT Electrical System Panel, P18-3

Row	Col	Number	Name
E	4	(C00700)	HEATERS DRAIN MAST AIR

PARTIAL SIGN OFF STATUS:

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SUBTASK 32-21-00-710-003

(11) Do this task: Operational Test for the Nose Landing Gear, TASK 32-33-00-710-801.

NOTE: It is optional to perform the Nose Landing Gear Test - Component Replacement, TASK 32-33-00-710-802 instead of the operational test if a hydraulic cart is not available.

INDEPENDENT INSPECTION 27/3/25 0930 GAT 082

SUBTASK 32-21-00-710-004

(12) Do this task: Nose Gear Manual Extension System Test - Airplane on Jacks, TASK 32-35-00-730-801.

CR-2

SUBTASK 32-21-00-820-001

(13) Adjust the NWS-A/B cables [49] (TASK 32-51-00-820-802).

SUBTASK 32-21-00-710-005

(14) Do this task: Nose Wheel Steering System Test, TASK 32-51-00-700-801.

CR-2 27/3/25 0945

SUBTASK 32-21-00-860-008

(15) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

INDEPENDENT INSPECTION 28/3/25 1000 GAT 082

SUBTASK 32-21-00-580-006

(16) If airplane was lifted, do this task: Lower the Airplane Off the Jacks, TASK 07-11-01-580-816.

SUBTASK 32-21-00-860-009

(17) Do this task: Lower the Airplane Nose Off of the Jack, TASK 07-11-21-580-802.

SUBTASK 32-21-00-200-001

(18) Check the pressure and extension of the shock strut.

(a) Measure the pressure and extension of the shock strut.

1) Refer to the servicing placard and make sure that the pressure you measure is still correct for the extension of the shock strut (TASK 12-15-41-610-801).

2) If the pressure is not correct for the extension of the shock strut, fully service the shock strut (TASK 12-15-41-610-802).

NOTE: If the pressure is not correct for the extension of the shock strut, the shock strut does not have the proper amount of fluid.

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Item: _____ Completed through item: _____ Sign: _____

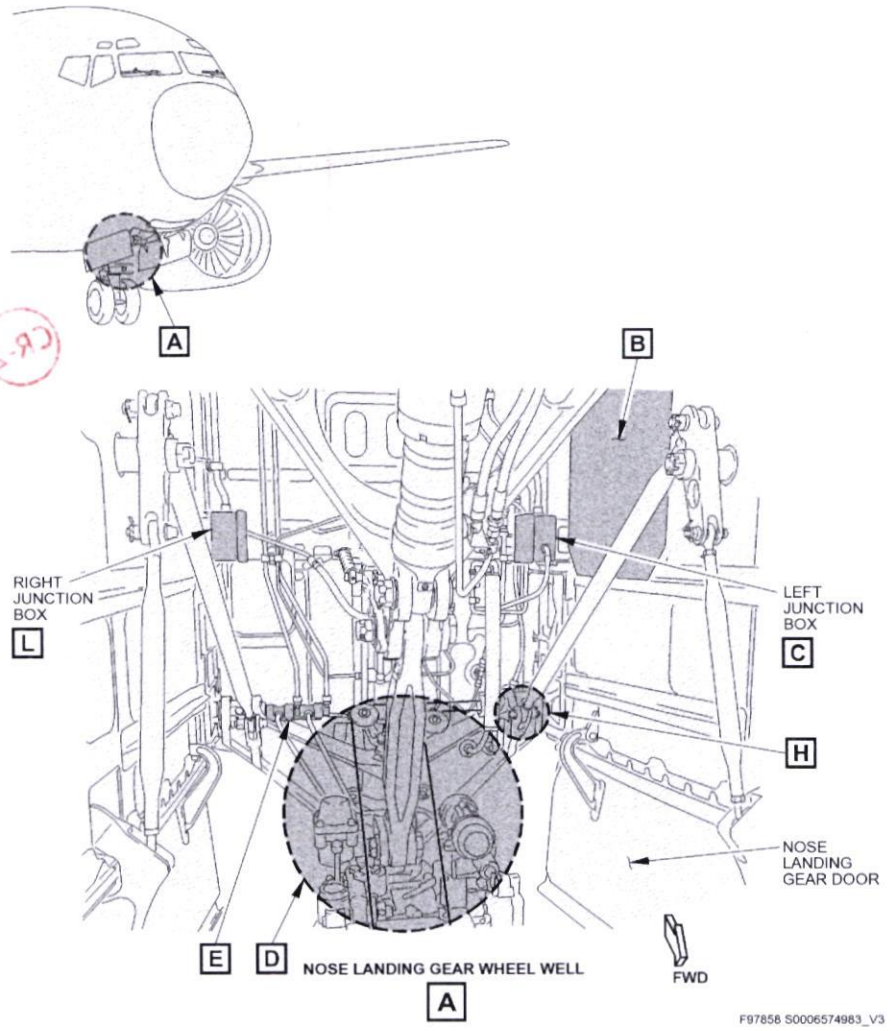

Figure 401. Nose Landing Gear Installation - Sheet 1
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

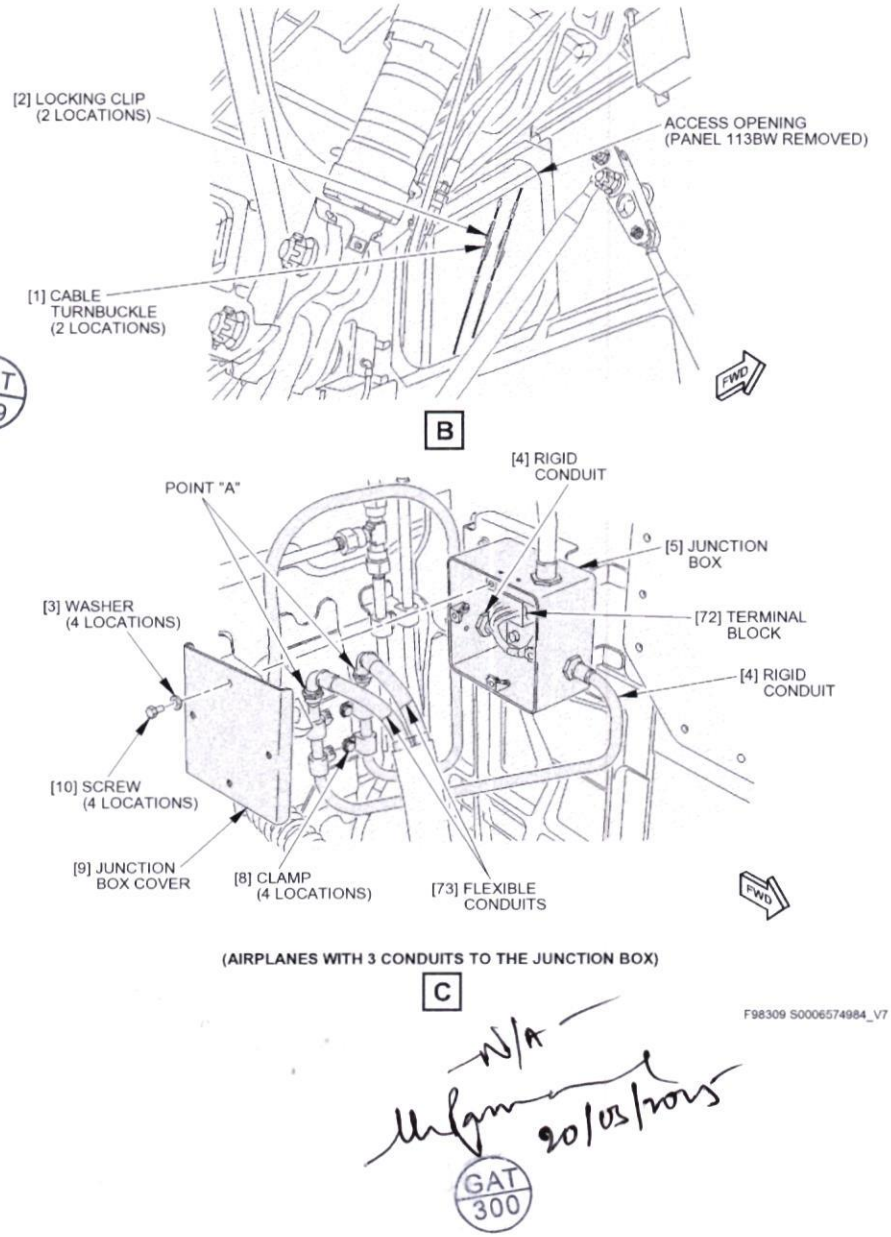

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Figure 401. Nose Landing Gear Installation - Sheet 2
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

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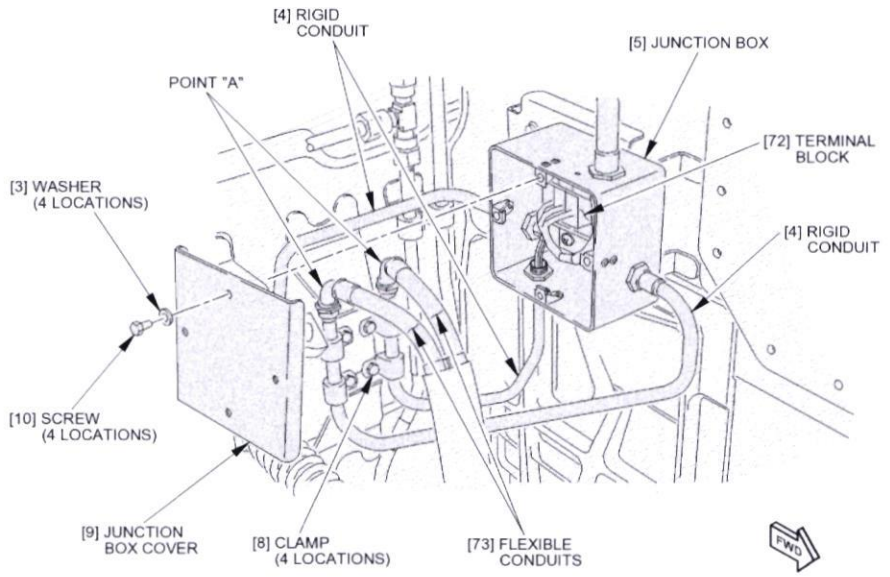
Item: _____ Completed through item: _____ Sign: _____



(AIRPLANES WITH 4 CONDUITS TO THE JUNCTION BOX)

C

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Figure 401. Nose Landing Gear Installation - Sheet 3
TASK 32-21-00-000-801

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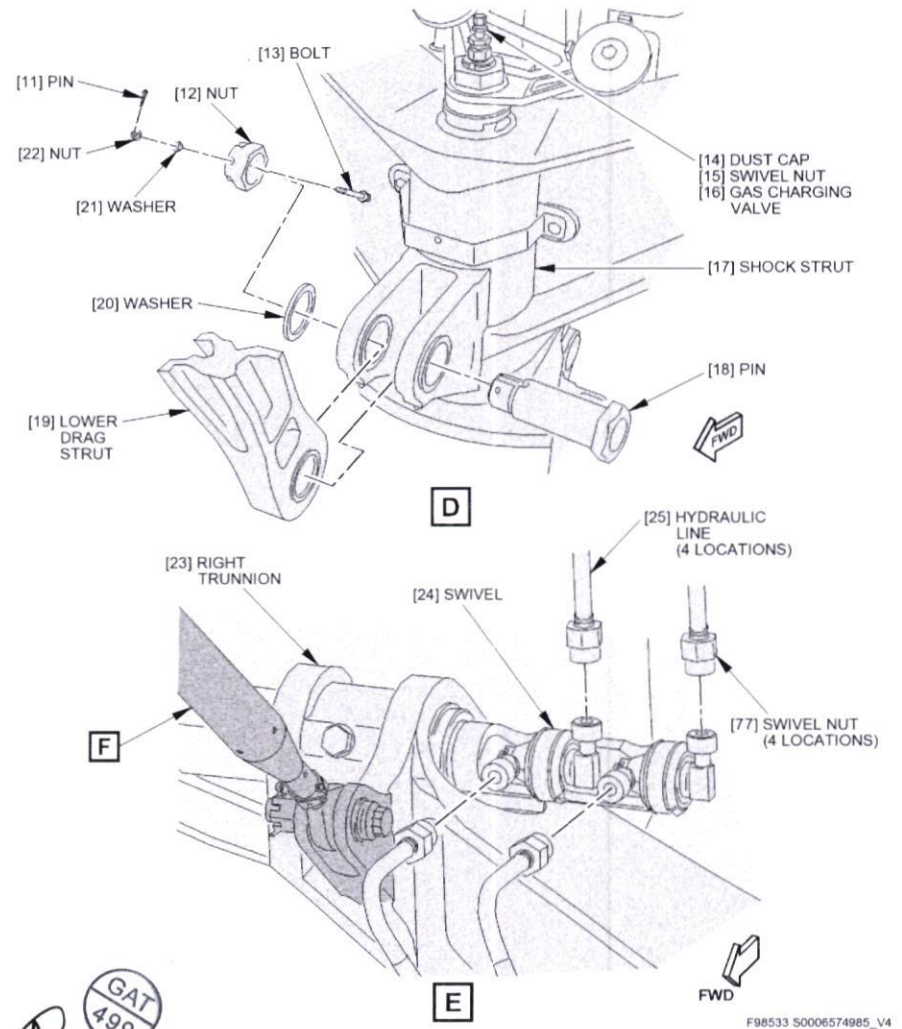
Item: _____ Completed through item: _____ Sign: _____



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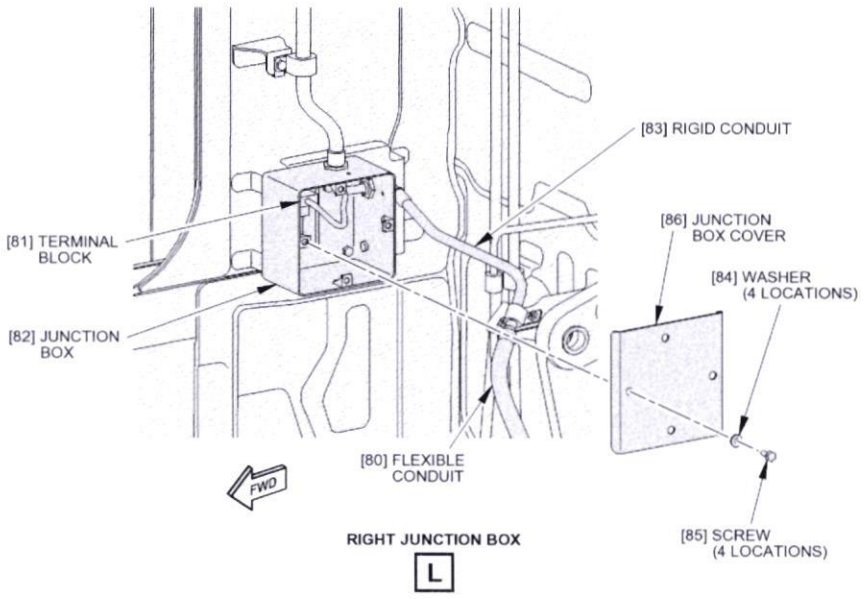
2013/25

Figure 401. Nose Landing Gear Installation - Sheet 4
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



3022968 S0000795208_V1

20/03/2025

GAT 300

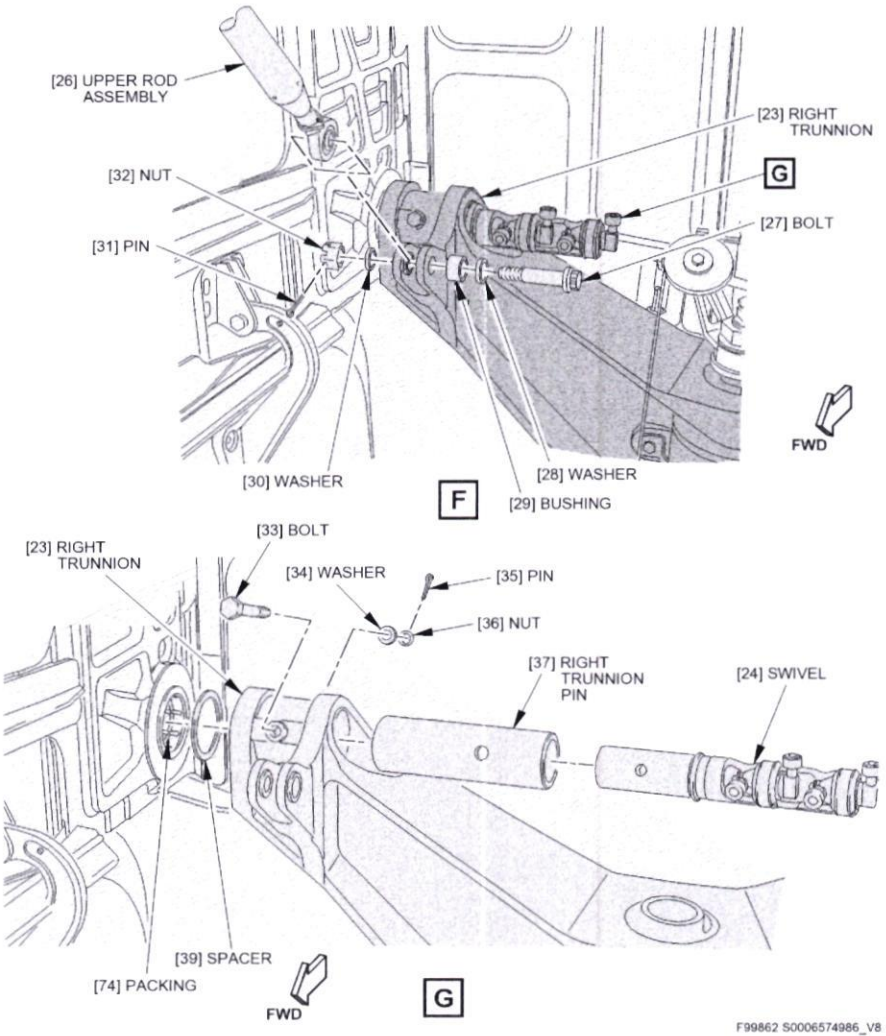
Figure 401. Nose Landing Gear Installation - Sheet 5
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033



GAT 499

20/3/25

Figure 401. Nose Landing Gear Installation - Sheet 6
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

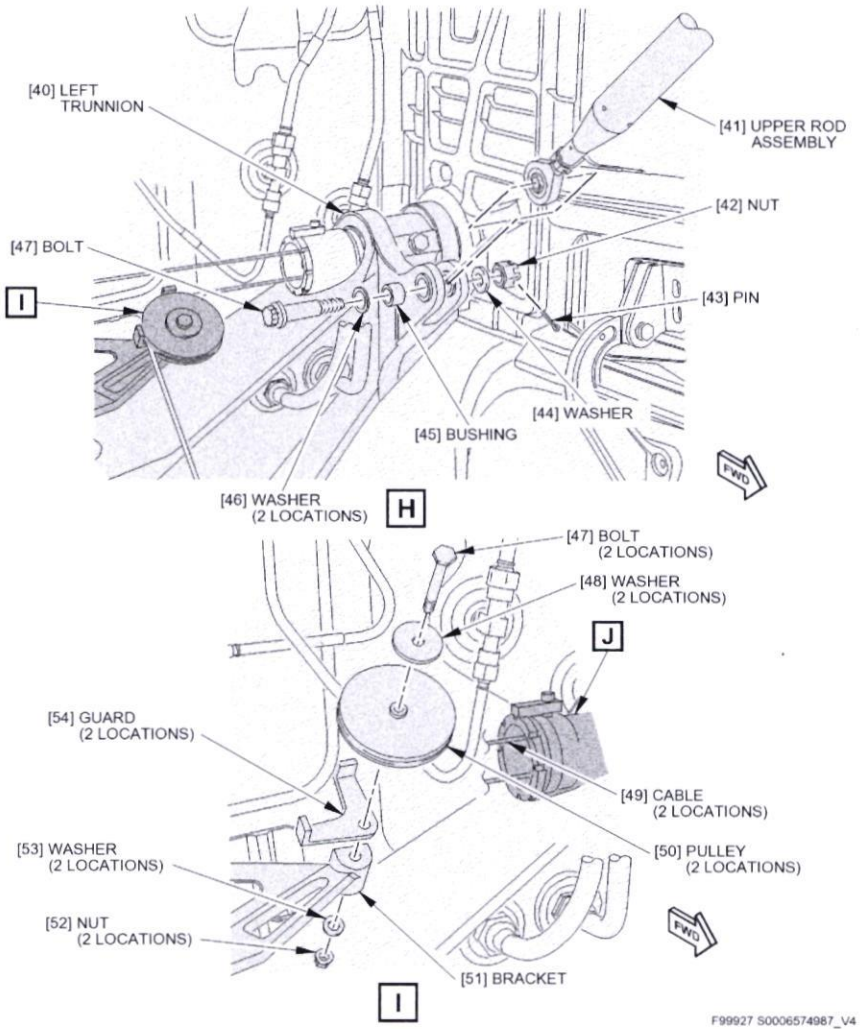


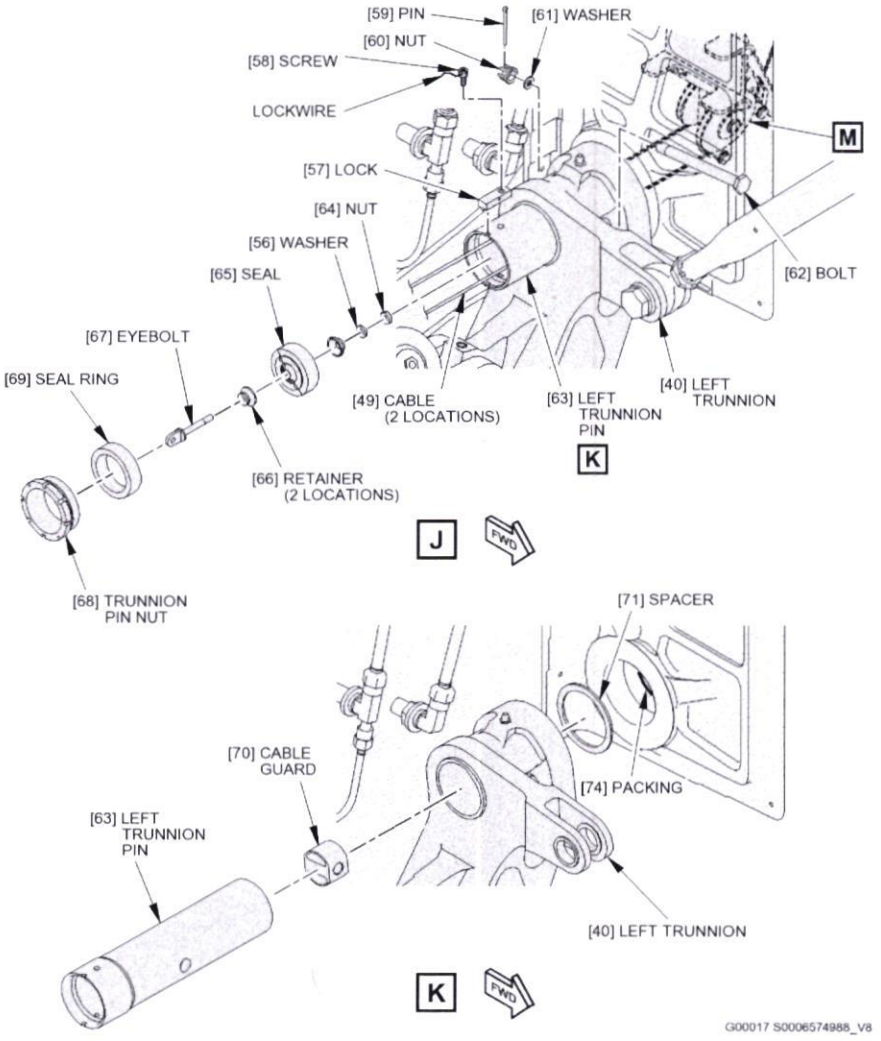
Figure 401. Nose Landing Gear Installation - Sheet 7
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033



GAT 499

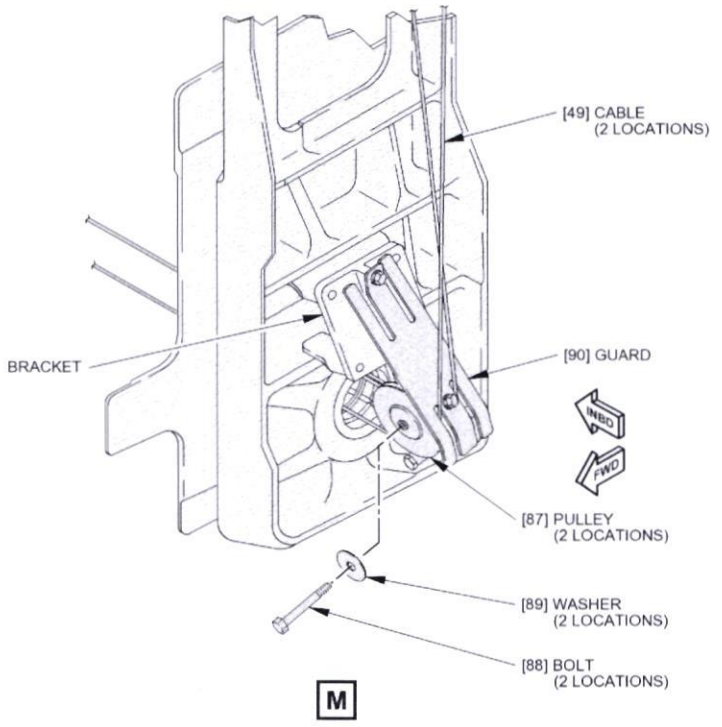
2013/25

Figure 401. Nose Landing Gear Installation - Sheet 8
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



GAT 499

2013125

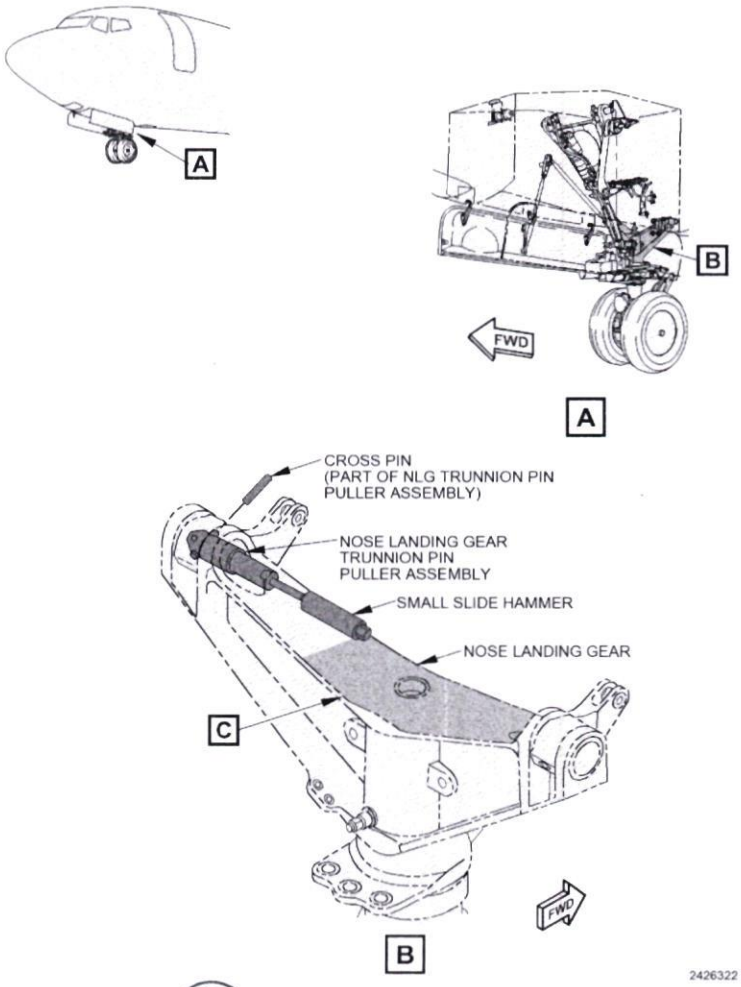
TAR 13388

Figure 401. Nose Landing Gear Installation - Sheet 9
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



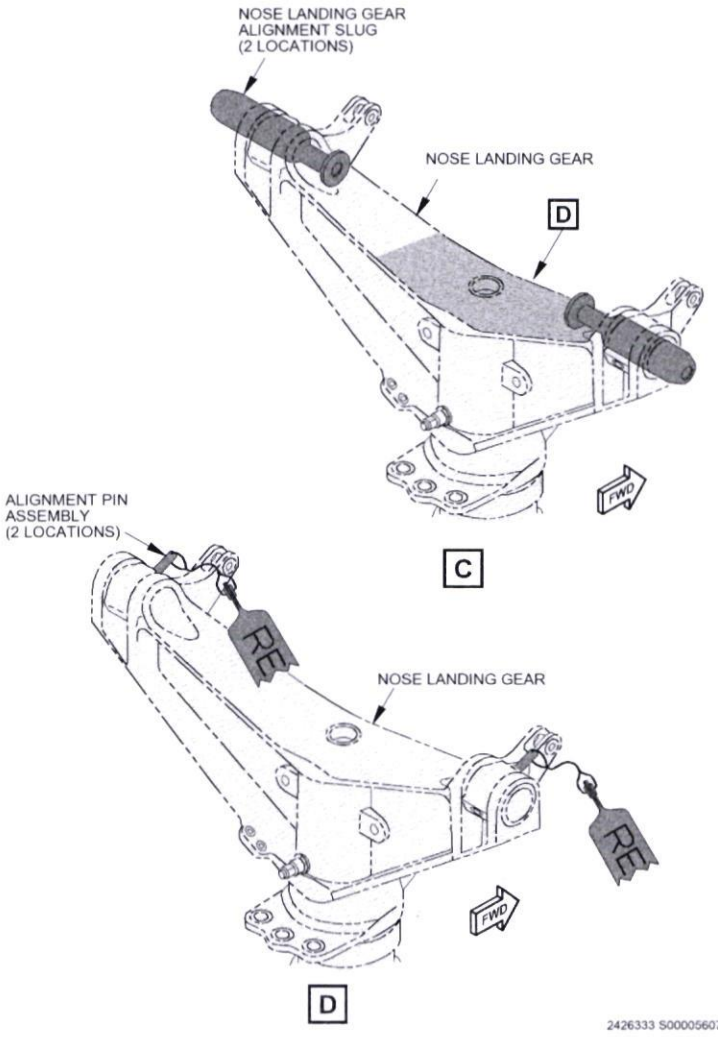
GAT 499

Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment - Sheet 1
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



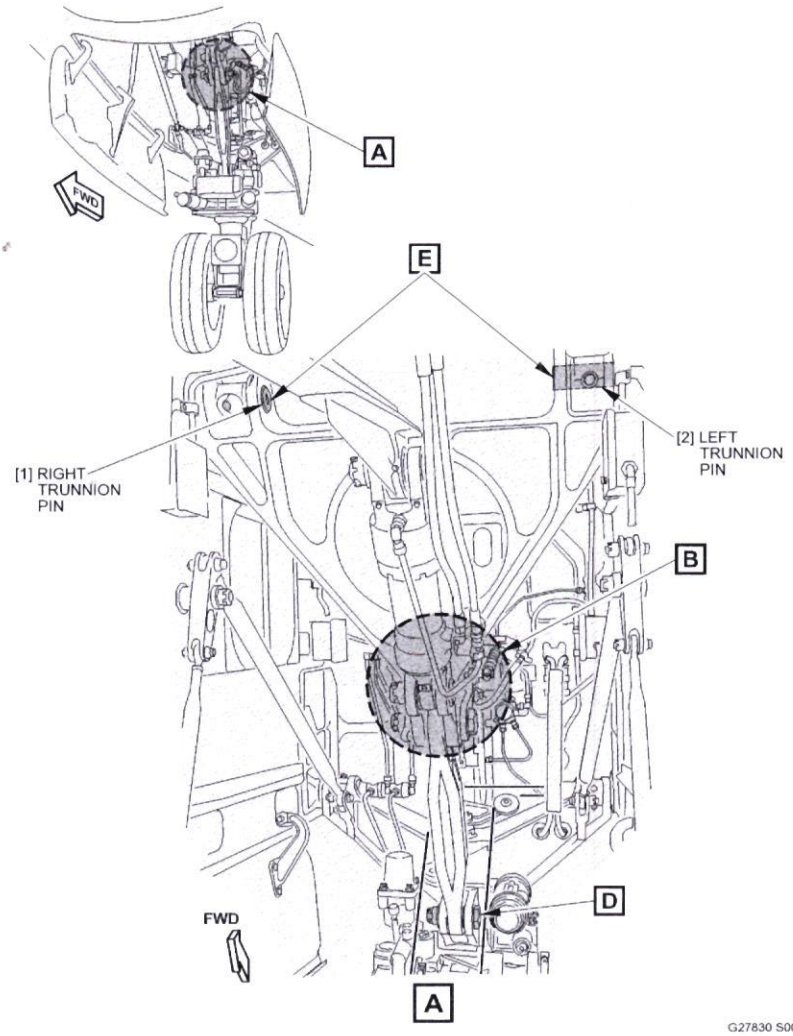
GAT 499

Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment - Sheet 2
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



GAT 499

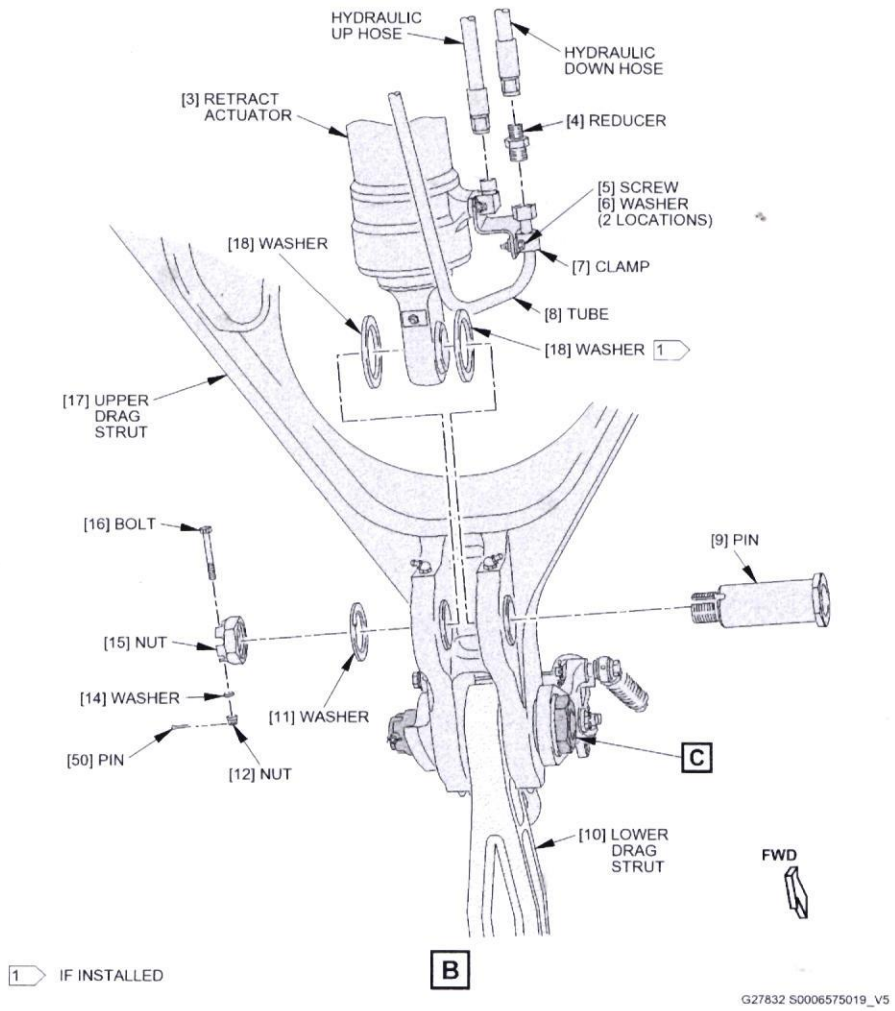
Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 1
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES



GAT 499

20/3/25

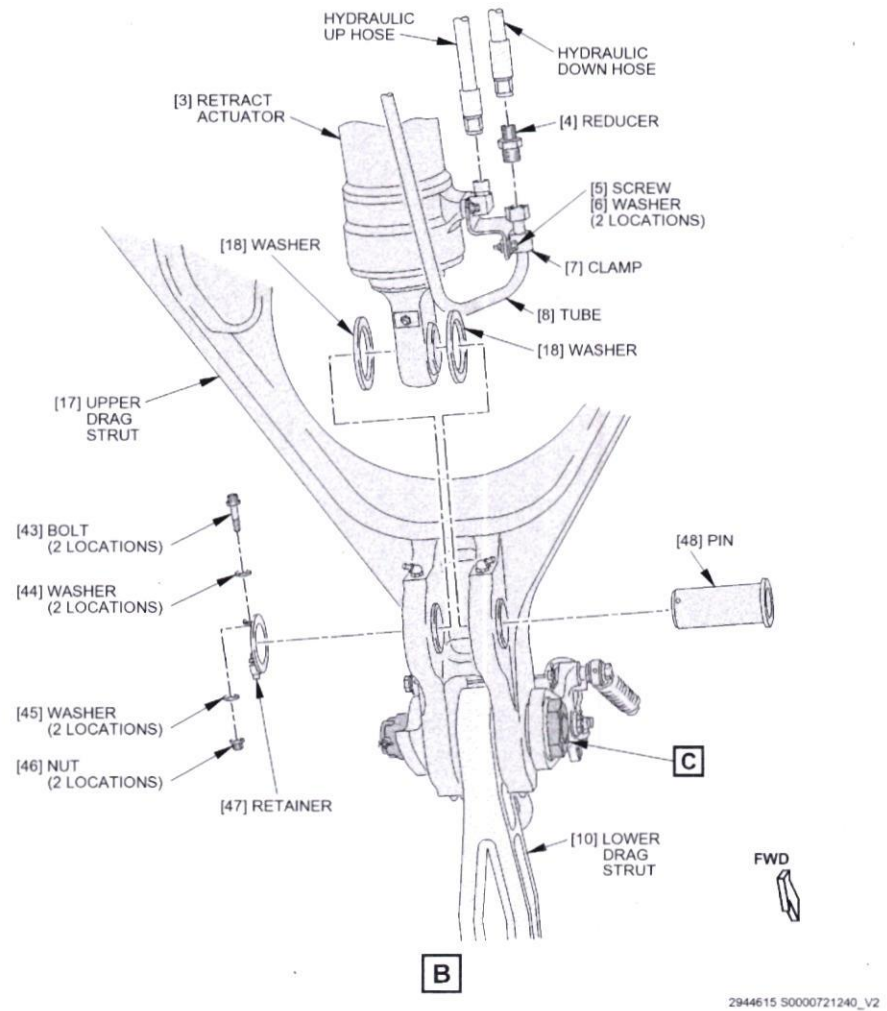
Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 2
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES



N/A

GAT 499

2013125

Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 3
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

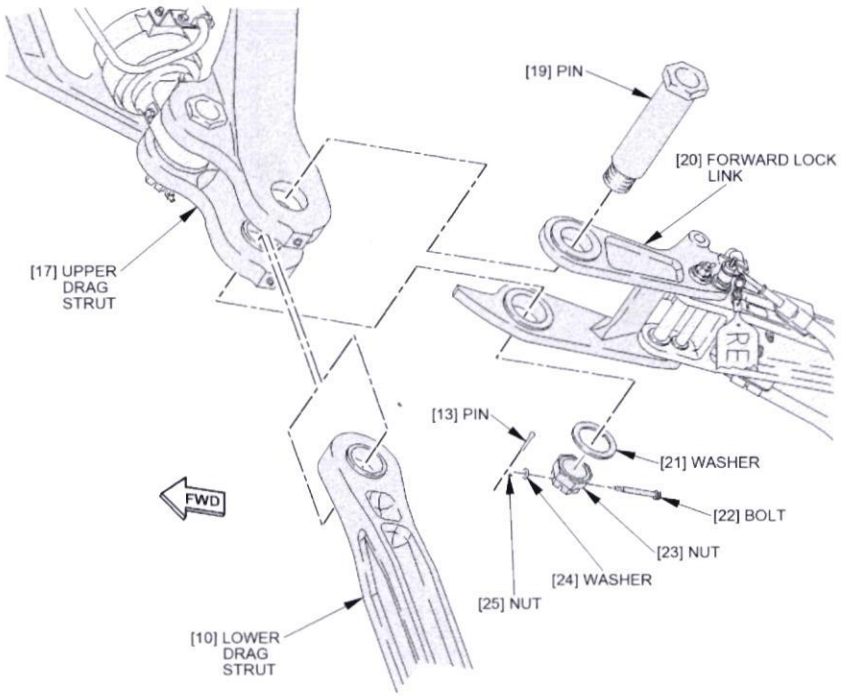
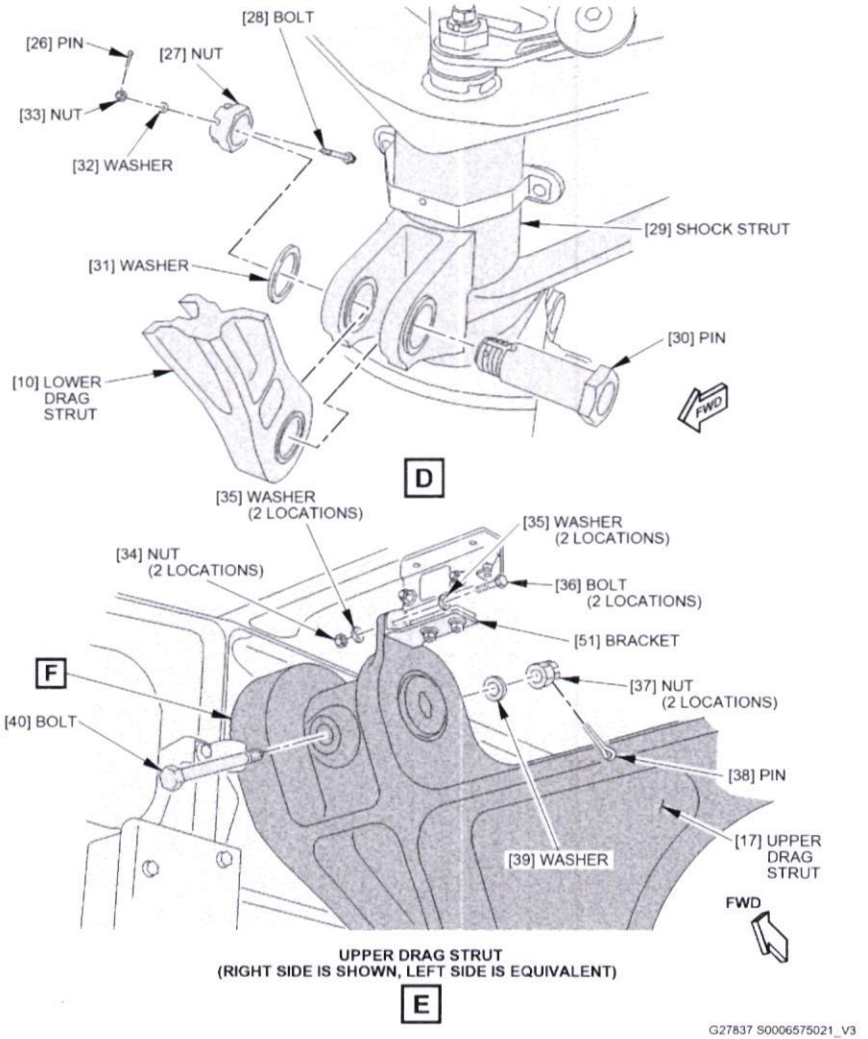


Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 4
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



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20/3/25

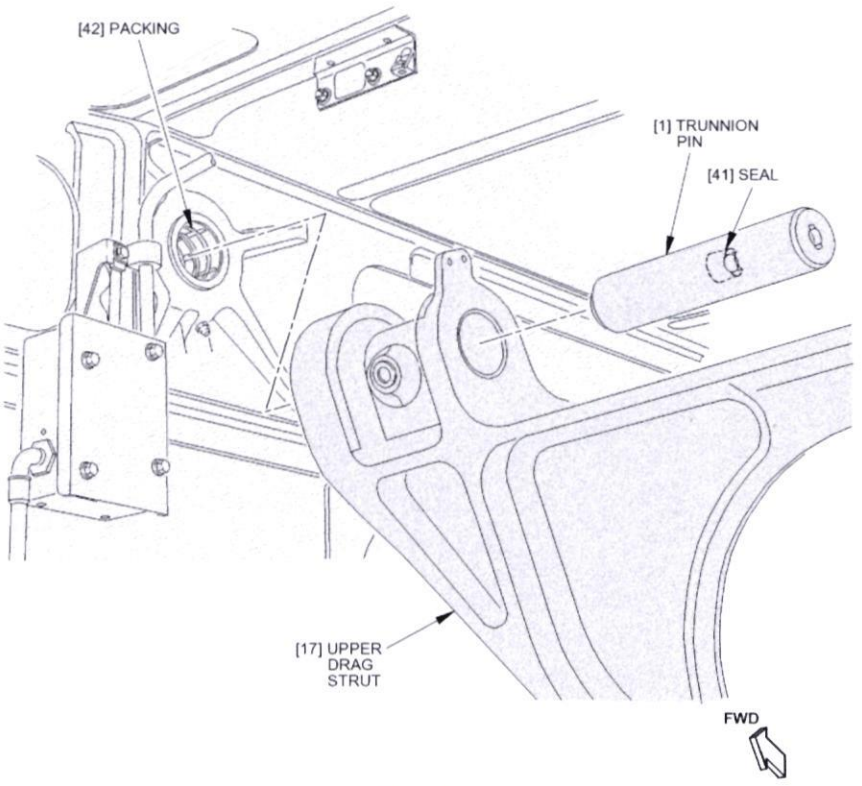
Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 5
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033



UPPER DRAG STRUT
(RIGHT SIDE IS SHOWN, LEFT SIDE IS EQUIVALENT)

F

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GAT 499

20/3/25

Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 6
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

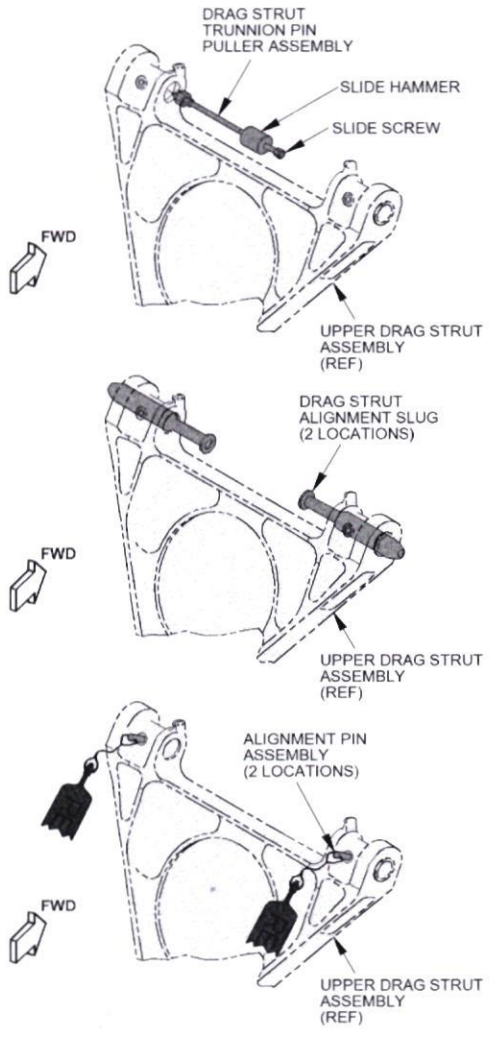
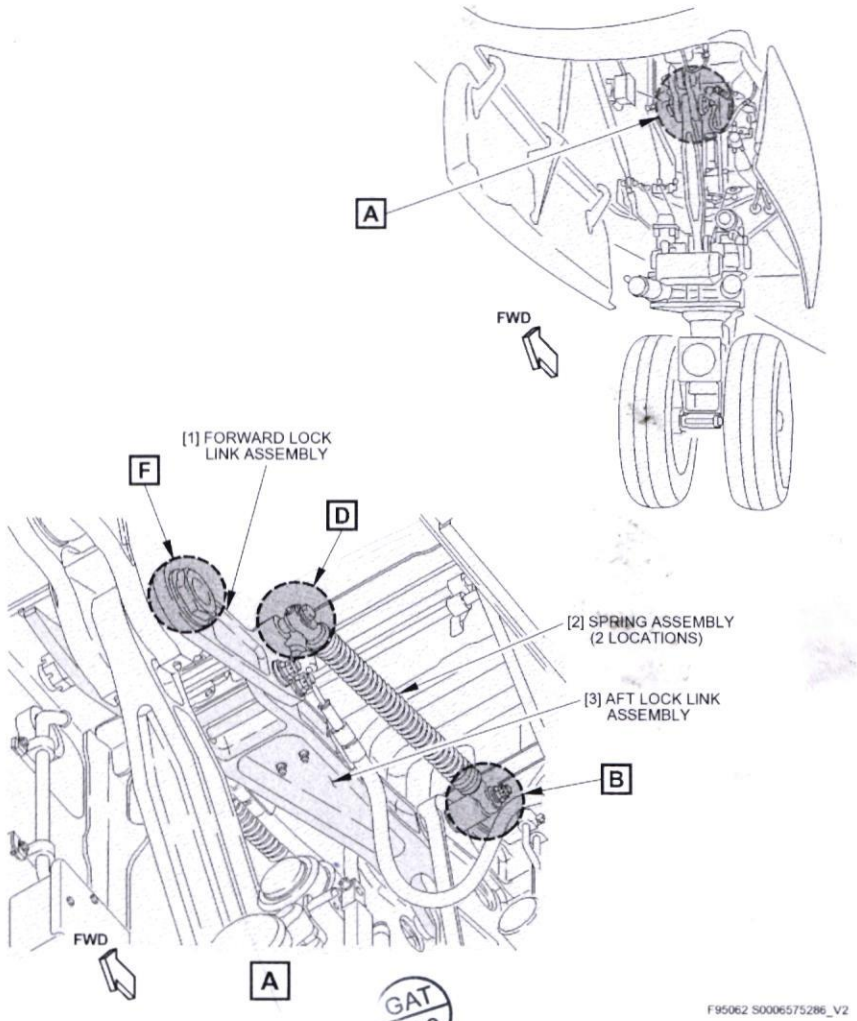


Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment - Sheet 1
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



GAT 499

2013/25

Figure 401. Nose Gear Lock Mechanism Installation - Sheet 1
TASK 32-33-51-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

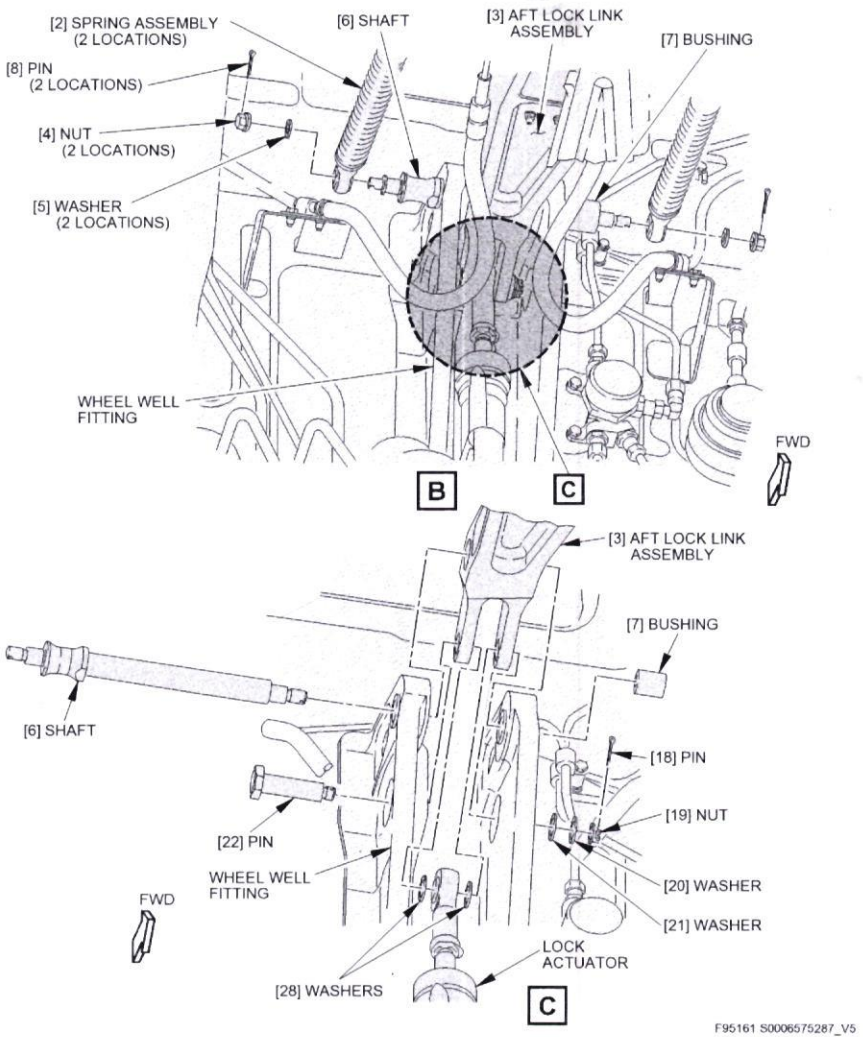
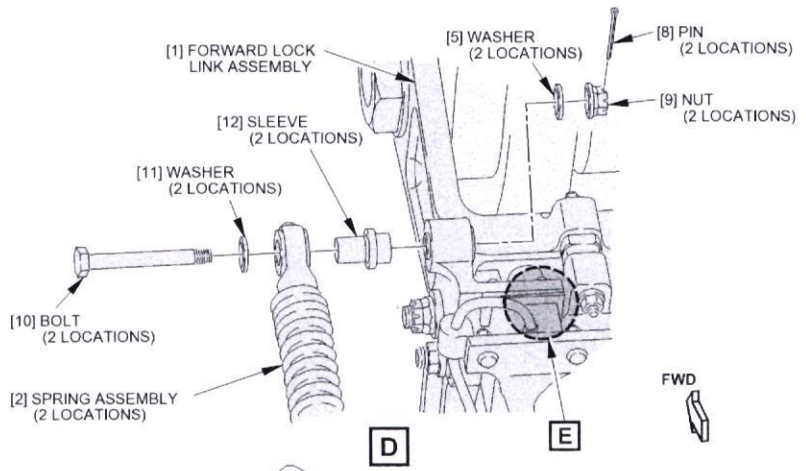


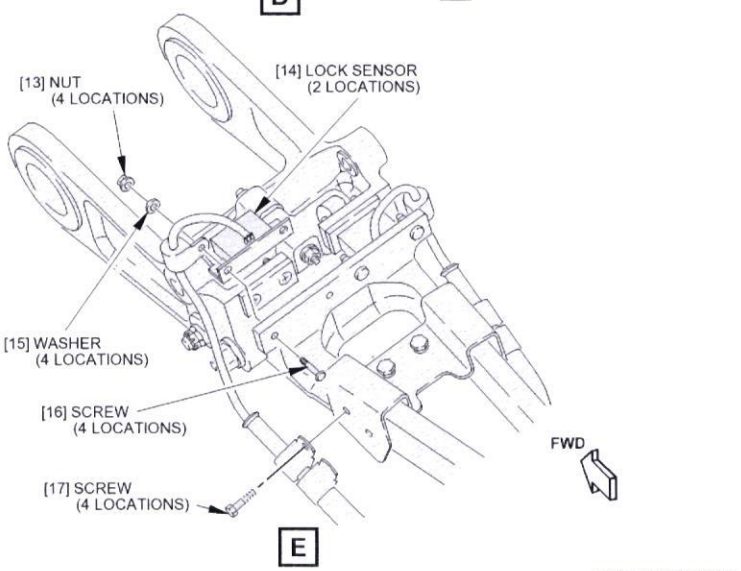
Figure 401. Nose Gear Lock Mechanism Installation - Sheet 2
TASK 32-33-51-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____





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20/3/25

Figure 401. Nose Gear Lock Mechanism Installation - Sheet 3
TASK 32-33-51-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

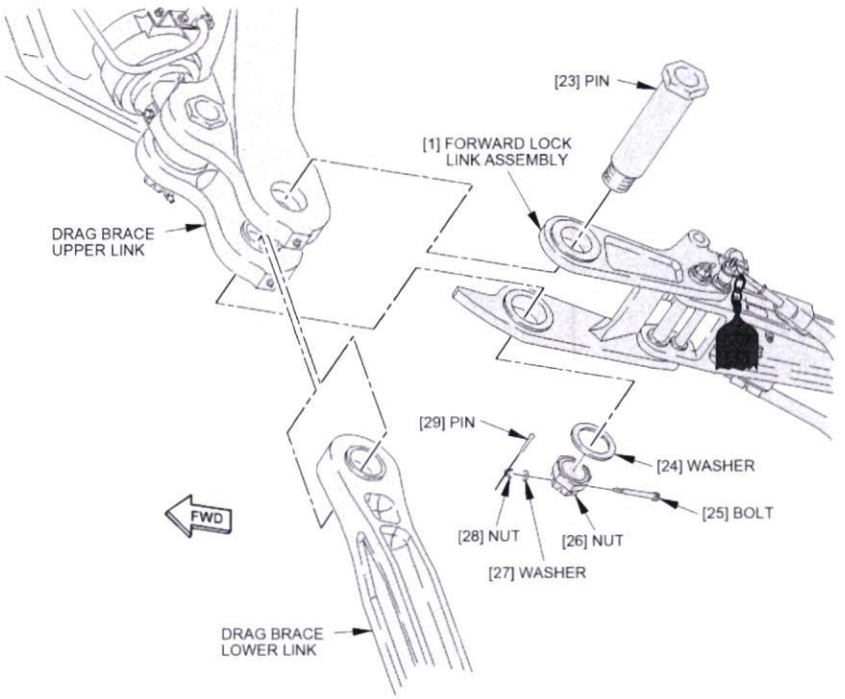
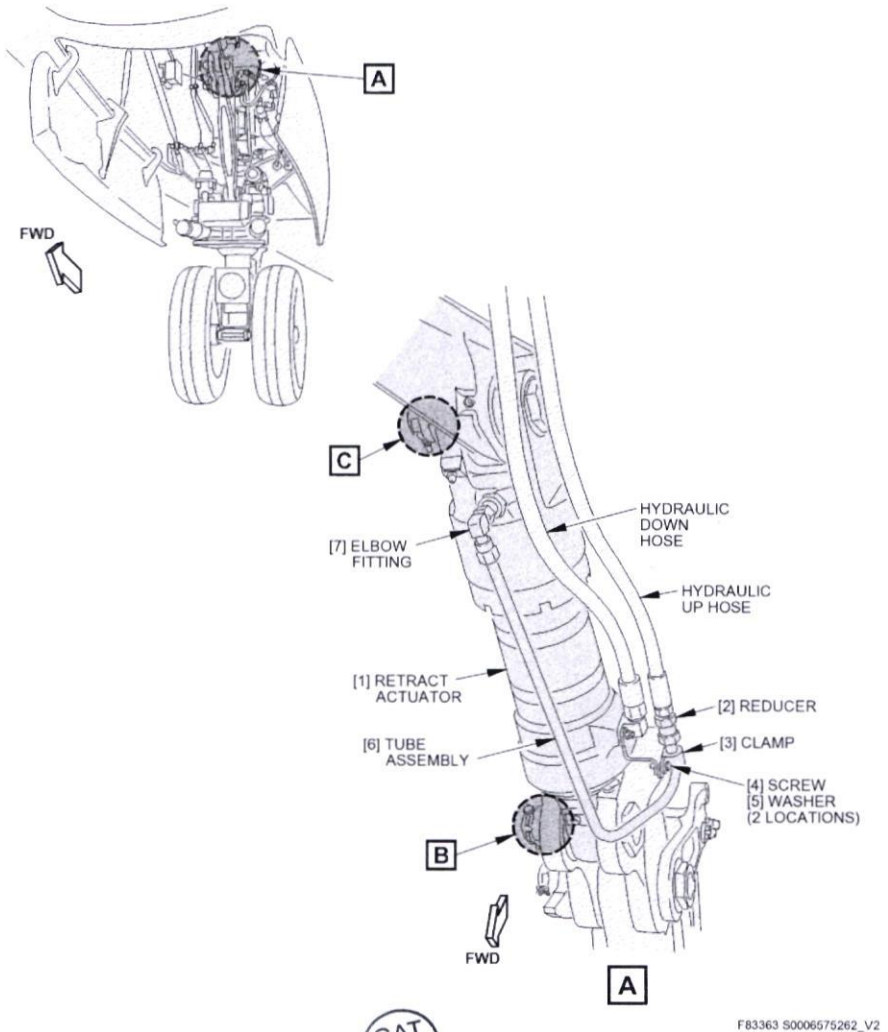


Figure 401. Nose Gear Lock Mechanism Installation - Sheet 4
TASK 32-33-51-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



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2013125

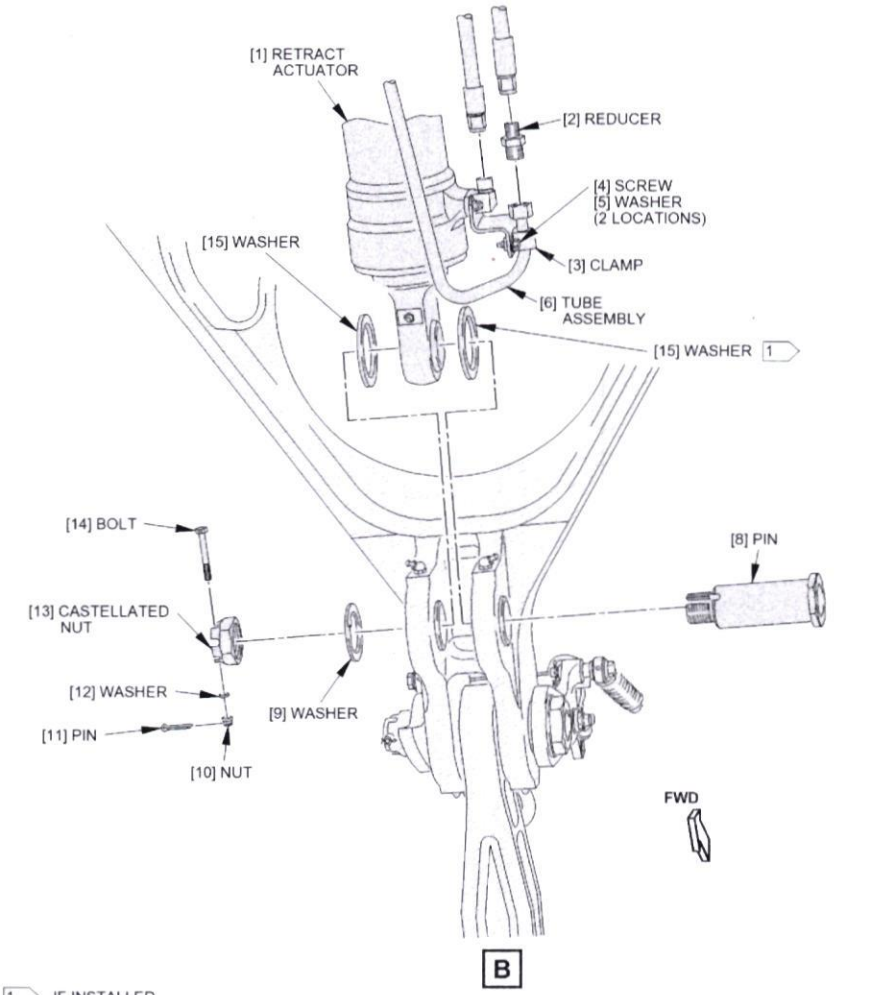
Figure 401. Nose Gear Retract Actuator Installation - Sheet 1
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES



1 IF INSTALLED

F84023 S0006575263_V3

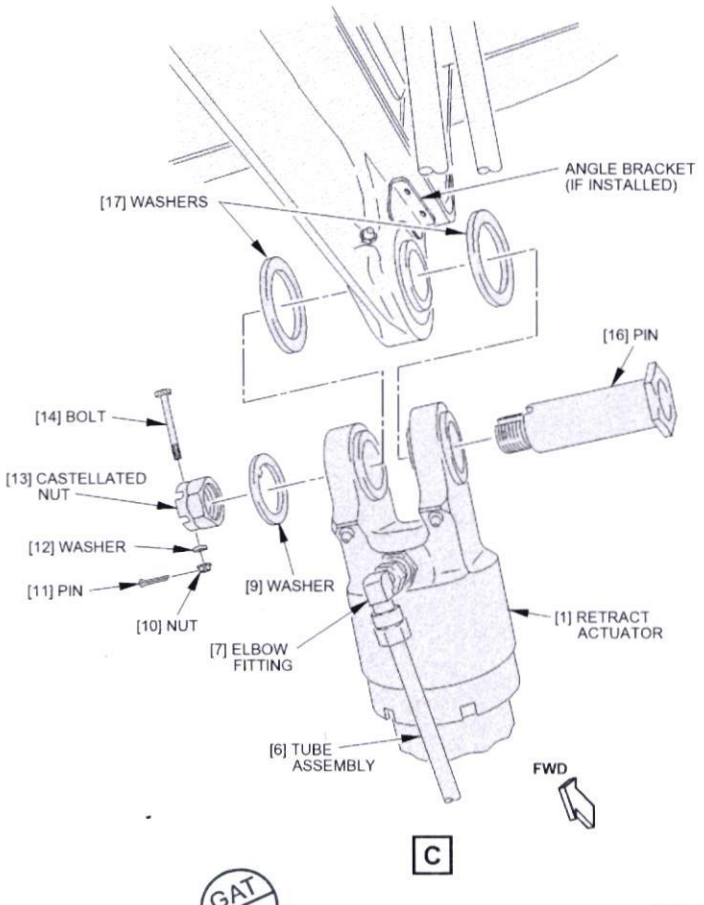
Figure 401. Nose Gear Retract Actuator Installation - Sheet 2
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES



GAT 499

Figure 401. Nose Gear Retract Actuator Installation - Sheet 3
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

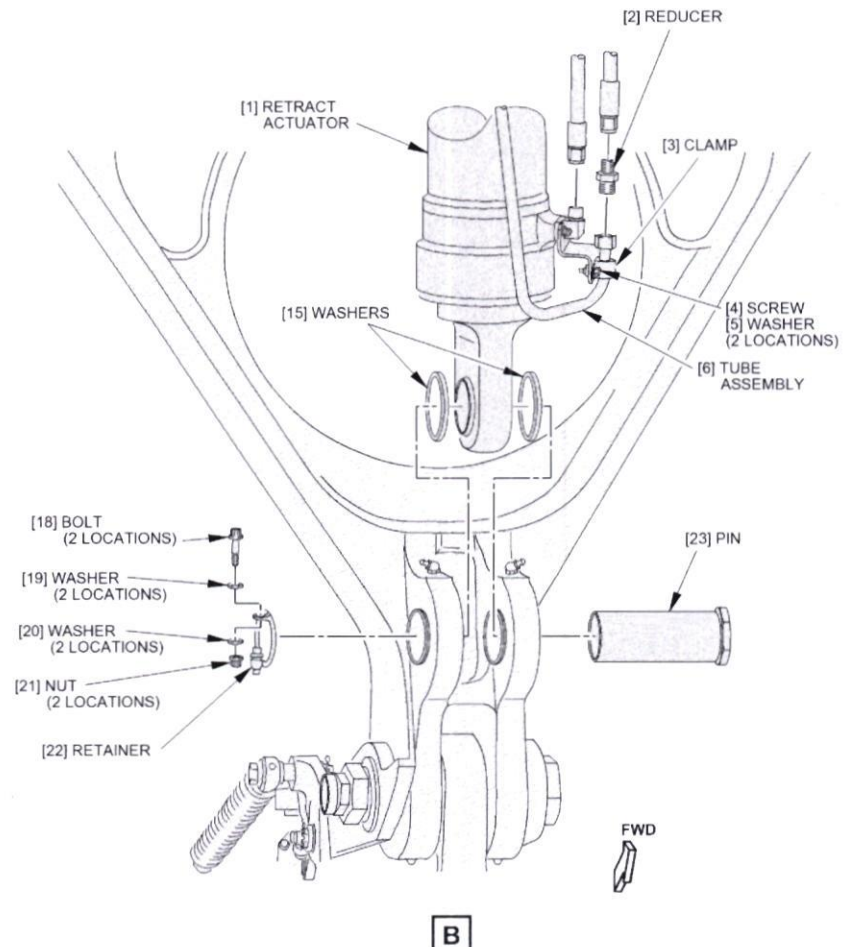


Figure 401. Nose Gear Retract Actuator Installation - Sheet 4
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

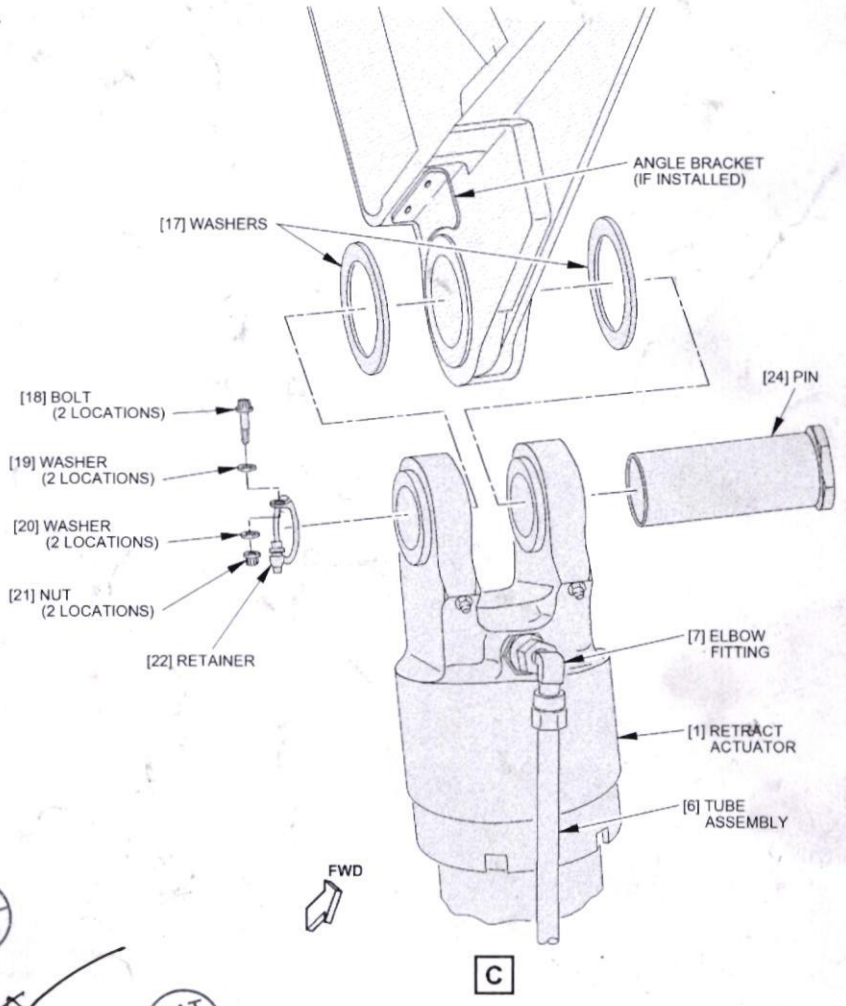


Figure 401. Nose Gear Retract Actuator Installation - Sheet 5
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

		Regulations ☑ DGCA CAR 145 ☑ EASA Part 145 ☑ FAA Part 145 ☐ Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
MS29513-031		PACKING-PREFORMED,HYDROCARBON	Expendable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-081684-0521 / 67830	100.00 EA	3P0000850421	GR21/001179/0521	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
BOEING DISTRIBUTION SERVICES INC.	MAT CERT.	25/01/2021	31/03/2036	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S /01 /040808D

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1047 /Aneesh K		18/05/2021
SIGNATURE		**STAMP**
		GAT 287

21

CUSTOMS INVOICE/PACKING SHEET

6456773-00

SHIPPER: Proponent MAIN whse
3120 ENTERPRISE ST.
BREA,CA 92821 US

UPC VENDOR	INVOICE NO.	ON DOCK
000000	K5136R	04/09/21
PROMISED	REQUEST	SHIPPED
04/12/21	04/12/21	
CUSTOMER P.O.		CUSTOMER RELEASE
3P0000850421		FA6X32

Cust#: 41513
SOLD TO: Boeing Distribution Svcs BDSI

SHIP TO: GMR AIR CARGO AND AEROSPACE ENG.LTD
PLOT NO.1, GMR AEROSPACE PARK
GMR HYDERABAD AVIATION SEZ LTD

HYDERABAD, IN 500 108 IN

CORRESPONDENCE TO: GMR AIR CARGO AND AEROSPACE ENG.LTD
PLOT NO.1, GMR AEROSPACE PARK
GMR HYDERABAD AVIATION SEZ LTD

HYDERABAD ,IN IN 500 108

, US

Pref. Routing Routine: DHL INTL COLL

P.O. NUMBER	ITEM NO.	PART NUMBER	ICN No.	QTY	UOM	UNIT PRICE	TOTAL VALUE	COUNTRY OF ORIGIN	QUANTITY ORDERED	QUANTITY B.O.	QUANTITY SHIPPED
00LYX24	2	MS29513-031		100.00	EA	0.30	30.00		100.00	0.00	100.00
		Export ECCN: EAR99									
		Desc: PACKING PCAT: S Sch B 4016930000									
			700612	100.00		Cure: 1Q2021		US			
		MFR- Name: PARCO INC. MFR-Product: 0568-031 Revision: C MFR- Batch: 67830 ITEM(S) IDENTIFIED HEREIN CONFORM TO AN ESTABLISHED INDUSTRY GOVERNMENT OR COMMERCIAL STANDARD AND ARE BEING SHIPPED BY PROPONENT ON BEHALF OF BOEING DISTRIBUTION SERVICES INC. S/L: 15 YRS PER AS5316 INSP BY: Yesenia Rodriguez 04/09/2021									

TOTAL BOX VALUE:

FOR CUSTOMS PURPOSES ONLY. DO NOT PAY FROM THIS DOCUMENT.
NLR unless otherwise advised in body of document

BOEING
Boeing Distribution Services Inc.

CERTIFICATE OF CONFORMANCE

WE HEREBY CERTIFY THAT THE PRODUCT SUPPLIED IS NEWLY MANUFACTURED, CONFORMS TO THE APPROVED DESIGN DATA, AND MEETS ALL REQUIREMENTS OF THE APPLICABLE PURCHASE ORDER. EVIDENCE OF CONFORMANCE IS ON FILE AND AVAILABLE FOR REVIEW UPON REQUEST.

Thank You For This Order

GAR Aero Technic

STORE ACCEPTANCE TAG

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others

_____ (Specify)

Description:	PACKING	GRN :	GR09/003416/1019
Part no :	MS29513-033	Cond :	NEW
Serial No :		Location :	01/~~01~~ 030702B
Batch/ Lot No :	0080301071 / LNG-025969-1019		
Release Note No:	01216959	Date:	03-10-2019
Supplier:	BOEING		
P.O. No.:	9P0014321019	Date:	03-10-2019
Quantity:	100	UOM:	EA
Storage Requirements:	—	Expire On :	31/12/2032
Signature of Inspector:	Jayanta N	Date:	11-10-2019
		INSPECTOR STAMP	GAT 206

Acceptance

TAG No. GAT/MS/005 Issue Initial, Rev 1, Dated 09/07/2019

Authority/Country:
FA UNITED STATES

AUTHORIZED RELEASE CERTIFICATE

FA orm 8130-3, AIRWORTHINESS APPROVAL TAG

01216959

4. Organization Name and Address:
THE BOEING COMPANY,1901 OAKESDALE AVE SW, M/C 33-112 RENTON,WA 98057-2623 P.C. #700

5. Work Order/Contract/Invoice Number:
1157497

6. Item:	7. Description:		8. Part Number:	9. Quantity	10. Serial Number:	11. Status/Work:
000001	WIRE ASSY	8000929518	D72D70337-913	5 EA	N/A	NEW
000002	PACKING	8000929519	MS29513-033	100 EA	N/A	NEW
000003	PLUG BUTTON	8000929520	411A2226-5A	50 EA	N/A	NEW
000004	RUB STRIP	8000929521	114A1804-1	1 EA	N/A	NEW

12. Remarks: 5002 1CM

Page 1 of 1

SHIPMENT:1157497

AIRWORTHINESS APPROVAL.

13a Certifies the item identified above were manufactured in conformity to:

☒ Approved design data and are in a condition for safe operation.
☐ Non-approved design data specified in Block 12.

14a ☐ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. SIGNATURE:	13c. Approval/Authorization No:	14b. SIGNATURE:	14c. Approval/Authorization No:
David Cho	ODA-300064-NM		
13d. Name (Typed or Printed): DAVID CHO	13e. Date (dd/mmm/yyyy): 03/OCT/2019	14d. Name (Typed or Printed):	14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airwortiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [ ] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC03A04P		PIN	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-240018-0424 / 03744147	500.00 EA	4P0051780324	GR24/000359/0424	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING	03744147	04/04/2024	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

02/041009A H/S 02/021007A

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1538 /Ramesh Manchu		13/04/2024
SIGNATURE		**STAMP**
		GAT 4DB

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

Authority/Country:
FAA/UNITED STATES

AUTHORIZED RELEASE CERTIFICATE

FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG

03744147

4. Organization Name and Address:
THE BOEING COMPANY,737 LOGAN AVE N, RENTON,WA 98057-0000 PC #700

5. Work Order/Contract/Invoice Number:
3540561

6. Item:	7. Description:		8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
000001	PIN, COTTER (SPLIT)	8008218293	BACP18BC03A04P	500 EA	N/A	NEW
000002	PACKING, O-RING	8008218294	M25988/4-906	30 EA	N/A	NEW

12. Remarks: 5007 1GM

Page 1 of 1

SHIPMENT:3540561

AIRWORTHINESS DOCUMENT.

13a. Certifies the item identified above were manufactured in conformity to:

☒ Approved design data and are in a condition for safe operation.
☐ Non-approved design data specified in Block 12.

14a. ☐ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No: PC 700	14b. Authorized Signature:	14c. Approval/Certificate No:
13d. Name (Typed or Printed): GHASSAN M KHAN	13e. Date (dd/mmm/yyyy): 04/APR/2024	14d. Name (Typed or Printed):	14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [ ] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC03A10P		PIN	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-249037-0524 / 03817337	300.00 EA	4P0003790524	GR24/001463/0524	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING	03817337	13/05/2024	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 02 / 030809B

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1538 /Ramesh Manchu		20/05/2024
SIGNATURE		**STAMP**

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

1. Approving Civil Aviation Authority/Country: FAA/UNITED STATES	2. **AUTHORIZED RELEASE CERTIFICATE** FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	3. Form Tracking Number: 03817337

4. Organization Name and Address: THE BOEING COMPANY,737 LOGAN AVE N, RENTON,WA 98057-0000 – PC #700	5. Work Order/Contract/Invoice Number: 3609669

6. Item:	7. Description:		8. Part Number:	9. Quantity	10. Serial Number:	11. Status/Work:
000001	PIN, COTTER (SPLIT)	8008388199	BACP18BC03A10P	300 EA	N/A	NEW
000002	PIN, COTTER (SPLIT)	8008388200	BACP18BC02A06P	1000 EA	N/A	NEW

12. Remarks: 5009 1GM

SHIPMENT:3609669

SUPPLEMENTAL SHIPMENT - AIRWORTHINESS DOCUMENT

13a. Certifies the item identified above were manufactured in conformity to:

☒ Approved design data and are in a condition for safe operation.
☐ Non-approved design data specified in Block 12.

14a. ☐ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No: PC 700	14b. Authorized Signature:	14c. Approval/Certificate No:
13d. Name (Typed or Printed): LARRY W. GAC	13e. Date (dd/mmm/yyyy): 13/MAY/2024	14d. Name (Typed or Printed):	14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

BOEING

X37969 NEW (30 JAN 2017)

GMR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
CORBAN 27L		CORROSION INHIBITANT	Consumable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-329790-0225 / 24031A/24051	5.00 CAN	4P0028580125	GR24/009377/0225	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
TENTACLE AEROLOGISTIX PRIVATED LIMITED	ZIPCHEMCOC/258510	28/05/2024	28/05/2027	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 09/040502X

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

STORE B/W 4° - 37°C

INSPECTED BY		INSPECTED DATE
MR1591 /Sunil kumar Puli		04/02/2025
SIGNATURE		**STAMP**
		GAT 456

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

ZIP-CHEM PRODUCTS
A DIVISION OF ANDPAK, INC
400 JARVIS DRIVE
MORGAN HILL, CA. 95037
(408) 782-2335

Certification of Conformance

We hereby certify that the goods supplied on this purchase order were produced in compliance with the requirements of the Fair Labor Standards Act, as ammended, and of regulations and orders of the United States Department of Labor issued thereunder. Materials furnished have been manufactured and/or packaged in accordance with all applicable instructions and specifications.
The user shall determine the suitability of this product for its intended use and user assumes all risk and liability in connection therewith. Certification, test reports or any applicable documents pertaining to this purchase order are kept on file and available for review at Zip-Chem Products.

Date shipped:	MAY 28, 2024	Batch#:	24031A/24051
To:	BOEING DISTRIBUTION INC.	Date of manufacture:	01/2024
Cust. P.O.#:	46881979	Job Number:	258510
Product:	Cor-Ban 27L CIC (009404)	Date of expiration:	05/28/2027
Net contents:	PINT	Quantity shipped:	25 CASES

ACCEPTANCE TEST DATA

TEST PROPERTY	REQUIREMENT	TEST METHOD	RESULTS
Consistency, 77° F (25 C), ± 5° F and 50± 10% relative humidity. Consistency, 130° F (54 C)	Smooth in texture, free of entrapped air when examined visually. No flow at 130 F (54 C).	Visual inspection of texture and high temperature flow test	Pass
Appearance	Shall spread smoothly to result in an even film and have a uniform composition, with the same physical properties throughout. No lumps or separated material shall be present. Color shall be Amber / Tan.	4.8.7 (GMS 4109)	Pass
Odor	No rancid, perfume or alcohol odor.	ASTM D 4339	Pass
Application Properties	A minimum of 15 grams per minute at 40° F (4 C).	SLP/CORBAN27L/APP	149
Nonvolatile Content	92.77% ± 5% (88.13 – 97.41) (per BMS 3-38)	ASTM D 1644 Test Method A	91.76%
Preload Test	The average readings must be between 4,940 and 5,460 pounds. All readings must be between 4,800 and 5,800 pounds. (Avg)	SLP/CORBAN27L/LOAD	5210
Nonvolatile Content	90% minimum (per MEP 09-075)	SLP-CORBAN27L-NVE	92

This product is qualified to BMS 3-38B, HMS 20-1267/2372, MEP 09-075F, LGQP 5010E Ty V, CSNP038H Ty X, CML 05-133, GMS 4109, TY. NC1, Rev. C and is referenced under DPM 6498. The specific batch stated above, fully meets the requirements of these specifications.
Shelf life: 3 years from the DOS unless otherwise indicated above, when properly stored.
Storage: Material should be stored at 40 to 100 degrees Fahrenheit in the original unopened containers.

QA 1 MEPAK

ZIP-CHEM PRODUCTS
MIKE SMITH,
QUALITY MANAGER

CERTIFIED TRUE COPY

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
SHC 100		MOBIL GREASE SHC 100 (2KG/CAN)	Consumable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-220532-1223 / 70436828T	8.00 KG	4P0043621123	GR23/005075/1223	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
TENTACLE AEROLOGISTIX PRIVATED LIMITED	70436882	07/12/2022	07/12/2028

WAREHOUSE / ZONE / BIN / RACK DETAILS

~~LM-HYD-SER~~ H/6/09/041009X

STORAGE REMARKS

FOLLOW GENERAL STORAGE CONDITIONS

INSPECTED BY		INSPECTED DATE
MR1144 /Manmohan Dash		07/12/2023
SIGNATURE		STAMP
		GAT 351

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

Certificate of Analysis

ExxonMobil

145424.

Beaumont Lube Blending Plant
2805 Sycamore St. Beaumont, Texas 77701
Date: 12/14/2022

Product: MOBIL AVIATION GREASE SHC 100

Batch Number	70436828T	Quantity / Package	TIN-FINISH
Order Key	70436882	Manufacture Date	12/07/2022
Export# / P.O.#		Destination	
Yardmark	10122M08A	Reference #	
CMCS Code/ Prod#	201550402010	T/C or T/T	CS 245/TIN 3

Test Description	Method	Test Result
Odor	OLFACTORY	PASS
Texture	VISUAL	SMOOTH
Color	VISUAL	RED
Dropping Point, C	ASTM D2265	292
Four-Ball Wear Test, Scar Diameter, mm	ASTM D2266	0.6
Penetration, 60X, 0.1 mm	ASTM D217	295
Dirt, # particles 125u or larger	FTM 3005	0
Low Temperature Torque, Starting @ -54 C, Nm	ASTM D1478	0.6
Low Temp Torque, Running@-54 C, 60 min, Nm	ASTM D1478	0.2
Pen Worked X 100,000, 1/16" holes, 0.1 mm	FTM 313	321
Dirt, # Particles 25 - 124u	FTM 3005	0
Copper Strip Corrosion, 24h at 100C	ASTM D4048	1A
Oil Separation, 30 h @ 177 C, mass%	ASTM D6184	6.1

This material meets the ExxonMobil Sales specification established for this product and has been produced in a facility complying with the requirements of the ISO 9001 certified Global Product Integrity Management System (GPIMS). Test results on this certificate represent the most recent inspections done on this product for the stated characteristics and may be based on tank certification, manufacturing data, periodic testing and / or most recent product restock.

Direct Inquiries to:
Lorynda Clopton, Laboratory Supervisor
Phone: 1-409-240-4388
Fax: 1-409-240-8479

This document is electronically generated and does not require a physical signature to be valid.

ABSC, DUNLOP, GOODRICH, HONEYWELL, SAFRAN LANDING SYSTEMS, PARKER/CLEVELAND Wheel Bearing Approved.

Shelf life is 6 years from manufacturing date providing the container has remained unopened, undamaged, and was stored indoors.

Tests conducted according to ASTM Standard Test Methods are routinely verified to be in compliance with the latest published versions. Minor changes may be made where they have no material impact on test schedules and are necessitated by reasons such as safety, environmental standards, and method effectiveness.

BMT4514336_CofA

CERTIFIED TRUE COPY

GAR AERO TECHNIC

Regulations
- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
CORBAN 27L		CORROSION INHIBITANT	Consumable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-329790-0225 / 24031A/24051	5.00 CAN	4P0028580125	GR24/009377/0225	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
TENTACLE AEROLOGISTIX PRIVATED LIMITED	ZIPCHEMCOC/258510	28/05/2024	28/05/2027	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 09/040502X

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

STORE B/W 4°-37°C

INSPECTED BY		INSPECTED DATE
MR1591 /Sunil kumar Puli		04/02/2025
SIGNATURE		**STAMP**
		GAT 456

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

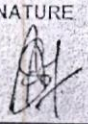
PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
AEROSHELL GREASE 33		SYNTHETIC UNIVERSAL AIRFRAME GREASE (CAN=3	Consumable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-251169-0624 / 12119150	90.00 KG	4P0003930524	GR24/001883/0624	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
TEAM AVIATION	SHELL COA	27/04/2023	04/04/2029	

WAREHOUSE / ZONE / BIN / RACK DETAILS

~~LM-HYD-3ER~~

STORAGE REMARKS

0°C TO 50°C

INSPECTED BY		INSPECTED DATE
MR1427 /Sharath Kumar Ayitha		03/06/2024
SIGNATURE		**STAMP**
		GAT 458

Z042-CH4500 Bern LubProd
Steigerhubelstrasse 8
Bern
3008
Switzerland

Certificate of Analysis

Material	550043673	Date Manufactured	05 Apr 2023
Material Description	AeroShell Grease 33 4*3kg	Date Tested	17 Apr 2023
Batch Number	12119150		

Test Description	Result	Unit	Method
Aspect/Consistency/Texture	Smooth	-	VISUAL
Bluegreen Colour	Bluegreen equal to standard	-	VISUAL
CU Corrosion 24h/100°C / 1b	1b	-	ASTM D4048
Dirt Count 25-74 µm (Part/ml)	0	******	FED-STD-791 3005
Dirt Count 75-124 µm (Part/ml)	0	******	FED-STD-791 3005
Dropping point	229,0	CEL	IP 396
Evaporation Loss 22h 100°C	0,98	%(m)	ASTM D2595
Four Ball Weld Load	500,0	kg	ASTM D2596
Low Temp Torque -73°C - Run	0,055	N/m	ASTM D1478
Low Temp Torque -73°C - Start	0,515	N/m	ASTM D1478
Load Wear Index	78,2	kg	ASTM D2596
Odour Aeroshell Grease	Correspond to Reference	-	NONE
Oil Separation 30h 100°C	1,9	%(m)	ASTM D6184
Oxidation Stability 100h 99°C	16,0	kPa	ASTM D942
Penetration Unworked 25°C	281	dmm	ASTM D217
PEN_WORKED_25C	290	dmm	ASTM D217
Pen Worked 100000 25°C	301	dmm	ASTM D217
Pen Worked 100000 25°C FTM	303	dmm	FED-STD-791 313
Work Stability 10% H2O (DEF) 05-50/P63	25	dmm	DEF. STAN.
Rust Test	Pass	-	ASTM D1743
Water Washout 1h 38°C	3,3	%(m)	ASTM D1264

Date 27 Apr 2023
Certified by Andreas Sigrist
Preferred contact For any queries, please contact Shell Customer Service Center in your region.

Shell Lubricants Switzerland maintains Control Systems to ensure product supplied meets required specifications. Certified to ISO 9001 / 14001 / 45001. Approved to AMS 3052, AIMS 09-06-002, IPS 09-06-002-03, MIL-PRF-23827C Type 1 and BMS 3-33C Type 1. Document is valid without a signature.
Shelf life of AeroShell Grease is 6 years from date of manufacture. The validity of shelf life is based on the container remaining unopened, undamaged and stored under shelter at constant ambient temperature condition.

For TEAM AVIATION INDIA PVT. LTD.
RAJEEV RANJAN SHARMA
MANAGER QUALITY CONTROL

GMR Aero Technic

Regulations

- [x] DGCA CAR 145
- [ ] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC03A10P		PIN	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-249037-0524 / 03817337	300.00 EA	4P0003790524	GR24/001463/0524	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING	03817337	13/05/2024	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 02 / 030809B

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1538 /Ramesh Manchu		20/05/2024
SIGNATURE		**STAMP**

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

1. Approving Civil Aviation Authority/Country: FAA/UNITED STATES	2. **AUTHORIZED RELEASE CERTIFICATE** FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	3. Form Tracking Number: 03817337

4. Organization Name and Address: THE BOEING COMPANY,737 LOGAN AVE N, RENTON,WA 98057-0000 – PC #700	5. Work Order/Contract/Invoice Number: 3609669

6. Item:	7. Description:		8. Part Number:	9. Quantity	10. Serial Number:	11. Status/Work:
000001	PIN, COTTER (SPLIT)	8008388199	BACP18BC03A10P	300 EA	N/A	NEW
000002	PIN, COTTER (SPLIT)	8008388200	BACP18BC02A06P	1000 EA	N/A	NEW

12. Remarks: 5009 1GM

SHIPMENT:3609669

SUPPLEMENTAL SHIPMENT - AIRWORTHINESS DOCUMENT

13a. Certifies the item identified above were manufactured in conformity to:

☒ Approved design data and are in a condition for safe operation.
☐ Non-approved design data specified in Block 12.

14a. ☐ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No: PC 700	14b. Authorized Signature:	14c. Approval/Certificate No:
13d. Name (Typed or Printed): LARRY W. GAC	13e. Date (dd/mmm/yyyy): 13/MAY/2024	14d. Name (Typed or Printed):	14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

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BOEING

X37065 NEW (30 JAN 2017)

LNG-103571-0517

GMR Aero Technic

STORE ACCEPTANCE TAG

Description: CLIP APPL FOR E50/B737 GRN: 07/000658/0517

Part no.: BACC15AJ2 Cond: NEW / ~~REPAIRED~~

Serial No.: — Location: 02/~~0406048~~ 0406062

Batch/ Lot No.: N/A C of C CASE No.:

Release Note No.: 09186822 Date: 25/5/17

Supplier: BOEING.

P.O. No.: 7P0003390517 Date: 25/5/17

Quantity: 100 UOM: EA

Shelf Life: — Expire On : —

Signature of Inspector: [signature] Date: 30/5/17

INSPECTOR STAMP: GAT 138

ACCEPTANCE

Regulations

- ☑ CAR 145
- ☑ EASA Part 145
- ☐ FAA Part 145
- ☐ Others (Specify)

Tag No. GAT/MS/005, Issue Initial, Rev 1, Dated 03 Nov 2015

verified, Qty - 190

1. Approving Civil Aviation Authority/Country: FAA/UNITED STATES

2.

AUTHORIZED RELEASE CERTIFICATE

FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG

3. Form Tracking Number: 09186822

4. Organization Name and Address: The Boeing Company
1901 Oakesdale Ave SW, M/C 34-02 Renton, WA 98057 2623 P.C. #700

5. Work Order/Contract/Invoice Number: GH5993

6. Item:	7. Description:		8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
0001	BOLT	CQI83U	BACB30NM4HK11	29 EA	N/A	NEW
0002	CLIP	DQI83U	BACC15AJ2	100 EA	N/A	NEW
0003	TERMINAL	EQI83U	BACT12AC48	100 EA	N/A	NEW

12. Remarks: 1GM LDC

PAGE 1 of 1

OVERBOX NO: GH5993

SUPPLEMENTAL SHIPMENT - AIRWORTHINESS APPROVAL

13a. Certifies the items identified above were manufactured in conformity to:

- [X] Approved design data and are in a condition for safe operation.
- [ ] Non-approved design data specified in Block 12.

14a. [ ] 14 CFR 43.9 Return to Service [ ] Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14 Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature: Raymond Owens

13c. Approval/Authorization No.: ODA-300064-NM

13d. Name (Typed or Printed): RAMOND OWENS

13e. Date (dd/mmm/yyyy): 25 MAY 2017

14b. Authorized Signature:

14c. Approval/Certificate No.:

14d. Name (Typed or Printed):

14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

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Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

* Duplicate Tag *

GMR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC03A08P		PIN	Expendable	Serviceable

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-006731-0619 / CSMS102808A	700.00 EA		GR08/000403/0518	OWNED

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING	N/A	26/04/2018	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S /02 /040308B

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1591/ Sunil Kumar Puli		15/11/2024
SIGNATURE		**STAMP**
		GAT 456

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

P.O. Box 025263, Miami, FL 33102-5263 • Tel: 305.925.2600 • Fax: 305.507.7191
Plant Location: 10000 N.W. 15th Terrace, Miami, FL 33172 • SITA: MIAMMCR
www.KLXaerospace.com

Shipped From: 10000 N W 15TH TERRACE, MIAMI, FL 33172

Material Certification

The items set forth on the purchase order referred to below have been visually inspected and the dimensions thereof have been measured by us, and based on the aforesaid, as well as the representation made to us by the manufacturers of the items subject of such purchase order, we hereby certify that such items are in conformity with all current governmental and manufacturer's requirements, specifications, drawings, and conform to the purchase order requirements. Said items are in new condition and have not been obtained from any U.S. Government or Military source and are traceable to KLX Aerospace Solutions.

FIRM: GMR AERO TECHNIC LIMITED

PURCHASE ORDER#: 8P0001420418

LINE#	QUANTITY	U/M	PART NUMBER	CUST REF#	LOT NUMBER	MANUFACTURER	CCODE	REV	EFF Date	EXP DATE
5	700	EA	BACP10BCD3A08F		C8MB102908A	WESTERN WIRE/FENTON	65029	D		

04/26/18

Jason Lewis
Senior Director, Global Quality

Inv # 04/26/18

GMR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BMS13-54 GRD TY3 CL1		LACING TAPE, FINISH C, WHITE, WIDTH; 0.075", THK; 0.012"	Consumable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-341164-0325 / 20102677-01-01	1.00 FT	4P0034160325	GR24/011150/0325	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
Aerospace Reliance	20102677	07/11/2024	31/10/2034	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 01/03/04/13

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1427 /Sharath Kumar Ayitha		15/03/2025
SIGNATURE		**STAMP**
		GAT 458

western filament, inc.
630 Hollingsworth St. - Grand Junction, CO 81505 - USA

CERTIFICATE OF CONFORMANCE

THIS IS TO CERTIFY THAT THE FOLLOWING PRODUCT, WHICH WE ARE DELIVERING AGAINST:

PURCHASE ORDER NUMBER: PO4000772
WORK ORDER NUMBER: 20102677 LN: 01 DL: 01
CERTIFIED ON: 11/07/2024

CONFORMS TO THE STANDARDS, SPECIFICATIONS, AND/OR DRAWINGS LISTED.

PART NUMBER:	HOF40RWHTR
DESCRIPTION:	40HOF17R - WHITE 500 YD W/TRACER
SPECIFICATION/DRAWING:	BMS 13-54 REV. H
	BMS13-54H-GD-TIII-C1-FC-S75/12-CWH-00768
	RMS: 5305100070
LOT #:	20102677-01-01
D.O.M. (MMYY):	1024
SHELF LIFE:	10 YEARS
D.O.E. (MMYY):	1034
QUANTITY:	40
U.O.M.:	EA

E. Moorey

E. MOOREY
QUALITY ASSURANCE
WESTERN FILAMENT, INC.

GMR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ..... (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
MS20995NC20		WIRE	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-324022-0125 / 43791	2.00 ROLL	4P0021991124	GR24/008718/0125	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
DIVYANSHI AVIATION SERVICES PVT LTD	15920-MAL	02/02/2024	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1427 /Sharath Kumar Ayitha		15/01/2025

SIGNATURE		STAMP
		GAT 458

MALIN CO.
Wire Specialists Since 1884.

5400 Smith Road · Brook Park, OH 44142 · Phone: 216-267-9080 · Fax: 216-267-9077 · Toll Free: 800-967-9697

CERTIFICATE OF COMPLIANCE

To: BOEING DISTRIBUTION, INC
PO#46792290001
PN#MS20995NC20MON-05-0201-1bkc
PACKING SLIP#15920-MAL
.0201 MONEL #24 1LB. RED CAN MS20995NC20

Lot: 43791
Mill Order: L0264030
Type: .0201 MONEL #24
ASTM B164

THIS IS TO CERTIFY THAT THE MATERIAL SHIPPED ON THE ABOVE PURCHASE ORDER COMPLIES WITH THE SPECIFICATIONS NOTED HEREON:

PHYSICALS:

AVERAGE SIZE	TENSILE	YIELD	ELONG % IN 10"	REDUCTION IN AREA	WRAP/BEND TEST
.02	77,762 (PSI)	35,169	34.41	79.25	PASS

CHEMISTRY: (43791)

C	Mn	Si	S	P	Cr	Ni	Mo	Cu
0.0600	1.0100	0.1800	0.0036	0.0010		64.500	0.0050	32.340

N	Co	Ti	Al	Fe	Bal
	0.0100	0.0970	0.2030	1.3900	

OTHER:

Country of Origin: FR (FRANCE)

HEAT NUMBER
77698

NOTES:

Printed: 02/02/2024

Contact: Tracy Byrne x204

Authorized Signature

J Hauck S Schall CSR

www.malinco.com

DA-True Certified Copy

GMR Aero Technic

Regulations

- [x] DGCA CAR 145
- [ ] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC01A06P		PIN	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-252094-0624 / 03868589	1000.00 EA	4P0006020624	GR24/002180/0624	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING	03868589	07/06/2024	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 02/060105A

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1538 /Ramesh Manchu		12/06/2024

SIGNATURE		STAMP
		GAT

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

1. Approving Civil Aviation Authority/Country: FAA/UNITED STATES	2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	3. Form Tracking Number: 03868589
4. Organization Name and Address: THE BOEING COMPANY,737 LOGAN AVE N, RENTON,WA 98057-0000 PC #700		5. Work Order/Contract/Invoice Number: 3658857

6. Item:	7. Description:	8. Part Number:	9. Quantity	10. Serial Number:	11. Status/Work:
000001	PIN, COTTER (SPLIT) 8008506049	BACP18BC01A06P	1000 EA	N/A	NEW

12. Remarks: 5008 1GM

Page 1 of 1

SHIPMENT:3658857

SUPPLEMENTAL SHIPMENT - AIRWORTHINESS DOCUMENT

13a. Certifies the item identified above were manufactured in conformity to: ☒ Approved design data and are in a condition for safe operation. ☐ Non-approved design data specified in Block 12.		14a. ☐ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12 Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.	
13b. Authorized Signature: LARRY W GAC	13c. Approval/Authorization No: PC 700	14b. Authorized Signature:	14c. Approval/Certificate No:
13d. Name (Typed or Printed): LARRY W GAC	13e. Date (dd/mmm/yyyy): 07/JUN/2024	14d. Name (Typed or Printed):	14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

EMASGAR Aero Technic

STORE ACCEPTANCE TAG

Description: Cotter Pin. GRN: 02/000792/0712

Part no. BACP18BC03C10P Cond: NEW / REPAIRED

Serial No.: — Location: 02/660401B

Batch/ Lot No.: — C of C CASE No.:

Release Note No.: 06573329 Date: 3/7/2012

Supplier: Boeing

P.O. No.: PO702/000277/0612 Date: 28/06/2012

Shelf Life : — Quantity: 200 UOM: EA

Signature of Inspector: Nsingh Date: 14/7/2012

MGA STAMP 024

Regulations
- [x] CAR 145
- [x] EASA Part 145
- [ ] FAR Part 145
- [ ] DCA, Malaysia

FAA FORM 8130 This document has a light blue background, "COPY" message signals a replication. 1GM DDC

AUTHORIZED RELEASE CERTIFICATE

FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG

1. Approving National Aviation Authority/Country:	2.	3. Form Tracking Number:
FAA/UNITED STATES		06573329
		5. Work Order/Contract/Invoice Number: US8783

4. Organization Name and Address: The Boeing Company, 1301 Second Avenue, Seattle, Washington 98101 USA Mail Code 34-02, Production Certificate #700

6. Item:	7. Description:		8. Part Number:	9. Eligibility:*	10. Quantity:	11. Serial/Batch Number:	12. Status/Work:
0001	COLLAR P	AKU93U	BACC30BH6	N/A	50 EA	N/A	NEW
0002	BOLT	BKU93U	BACB30NZ5K3	N/A	100 EA	N/A	NEW
0003	BOLT	FKU93U	BACB30NZ6K4Y	N/A	100 EA	N/A	NEW
0004	PIN COTT	QKU95P	BACP18BC03C10P	N/A	200 EA	N/A	NEW
0005	BOLT	XKU93T	BACB30NX5HK5	N/A	100 EA	N/A	NEW
0006	COLLAR P	ZKU93T	BACC30BH5	N/A	50 EA	N/A	NEW

13. Remarks

SUPPLEMENTAL SHIPMENT EXPORT AIRWORTHINESS APPROVAL
THIS PART(S) MEETS THE SPECIAL REQUIREMENTS OF INDIA

14. Certifies the items identified above were manufactured in conformity to:

- [X] Approved design data and are in a condition for safe operation.
- [ ] Non-approved design data specified in Block 13.

19. [ ] 14 CFR 43.9 Return to Service [ ] Other regulation specified in Block 13

Certifies that unless otherwise specified in Block 13, the work identified in Block 12 and described in Block 13 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for for return to service.

15. Authorized Signature:	16. Approval/Authorization No.: ODA-300064-NM	20. Authorized Signature:	21. Approval/Certificate No.:
17. Name (Typed or Printed): JU GEORGE	18. Date (m/d/y): Jul 03 2012	22. Name (Typed or Printed):	23. Date (m/d/y):

User/Installer Responsibilities

is important to understand that the existence of this document alone does not automatically constitute authority to install the part/component/assembly.

here the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in, Block 1, s essential that the user/installer ensures that his/her airworthiness authority accepts parts/components/assemblies from the airworthiness authority of the country specified in Block 1.

tements in Blocks 14 and 19 do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the ional regulations by the user/installer before the aircraft may be flown.

8130-3 (06-01) *Installer must cross-check eligibility with applicable technical data NSN: 0052-00-012-9005

GMR Aero Technic

STORE ACCEPTANCE TAG

Description:	PIN- SPLIT	GRN :	GR08/001912/0918
Part no :	BACP18BC03A06P	Cond :	NEW/~~REPAIRED~~
Serial No :		Location :	~~07/020/2016~~ 02/04/0205D
Batch/ Lot No :	CDMS090616A-5 / LNG-149973-0918	C of C CASE No. :	
Release Note No:	MAT CERT	Date:	27-08-2018
Supplier:	KLX AEROSPACE SOLUTIONS		
P.O. No.:	8P0007530818		
Quantity:	500	Date:	22-08-2018
		UOM:	EA
Shelf Life	Not Available	Expire On :	Not Available
Signature of Inspector:	Girinela	Date:	13/09/2018

INSPECTOR STAMP GAT 206

Acceptance

Regulations
- [x] CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others

______ (Specify)

Tag No. GAT/MS/005, Issue Initial, Rev Dated 18/08/2016

P.O. Box 025263, Miami, FL 33102-5263 • Tel: 305.925.2600 • Fax: 305.507.7191
Plant Location: 10000 N.W. 15th Terrace, Miami, FL 33172 • SITA: MIAMMCR
www.KLXaerospace.com

Shipped From: 10000 N W 15TH TERRACE, MIAMI, FL 33172

Material Certification

The items set forth on the purchase order referred to below have been visually inspected and the dimensions thereof have been measured by us, and based on the aforesaid, as well as the representation made to us by the manufacturers of the items subject of such purchase order, we hereby certify that such items are in conformity with all current governmental and manufacturer's requirements, specifications, drawings, and conform to the purchase order requirements. Said items are in new condition and have not been obtained from any U.S. Government or Military source and are traceable to KLX Aerospace Solutions.

FIRM: GMR AERO TECHNIC LIMITED

PURCHASE ORDER#: 8P0007530818

LINE#	QUANTITY	U/M	PART-NUMBER	CUST REF#	LOT-NUMBER	MANUFACTURER	CCODE	Eff Date	EXP DATE
17	500	EA	BACP18BC03A06P		CDMS090616A-5	WESTERN WIRE/FENTON	65029		

08/27/18

Jason Lewis
Senior Director, Global Quality

Inv # 08/27/18

(1)

CIVIL AIRCRAFT PARTS

Notes :

STORE ACCEPTANCE TAG

~~A10603077A~~ 02/030302A

GRN : 02/000807/0712

MASGAR Aero Technic

Description: PIN

Part no. BACP18BC02C04P Cond: NEW / ~~REPAIRED~~

Serial No.: —— Location: ~~02030302A~~ 12/060807A

Batch/ Lot No.: D59947 C of C CASE No.:

Release Note No.: SIN-2012/0585 Date: 06/07/2012

Supplier: AEROTECHNIC

P.O. No.: ~~P0772~~ / POTT/000100/1011 Date: 01/10/2011

Shelf Life : N/A Quantity: 100 UOM: EA

Signature of Inspector: Date: 16/07/2012

Regulations

- ☑ CAR 145
- ☑ EASA Part 145
- ☐ ~~FAR Part 145~~
- ☐ ~~DCA, Malaysia~~

IN MGAT 025

Tag no.: MGAT/MT/005

ACCEPTANCE

Hyderabad 500 409, Andhra Prade

India

Tel :

Fax :

E-Mail :

erotechnic

CERTIFICATE OF CONFORMITY

CHNIC ASIA PTE LTD
le Road
Bright Chambers
967 Singapore FTZ Singapore
ST Registration Num' # 20-0301385-E
)0301385E
l : +84-8384842 , Fax : +84-838483894
vw.aerotechnic-group.com

Invoice #	Date
SIN-2012/0585	06 Jul 12

Ref	MAS GMR Aero Technic Ltd. (GMR)
Requisition #	REQ-SIN003783 (Critical)
Client's PO #	POTT-000100-1011
Bill to	MAS GMR Aero Technic Limited,

Plot No 1, C/o GMR Hyderabad Aviation SEZ Limited
Rajiv Gandhi International Airport, Shamshabad,
Hyderabad 500 409, Andhra Pradesh,
India
Tel :
Fax :
E-Mail :
VAT # :

Ship to : MAS GMR Aero Technic Limited,
No 1, C/o GMR Hyderabad Aviation SEZ Limited
jiv Gandhi International Airport, Shamshabad,
derabad 500 409, Andhra Pradesh,
ia

ail :

owed up by : THAO LE THI THANH
admin2@aerotechnicasia.com

nent terms : 30 Days

Due date : 05 Aug 2012

ition # : REQ-SIN003783 -- Client's PO # : POTT-000100-1011

PART NUMBER	DESCRIPTION	QTY	UOM	COND		
HACP18RC02C04P	PIN	100.00	each	NE		

Characteristics :

- Job Number : PO-SIN007324

IRCRAFT PARTS

ORIGINAL

Asia Pte Ltd attests that the parts/materials referenced in this document are in new or overhauled or repaired condition erwise stated), and were purchased from an approved source (EASA, FAA, TC, Military Specification or Commercial his is substantiated by documentary records from the source of supply maintained on file. To the best of our knowledge, terials referenced in this document, were not obtained from, nor operated by, any United States or other government, or ce. The parts/materials have not been subject to severe stress or heat (as in a major engine failure, accident or fire) Asia Pte Ltd makes no independent representation that the parts/materials are airworthy or acceptable for installation inations are to be made by the installer based on inspection of the parts and the documentation forwarded by sia Pte Ltd.

CONDITIONS OF SALE	SIGNATURE	ABBREVIATION INDEX
es supplied are subject to Aerotechnic Asia Pte Ltd standard terms and conditions which will be supplied on request. All goods are shipped at addressee's own risk. In s damaged in transit, client must process a claim with the relevant carrier. Unless yments are to be made prompt cash, without discount. Any possible dispute will n of the Magistrate court of Singapore. the goods listed herein will remain Aerotechnic's Asia Pte Ltd ownership until lient. For low value standard parts, delivered quantities are within a tolerance of ± counts shall bear interest at prime interest rate plus two per cent (2%). These gy or software were exported from the United States in accordance with the Regulation. Diversion contrary to U.S. Law is prohibited.		AR As Removed NE Factory New NS New Surplus OH Overhaul RP Repaired SV Servi...

MASGAR Aero Technic

STORE ACCEPTANCE TAG

Description: cotter pin GRN: 445-1111

Part no. BACP18BC02C06P Cond: NEW / REPAIRED

Serial No.: — Location: ~~19/0003078~~ 02/0402090 ~~02/0402025A~~

Batch/ Lot No.: BA040811A C of C CASE No.:

Release Note No.: CofC 784096 Date: 24-10-2011

Supplier: Avio Diepen

P.O. No.: PTT-000093-0911 Date: 28/9/2011

Shelf Life: — Quantity: 1000 UOM: each

Signature of Inspector: [signature] Date: 1/12/2011

INSPECTION STAMP: MGAT 024

Regulations
- [x] CAR 145
- [x] EASA Part 145
- [ ] FAR Part 145
- [ ] DCA, Malaysia

Tag no.: MGAT/MT/005

07/20/11

CUSTOMS INVOICE/PACKING SHEET

SHIPPER: KAPCO
1 ESSEX INDUSTRIAL PARK
ESSEX, CT 06426

UPC VENDOR	INVOICE NO.	ON DOCK
000000	2873241-00	
PROMISED	REQUEST	SHIPPED
07/19/11	07/19/11	
CUSTOMER P.O.		

2873241 00

Cust#: 1265
SOLD TO: AVIO DIEPEN BV
PO BOX 566
AN ALPHEN AAN DERIJN

THE NETHERLANDS, 2400 NL

VAT# NL001512341B01

SHIP TO: Avio-Diepen, Inc.
****Cross Dock*****
561 South Parkway Suite 500

Atlanta, GA 30349 US

Pref. Routing FED EX GRD

FORWARD TO:

P.O. NUMBER	ITEM NO.	PART NUMBER	ICN No.	QTY	UOM	UNIT PRICE	TOTAL VALUE	COUNTRY OF ORIGIN	QUANTITY ORDERED	QUANTITY B.O.	QUANTITY SHIPPED
		Dwgt: DSS090110A									
		CL: UNLTD									
		INSP BY: C. BRAWN 07/20/2011									
40108701	4	BACP18BC02C06P		1000.00	EA	[redacted]	[redacted]		1000.00	0.00	1000.00
		Desc: PIN COTTER PIN									
		MAT: B81									
		License Required									
		MFR- Name: WESTERN WIRE PRODUCTS COMPANY	253742	1000.00		D/M: 2Q2011		US			
		Division: PIN									
		Dwgt: BA040811A									
		CL: UNLTD									
		INSP BY: C. BRAWN 07/20/2011									
40108701	5	BACP18BC02C08P		896.00	EA	[redacted]	[redacted]		896.00	0.00	896.00
		Desc: PIN COTTER PIN									

THIS IS A TRUE COPY CERTIFIED BY: qc101

The merchandise listed has been produced in accordance with Fair Labor Standards Act of 1938 as amended. No claims allowed unless made within ten (10) dys after receipts of Goods and in no case shall the liability assumed by us under the guarantees either expressed or implied, exceed the face value of the invoice for the merchandise in question.

TOTAL BOX VALUE:

PAGE 2

These commodities, technology or software were exported from the United States in accordance with the Export Administration Regulations. Diversion contrary to U.S. Law is prohibited.

SHIPPED UNDER NLR HS#8803.30.0010- ECCN: 9A991.D

FOR CUSTOMS PURPOSES ONLY. DO NOT PAY FROM THIS DOCUMENT.

KAPCO

CERTIFICATE OF CONFORMANCE

WE HEREBY CERTIFY THAT THE PRODUCT SUPPLIED IS NEWLY MANUFACTURED, CONFORMS TO THE APPROVED DESIGN DATA, AND MEETS ALL REQUIREMENTS OF THE APPLICABLE PURCHASE ORDER. EVIDENCE OF CONFORMANCE IS ON FILE AND AVAILABLE FOR REVIEW UPON REQUEST.

TIM GILL
CORPORATE VICE PRESIDENT OF QUALITY

Thank You For This Order

PACKING LIST