


flydubai Dubai	flydubai	W/O 2921939 Working Copy	Barcode  WO2921939	Registration A6FET 73NG 738
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Type S SCHED	Origin V Replacement Event	ATA 32 LANDING GEAR	Position NLG NOSE LANDING GEAR	Zone	Area
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Partnumber 162A1100-14	Description NLG COMPONENT INSTL	Serialnumber GK15668Y5241
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Type	Reference	Description
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Due Date 19.Jan.2025	Due at TAH	Due at TAC 18000
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Description Step 1	Udit Bansal (F204850), 09.Jun.2022	Action Step 1-1
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REPLACE PN:162A1100-14 SN:GK15668Y5241 (NLG COMPONENT INSTL)

REPLACE PN:162A1100-14 SN:GK15668Y5241 (NLG COMPONENT INSTL)
 INSTALLED AT: NLG(NOSE LANDING GEAR)
 INFO/TRIGGERED BY: 162A1120-8/ZKM1956 RS(RESTORE) RESTORE / 162A1120-8 RESTORE

RECORD THE FOLLOWING:
 OFF : P/N: _____ S/N: _____
 ON : P/N: _____ S/N: _____

REPLACED NLG AS PER MPD 32-090-00-01
 REV. 41 DT- OCT. 17, 2024
 P/N:162A1100-14
 SN: GK15668Y5241 (NLG COMPONENT INSTL)

INSTALLED AT: NLG (NOSE LANDING GEAR)

RECORDED THE FOLLOWING:
 OFF: P/N: 162A1100-14
 S/N: GK15668Y5241

ON: P/N: 162A1100-14
 S/N: GK15614Y5187

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 02/01/2025

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Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Description Step 2	Munal Rao (F204849), 28.Oct.2024	Action Step 2-1
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NLG STEERING VALVE ASSY REPLACEMENT ALONG WITH NLG

CARRY OUT REPLACEMENT OF NLG STEERING VALVE ASSY P/N: 383900-1011. REFER AMM 32-51-11.

RECORD THE FOLLOWING:
 OFF : P/N: _____ S/N: _____
 ON : P/N: _____ S/N: _____

CARRIED OUT REPLACEMENT OF NLG STEERING VALVE ASSY

P/N: 383900-1011. AS PER AMM 32-51-11.REV 85 DTD 15 OCT 2024

RECORDED THE FOLLOWING:
 OFF: P/N: 383900-1011
 S/N: 5860A

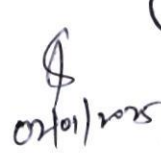



ON: P/N: 383900-1011
 S/N: 5819A

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flydubai	flydubai	W/O	Barcode	Registration
	Dubai	2921939 Working Copy	 W02921939	A6FET 73NG 738

Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Description Step 3	Munal Rao (FZ04849), 28.Oct.2024	Action Step 3-1
<p>NLG TAXI LIGHT REPLACEMENT ALONG WITH NLG</p> <p>CARRY OUT REPLACEMENT OF NLG TAXI LIGHT ASSY P/N: 50-0199-13. REFER AMM TASK 32-21-00-000-801. RECORD THE FOLLOWING:</p> <p>OFF : P/N: _____ S/N: _____</p> <p>ON: P/N: _____ S/N: _____</p> <p>NOTE P/N: 50-0199-15 CANNOT BE INSTALLED AS A REPLACEMENT.</p>		
<p>CARRIED OUT REPLACEMENT OF NLG TAXI LIGHT ASSY</p> <p>P/N: 50-0199-13. AS PER AMM TASK 32-21-00-000-801 REV 85 DATED 15 OCT 2024.</p> <p>RECORDED THE FOLLOWING:</p> <p>OFF: P/N: 50-0199-13 S/N: 6880</p> <p>ON: P/N: 50-0199-13 S/N: 6847</p> <p>NOTE: NOTED.</p> <p><i>02/10/2025</i>  </p> <p><i>02/10/2025</i>  </p>		

Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Description Step 4	Munal Rao (FZ04849), 28.Oct.2024	Action Step 4-1
<p>NLG RETRACT ACTUATOR REPLACEMENT ALONG WITH NLG</p> <p>CARRY OUT REPLACEMENT OF NLG RETRACT ACTUATOR P/N: 273A1101-2. REFER AMM 32-33-11.</p> <p>RECORD THE FOLLOWING:</p> <p>OFF : P/N: _____ S/N: _____</p> <p>ON: P/N: _____ S/N: _____</p>		
<p>CARRIED OUT REPLACEMENT OF NLG RETRACT ACTUATOR</p> <p>P/N: 273A1101-2 AS PER AMM 32-33-11 REV 85 DATED 15 OCT 2024.</p> <p>RECORDED THE FOLLOWING:</p> <p>OFF: P/N: 273A1101-2 S/N: 1101/5701 ON: P/N: 273A1101-2 S/N: 1101/5647</p> <p><i>02/10/2025</i>  </p>		

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Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Description Step 5 Munal Rao (F204849), 28.Oct.2024 Action Step 5-1

LH STEERING ACTUATOR REPLACEMENT ALONG WITH NLG

CARRY OUT REPLACEMENT OF THE FOLLOWING NLG STEERING ACTUATORS AS PER AMM 32-51-51.

1) NLG STEERING ACTUATOR ASSY LH P/N : 275A1101-9.


RECORD THE FOLLOWING:
 OFF : P/N: _____ S/N: _____
 ON : P/N: _____ S/N: _____


CARRIED OUT REPLACEMENT OF THE FOLLOWING NLG STEERING ACTUATORS AS PER AMM 32-51-51 REV 85 DATED 15 OCT 2024.

1) NLG STEERING ACTUATOR ASSY LH P/N : 275A1101-9.

RECORDED THE FOLLOWING:
 OFF: P/N: 275A1101-9
 S/N: TASV11093

ON: P/N: 275A1101-9
 S/N: TASV11075


02/10/2025



Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Description Step 6 Munal Rao (F204849), 28.Oct.2024 Action Step 6-1

RH STEERING ACTUATOR REPLACEMENT ALONG WITH NLG

CARRY OUT REPLACEMENT OF THE FOLLOWING NLG STEERING ACTUATORS AS PER AMM 32-51-51.

1) RH NLG STEERING ACTUATOR ASSY RH P/N : 275A1101-10.


RECORD THE FOLLOWING:
 OFF : P/N: _____ S/N: _____
 ON : P/N: _____ S/N: _____


CARRIED OUT REPLACEMENT OF THE FOLLOWING NLG STEERING ACTUATORS AS PER AMM 32-51-51 REV 85 DTD 15 OCT 2024

1) RH NLG STEERING ACTUATOR ASSY RH P/N: 275A1101-10.

RECORDED THE FOLLOWING:
 OFF: P/N: 275A1101-10
 S/N: TASV11080



ON: P/N: 275A1101-10
 S/N: TASV8326.


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Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate


Description Step 7		Munal Rao (FZ04849), 28.Oct.2024	Action Step 7-1
FIRST INSPECTION OF THE NLG STEERING ACTUATOR INST. CARRY OUT FIRST INSPECTION FIRST INSPECTION OF THE NLG STEERING ACTUATOR INSTALLATION. DATE: _____ TIME OF INSPECTION : _____ UTC		CARRIED OUT THE FIRST INSPECTION OF THE NLG STEERING ACTUATOR INSTALLATION. DATE: 02/01/2025 TIME ON INSP: 0845 UTC   02/01/2025	

Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

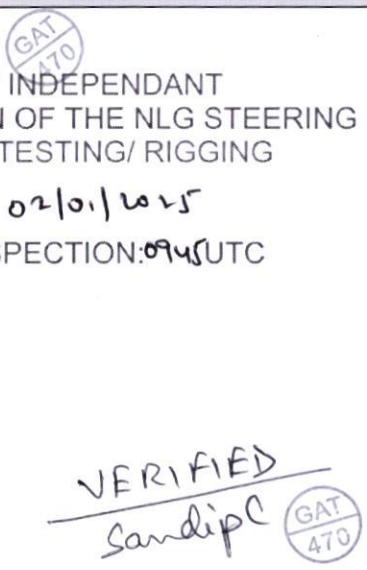
Description Step 8		Munal Rao (FZ04849), 28.Oct.2024	Action Step 8-1
INDEPENDENT INSPECTION OF THE NLG STEERING ACTUATOR INST. CARRY OUT INDEPENDENT INSPECTION OF THE NLG STEERING ACTUATOR INSTALLATION. DATE: _____ TIME OF INSPECTION : _____ UTC		CARRIED OUT INDEPENDENT INSPECTION OF THE NLG STEERING ACTUATOR INSTALLATION. DATE: 02/01/2025 TIME OF INSPECTION: 0900 UTC VERIFIED <u>Sandip l</u> 	

flydubai	flydubai	W/O	Barcode	Registration
	Dubai	2921939	 W02921939	A6FET 73NG 738

Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

Description Step 9		Munal Rao (FZ04849), 28.Oct.2024	Action Step 9-1
FIRST INSPECTION OF THE NLG STEERING ACTUATOR TESTING/RIGGING			
<p>CARRY OUT FIRST INSPECTION OF THE NLG STEERING ACTUATOR TESTING/ RIGGING .</p> <p>DATE: _____</p> <p>TIME OF INSPECTION : _____ UTC</p>		<p>CARRIED OUT FIRST INSPECTION OF THE NLG STEERING ACTUATOR TESTING/ RIGGING</p> <p>DATE: 02/01/2025</p> <p>TIME OF INSPECTION: 0930 UTC</p>	
			



Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate


Description Step 10		Munal Rao (FZ04849), 28.Oct.2024	Action Step 10-1
INDEPENDANT INSPECTION OF THE NLG STEERING ACTUATOR TESTING/RIGGING			
<p>CARRY OUT INDEPENDANT INSPECTION OF THE NLG STEERING ACTUATOR TESTING/ RIGGING .</p> <p>DATE: _____</p> <p>TIME OF INSPECTION : _____ UTC</p>		<p>CARRY OUT INDEPENDANT INSPECTION OF THE NLG STEERING ACTUATOR TESTING/ RIGGING</p> <p>DATE: 02/01/2025</p> <p>TIME OF INSPECTION: 0945UTC</p>	
			

flydubai	flydubai	W/O	Barcode	Registration
	Dubai	2921939 Working Copy	 WO2921939	A6FET 73NG 738

Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate

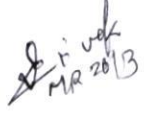

Workorder Transfer						
Flight Hours	Flight Cycles	Days	Work deferred in accordance with	Limitation / Downgrades	Date	
Due at TAH	Due at TAC	Due Date				Stamp / Sign
	18000	19.Jan.2025				

Released To Service	
Certifies that the work specified except as otherwise specified was carried out in accordance with CAR 145 and in respect to that work the aircraft /aircraft component is considered ready for release to service	Date: 02/01/2025 Stamp / Sign:  

DOCUMENT CHECKED BY: PLANNER MR No: MR1832 SIGN (INITIALS):-  DATE:- 07-01-25

Task Card no: 32-090-00 / ITEM-1	Task Card Title: RESTORE THE NOSE LANDING GEAR ASSEMBLY.	A/C Area: N/A
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Part Number: 162A1100-14	Serial Number: GK15668Y5241	Description: NLG COMPONENT INSTL	Pos: NLG
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All system components and attachments (CMM 32-21-16) NOTE: TASK COMPONENT RELATED	 MR 2013	 02/01/2025
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TASKCARD FINDING(Tick applicable Box): YES NO
 (Please enter Taskcard Reference also on Workorder)
 FINDING WORKORDER NO. NIL

If Job Stopped or interrupted :
 Task Card Performed Up to Page : _____ Step: _____
 Remarks:

If Job Stopped or interrupted :
 Task Card Performed Up to Page : _____ Step: _____
 Remarks:

Manhours used to perform taskcard : _____


Station GARAGE/H-10	Date 02/01/2025	Accomplished 	
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CAT C Verification 	
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FEELER GAUGE
 P/N # 184-3135
 S/N # 2208 225510
 DUE # 22/08/2025.

DOCUMENT CHECKED BY:
PLANNER MR No:- MR1832
SIGN (INITIALS):- h
DATE:- 02-01-25

CR-2

 737-600/700/800/900	32-090-00-01 Version: 41	Page 1 of 84
	Restore Nose Landing Gear	
Type: Routine Card	ATA: 32--	Flow: -
		Work Area: -

EVENT DETAILS

Tail #	Work Order	Base	Start	End
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REQUIREMENT 32-090-00

PERIODICITY

Version	Threshold	Interval	Limit	Threshold Date	Limit Date
1.1	18000 FC	18000 FC			
1.2	10 YR	10 YR			

Whichever comes first

SOURCES

Type	Reference	Title	Version	Version Date	Originator	Mandatory
MPD	32-090-00				BOEING	true
MRB	D626A001-MRBR		5	Mar2003	BOEING	true

INTERVAL NOTES: Whichever comes first.

AIRPLANE NOTES:

ENGINE NOTES:

ACCESS NOTES:

SPECIAL NOTES:

ENGINEERING COMMENTS:

ZONES: 000, 113, 114, 115, 116, 710, 713, 730, 740
 ACCESS: 711AL, 712AR, 113AC, 113AW, 113BW, 114AC, 114AW, 114BW
 SKILLS:
 CERTIFICATIONS:
 CONDITIONS:

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02/01/25

ANY FINDINGS	
YES	<input checked="" type="checkbox"/> NO
IRC No. : <u>NIL</u>	

PARTS

Operator Part No	Specification	Qty	Unit of Measure	Type	Description	Manufacturer
C00308	MIL-C-11796	1		Consumable	Compound - Corrosion Preventive, Petrolatum Hot Application	81205
C00528	MIL-C-11796 CLASS III	1		Consumable	Compound - Corrosion Preventive, Petroleum Hot Application (Soft Film)	81205

Rev. # 41



Rev Date: Oct 17, 2024 PDT

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24/01/2024

S. L. M. 1957

GCAA APPROVAL No : UAE.145.1232

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02/01/2025

flydubai

737-600/700/800/900

32-090-00-01 Version: 41

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Restore Nose Landing Gear

Type:	Routine Card	ATA:	32--	Flow:-	Work Area:-
D00013	MIL-PRF-23827 1 (NATO G-354) (SUPERSEDE S MIL- G-23827)			Consumable	Grease - Aircraft And Instrument Grease 81205
D00013	MIL-PRF-23827 1 (NATO G-354) (S			Consumable	Grease - Aircraft And Instrument Grease 81205
D00153	BMS3-11 TYPE 1 IV			Consumable	Fluid - Hydraulic Fluid, Fire Resistant 81205
D00633	BMS3-33 1			Consumable	Grease - Aircraft General Purpose 81205
G00018	A-A-59503 1 TYPE I GRADE B, MIL- PRF-27401 TYPE I GRADE A			Consumable	Nitrogen - Gaseous, Pressurizing, 99.5 Percent Pure 81205
G00018	A-A-59503 1 TYPE I GRADE B, MIL			Consumable	Nitrogen - Gaseous, Pressurizing, 99.5 Percent Pure 81205
G50136	BMS3-38 1			Consumable	Compound - Corrosion Inhibiting, Non-drying 81205
G50225	NASM20995 1			Consumable	Lockwire - MS20995C20, Corrosion Resistant Steel - 0.020 Inch (0.508 mm) Diameter Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L 81205
G50237	BMS3-38 1			Consumable	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L 81205

TOOLS

Operator Part No	Mfr Part No	Qty	Description	Manufacturer
COM-4938	-	1	Insertion/Removal - contact (size 16, backshell side) (Part #: M81969/1-03, Supplier: 11139, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -ALL) (Part #: M81969/14-03, Supplier: 11139, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -ALL) (Part #: MS3447-20, Supplier: 58960, A/P Effectivity: 737-ALL)	81205
SPL-10305	-	1	Outrigger Equipment - Nose Landing Gear Installation/Removal	81205

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Handwritten signatures and dates: 21/12/2024, MR1457

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____
Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

Restore Nose Landing Gear

Type: Routine Card ATA: 32-- Flow: - Work Area: -

			(Part #: C32049-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	
SPL-14475	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-6 NLG Trunnion Pin Puller Assy is included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14476	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-20 Small Slide Hammer with CG240-8 Side Screw are included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14477	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-2 NLG Alignment Slugs are included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14478	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-5 Drag Strut Trunnion Pin Puller Assy is included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14479	-	1	R/I Equipment - Trunnion Pin, NLG (Note: CG240-9 Slide Hammer and CG240-8 Slide Screw are included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14480	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-4 Drag Strut Alignment Slug is included in C32033-1 kit) JXB ALL; 737-800 (Part #: C32033-1, Supplier: 81205)	81205
SPL-14481	-	1	R/I Equipment - Trunnion Pin, NLG (Note: C32033-3 Alignment Pin Assy is included in C32033-1 kit)	81205
SPL-1521	-	1	Tool - Strut Inflation, Landing Gear (Part #: F70200-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1559	-	1	Adjustable Spanner Wrench (2.00 to 4.00 Inch Dia. Retainer) JXB ALL; 737-800 (Part #: F72959-34, Supplier: 81205) (Part #: F72959-35, Supplier: 81205) (Part #: F72959-36, Supplier: 81205) (Part #: F72959-41, Supplier: 81205) (Opt Part #: F72959-5, Supplier: 81205) (Opt Part #: F72959-6, Supplier: 81205) (Opt Part #: F72959-7, Supplier: 81205)	81205

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MR1957

[Signature]
24/12/2024
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PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



Restore Nose Landing Gear

Type: Routine Card ATA: 32-- Flow: - Work Area: -

(Opt Part #: F72959-8, Supplier: 81205)			
SPL-1821	-	1	Wrench - Spanner, Nose Landing Gear Trunnion Pin Retainer Nut 81205
(Part #: F72959-40, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)			
SPL-1866	-	1	Expander Set - Spring, Main Landing Gear 81205
(Part #: C32014-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)			
SPL-1871	-	1	Strap - Retention, NLG/MLG Inner Cylinder 81205
(Part #: C32030-10, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)			
STD-6545	-	1	Rope - Nylon 81205

REFERENCES

Reference Number	Type	Title	Version Number	Version Date
07-11-01-580-815	AMM:task	Lift the Airplane with the Jacks	-	-
07-11-01-580-816	AMM:task	Lower the Airplane Off the Jacks	-	-
07-11-03-201	AMM:pgblk	JACK AIRPLANE AXLES - MAINTENANCE PRACTICES	-	-
07-11-03-580-802	AMM:task	Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E	-	-
07-11-21-580-801	AMM:task	Lift the Airplane Nose with the Nose Jack at Jack Point D	-	-
07-11-21-580-802	AMM:task	Lower the Airplane Nose Off of the Jack	-	-
10-11-05-201	AMM:pgblk	CHOCK INSTALLATION	-	-
12-12-00-610-801	AMM:task	Hydraulic Reservoir Servicing	-	-
12-15-41-610-801	AMM:task	Nose Landing Gear Shock Strut Fluid Check	-	-
12-15-41-610-802	AMM:task	Nose Landing Gear Shock Strut Servicing	-	-
12-15-41-610-805	AMM:task	Nose Landing Gear Shock Strut Servicing, Airplane on Jacks	-	-
12-21-21-640-801	AMM:task	Nose Landing Gear Upper End Components Servicing	-	-
12-21-21-640-802	AMM:task	Nose Landing Gear Lower End Components Servicing	-	-
29-09-00-860-802	AMM:task	Hydraulic Reservoirs Depressurization	-	-
29-11-00-860-801	AMM:task	Hydraulic System A or B Pressurization	-	-
29-11-00-860-805	AMM:task	Hydraulic System A or B Power Removal	-	-

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S. W. H. 1952

[Signature]
24/11/2020

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____
Item: _____ Completed through item: _____ Sign: _____




737-600/700/800/900

Restore Nose Landing Gear

Type:Routine Card	ATA:32--	Flow:-	Work Area:-
32-00-01-080-801	AMM:task	Landing Gear Downlock Pins Removal	- -
32-00-01-480-801	AMM:task	Landing Gear Downlock Pins Installation	- -
32-33-00-710-801	AMM:task	Operational Test for the Nose Landing Gear	- -
32-33-00-710-802	AMM:task	Nose Landing Gear Test - Component Replacement	- -
32-33-52-000-801	AMM:task	Nose Gear Lock Spring Removal	- -
32-33-52-400-801	AMM:task	Nose Gear Lock Spring Installation	- -
32-35-00-730-801	AMM:task	Nose Gear Manual Extension System Test - Airplane on Jacks	- -
32-45-21-000-801	AMM:task	Nose Landing Gear Wheel and Tire Assembly Removal	- -
32-45-21-400-801	AMM:task	Nose Landing Gear Wheel and Tire Assembly Installation	- -
32-51-00-700-801	AMM:task	Nose Wheel Steering System Test	- -
32-51-00-820-802	AMM:task	Nose Wheel Steering System Adjustment	- -
32-51-31-000-802	AMM:task	Removal of the Steering System Cables for the Nose Gear	- -
32-51-31-400-802	AMM:task	Installation of the Steering System Cables for the Nose Gear	- -
32-61-41-400-801	AMM:task	Nose Landing Gear Lock Sensor Installation	- -
32-61-51-400-801	AMM:task	Nose Landing Gear Down Position Sensor Clearance Adjustment	- -
33-45-01-000-801	AMM:task	Taxi Light Housing Assembly - Removal	- -
33-45-01-400-801	AMM:task	Taxi Light Housing Assembly - Installation	- -
53-14-01-020-801	AMM:task	Nose Wheel Well Access Panels - Removal	- -
53-14-01-420-801	AMM:task	Nose Wheel Well Access Panels - Installation	- -

NOTES: Restore the nose landing gear assembly.

Handwritten signature and date: 24/11/2024

Handwritten signature and date: S. Singh 24/11/2024

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

JXB ALL

TASK 32-21-00-000-801

1. Nose Landing Gear - Removal (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)
07-11-03 P/B 201	JACK AIRPLANE AXLES - MAINTENANCE PRACTICES
07-11-03-580-802	Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E (P/B 201)
07-11-21-580-801	Lift the Airplane Nose with the Nose Jack at Jack Point D (P/B 201)
10-11-05 P/B 201	CHOCK INSTALLATION
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-45-21-000-801	Nose Landing Gear Wheel and Tire Assembly - Removal (P/B 401)
32-51-31-000-802	Nose Gear Steering System Cables Removal (P/B 401)
33-45-01-000-801	Taxi Light Housing Assembly - Removal (P/B 201)
53-14-01-020-801	Nose Wheel Well Access Panels - Removal (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
COM-4938	Insertion/Removal Tool - Contact (size 16, backshell side) 737-800 (Part #: M81969/1-03, Supplier: 11851) (Part #: M81969/14-03, Supplier: 11139)
SPL-1559	Adjustable Spanner Wrench (2.00 to 4.00 Inch Dia. Retainer) 737-800 (Part #: F72959-34, Supplier: 81205) (Part #: F72959-35, Supplier: 81205)

MECH	INSP
S.L	24/11/24
S.L	24/11/24
S.L	24/11/24
S.L	24/11/24

GAT 171

GAT 171

GAT 171

GAT 171

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card	ATA: 32--	Flow: -	Work Area: -
			MECH INSP
	(Part #: F72959-36, Supplier: 81205)		<div style="border-left: 1px solid black; border-right: 1px solid black; height: 100%; position: relative;"> } <div style="position: absolute; top: 10%; left: 10%; font-size: 1.2em;"> S.M. 24/12/24 </div> <div style="position: absolute; top: 45%; left: 10%; border: 1px solid black; border-radius: 50%; padding: 2px;"> GAT 171 </div> </div>
	(Part #: F72959-41, Supplier: 81205)		
	(Opt Part #: F72959-5, Supplier: 81205)		
	(Opt Part #: F72959-6, Supplier: 81205)		
	(Opt Part #: F72959-7, Supplier: 81205)		
	(Opt Part #: F72959-8, Supplier: 81205)		
SPL-1821	Adjustable Spanner Wrench (0.75 to 2.00 Inch Dia. Retainer, 0.12 x 0.12 Key Arm)		
	737-800		
	(Part #: F72959-40, Supplier: 81205)		
	(Opt Part #: F72959-4, Supplier: 81205)		
SPL-1871	Strap - Retention, NLG/MLG Inner Cylinder		
	737-800		
	(Part #: C32030 -31, Supplier: 81205)		
	(Opt Part #: C32030-10, Supplier: 81205)		
SPL-10305	Outtrigger Equipment - Nose Landing Gear Installation/ Removal		
	737-800		
	(Part #: C32049-1, Supplier: 81205)		
SPL-14475	NLG Trunnion Pin Puller Assy, C32033-6 (included in C32033 Eqpt)		
	737-800		
	(Part #: C32033-1, Supplier: 81205)		
SPL-14476	Small Slide Hammer, C32033-20 (included in C32033 Eqpt)		
	737-800		
	(Part #: C32033-1, Supplier: 81205)		
STD-6545	Rope - Nylon		
C. Location Zones			
Zone	Area		
115	Nose Landing Gear Wheel Well - Left		
116	Nose Landing Gear Wheel Well - Right		
713	Nose Landing Gear		

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card ATA: 32-- Flow: - Work Area: -

	MECH	INSP
D 15 (C01401) LANDING GEAR AIR/GND RELAY	S.L. MR 1952	24/12/24
D 16 (C01432) LANDING GEAR ALTN EXTEND SOL		
D 17 (C01027) LANDING GEAR NOSE GEAR STEER		
D 18 (C00451) LANDING GEAR AURAL WARN		

SUBTASK 32-21-00-020-002

- (5) Disconnect the NWS-A/B cables [49] (TASK 32-51-31-000-802).
 - (a) Install the rig pin NS2 in the drum for the Captain's control wheel.
 - (b) To get access to the cable turnbuckle [1], do this step:
 - 1) Open this access panel: (TASK 53-14-01-020-801)

Number Name/Location

113BW Forward Nose Wheel Well Panel

- (c) Remove the locking clips [2] for the cable turnbuckles [1] (View B, Figure 401).
- (d) Loosen the tension at the cable turnbuckles [1].
- (e) Disconnect the NWS-A/B cables [49] at the cable turnbuckles [1].
- (f) Remove the bolts [88], washers [89], and pulleys [87] from the bracket that is outboard of the left trunnion pin [63].
 - 1) Remove the NWS-A/B cables [49] from the guard [90] and grooves of the pulleys [87].
- (g) Attach the identification tags to the NWS-A/B cables [49].

SUBTASK 32-21-00-020-003

- (6) Do this task to disconnect the flexible conduits [73] (View C, Figure 401).
 - (a) Remove the screws [10] and washers [3] from the junction box cover [9].
 - (b) Remove the junction box cover [9] from the junction box [5].
 - (c) Disconnect the terminal block [72] from the junction box [5].
 - 1) Pull the terminal block [72] out to get access to the wires during the removal.
 - (d) Identify the wires in the terminal block [72] that go to the landing gear and label them or make a drawing to be used during installation.
 - (e) Use a contact insertion / removal tool, COM-4938, tool to remove the wires from the terminal block [72] inside the junction box [5].
 - (f) Attach a string to each wire removed from the terminal block [72].
 - 1) Make sure that the length of string is sufficient to go through the rigid conduit [4] to point "A".
 - (g) If it is necessary, remove the lockwire from the flexible conduits [73].
 - (h) Disconnect the flexible conduits [73] at point "A" and pull the wires through.
 - 1) Do not pull the string all the way through the conduit.

NOTE: The string is necessary to pull the wires back through the rigid conduit on installation.

- (i) Disconnect the string from the wires.

SUBTASK 32-21-00-020-021

- (7) Do this task to disconnect the flexible conduit [80] (View L, Figure 401).
 - (a) Remove the screws [85] and washers [84] from the junction box cover [86].

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



Handwritten notes and stamps on the right side of the page. Includes 'S.L. MR 1952' written vertically, and several circular stamps with 'GAT' and numbers '171' and '128'. There are also handwritten dates like '24/12/24' and '24/12/23'.

737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

- (b) Remove the junction box cover [86] from the junction box [82].
 - (c) Disconnect the terminal block [81] from the junction box [82].
 - 1) Pull the terminal block [81] out to get access to the wires during the removal.
 - (d) Identify the wires in the terminal block [81] that go to the landing gear and label them or make a drawing to be used during installation.
 - (e) Use a contact insertion / removal tool, COM-4938, tool to remove the wires from the terminal block [81] inside the junction box [82].
 - (f) Attach a string to each wire removed from the terminal block [81].
 - 1) Make sure that the length of string is sufficient to go through the rigid conduit [83] to the flexible conduit [80].
 - (g) If it is necessary, remove the lockwire from the flexible conduit [80].
 - (h) Disconnect the flexible conduit [80] from the rigid conduit [83] and pull the wires through.
 - 1) Do not pull the string all the way through the conduit.
- NOTE: The string is necessary to pull the wires back through the rigid conduit on installation.**
- (i) Disconnect the string from the wires.

MECH INSP

[Handwritten signatures and dates in the MECH/INSP columns]

24/12/24

24/12/24

24/12/24

GAT 128

GAT 128

GAT 128

JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053

SUBTASK 32-21-00-000-001

- (8) If it is necessary, remove the taxi light, do this task: Taxi Light Housing Assembly - Removal, TASK 33-45-01-000-801.

JXB ALL

F. Nose Landing Gear Removal

SUBTASK 32-21-00-020-015

- (1) Disconnect the hydraulic lines [25] (View E, Figure 401).
 - (a) Loosen the swivel nuts [77] to disconnect the hydraulic lines [25] from the swivel [24].
 - (b) Put caps on the hydraulic lines [25].
 - (c) Install plugs on each of the hydraulic ports of the swivel [24].
 - (d) Put tags to identify the swivel ports and hydraulic lines [25] that will prevent the crossing of the lines during the installation.

SUBTASK 32-21-00-020-006

- (2) Disconnect the right upper rod assembly [26] (View F, Figure 401).
 - (a) Remove the pin [31], nut [32], and washer [30].
 - (b) Remove the bolt [27], washer [28], and bushing [29] from the clevis of the right trunnion [23].
 - (c) Use a rope to hold the upper rod assembly [26] away from the work area.

SUBTASK 32-21-00-020-007

- (3) Disconnect the left upper rod assembly [41] (View H, Figure 401).
 - (a) Remove the pin [43], nut [42], and washer [44].
 - (b) Remove the bolt [47], washer [46], and bushing [45] from the clevis of the left trunnion [40].
 - (c) Use a rope to hold the upper rod assembly [41] away from the work area.

SUBTASK 32-21-00-020-008

[Handwritten signatures and dates in the MECH/INSP columns]

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24/11/24

24/11/24

GAT 171

GAT 171

GAT 171

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

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737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

- (4) Remove each pulley [50] from the bracket [51] (View I, Figure 401).
 - (a) Remove the nut [52] and washer [53] from the bolt [47].
 - (b) Remove the bolt [47] and washer [48] that hold the pulley [50] and cable guard [54] from the bracket [51].
 - (c) Remove the washer [48] from the bolt [47].
 - (d) Remove the pulley [50] and cable guard [54] from the bracket [51].
 - (e) Remove the NWS-A/B cables [49] from the groove of the pulley [50].
- SUBTASK 32-21-00-020-009
- (5) Remove the seal [65] (View J, Figure 401).
 - (a) Remove the lockwire from the screw [58] and lock [57].
 - (b) Remove the screw [58] that holds the lock [57] to the left trunnion pin [63].
 - (c) Remove the lock [57] from the left trunnion pin [63].
 - (d) Use the spanner wrench, SPL-1821 (face lug adapter), or spanner wrench set, SPL-1559, to remove the trunnion pin nut [68] from the left trunnion pin [63].
 - (e) Pull the seal ring [69] and seal [65] from the left trunnion pin [63].
 - 1) If it is difficult to pull the seal [65], hold the eyebolt [67] and pull the seal [65] out of the left trunnion pin [63].
 - (f) Remove the nut [64], washer [56], and inboard retainer [66] from the eyebolt [67].
 - (g) Remove the eyebolt [67] from the left trunnion pin [63].
 - (h) Remove the outboard retainer [66] from the eyebolt [67].
 - (i) Isolate the segments of the seal [65] from the NWS-A/B cables [49].
 - (j) Pull the NWS-A/B cables [49] from the trunnion pin nut [68], seal ring [69], and left trunnion pin [63].
 - (k) Attach identification tags to the NWS-A/B cables [49].
 - (l) Wind the NWS-A/B cables [49] into a coil and attach it to the trunnion of the gear with a tape.

SUBTASK 32-21-00-020-010

WARNING: DO NOT REMOVE THE VALVE BODY UNTIL YOU DEFLATE THE SHOCK STRUT FULLY. THE AIR PRESSURE CAN BLOW THE VALVE BODY OUT AND CAUSE INJURIES TO PERSONNEL.

- (6) Deflate the shock strut [17] (View D, Figure 401).
 - (a) Remove the dust cap [14] for the gas charging valve [16].
 - (b) Loosen the swivel nut [15] a maximum of two turns.
 - (c) Deflate the shock strut slowly to prevent the leakage of the fluid through the gas valve.
 - NOTE: Fluid in the shock strut will have bubbles when the pressure is released.**
 - (d) Loosen the swivel nut [15] fully when all of the pressure in the shock strut [17] is released.

SUBTASK 32-21-00-480-007

- (7) Install the retention strap, SPL-1871, with the marker facing up to hold the shock strut [17] in its compressed position.

SUBTASK 32-21-00-580-002

- (8) Do this task: Lift the Airplane Nose with the Nose Jack at Jack Point D, TASK 07-11-21-580-801.

MECH	INSP
S.H.	Q. Subh
S.H.	Q. Subh
S.H.	Q. Subh
S.H.	Q. Subh
S.H.	Q. Subh
S.H.	Q. Subh
S.H.	Q. Subh
S.H.	Q. Subh
S.H.	Q. Subh
S.H.	Q. Subh



PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

(a) Make sure that the chocks are installed at the main landing gear after the nose landing gear is lifted.

SUBTASK 32-21-00-580-005

(9) If it is necessary, do this task: Lift the Airplane with the Jacks, TASK 07-11-01-580-815.

SUBTASK 32-21-00-020-017

(10) Do this task: Nose Landing Gear Wheel and Tire Assembly - Removal, TASK 32-45-21-000-801.

SUBTASK 32-21-00-020-016

(11) Disconnect the lower drag strut assembly [19] (View D, Figure 401).

(a) Remove the pin [11], nut [22], washer [21], and bolt [13] from the pin [18].

(b) Remove the nut [12] and washer [20] from the pin [18].

(c) Hold the lower drag strut assembly [19] in position and remove the pin [18].

(d) Use a STD-6545 nylon rope, to hold the lower drag strut assembly [19] away from the work area.

NOTE: When the pin is removed the lower drag strut will move forward.

SUBTASK 32-21-00-580-003

(12) If it is necessary to lift the gear with an axle jack to reduce the weight of the gear when the left trunnion pin [63] and right trunnion pin [37] are removed, do this task: Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E, TASK 07-11-03-580-802.

SUBTASK 32-21-00-020-011

(13) Remove the right trunnion pin [37] (View G, Figure 401).

(a) Remove the pin [35], nut [36], washer [34], and bolt [33].

(b) Remove the swivel [24] from the right trunnion pin [37].

(c) Use the NLG Trunnion Pin puller, C32033-6, SPL-14475, and small slide hammer, C32033-20, SPL-14476, to remove the right trunnion pin [37] from the right trunnion [23].

1) Make sure that the spacer [39] that is between the sidewall of the nose wheel well and right trunnion [23] is caught.

SUBTASK 32-21-00-020-012

(14) Remove the left trunnion pin [63] (View J, Figure 401).

(a) Remove the pin [59], nut [60], washer [61], and bolt [62].

(b) Use the trunnion NLG Trunnion Pin puller, C32033-6, SPL-14475, and small slide hammer, C32033-20, SPL-14476, to remove the left trunnion pin [63] from the left trunnion [40].

1) Make sure that the spacer [71] that is between the sidewall of the nose wheel well and left trunnion [40] is caught (View K, Figure 401).

(c) Remove the cable guard [70] from the left trunnion pin [63].

SUBTASK 32-21-00-860-005

(15) Measure the thickness of the spacer [39] and spacer [71] that were between the sidewalls of the wheel well and nose landing gear trunnion.

(a) Record the measurement of the washer stack for use when installing the washer stack.

LH - 2.18MM RH - 1.95MM

NOTE: The thickness of the washer stacks are special for each airplane.

MECH	INSP	GAT 171
<i>[Signature]</i>	<i>[Signature]</i>	<i>[Signature]</i>
<i>[Signature]</i>	<i>[Signature]</i>	<i>[Signature]</i>
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737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

SUBTASK 32-21-00-020-013

- (16) Remove the nose landing gear from the airplane.
 - (a) Move the nose landing gear forward in the wheel well.
 - (b) Turn the gear approximately 90 degrees such that it is clear of the wheel well sidewalls and nose wheel well doors.
 - (c) Lower the nose landing gear at point E until it clears the nose wheel well (PAGEBLOCK 07-11-03/201).
 - (d) If it is necessary, use a dolly or the Outrigger R/I Equipment, SPL-10305, to transport the nose landing gear.
 - 1) When the Outrigger R/I Equipment, SPL-10305, is installed and will be left unattended, chock the nose landing gear tires and lock the caster brake (PAGEBLOCK 10-11-05/201).

Figure 401. Nose Landing Gear Installation

- Sheet 1 - Effectivity: JXB ALL
- Sheet 2 - Effectivity: JXB ALL
- Sheet 3 - Effectivity: JXB ALL
- Sheet 4 - Effectivity: JXB ALL
- Sheet 5 - Effectivity: JXB ALL
- Sheet 6 - Effectivity: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033
- Sheet 7 - Effectivity: JXB ALL
- Sheet 8 - Effectivity: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033
- Sheet 9 - Effectivity: JXB ALL

Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment

- Sheet 1 - Effectivity: JXB ALL
- Sheet 2 - Effectivity: JXB ALL

MECH	INSP
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[Signature]	[Signature]

GAT 171

GAT 171

GAT 171

DILIMASIE CAULPER
 P/N: 500-197-30
 S/N: 2209206611
 CALDUE: 05/04/2025

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 24/12/24

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____



737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

JXB ALL

TASK 32-21-21-000-801

2. Nose Landing Gear Drag Strut Removal (Figure 401) (Figure 402)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-03-580-802	Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E (P/B 201)
07-11-21-580-801	Lift the Airplane Nose with the Nose Jack at Jack Point D (P/B 201)
10-11-05 P/B 201	CHOCK INSTALLATION
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-080-801	Landing Gear Downlock Pins Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-33-52-000-801	Nose Gear Lock Spring Removal (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1871	Strap - Retention, NLG/MLG Inner Cylinder 737-800 (Part #: C32030 -31, Supplier: 81205) (Opt Part #: C32030-10, Supplier: 81205)
SPL-14478	Drag Strut Trun Pin Puller Assy, C32033-5 (included in C32033 Eqpt) 737-800 (Part #: C32033-1, Supplier: 81205)
SPL-14479	Slide Hammer, C32033-22, with Slide Screw, C32033-21 (included in C32033 Eqpt) 737-800 (Part #: C32033-1, Supplier: 81205)

JXB ALL

C. Location Zones

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

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<i>[Signature]</i>	<i>[Signature]</i>
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[Circular stamps: GAT 171]

737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

Zone	Area
115	Nose Landing Gear Wheel Well - Left
116	Nose Landing Gear Wheel Well - Right
713	Nose Landing Gear

MECH	INSP
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<i>[Signature]</i>	<i>[Signature]</i>
<i>[Signature]</i>	<i>[Signature]</i>
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<i>[Signature]</i>	<i>[Signature]</i>
<i>[Signature]</i>	<i>[Signature]</i>
<i>[Signature]</i>	<i>[Signature]</i>



D. Prepare for the Removal

SUBTASK 32-21-21-480-001

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT. THIS CAN CAUSE INJURIES TO PERSONS, AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed on all the landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-21-21-580-001

(2) Do this task: Lift the Airplane Nose with the Nose Jack at Jack Point D, TASK 07-11-21-580-801.

NOTE: Lower the jack that supports the tail of the airplane while you lift the nose landing gear.

SUBTASK 32-21-21-580-002

(3) If it is necessary to lift the gear with an axle jack to reduce the weight of the gear when you remove the right trunnion pin [1] and left trunnion pin [2], do this task: Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E, TASK 07-11-03-580-802.

SUBTASK 32-21-21-860-001

(4) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-21-21-860-002

(5) Open these circuit breakers and install safety tags: CAPT Electrical System Panel, P18-3

Row	Col	Number	Name
JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053			

D	14	(C00123)	EXT LIGHTING NOSE GEAR TAXI
---	----	----------	-----------------------------

JXB ALL

F/O Electrical System Panel, P6-3

Row	Col	Number	Name
B	17	(C00129)	LANDING GEAR LATCH & PRESS WARN
C	15	(C01355)	LANDING GEAR AIR/GND SYS 2
C	16	(C01356)	LANDING GEAR AIR/GND SYS 1

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



737-600/700/800/900

Restore Nose Landing Gear

Type:	Routine Card		ATA:32--	Flow:-	Work Area:-	MECH	INSP
C	18	(C01398)		LANDING GEAR TAKEOFF WARNING CUTOFF	}		
D	1	(C01399)		PSEU PRI			
D	2	(C01400)		PSEU ALTN			
D	15	(C01401)		LANDING GEAR AIR/GND RELAY			
D	16	(C01432)		LANDING GEAR ALTN EXTEND SOL			
D	17	(C01027)		LANDING GEAR NOSE GEAR STEER			
D	18	(C00451)		LANDING GEAR AURAL WARN			
SUBTASK 32-21-21-480-002							
(6) Install chocks around the tires of all the landing gear (PAGEBLOCK 10-11-05/201).							
SUBTASK 32-21-21-020-001							
(7) Do this task: Nose Gear Lock Spring Removal, TASK 32-33-52-000-801.							
SUBTASK 32-21-21-080-001							
(8) Remove the downlock pin for the nose landing gear, do this task: Landing Gear Downlock Pins Removal, TASK 32-00-01-080-801.							
E. Nose Landing Gear Drag Strut Removal							
SUBTASK 32-21-21-020-002							
(1) If necessary, disconnect the hydraulic hose connection from the rod end of the retract actuator [3].							
NOTE: These steps are only necessary if the retract actuator will be removed.							
(a) Disconnect the hydraulic UP-hose at the bracket on the retract actuator [3].							
(b) Disconnect the hydraulic DOWN-hose at the fitting on the retract actuator [3].							
(c) Install caps on the hydraulic hoses and the fittings on the retract actuator [3].							
JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES							
SUBTASK 32-21-21-020-003							
(2) Disconnect the rod end of the retract actuator [3].							
(a) Remove the cotter pin [50] from the bolt [16].					}		
(b) Remove the nut [12] and washer [14] from the bolt [16].							
(c) Remove the bolt [16] from the pin [9].							
(d) Remove the nut [15] and washer [11] from the pin [9].							
(e) Remove the pin [9] that holds the rod end of the retract actuator [3] and washers [18] from the upper drag strut [17].							
JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES							
SUBTASK 32-21-21-020-014							
(3) Disconnect the rod end of the retract actuator [3].							
(a) Remove the nuts [46], washers [45], washers [44], bolts [43], and retainer [47] from the pin [48].							
(b) Remove the pin [48] that holds the rod end of the retract actuator [3] and washers [18] from the upper drag strut [17].							

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JXB ALL

SUBTASK 32-21-21-020-004

(4) Disconnect the forward lock link [20].

NOTE: This step will also disconnect the lower drag strut [10] from the upper drag strut [17].

- (a) Remove the cotter pin [13] from the bolt [22].
- (b) Remove the nut [25] and washer [24] from the bolt [22].
- (c) Remove the bolt [22] from the pin [19].
- (d) Remove the nut [23] and washer [21] from the pin [19].
- (e) Remove the pin [19] to disconnect the forward lock link [20] from the upper drag strut [17] and lower drag strut [10].

SUBTASK 32-21-21-020-005

WARNING: MAKE SURE THAT YOU FULLY EXTEND THE SHOCK STRUT FOR THE NOSE LANDING GEAR BEFORE YOU DISCONNECT THE DRAG STRUT. IF YOU DO NOT FULLY EXTEND THE SHOCK STRUT, IT CAN MOVE QUICKLY AND CAN CAUSE INJURY TO PERSONNEL AND DAMAGE TO EQUIPMENT.

(5) Disconnect the lower drag strut [10] from the shock strut [29].

- (a) Remove the cotter pin [26] from the bolt [28].
- (b) Remove the nut [33] and washer [32] from the bolt [28].
- (c) Remove the bolt [28] from the pin [30].
- (d) Remove the nut [27] and washer [31] from the pin [30].
- (e) Remove the pin [30] that holds the lower drag strut [10] from the shock strut [29].

SUBTASK 32-21-21-480-003

(6) Use the retention strap, SPL-1871, to hold the shock strut [29] in the compressed position.

SUBTASK 32-21-21-020-006

(7) Disconnect the upper drag strut [17] from the bracket [51].

NOTE: This step will also remove the bracket, that contains the down position sensor target. Keep the bracket for the installation procedure.

- (a) Remove the nut [34] and washer [35] from the bolt [36].
- (b) Remove the bolt [36] and washer [35] to disconnect the upper drag strut [17] from the bracket [51].
- (c) Remove the washer [35] from the bolt [36].

SUBTASK 32-21-21-020-007

(8) Remove the right trunnion pin [1] and left trunnion pin [2].

- (a) Remove the cotter pin [38] from the bolt [40].
- (b) Remove the nut [37] and washer [39] from bolt [40].
- (c) Remove the bolt [40] that hold the right trunnion pin [1] and left trunnion pin [2] to the sidewall of the nose wheel well.
- (d) Remove the right trunnion pin [1] and left trunnion pin [2] to disconnect the upper drag strut [17] from the sidewall of the nose wheel well.

NOTE: Use the Drag Strut Trunnion Pin puller, C32033-5, SPL-14478, and slide hammer, C32033-22, with slide screw, C32033-21,

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Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

SPL-14479, to remove the trunnion pins from inboard side of the upper drag strut.

JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

(e) Remove and discard the packing [42].

JXB ALL

SUBTASK 32-21-21-020-008

(9) Remove the upper drag strut [17] from the airplane.

Figure 401. Nose Landing Gear Drag Strut Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

Sheet 3 - Effectivity: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

Sheet 4 - Effectivity: JXB ALL

Sheet 5 - Effectivity: JXB ALL

Sheet 6 - Effectivity: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment

Sheet 1 - Effectivity: JXB ALL

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52	24/11/2024
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GAT 171

24/11/2024

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Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

JXB ALL

TASK 32-33-51-000-801

3. Nose Landing Gear Lock Mechanism Removal (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-21-580-801	Lift the Airplane Nose with the Nose Jack at Jack Point D (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1866	Expander Set - Spring, Main Landing Gear and Nose Landing Gear 737-800 ✓ (Part #: C32014-29, Supplier: 81205) (Opt Part #: C32014-1, Supplier: 81205) (Opt Part #: C32014-20, Supplier: 81205)

JXB ALL

C. Location Zones

Zone	Area
113	Area Above and Outboard of Nose Landing Gear Wheel Well - Left
114	Area Above and Outboard of Nose Landing Gear Wheel Well - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors

D. Prepare for the Removal

SUBTASK 32-33-51-480-001

WARNING: MAKE SURE THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR COULD RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

MECH	INSP
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S.L.	Q. [Signature]
S.L.	Q. [Signature]
S.L.	Q. [Signature]
S.L.	Q. [Signature]
S.L.	Q. [Signature]

GAT 171 (multiple stamps)

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Item: _____ Completed through item: _____ Sign: _____



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Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

(6) Remove the bushing [7] and shaft [6] to disconnect the aft lock link assembly [3] from the wheel well fitting (Detail C).

SUBTASK 32-33-51-020-007

(7) Remove the lock mechanism as a complete assembly.

Figure 401. Nose Gear Lock Mechanism Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

Sheet 4 - Effectivity: JXB ALL

MECH	INSP
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Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

JXB ALL

TASK 32-33-11-000-801

4. Nose Gear Retract Actuator Removal (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
10-11-05 P/B 201	CHOCK INSTALLATION
29-09-00-860-802	Hydraulic Reservoirs Depressurization (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)

B. Location Zones

Zone	Area
113	Area Above and Outboard of Nose Landing Gear Wheel Well - Left
114	Area Above and Outboard of Nose Landing Gear Wheel Well - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors
730	Subzone - Left Main Landing Gear and Landing Gear Doors
740	Subzone - Right Main Landing Gear and Landing Gear Doors

C. Prepare for the Removal

SUBTASK 32-33-11-480-001

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT. THIS CAN CAUSE INJURIES TO PERSONS, AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed in the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-33-11-860-001








(2) For hydraulic system A, do this task: Hydraulic Reservoirs Depressurization, TASK 29-09-00-860-802.

SUBTASK 32-33-11-580-001

(3) Install chocks around the tires of the nose landing gear (CHOCK INSTALLATION, PAGEBLOCK 10-11-05/201).

D. Nose Gear Retract Actuator Removal

NOTE: The pin [8] or pin [16] together with pin [11], nut [10], washer [12], bolt [14], castellated nut [13] and washer [9] can replace or be replaced by pin [23] or pin [24] together with nuts [21], washers [20] and washers [19], bolts [18], and retainer [22].

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Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

SUBTASK 32-33-11-020-001

(1) Disconnect the hydraulic UP-hose at the bracket on the retract actuator [1].

SUBTASK 32-33-11-020-002

(2) Disconnect the hydraulic DOWN-hose at the fitting on the retract actuator [1].

SUBTASK 32-33-11-480-002

(3) Install caps on the hydraulic hoses and the fittings on the retract actuator [1].

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

SUBTASK 32-33-11-020-003

(4) Hold the retract actuator [1] in its position and disconnect the rod end of the retract actuator [1] from the drag brace (Detail B):

(a) Remove the pin [11], the nut [10], the washer [12], and the bolt [14] from the castellated nut [13] and the pin [8].

(b) Remove the castellated nut [13], the washer [9], the pin [8] and the washers [15] from the airplane.

SUBTASK 32-33-11-020-004

(5) Hold the retract actuator [1] in its position and disconnect the head end of the retract actuator [1] from the top fitting (Detail C):

(a) Remove the pin [11], the nut [10], the washer [12], and the bolt [14] from the castellated nut [13] and the pin [16].

(b) Remove the castellated nut [13], the washer [9], the pin [16] and the washers [17] from the airplane.

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

SUBTASK 32-33-11-020-008

(6) Hold the retract actuator [1] in its position and disconnect the rod end of the retract actuator [1] from the drag brace (Detail B):

(a) Remove the bolts [18], the washers [19], the washers [20], the nuts [21] and the retainer [22] from the pin [23].

(b) Remove the washers [15] and the pin [23] from the airplane.

SUBTASK 32-33-11-020-009

(7) Hold the retract actuator [1] in its position and disconnect the head end of the retract actuator [1] from the top fitting (Detail C):

(a) Remove the bolts [18], the washers [19], the washers [20], the nuts [21], and the retainer [22] from the pin [24].

(b) Remove the washers [17] and the pin [24] from the airplane.

JXB ALL

SUBTASK 32-33-11-020-005

(8) Remove the retract actuator [1] from the airplane.

SUBTASK 32-33-11-020-006

(9) If the replacement retract actuator [1] does not have the tube assembly installed, remove these parts from the retract actuator [1] you removed:

(a) The reducer [2].

(b) The screws [4], the washers [5], and the clamp [3].

(c) The tube assembly [6].

(d) The elbow fitting [7].

MECH INSP

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GAT 171

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Restore Nose Landing Gear

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SUBTASK 32-33-11-480-003

(10) Install a plug in the extend port of the retract actuator [1].

Figure 401. Nose Gear Retract Actuator Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

Sheet 3 - Effectivity: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

Sheet 4 - Effectivity: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

Sheet 5 - Effectivity: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

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Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

JXB ALL

TASK 32-33-11-400-801

5. Nose Gear Retract Actuator Installation (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-21-580-801	Lift the Airplane Nose with the Nose Jack at Jack Point D (P/B 201)
07-11-21-580-802	Lower the Airplane Nose Off of the Jack (P/B 201)
12-12-00-610-801	Hydraulic Reservoir Servicing (P/B 301)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-080-801	Landing Gear Downlock Pins Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)

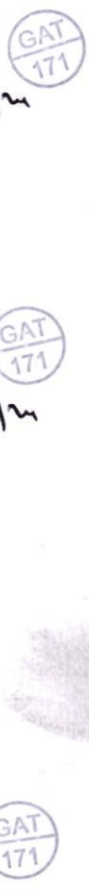
B. Consumable Materials

Reference	Description	Specification
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827 (NATO G-354) (Supersedes MIL-G-23827)
D00153	Fluid - Hydraulic Fluid, Fire Resistant (Interchangeable And Intermixable With BMS 3-11 Type V)	BMS3-11 Type IV
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

C. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
1	Retract actuator	32-33-11-01-036	JXB ALL

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Restore Nose Landing Gear

Type: Routine Card ATA: 32-- Flow: -- Work Area: --

11 Pin 32-33-11-01-017 JXB 004, 005, 007-009

D. Location Zones

Zone	Area
113	Area Above and Outboard of Nose Landing Gear Wheel Well - Left
114	Area Above and Outboard of Nose Landing Gear Wheel Well - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors
730	Subzone - Left Main Landing Gear and Landing Gear Doors
740	Subzone - Right Main Landing Gear and Landing Gear Doors

E. Nose Gear Retract Actuator Installation

NOTE: The pin [8] or pin [16] together with pin [11], nut [10], washer [12], bolt [14], castellated nut [13] and washer [9] can replace or be replaced by pin [23] or pin [24] together with nuts [21], washers [20] and washers [19], bolts [18], and retainer [22].

SUBTASK 32-33-11-080-001

(1) Remove the plugs from the port and the fitting on the retract actuator [1].

SUBTASK 32-33-11-020-007

- (2) If the replacement retract actuator [1] does not have the tube assembly [6] installed, install these parts from the retract actuator [1] you removed:
- (a) The elbow fitting [7].
 - (b) The tube assembly [6].
 - (c) The screws [4], washers [5], and clamp [3].
 - (d) The reducer [2].

SUBTASK 32-33-11-980-001

(3) Make sure that the retract actuator [1] is in the retracted position.

SUBTASK 32-33-11-610-001

- (4) Make sure that the retract actuator [1] is full of D00153 hydraulic fluid .
- (a) Put a cap on the hydraulic fitting on the retract actuator [1] where the DOWN-hose will attach.
 - (b) Put a cap on the fitting on the retract actuator [1] where the UP-hose will attach.

SUBTASK 32-33-11-600-001

(5) Prepare the retract actuator [1] for installation on the top fitting:

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

(a) Apply D00633 grease , or D00013 grease , to the bushings of the retract actuator [1] and the shaft of the pin [16].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN,

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Work Area: -

IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(b) Apply G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to the threads and thread reliefs of the castellated nut [13] and to the faces of the washer [9].

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

(c) Apply D00633 grease, or D00013 grease, to the bushings of the retract actuator [1] and the shaft of the pin [24].

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

SUBTASK 32-33-11-420-014

(6) Hold the retract actuator [1] in its position and do these steps to connect the head end of the retract actuator [1] to the top fitting (View C, Figure 401):

(a) Install the pin [16] and washers [17].

NOTE: The head end of pin must be on the same side as the hydraulic hoses to prevent physical interference.

JXB ALL

1) If it is necessary, trim the washers [17] around the entire circumference to prevent part interference with the angle bracket.

NOTE: This step is optional.

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

- (b) Install the washer [9] and castellated nut [13] on the pin [16].
- (c) Tighten the castellated nut [13] to 50 ft-lb (67.8 N·m) - 58 ft-lb (78.6 N·m).
- (d) Loosen the castellated nut [13] to the nearest castellation and install the bolt [14], washer [12], and nut [10] on the castellated nut [13] and pin [16].
- (e) Install the pin [11] on the nut [10] and bolt [14].
- (f) Wipe of excess G50237 Cor-Ban 27L Compound, and G50136 corrosion inhibiting material.

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

SUBTASK 32-33-11-420-005

(7) Hold the retract actuator [1] in its position and do these steps to connect the head end of the retract actuator [1] to the top fitting (View C, Figure 401):

(a) Install the pin [24] and washers [17].

NOTE: The head end of pin must be on the same side as the hydraulic hoses to prevent physical interference.

JXB ALL

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Restore Nose Landing Gear

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Flow: -

Work Area: -

1) If it is necessary, trim the washers [17] around the entire circumference to prevent part interference with the angle bracket.

NOTE: This step is optional.

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

- (b) Install the retainer [22] with the flat surface touching the retract actuator [1] on the pin [24].
- (c) Install the bolts [18], washers [19], washers [20], and nuts [21] on the retainer [22] and pin [24].

JXB ALL

SUBTASK 32-33-11-420-011

- (8) Prepare the retract actuator [1] for installation on the drag brace:

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

- (a) Apply D00633 grease , or D00013 grease , to the bushings of the retract actuator [1] and the shaft of the pin [8].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: REMOVE UNWANTED CORROSION-INHIBITING COMPOUND FROM SURFACES WHICH WILL BE LUBRICATED. IF YOU APPLY CORROSION-INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT CAN OCCUR.

- (b) Apply G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to the threads and thread reliefs of the pin [8] and to the faces of the washers [15].

JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

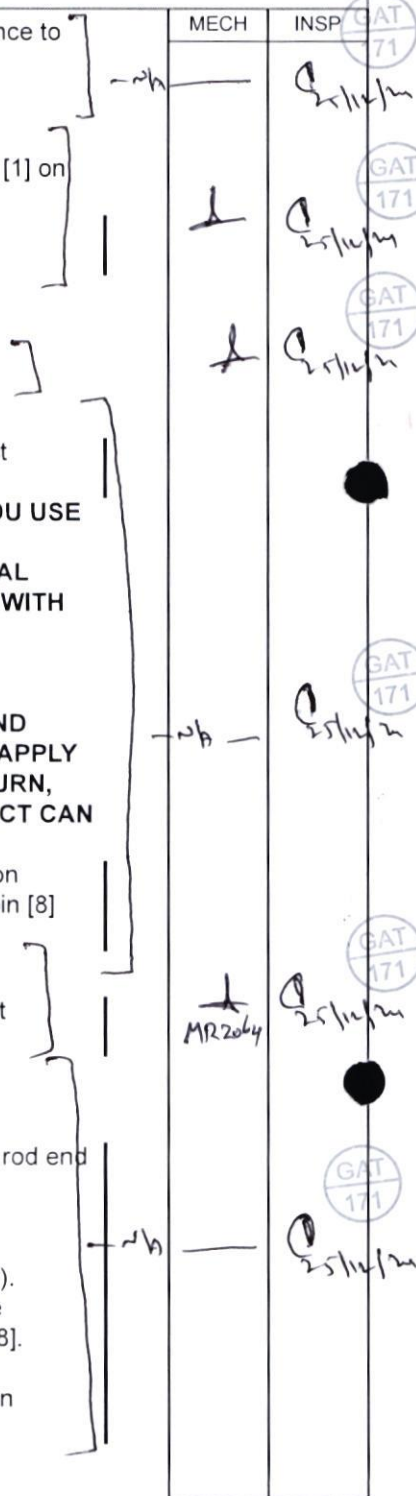
- (c) Apply D00633 grease , or D00013 grease , to the bushings of the retract actuator [1] and the shaft of the pin [23].

JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES

SUBTASK 32-33-11-420-015

- (9) Hold the retract actuator [1] in its position and do these steps to connect the rod end of the retract actuator [1] to the drag brace (View B, Figure 401):

- (a) Install the pin [8] and washers [15].
- (b) Install the washer [9] and castellated nut [13] on the pin [8].
- (c) Tighten the castellated nut [13] to 50 ft-lb (67.8 N·m) - 58 ft-lb (78.6 N·m).
- (d) Loosen the castellated nut [13] to the nearest castellation and install the bolt [14], washer [12], and nut [10] on the castellated nut [13] and pin [8].
- (e) Install the pin [11] on the nut [10] and bolt [14].
- (f) Wipe of excess G50237 Cor-Ban 27L Compound , and G50136 corrosion inhibiting material .



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Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

JXB ALL

TASK 32-33-51-400-801

6. Nose Gear Lock Mechanism Installation (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-61-41-400-801	Nose Landing Gear Lock Sensor Installation (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1866	Expander Set - Spring, Main Landing Gear and Nose Landing Gear
737-800	
	(Part #: C32014-29, Supplier: 81205)
	(Opt Part #: C32014-1, Supplier: 81205)
	(Opt Part #: C32014-20, Supplier: 81205)

JXB ALL

C. Consumable Materials

Reference	Description	Specification
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827 (NATO G-354) (Supersedes MIL-G-23827)
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38

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737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

G50237

Compound - Corrosion Inhibiting, Non-drying - BMS3-38
Cor-Ban 27L

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
1	Link assembly	32-33-51-02-120	JXB ALL
2	Spring assembly	32-33-51-02-030	JXB ALL
3	Link assembly	32-33-51-02-155	JXB ALL
8	Pin	32-21-21-02-045	JXB ALL
18	Pin	32-33-21-01-016	JXB ALL
29	Pin	32-21-21-02-075	JXB ALL

E. Location Zones

Zone	Area
113	Area Above and Outboard of Nose Landing Gear Wheel Well - Left
114	Area Above and Outboard of Nose Landing Gear Wheel Well - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors

F. Procedure

SUBTASK 32-33-51-420-006

(1) If you replace one of the lock links, you must install or remove the shims on the completed lock mechanism assembly to adjust the overcenter dimension.

NOTE: The overcenter dimension is set on the assembly.

SUBTASK 32-33-51-420-001

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: REMOVE UNWANTED CORROSION-INHIBITING COMPOUND FROM SURFACES WHICH WILL BE LUBRICATED. IF YOU APPLY CORROSION-INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT CAN OCCUR.

(2) Do the steps that follow to connect the aft lock link assembly [3] to the fitting on the aft wheel well wall (Figure 401, View B, C):

(a) Apply a thin layer of G50237 Cor-Ban 27L Compound to the threads and thread reliefs of the shaft [6] and the nuts [4].

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Work Area: -

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(f) Apply a thin layer of G50237 Cor-Ban 27L Compound, to the threads of the nut [28], the faces of the washer [27], and the new pin [29].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(g) Install the bolt [25], washer [27], and nut [28].

(h) Torque the nut [28] to 18 in-lb (2.0 N·m) – 25 in-lb (2.8 N·m) above the run-on torque.

1) If it is necessary, loosen the nut [28] to align the nearest castellation with the hole in the bolt [25].

(i) Install the new pin [29].

SUBTASK 32-33-51-420-003

(4) Do the steps that follow to connect the aft lock link assembly [3] of the lock mechanism to the rod end of the lock actuator (Figure 401, View C):

(a) Apply D00633 grease, or D00013 grease (Optional), to the bore of the rod end and aft lock link assembly [3], and the pin [22].

(b) Install the pin [22], washer [20], washer [21], two washers [28], and nut [19].

(c) Torque the nut [19] to 18 ft-lb (24.4 N·m) - 24 ft-lb (32.5 N·m). *20 ft-lb*
 1) If it is necessary, loosen the nut [19] to align the nearest castellation with the hole on the pin [22].

(d) Install the new pin [18].

SUBTASK 32-33-51-420-004

(5) Do these steps to connect the spring assembly [2] to the forward lock link assembly [1] and the wheel well fitting (Figure 401, View B, D):

(a) Remove the washers [5] and nuts [9] that attach the spring assembly [2] to the forward lock link assembly [1].

(b) Remove the spring assembly [2], bolts [10], washers [11], and sleeves [12] from the link assembly [1].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(c) Apply a thick layer of G50237 Cor-Ban 27L Compound to the shank and threads of the bolts [10].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(d) Apply a thin layer of G50237 Cor-Ban 27L Compound, to the threads of the nuts [9], the faces of the washers [5] and sleeves [12], and the new pins [8].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(e) Apply D00633 grease, to the chrome plated surfaces of the sleeves [12] and the bolts [10].

MECH INSP

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INDEPENDENT INSPECTION

GAT 470
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INDEPENDENT INSPECTION

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Work Area: -

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(f) Apply a thin layer of G50237 Cor-Ban 27L Compound to the threads and thread reliefs of the nuts [4], the faces of the washers [5], and the pins [8].

NOTE: If G50237 Cor-Ban 27L Compound is not available, you can use G50136 corrosion inhibiting material as an approved alternate.

(g) Apply D00633 grease, to the chrome plated surfaces of the shaft [6].

(h) Use the expander set, SPL-1866 to extend the spring assembly [2].

(i) Connect the spring assembly [2] to the forward lock link assembly [1] (Figure 401, View D):

- 1) Put the spring assembly [2] in its location at the forward lock link assembly [1].
- 2) Install the bolts [10], sleeves [12], and the washers [11] to connect the spring assembly [2] to the forward lock link assembly [1].
- 3) Install the washers [5] and the nuts [9] on the bolts [10].
- 4) Torque the nuts [9] to 20 in-lb (2.3 N·m) – 24 in-lb (2.7 N·m) above the run-on torque.

CAUTION: POINT THE PIN DOWN DURING INSTALLATION TO KEEP RISK OF CHAFING DAMAGE TO CONDUIT TO A MINIMUM.

a) Install the new pins [8].

(j) Connect the spring assembly [2] to the shaft [6] in the wheel well fitting (Figure 401, view B):

- 1) Install the shaft [6] in the wheel well fitting.
- 2) Put the spring assembly [2] in its location on the shaft [6].
- 3) Install the washers [5] and nuts [4] on the shaft [6].
- 4) Torque the nuts [4] to 20 in-lb (2.3 N·m) – 24 in-lb (2.7 N·m) above the run-on torque.

a) Install the new pins [8].

(k) Remove the expander set, SPL-1866 from spring assembly [2].

SUBTASK 32-33-51-420-005

(6) Do the steps that follow to install the lock sensor [14] on the aft lock link assembly [3] (Figure 401, View E):

(a) Put the lock sensors [14] on the aft lock link assembly [3].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(b) Apply a thin layer of G50237 Cor-Ban 27L Compound to the threads and thread reliefs of the screws [16], and the faces of the washers [15].

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	GAT 178 25/12/24

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737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

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TASK 32-21-21-400-801

7. **Nose Landing Gear Drag Strut Installation** (Figure 401) (Figure 402)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-21-580-802	Lower the Airplane Nose Off of the Jack (P/B 201)
12-21-21-640-801	Nose Landing Gear Upper End Components Servicing (P/B 301)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
32-33-00-710-801	Operational Test for the Nose Landing Gear (P/B 501)
32-33-00-710-802	Nose Landing Gear Test - Component Replacement (P/B 501)
32-33-52-400-801	Nose Gear Lock Spring Installation (P/B 401)
32-35-00-730-801	Nose Gear Manual Extension System Test - Airplane on Jacks (P/B 501)
32-51-00-700-801	Nose Wheel Steering System Test (P/B 501)
32-61-51-400-801	Nose Landing Gear Down Position Sensor Clearance Adjustment (P/B 501)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-14480	Drag Strut Align Slug, C32033-4 (included in C32033 Eqpt) 737-800 (Part #: C32033-1, Supplier: 81205)
SPL-14481	R/I Equipment - Trunnion Pin, NLG (Alignment Pin Assy is included in kit) 737-800 (Part #: C32033-1, Supplier: 81205)

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C. Consumable Materials

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737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

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Reference	Description	Specification
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
13	Pin	32-21-21-02-075	JXB ALL
26	Pin	32-21-00-02-005	JXB ALL
38	Pin	32-21-21-02-005	JXB ALL
42	Packing	32-21-21-02-022	JXB ALL
50	Pin	32-33-11-01-017	JXB 004, 005, 007-009

E. Location Zones

Zone	Area
115	Nose Landing Gear Wheel Well - Left
116	Nose Landing Gear Wheel Well - Right
713	Nose Landing Gear

F. Prepare for the Installation

SUBTASK 32-21-21-620-001

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: MAKE SURE THAT YOU REMOVE UNWANTED CORROSION INHIBITING COMPOUND FROM SURFACES THAT YOU WILL LUBRICATE. IF YOU APPLY CORROSION INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT COULD OCCUR.

- (1) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:
 - (a) The shank and the threads of the bolts [40].

- NOTE:** Apply a thick layer of the compound to the bolts [40].
 - (b) The threads and thread reliefs of the nuts [37].

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Restore Nose Landing Gear

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- (c) The faces of the washers [39].
- (d) The cotter pin [38].

SUBTASK 32-21-21-160-001

- (2) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

SUBTASK 32-21-21-640-001

- (3) Lubricate the chrome plated surfaces of the right trunnion pin [1] and left trunnion pin [2] with D00633 grease .

SUBTASK 32-21-21-160-002

- (4) Remove all unwanted D00633 grease .

G. Nose Landing Gear Drag Strut Installation

SUBTASK 32-21-21-560-001

- (1) Put the upper drag strut [17] in its position for installation on the sidewall of the nose wheel well.

SUBTASK 32-21-21-580-003

- (2) Align the upper drag strut [17] with the mating hole in the sidewall of the nose wheel well.

SUBTASK 32-21-21-420-001

- (3) Install the right trunnion pin [1] and left trunnion pin [2].

- (CR-2) (a) Make sure that the trunnion pin seal [41] is installed in the trunnion pin.

JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

- (b) Apply D00633 grease , and install new packing [42] inside the nose wheel well drag brace bushing.

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- (c) Use the Drag Strut alignment slug, C32033-4, SPL-14480, to align the upper drag strut [17] and sidewall of the nose wheel well.

- (d) Put the right trunnion pin [1] and left trunnion pin [2] through the upper drag strut [17] and sidewall of the nose wheel well.

- 1) Make sure that the hole in the right trunnion pin [1] and left trunnion pin [2] for the bolts [40] are aligned.

- a) Use nose landing gear trunnion pin alignment pin assembly, SPL-14481.

- (e) Put the bolt [40] through the right trunnion pin [1] and left trunnion pin [2].

- (CR-2) (f) Install the washer [39] and nut [37] on the bolt [40].

- (g) Torque the nut [37] to 20 in-lb (2.3 N·m) - 24 in-lb (2.7 N·m) more than the run-on torque. 20 in-lb

- (CR-2) (h) If it is necessary, loosen the nut [37] to the nearest castellation to align the holes for the cotter pin [38].

- (i) Install the cotter pin [38] in the bolt [40].

- (j) Rotate seal [41] as required to fully seat seal against crossbolt.

CAUTION: WEAR SAFETY GLASSES DURING APPLICATION, AND IF APPLICATION OF SOLVENT RESULTS IN IRRITATION, USE AN AIR PURIFYING ORGANIC VAPOR CARTRIDGE RESPIRATOR. USE NITRILE GLOVES AND WASH YOUR HANDS WITH MOISTURE CREAM AFTER APPLICATION. STORE DPM 6380-4

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Rev # 41



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Rev Date: Oct 17, 2024 PDT

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Restore Nose Landing Gear

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**CLEANER/SOLVENT IN FLAMMABLE RESISTANT CONTAINERS
IN QUANTITIES NO LARGER THAN 1 GALLON.**

(k) Reapply G50237 Cor-Ban 27L Compound, as required to coat the interior surface of the trunnion pin.

SUBTASK 32-21-21-420-002

- (4) Connect the upper drag strut [17] to the bracket [51].
 - (a) Put the washer [35] on the bolt [36].
 - (b) Align the hole in the upper drag strut [17] to the hole in the bracket [51].
 - (c) Put the bolt [36] through the bracket [51] and upper drag strut [17].
 - (d) Install the washer [35] and nut [34] on the bolt [36].

SUBTASK 32-21-21-620-002

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: MAKE SURE THAT YOU REMOVE UNWANTED CORROSION INHIBITING COMPOUND FROM SURFACES THAT YOU WILL LUBRICATE. IF YOU APPLY CORROSION INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT COULD OCCUR.

- (5) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:
 - (a) The threads and thread reliefs of the pin [30] and bolt [28].
 - (b) The threads of the nut [27] and nut [33].
 - (c) The faces of the washers [31] and washer [32].
 - (d) To the cotter pin [26].

SUBTASK 32-21-21-160-003

(6) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound.

SUBTASK 32-21-21-640-002

(7) Lubricate the shank of the pin [30] and bolt [28] with D00633 grease.

SUBTASK 32-21-21-160-004

(8) Remove all unwanted D00633 grease.

SUBTASK 32-21-21-420-003

- (9) Connect the lower drag strut [10].
 - (a) Make sure that the drag brace lubrication fittings are on the right side.
 - (b) Put the lower drag strut [10] in its position on the shock strut [29].
 - (c) Put the pin [30] through the shock strut [29] and lower drag strut [10].
 - (d) Install the washer [31] and nut [27] on the pin [30].
 - (e) Torque the nut [27] to 600 in-lb (67.8 N·m) - 700 in-lb (79.1 N·m).
 - (f) Loosen the nut [27] to the nearest castellation to align the holes for the bolt [28].
 - (g) Install the bolt [28] in the pin [30].
 - (h) Install the washer [32] on the bolt [28].
 - (i) Install the nut [33] on the bolt [28].

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737-600/700/800/900

Restore Nose Landing Gear

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(j) If it is necessary, loosen the nut [33] to the nearest castellation to align the holes for the cotter pin [26].

(k) Install the cotter pin [26] in the bolt [28].

SUBTASK 32-21-21-620-003

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(10) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:

- (a) The threads and thread reliefs of the pin [19] and bolt [22].
- (b) The threads of the nut [23] and nut [25].
- (c) The faces of the washers [21] and washer [24].
- (d) The cotter pin [13].

SUBTASK 32-21-21-160-005

(11) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

SUBTASK 32-21-21-420-009

(12) If nut and bolt is installed in the downlock pin hole, remove it and install a downlock pin in that location.

SUBTASK 32-21-21-420-004

(13) Connect the forward lock link [20] and lower drag strut [10].

NOTE: This step will also connect the lower drag strut [10] to the upper drag strut [17].

- (a) Put the lower drag strut [10] and forward lock link [20] in their positions on the upper drag strut [17].
- (b) Put the pin [19] through the forward lock link [20], upper drag strut [17], and lower drag strut [10].
- (c) Install the washer [21] and nut [23] on the pin [19].
- (d) Torque the nut [23] to 95 in-lb (10.7 N-m) - 115 in-lb (13.0 N-m) more than the run-on torque.
- (e) Loosen the nut [23] to the nearest castellation to align the hole for the bolt [22].
- (f) Install the bolt [22] in the pin [19].
- (g) Install the washer [24] and nut [25] on the bolt [22].
- (h) If it is necessary, loosen the nut [25] to the nearest castellation to align the holes for the cotter pin [13].
- (i) Install the cotter pin [13] in the bolt [22].

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737-600/700/800/900

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- (14) If the retract actuator was removed, hold the retract actuator [3] and do these steps to connect the actuator to the upper drag strut [17].
 - (a) Put the rod end of the retract actuator [3] and washers [18] in their position on the upper drag strut [17].
 - (b) Put the pin [9] through the upper drag strut [17], rod end of the retract actuator [3], and washers [18].
 - (c) Install the washer [11] and castellated nut [15] on the pin [9].
 - (d) Torque the nut [15] to 50 ft-lb (68 N·m) - 58 ft-lb (79 N·m).
 - (e) If it is necessary, loosen the nut [15] to the nearest castellation to align the hole for the bolt [16].
 - (f) Install the bolt [16] in the pin [9].
 - (g) Install the washer [14] and nut [12].
 - (h) If it is necessary, loosen the nut [12] to the nearest castellation to align the hole for the cotter pin [50].
 - (i) Install the cotter pin [50] in the bolt [16].

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JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES

SUBTASK 32-21-21-420-008

- (15) If the retract actuator was removed, hold the retract actuator [3] and do these steps to connect the actuator to the upper drag strut [17].
 - (a) Put the rod end of the retract actuator [3] and washers [18] in their position on the upper drag strut [17].
 - (b) Put the pin [48] through the upper drag strut [17], rod end of the retract actuator [3], and washers [18].
 - (c) Install the retainer [47] with the flat surface touching the upper drag strut [17] on the pin [48].
 - (d) Install the bolts [43], washers [44], washers [45], and nuts [46] on the retainer [47].

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JXB ALL

SUBTASK 32-21-21-080-002

- (16) If they were installed, remove the caps from the hydraulic hoses.

SUBTASK 32-21-21-420-006

- (17) If the retract actuator was removed, connect the hydraulic hoses to the retract actuator [3].

SUBTASK 32-21-21-720-001

- (18) Do a post installation check of the nose landing gear down position sensor (TASK 32-61-51-400-801).

NOTE: This step is necessary because the bracket [51], that contains the down position sensor target, was removed.

H. Put the Airplane Back to Its Usual Condition

SUBTASK 32-21-21-420-007

- (1) Do this task: Nose Gear Lock Spring Installation, TASK 32-33-52-400-801.

SUBTASK 32-21-21-610-001

- (2) Lubricate the upper components of the nose landing gear (TASK 12-21-21-640-801).

SUBTASK 32-21-21-860-005

- (3) Remove the safety tags and close these circuit breakers:

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 Item: 18 Completed through item: 18 Sign: [Signature] 27/12/24

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INDEPENDENT INSPECTION

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CAPT Electrical System Panel, P18-3

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Row Col Number Name
JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053

D 14 (C00123) EXT LIGHTING NOSE GEAR TAXI

JXB ALL

F/O Electrical System Panel, P6-3

Row Col Number Name
 B 17 (C00129) LANDING GEAR LATCH & PRESS WARN
 C 15 (C01355) LANDING GEAR AIR/GND SYS 2
 C 16 (C01356) LANDING GEAR AIR/GND SYS 1
 C 18 (C01398) LANDING GEAR TAKEOFF WARNING CUTOFF
 D 1 (C01399) PSEU PRI
 D 2 (C01400) PSEU ALTN
 D 15 (C01401) LANDING GEAR AIR/GND RELAY
 D 16 (C01432) LANDING GEAR ALTN EXTEND SOL
 D 17 (C01027) LANDING GEAR NOSE GEAR STEER
 D 18 (C00451) LANDING GEAR AURAL WARN

SUBTASK 32-21-21-860-003

(4) For hydraulic system A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-21-21-710-001

(5) Do this task: Operational Test for the Nose Landing Gear, TASK 32-33-00-710-801.

NOTE: It is optional to perform the Nose Landing Gear Test - Component Replacement, TASK 32-33-00-710-802 instead of the operational test if a hydraulic cart is not available.

SUBTASK 32-21-21-710-002

(6) Do this task: Nose Gear Manual Extension System Test - Airplane on Jacks, TASK 32-35-00-730-801.

SUBTASK 32-21-21-710-003

(7) Do this task: Nose Wheel Steering System Test, TASK 32-51-00-700-801.

SUBTASK 32-21-21-860-004

(8) Do this task: Lower the Airplane Nose Off of the Jack, TASK 07-11-21-580-802.

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TASK 32-21-00-400-801


8. Nose Landing Gear - Installation (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)
07-11-03-580-802	Lift the Airplane Nose Landing Gear with the Axle Jack at Jack Point E (P/B 201)
07-11-21-580-802	Lower the Airplane Nose Off of the Jack (P/B 201)
10-11-05 P/B 201	CHOCK INSTALLATION
12-15-41-610-801	Nose Landing Gear Shock Strut Fluid Check (P/B 301)
12-15-41-610-802	Nose Landing Gear Shock Strut Servicing, Airplane on the Ground (P/B 301)
12-15-41-610-805	Nose Landing Gear Shock Strut Servicing, Airplane on Jacks (P/B 301)
12-21-21-640-801	Nose Landing Gear Upper End Components Servicing (P/B 301)
12-21-21-640-802	Nose Landing Gear Lower End Components Servicing (P/B 301)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-33-00-710-801	Operational Test for the Nose Landing Gear (P/B 501)
32-33-00-710-802	Nose Landing Gear Test - Component Replacement (P/B 501)
32-35-00-730-801	Nose Gear Manual Extension System Test - Airplane on Jacks (P/B 501)
32-45-21-400-801	Nose Landing Gear Wheel and Tire Assembly - Installation (P/B 401)
32-51-00-700-801	Nose Wheel Steering System Test (P/B 501)
32-51-00-820-802	Nose Wheel Steering System Adjustment (P/B 501)
32-51-31-400-802	Nose Gear Steering System Cables Installation (P/B 401)
33-45-01-400-801	Taxi Light Housing Assembly - Installation (P/B 201)
53-14-01-420-801	Nose Wheel Well Access Panels - Installation (P/B 401)

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Restore Nose Landing Gear

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B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
COM-4938	Insertion/Removal Tool - Contact (size 16, backshell side) 737-800 (Part #: M81969/1-03, Supplier: 11851) (Part #: M81969/14-03, Supplier: 11139)
SPL-1521	Tool - Strut Inflation, Landing Gear 737-800 (Part #: F70200-35, Supplier: 81205) (Opt Part #: F70200-1, Supplier: 81205) (Opt Part #: F70200-14, Supplier: 81205) (Opt Part #: F70200-17, Supplier: 81205) (Opt Part #: F70200-18, Supplier: 81205)
SPL-1559	Adjustable Spanner Wrench (2.00 to 4.00 Inch Dia. Retainer) 737-800 (Part #: F72959-34, Supplier: 81205) (Part #: F72959-35, Supplier: 81205) (Part #: F72959-36, Supplier: 81205) (Part #: F72959-41, Supplier: 81205) (Opt Part #: F72959-5, Supplier: 81205) (Opt Part #: F72959-6, Supplier: 81205) (Opt Part #: F72959-7, Supplier: 81205) (Opt Part #: F72959-8, Supplier: 81205)
SPL-1821	Adjustable Spanner Wrench (0.75 to 2.00 Inch Dia. Retainer, 0.12 x 0.12 Key Arm) 737-800 (Part #: F72959-40, Supplier: 81205) (Opt Part #: F72959-4, Supplier: 81205)
SPL-1871	Strap - Retention, NLG/MLG Inner Cylinder 737-800

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- (Part #: C32030 -31, Supplier: 81205)
(Opt Part #: C32030-10, Supplier: 81205)
- SPL-10305 Outrigger Equipment - Nose Landing Gear Installation/
Removal
737-800
- (Part #: C32049-1, Supplier: 81205)
- SPL-14477 Alignment Pin Assy, C32033-2 (included in C32033 Eqpt)
737-800
- (Part #: C32033-1, Supplier: 81205)
- SPL-14481 R/I Equipment - Trunnion Pin, NLG (Alignment Pin Assy is
included in kit)
737-800
- (Part #: C32033-1, Supplier: 81205)

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C. Consumable Materials

Reference	Description	Specification
C00308	Compound - Corrosion Preventive, Petrolatum Hot Application	MIL-C-11796
C00528	Compound - Corrosion Preventive, Petroleum Hot Application (Soft Film)	MIL-C-11796 Class III
D00633	Grease - Aircraft General Purpose	BMS3-33
G00018	Nitrogen - Gaseous, Pressurizing, 99.5 Percent Pure	A-A-59503 Type I Grade B, MIL-PRF-27401 Type I Grade A
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38
G50225	Lockwire - MS20995C20, Corrosion Resistant Steel - 0.020 Inch (0.508 mm) Diameter	NASM20995
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

D. Expendables/Parts

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AMM Item	Description	AIPC Reference	AIPC Effectivity
11	Pin	32-21-00-02-005	JXB ALL
19	Strut assembly	32-21-21-02-155	JXB ALL
31	Pin	32-22-11-01A-005	JXB ALL
35	Pin	32-21-00-02-040	JXB ALL
43	Pin	32-22-11-01A-005	JXB ALL
59	Pin	32-21-00-02-040	JXB ALL
74	Packing	32-21-00-02-062	JXB ALL

E. Location Zones

Zone	Area
115	Nose Landing Gear Wheel Well - Left
116	Nose Landing Gear Wheel Well - Right
713	Nose Landing Gear

F. Access Panels

Number	Name/Location
113BW	Forward Nose Wheel Well Panel

G. Prepare for the Installation

SUBTASK 32-21-00-480-011

(1) If it is necessary, use a dolly or the Outrigger R/I Equipment, SPL-10305, to move the nose landing gear.

(a) When the Outrigger R/I Equipment, SPL-10305, is installed and will be left unattended, chock the nose landing gear tires and lock the caster brake (PAGEBLOCK 10-11-05/201).

SUBTASK 32-21-00-480-008

(2) Install the retention strap, SPL-1871, with the marker facing up to hold the shock strut [17] in its compressed position.

SUBTASK 32-21-00-620-001

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO

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Restore Nose Landing Gear

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THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

- (3) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:
- (a) The threads and thread reliefs of the bolt [62] and bolt [33]
 - (b) The threads of the nut [60] and nut [36]
 - (c) The faces of the washer [61] and washer [34].

SUBTASK 32-21-00-160-001

- (4) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate).

SUBTASK 32-21-00-640-002

- (5) Lubricate the following items with D00633 grease :
- (a) The shank of the bolt [62] and bolt [33]
 - (b) The shank of left trunnion pin [63] and right trunnion pin [37]
 - (c) The spacer [39] and spacer [71]
 - (d) The cable guard [70].

SUBTASK 32-21-00-160-002

- (6) Remove all unwanted D00633 grease .

H. Nose Landing Gear Installation

SUBTASK 32-21-00-560-002

- (1) You may optionally measure the net freeplay between the wheel well trunnion bushing faces and the strut trunnion bushing faces, before installation, as follows:
- NOTE: This measurement will assist in reduction of extra gear removals because the net freeplay requirement of 0.008 in. (0.20 mm) – 0.015 in. (0.38 mm) is not met after installation.**

- (a) Measure the distance between the wheel well trunnion bushing faces and make a record of it. [Value 1] _____.
- (b) Measure the distance over the strut trunnion bushing faces and make a record of it. [Value 2] _____.
- (c) Measure the thickness of the spacer [39] and spacer [71] removed as value 3. [Value 3] _____.
- (d) Subtract the three measurements. Value 1 - Value 2 - Value 3 = Value 4 _____.
- (e) Make sure that the value 4 equals 0.008 in. (0.20 mm) – 0.015 in. (0.38 mm) total axial freeplay.
- (f) If it is necessary, use a different spacer [39] and spacer [71] thickness to obtain the required freeplay dimension.

NOTE: It is recommended to make the spacer [39] and spacer [71] of equal thickness.

SUBTASK 32-21-00-560-001

- (2) Put the nose landing gear in its position for installation on the sidewall of the nose wheel well.

SUBTASK 32-21-00-580-004

- (3) Lift the nose landing gear with an axle jack until the right trunnion [23] and left trunnion [40] align with the mating hole in the sidewall of the nose wheel well (TASK 07-11-03-580-802).

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S-L	25/11/24

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SUBTASK 32-21-00-420-001

(4) Install the right trunnion pin [37] (View G, Figure 401).

(a) Put the spacer [39] in its position between the right trunnion [23] and the sidewall of the nose wheel well.

JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

(b) Apply D00633 grease , and install new packing [74] inside nose wheel well trunnion bushing.

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(c) Use the alignment pin assy, C32033-2, SPL-14477, to put the right trunnion pin [37] through the right trunnion [23], spacer [39], and sidewall of the nose wheel well.

(d) Use the nose landing gear trunnion pin alignment pin assembly, SPL-14481, to make sure that the hole in the right trunnion pin [37] for the bolt [33] aligns with the mating hole in the right trunnion [23].

(e) Put the swivel [24] through the right trunnion pin [37].

1) Make sure that the hole in the swivel [24] for the bolt [33] aligns with the mating hole in the right trunnion pin [37] and right trunnion [23].

(f) Put the bolt [33] through the right trunnion [23], right trunnion pin [37], and swivel [24].

(CR-2) (g) Install the washer [34] and nut [36] on the bolt [33].

(h) Tighten the nut [36] to 20 in-lb (2.3 N·m) - 24 in-lb (2.7 N·m) plus the run-on torque. 22 in-lb

(i) Install the cotter pin [35] in the bolt [33].

1) If it is necessary, loosen the nut [36] to the nearest castellation to align the holes for the cotter pin [35].

SUBTASK 32-21-00-420-002

(5) Install the left trunnion pin [63] (View K, Figure 401).

(a) Put the spacer [71] in its position between the left trunnion [40] and sidewall of the nose wheel well.

JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033

(b) Apply D00633 grease , and install new packing [74] inside nose wheel well trunnion bushing.

JXB ALL

(c) Use the alignment pin assy, C32033-2, SPL-14477, to put the left trunnion pin [63] through the left trunnion [40], spacer [71], and left sidewall of the nose wheel well.

(d) Use the nose landing gear trunnion pin alignment pin assembly, SPL-14481, to make sure that the hole in the left trunnion pin [63] for the bolt [62] aligns with the mating hole in the left trunnion [40].

(e) Put the cable guard [70] into the left trunnion pin [63].

1) Make sure that the hole in the cable guard [70] for the bolt [62] aligns with the mating hole in the left trunnion pin [63] and left trunnion [40].

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S.L MK1957	GAT 171 25/12/24

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737-600/700/800/900

Restore Nose Landing Gear

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- (f) Put the bolt [62] through the left trunnion [40], left trunnion pin [63], and cable guard [70].
- (g) Install the washer [61] and nut [60] on the bolt [62].
- (h) Tighten the nut [60] to 20 in-lb (2.3 N·m) - 24 in-lb (2.7 N·m) plus the run-on torque. *22 in-lb*
- (i) Install the cotter pin [59] in the bolt [62].
 - 1) If it is necessary, loosen the nut [60] to the nearest castellation to align the holes for the cotter pin [59].

SUBTASK 32-21-00-220-002

- (6) Measure the total axial freeplay in the trunnion.

NOTE: If the total axial freeplay is not correct, this will not cause any damage to the landing gear structure, but there may be a loud noise during landing gear extension/retraction.

- (a) Use a thickness gage to measure the clearance between the spacer [39], spacer [71], and bushing faces of the sidewall or the trunnion on the two sides of the nose landing gear.
- (b) Make sure that the combined clearance of the two trunnions is 0.008 in. (0.203 mm) - 0.015 in. (0.381 mm). *0.24 mm*
- (c) If it is necessary to get the correct clearance, adjust the spacer [39] and spacer [71] stack thickness.

SUBTASK 32-21-00-080-003

- (7) Remove the axle jack after the correct clearance has been achieved.

SUBTASK 32-21-00-420-003

- (8) Install each pulley [50] on the bracket [51] (View I, Figure 401).
 - (a) Put the NWS-A/B cables [49] in its position on the groove of the pulley [50].
 - (b) Put the cable guard [54] and pulley [50] in their position on the bracket [51].
 - (c) Install the washer [48] on the bolt [47].
 - (d) Put the bolt [47] through the pulley [50], cable guard [54], and bracket [51].
 - (e) Install the washer [53] and nut [52] on the bolt [47].

SUBTASK 32-21-00-420-004

CAUTION: MAKE SURE THAT THE STEERING CABLES DO NOT CROSS. CROSSED CABLES CAN INCREASE CABLE FRICTION WHICH CAN CAUSE DAMAGE TO THE CABLES.

- (9) Connect the NWS-A/B cables [49] (TASK 32-51-31-400-802).
 - (a) Make sure that the rig pin NS2 is in the drum for the Captain's control wheel.
 - (b) Put the NWS-A/B cables [49] through the inboard side of the trunnion pin nut [68], seal ring [69], and left trunnion pin [63].
 - 1) Make sure that the one NWS-A/B cable [49] goes over the trunnion pin bolt and one NWS-A/B cable [49] goes under the trunnion pin bolt where the cables go through the trunnion.
 - (c) For the pulleys [87] on the bracket that is outboard of the left trunnion pin [63], do these steps:
 - 1) Put the NWS-A/B cables [49] in the grooves of the left trunnion pin [63] and through the guard [90].
 - 2) Install the pulleys [87], bolts [88], and washers [89].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR

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SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

- (CR-2) a) Install the bolts [88] with G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (optional).
- b) Tighten the bolts [88] to 73 in-lb (8.2 N-m) - 77 in-lb (8.7 N-m). 75 in-lbs
- (d) Apply a thin layer of C00308 corrosion preventive compound, or C00528 compound, to the threads on the internal and external surfaces of the mating adjustable parts on the cable turnbuckles [1].
- (e) Connect the NWS-A/B cables [49] to the cable turnbuckles [1] (View B, Figure 401).
- (f) Remove the identification tags attached to the NWS-A/B cables [49].
- (g) Install the locking clips [2] on the cable turnbuckles [1].

SUBTASK 32-21-00-410-001

- (10) Install the seal [65] (View J, Figure 401).
 - (a) Assemble the seal [65] around the cables [49] at the inboard end of the left trunnion pin [63].
 - (b) Install one retainer [66] on the eyebolt [67].
 - (c) Put the eyebolt [67] through the seal [65].
 - (d) Install the other retainer [66], washer [56], and nut [64] on the eyebolt [67].
 - (e) Tighten the nut [64] sufficiently to clamp the seal [65].
 - (f) Loosen the nut [64] to approximately 1/4 turn until you can turn the eyebolt [67] with your fingers.
 - (g) Put the seal [65] into the left trunnion pin [63] until it touches the stop.
 - (h) Install the seal ring [69] into the left trunnion pin [63].

SUBTASK 32-21-00-620-002

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

- (11) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to the trunnion pin nut [68].

SUBTASK 32-21-00-420-005

- (12) Install the trunnion pin nut [68] (View J, Figure 401).
 - (a) Install the trunnion pin nut [68] on the inboard side of the left trunnion pin [63].

MECH	INSP
S.L.	GAT 171 26/12/24
S.L.	GAT 171 26/12/24 0615702
S.L. M121957	GAT 171 26/12/24
S.L.	GAT 171 26/12/24
S.L.	GAT 171 26/12/24
S.L.	GAT 171 26/12/24
S.L.	GAT 171 26/12/24
S.L. M122064	GAT 171 02/01/25

PARTIAL SIGN OFF STATUS:

Item: 8 Completed through item: 10, 11 Sign: [Signature] 20/12/24

Item: _____ Completed through item: _____ Sign: _____



737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

- (b) Use the spanner wrench, SPL-1821, or spanner wrench set, SPL-1559, to tighten the trunnion pin nut [68] to 100 in-lb (11.3 N·m) - 150 in-lb (16.9 N·m).
- (c) If it is necessary, loosen the trunnion pin nut [68] to align with the vernier of the lock holes.
- (d) Install the lock [57] on the inboard side of the left trunnion pin [63].
- (e) Install the screw [58] that holds the lock [57] to the left trunnion pin [63].
- (f) Install the lockwire on the screw [58].

SUBTASK 32-21-00-710-001

- (13) Make sure that the seal [65] will turn freely from the extended cable position of the gear to a simulated gear retracted cable position.

NOTE: This will verify that the cables will not become twisted when the gear is retracted.

SUBTASK 32-21-00-620-003

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

- (14) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:
 - (a) The threads and thread reliefs of the bolt [27] and the bolt [47]
 - (b) The threads of the nut [32] and the nut [42]
 - (c) The faces of the washer [28], washer [30], washer [44], and washer [46].

SUBTASK 32-21-00-160-005

- (15) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate).

SUBTASK 32-21-00-640-003

- (16) Lubricate the chrome plated surfaces of the bolt [27], bolt [47], bushing [29], and bushing [45] with D00633 grease .

SUBTASK 32-21-00-160-006

- (17) Remove all unwanted D00633 grease .

SUBTASK 32-21-00-420-006

- (18) Connect the left upper rod assembly [41] (View H, Figure 401).
 - (a) Put the upper rod assembly [41] in its position on the clevis of the left trunnion [40].
 - (b) Put the bushing [45] through the clevis of the left trunnion [40] and rod end of the upper rod assembly [41].
 - (c) Install the washer [46] on the bolt [47].

MECH	INSP
<i>[Signature]</i>	<i>[Signature]</i> 02/10/24
<i>[Signature]</i>	<i>[Signature]</i> 04/10/24
<i>[Signature]</i>	<i>[Signature]</i> 02/11/24
<i>[Signature]</i>	<i>[Signature]</i> 26/11/24
<i>[Signature]</i>	<i>[Signature]</i> 26/11/24
<i>[Signature]</i>	<i>[Signature]</i> 26/11/24
<i>[Signature]</i>	<i>[Signature]</i> 26/11/24
<i>[Signature]</i>	<i>[Signature]</i> 26/11/24
<i>[Signature]</i>	<i>[Signature]</i> 26/11/24

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



flydubai

737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA:32--

Flow:-

Work Area:-

INDEPENDENT INSPECTION
Sandipe
26/12/24
0715
GAT
ATD

INDEPENDENT INSPECTION
Sandipe
26/12/24
0725
GAT
ATD

INDEPENDENT INSPECTION
Sandipe
26/12/24
0805
GAT
ATD

INDEPENDENT INSPECTION
Sandipe
26/12/24
0815
GAT
ATD

(d) Put the bolt [47] through the clevis of the left trunnion [40] and rod end of the upper rod assembly [41].

(e) Install the washer [44] and nut [42] on the bolt [47].

(CR-2) (f) Tighten the nut [42] to 90 in-lb (10.2 N·m) - 125 in-lb (14.1 N·m). 100 in-lb

(g) Install the pin [43] in the bolt [47].

(CR-2) 1) If it is necessary, loosen the nut [42] to the nearest castellation to align the holes for the pin [43].

SUBTASK 32-21-00-420-007

(19) Connect the right upper rod assembly [26] (View F, Figure 401).

(a) Put the upper rod assembly [26] in its position on the clevis of the right trunnion [23].

(b) Put the bushing [29] through the clevis of the right trunnion [23] and rod end of the upper rod assembly [26].

(c) Install the washer [28] on the bolt [27].

(d) Put the bolt [27] through the clevis of the right trunnion [23] and rod end of the upper rod assembly [26].

(e) Install the washer [30] and nut [32] on the bolt [27].

(f) Tighten the nut [32] to 90 in-lb (10.2 N·m) - 125 in-lb (14.1 N·m). 100 in-lb

(g) Install the pin [31] in the nut [32].

1) If it is necessary, loosen the nut [32] to the nearest castellation to align the holes for the pin [31].

SUBTASK 32-21-00-420-008

(20) Connect the hydraulic lines [25] (View E, Figure 401).

(a) Remove the plugs from the hydraulic ports of the swivel [24].

(b) Remove the caps from the hydraulic lines [25].

(c) Connect the hydraulic lines [25] to the swivel [24].

(d) Remove the tags from the swivel [24] ports and hydraulic lines [25].

SUBTASK 32-21-00-620-004

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(21) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:

(a) The threads and thread reliefs of the pin [18]

(b) The threads of the nut [12]

(c) The faces of the washer [20] and washer [21].

SUBTASK 32-21-00-160-003

MECH	INSP	GAT
S.L	26/12/24 0710	GAT 171
S.L	26/12/24 0720	GAT 171
S.L	26/12/24 0800	GAT 171
S.L	26/12/24 0810	GAT 171
S.L	25/12/24 2620	GAT 171
S.L	25/12/24	GAT 171
S.L MK1952	25/12/24	GAT 171

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow:--

Work Area:--

(22) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate).

SUBTASK 32-21-00-640-004

(23) Lubricate the shank of the pin [18] and bolt [13] with D00633 grease .

SUBTASK 32-21-00-160-004

(24) Remove all unwanted D00633 grease .

SUBTASK 32-21-00-420-009

(25) Connect the lower drag strut assembly [19] (View D, Figure 401).

(a) Make sure that the drag brace lubrication fittings are on the right side.

(b) Put the lower drag strut assembly [19] in its position on the clevis of the shock strut [17].

(c) Put the pin [18] through the clevis of the shock strut [17] and the lower drag strut assembly [19].

(d) Install the washer [20] and nut [12] on the pin [18].

(e) Tighten the nut [12] to 600 in-lb (67.8 N·m) - 700 in-lb (79.1 N·m). *650 in-lb*

1) If it is necessary, loosen the nut [12] to the nearest castellation to align the holes for the bolt [13].

(f) Install the bolt [13] in the pin [18].

(g) Install the washer [21] on the bolt [13].

(h) Install the nut [22] on the bolt [13].

(i) Install the pin [11] in the bolt [13].

1) If it is necessary, loosen the nut [22] to the nearest castellation to align the holes for pin [11].

SUBTASK 32-21-00-420-012

(26) Do this task: Nose Landing Gear Wheel and Tire Assembly - Installation, TASK 32-45-21-400-801.

SUBTASK 32-21-00-420-011

(27) Do this task to connect the wires to the terminal block [72] (View C, Figure 401).

(a) For airplanes that have three rigid conduits on the junction box [5], open the tee fitting on the aft side of the junction box [5].

(b) Pull the wires through the conduit.

1) Attach the strings that hang out of the end of the rigid conduits [4] to the wires that hang out of the flexible conduits [73].

2) Pull the wires through the rigid conduits [4] to the junction box [5].

3) Connect the flexible conduits [73] after the clamps [8] at point "A".

4) If it is necessary, install G50225 MS20995C20 lockwire , on the flexible conduits [73].

NOTE: The jam nuts with safety wire holes drilled through the corners must not be lockwired.

5) Remove the strings attached to the wires.

(c) Connect the wires to the terminal block [72] with a contact insertion / removal tool, COM-4938 , tool using the drawing you made during the removal.

(d) Connect the terminal block [72] to the junction box [5].

(e) Install the screws [10] and washers [3] to close the junction box cover [9] on the junction box [5].

SUBTASK 32-21-00-420-017

MECH	INSP
S.L	GAT 171 25/12/24
S.L	GAT 171 25/12/24
S.L	GAT 171 25/12/24 0635
S.L MEMO	GAT 171 26/12/24
R	GAT 128 26/12/24
R	GAT 128 26/12/24
R	GAT 128 26/12/24

INDEPENDENT INSPECTION

GAT 470
0640
25/12/24

PARTIAL SIGN OFF STATUS:

Item: 8 Completed through item: H, 25 Sign: [Signature] *25/12/24*

Item: _____ Completed through item: _____ Sign: _____



737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

(28) Do this task to connect the wires to the terminal block [81] (View L, Figure 401).

(a) Pull the wires through the conduit.

1) Attach the strings that hang out of the end of the rigid conduit [83] to the wires that hang out of the flexible conduit [80].

2) Pull the wires through the rigid conduit [83] to the junction box [82].

3) Connect the flexible conduit [80] to the rigid conduit [83].

4) If it is necessary, install G50225 MS20995C20 lockwire, on the flexible conduit [80].

NOTE: The jam nuts with safety wire holes drilled through the corners must not be lockwired.

5) Remove the strings attached to the wires.

(b) Connect the wires to the terminal block [81] with a contact insertion / removal tool, COM-4938, tool using the drawing you made during the removal.

(c) Connect the terminal block [81] to the junction box [82].

(d) Install the screws [85] and washers [84] to close the junction box cover [86] on the junction box [82].

i. Put the Airplane Back to Its Usual Condition

SUBTASK 32-21-00-410-002

(1) Close this access panel:

(TASK 53-14-01-420-801)

Number	Name/Location
113BW	Forward Nose Wheel Well Panel

SUBTASK 32-21-00-080-004

(2) Remove the retention strap, SPL-1871, from the nose landing gear.

SUBTASK 32-21-00-610-003

(3) Lubricate the nose landing gear (TASK 12-21-21-640-801, TASK 12-21-21-640-802).

SUBTASK 32-21-00-610-001

(4) To fully service the shock strut with the airplane on jacks, hold the shock strut in a vertical position and do these steps to service the shock strut with fluid:

(a) Install on the shock strut a retention strap, SPL-1871.

(b) Do this task: Nose Landing Gear Shock Strut Servicing, Airplane on Jacks, TASK 12-15-41-610-805.

(c) Remove the retention strap, SPL-1871, from the nose landing gear.

SUBTASK 32-21-00-610-002

(5) To inflate the shock strut with the airplane on jacks, do these steps:

NOTE: To do these steps, you must have accomplished the steps to add fluid to the strut at the start of this task.

(a) Install the tool, SPL-1521, on the gas valve.

(b) Inflate the shock strut with G00018 nitrogen, until you reach the fully extended pressure that is on the servicing placard (TASK 12-15-41-610-805).

(c) Close the gas valve.

(d) Remove the tool, SPL-1521, from the gas valve.

(e) Install the cap on the gas valve.

MECH	INSP
h	26/12/24 GAT 128
h	26/12/24 GAT 128
h	26/12/24 GAT 128
h	02/04/25 GAT 171
h	26/11/24 GAT 171
h	26/11/24 GAT 171
h	26/11/24 GAT 171

JXB 004, 005, 007-009, 014, 026-031, 033, 036, 037, 040-051, 053

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

PW: QD2R200
 SN: 0219500576
 CALOUT: 26/08/2025

QD 1R50
 0918001437
 28/11/2024

INFL. TOOL
 PN: F70200-14
 SN: PL-005
 CALOUT: 17/09/2025

Repair
 26/12/24

26/12/24
 GAT 128

737-600/700/800/900

Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

INDEPENDENT INSPECTION
Sandipl
02/01/25
0840VRC
GAT 470

INDEPENDENT INSPECTION
Sandipl
02/01/25
0935VRC
GAT 470

SUBTASK 32-21-00-710-003

(11) Do this task: Operational Test for the Nose Landing Gear, TASK 32-33-00-710-801.

NOTE: It is optional to perform the Nose Landing Gear Test - Component Replacement, TASK 32-33-00-710-802 instead of the operational test if a hydraulic cart is not available.

SUBTASK 32-21-00-710-004

(12) Do this task: Nose Gear Manual Extension System Test - Airplane on Jacks, TASK 32-35-00-730-801.

SUBTASK 32-21-00-820-001

(13) Adjust the NWS-A/B cables [49] (TASK 32-51-00-820-802).

SUBTASK 32-21-00-710-005

(14) Do this task: Nose Wheel Steering System Test, TASK 32-51-00-700-801.

SUBTASK 32-21-00-860-008

(15) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-21-00-580-006

(16) If airplane was lifted, do this task: Lower the Airplane Off the Jacks, TASK 07-11-01-580-816.

SUBTASK 32-21-00-860-009

(17) Do this task: Lower the Airplane Nose Off of the Jack, TASK 07-11-21-580-802.

SUBTASK 32-21-00-200-001

(18) Check the pressure and extension of the shock strut.

(a) Measure the pressure and extension of the shock strut.

1) Refer to the servicing placard and make sure that the pressure you measure is still correct for the extension of the shock strut (TASK 12-15-41-610-801).

2) If the pressure is not correct for the extension of the shock strut, fully service the shock strut (TASK 12-15-41-610-802).

NOTE: If the pressure is not correct for the extension of the shock strut, the shock strut does not have the proper amount of fluid.

MECH INSP

X

X

X

X

X

X

X

X

GAT 171

GAT 171

GAT 171

GAT 171

GAT 171

GAT 171

GAT 171

GAT 171

GAT 171

GAT 171

GAT 171

PARTIAL SIGN OFF STATUS:

Item: 8 Completed through item: 5, 12 Sign: [Signature] 11/125

Item: _____ Completed through item: _____ Sign: _____



737-600/700/800/900

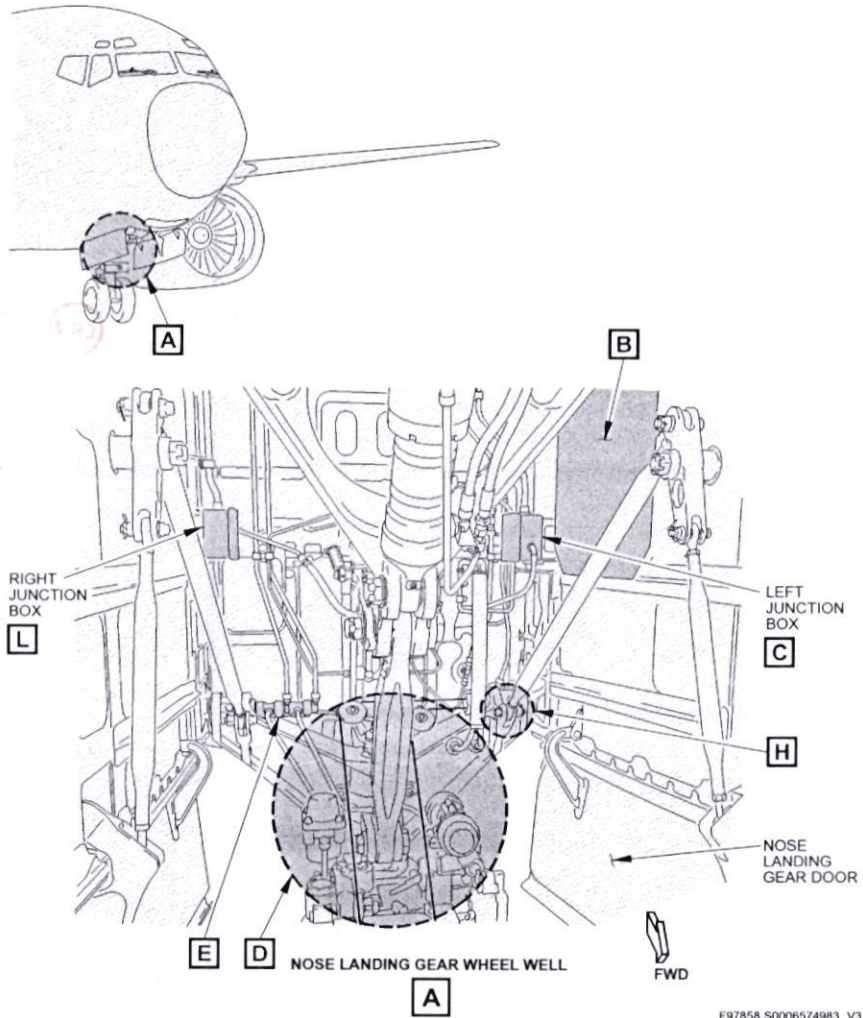
Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow:-

Work Area:-



F97858 S0006574983_V3

S. M. MR1917

GAT 171
25/12/2024

Figure 401. Nose Landing Gear Installation - Sheet 1
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



GCAA APPROVAL No : UAE.145.1232

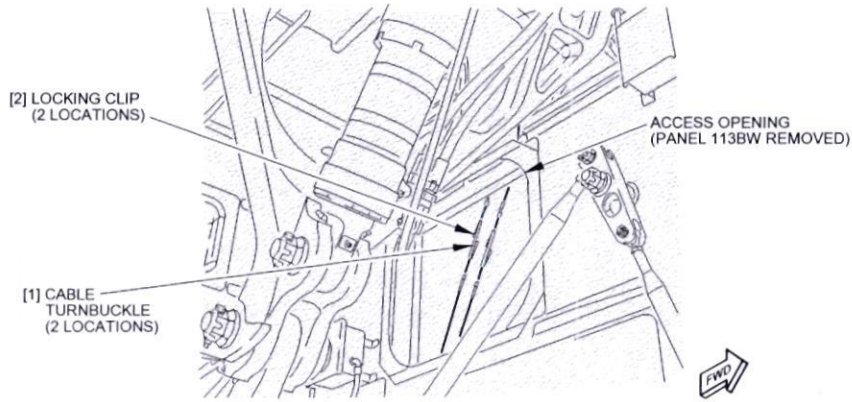
Restore Nose Landing Gear

Type: Routine Card

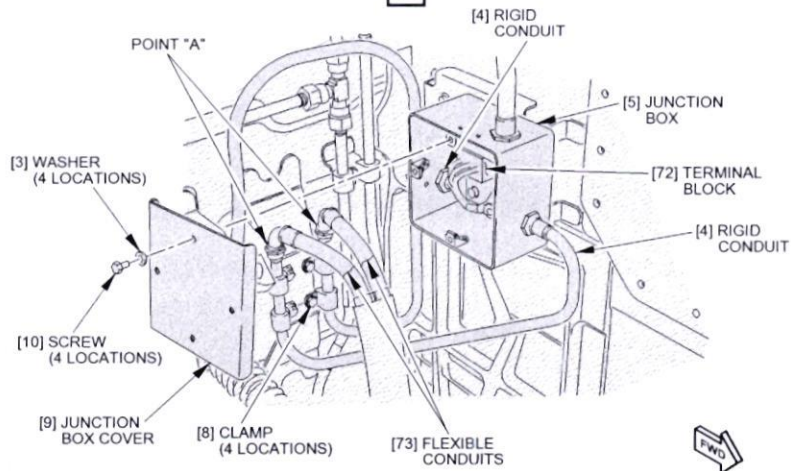
ATA: 32--

Flow:-

Work Area:-



B



(AIRPLANES WITH 3 CONDUITS TO THE JUNCTION BOX)

C

F98309 S0006574984_V7

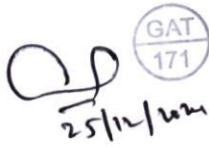


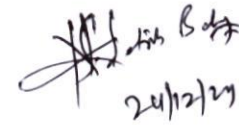
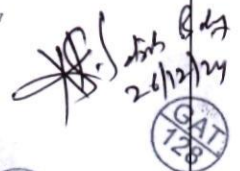






Figure 401. Nose Landing Gear Installation - Sheet 2

TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

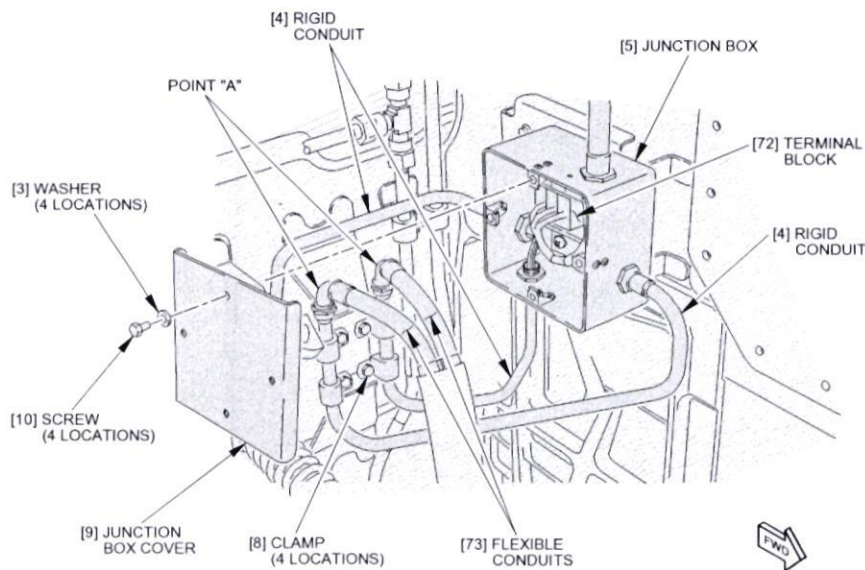
Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232



(AIRPLANES WITH 4 CONDUITS TO THE JUNCTION BOX)

C

J73703 50000177295_V4

NA

25/11/2024

NA

 24/12/24

Figure 401. Nose Landing Gear Installation - Sheet 3
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

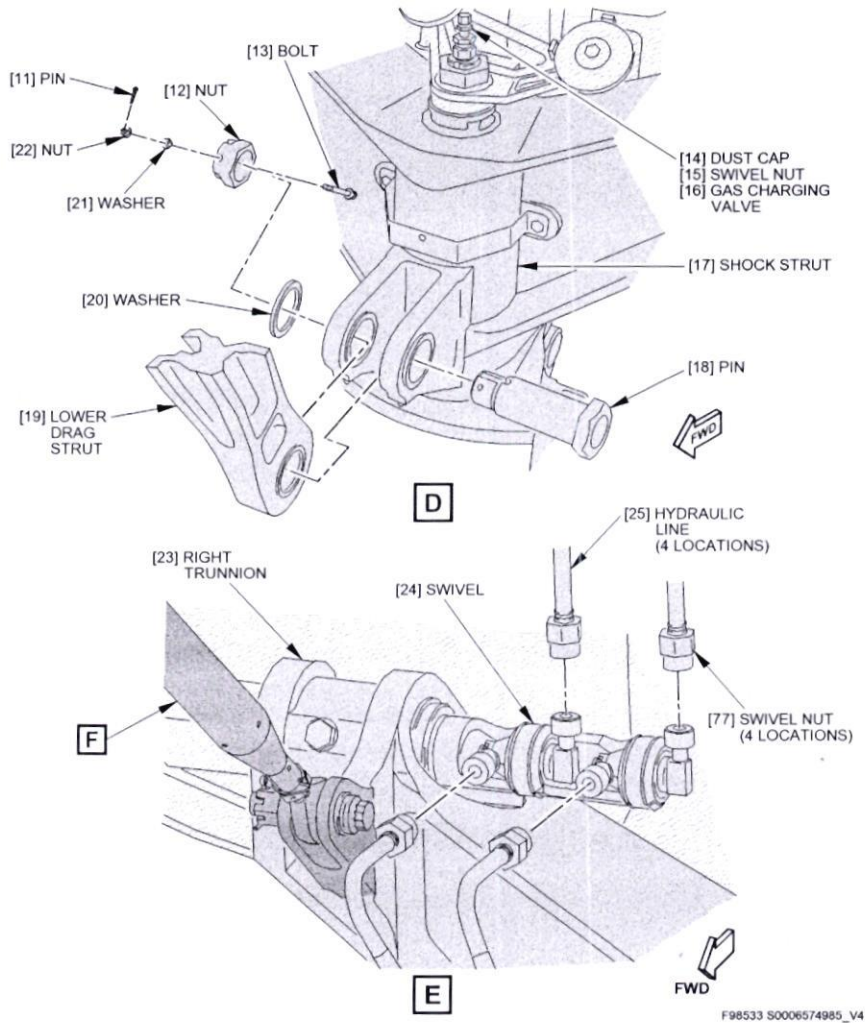
Rev # 41



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Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232



F98533 50006574985_V4

[Signature]
25/11/2024
GAT
171

S. King
ME1957

Figure 401. Nose Landing Gear Installation - Sheet 4
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

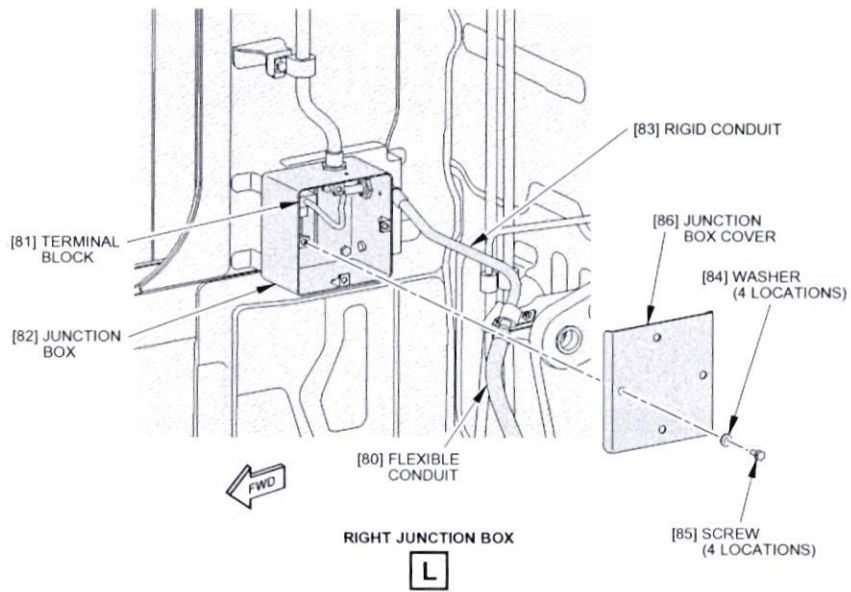
Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT





3022968 50000795208_V1

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Figure 401. Nose Landing Gear Installation - Sheet 5
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



* 3 2 - 0 9 0 - 0 0 - 0 1 *

GCAA APPROVAL No : UAE.145.1232

737-600/700/800/900

Restore Nose Landing Gear

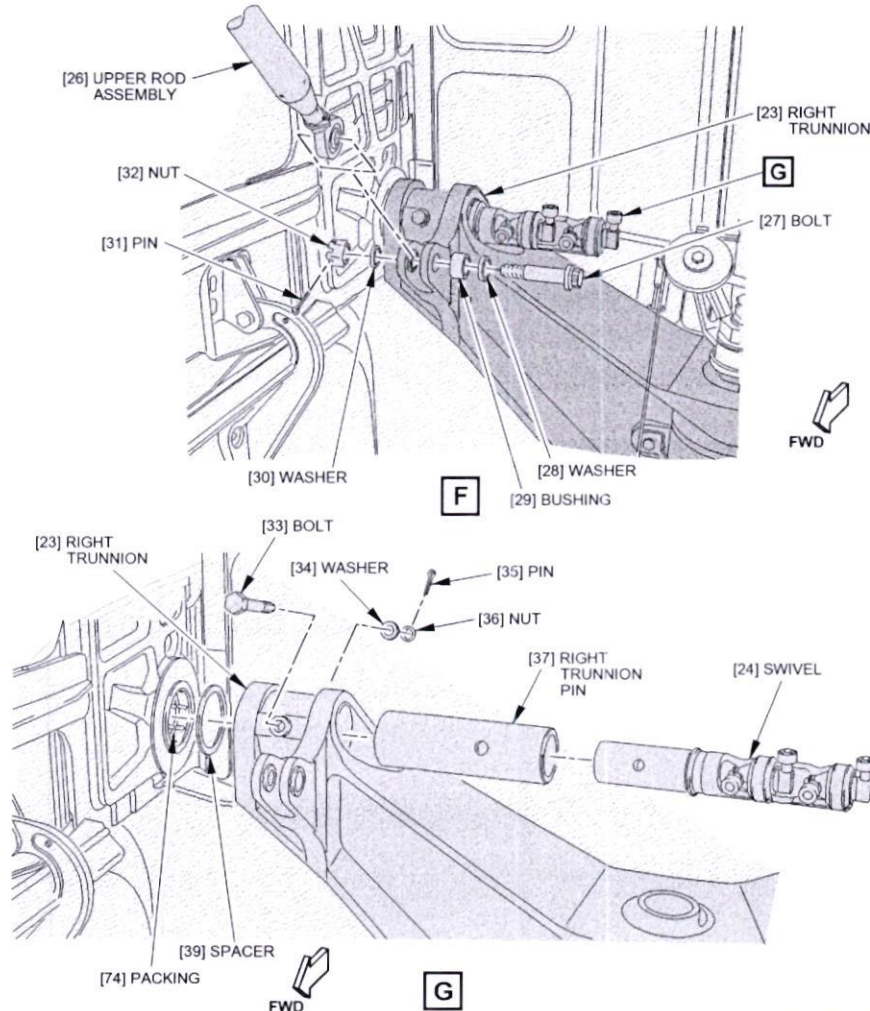
Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033



F99862 S0006574986_V8

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 25/11/2024
 GAT 171

Handwritten signature
 S. Kemp
 MK1952

Figure 401. Nose Landing Gear Installation - Sheet 6
 TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

737-600/700/800/900

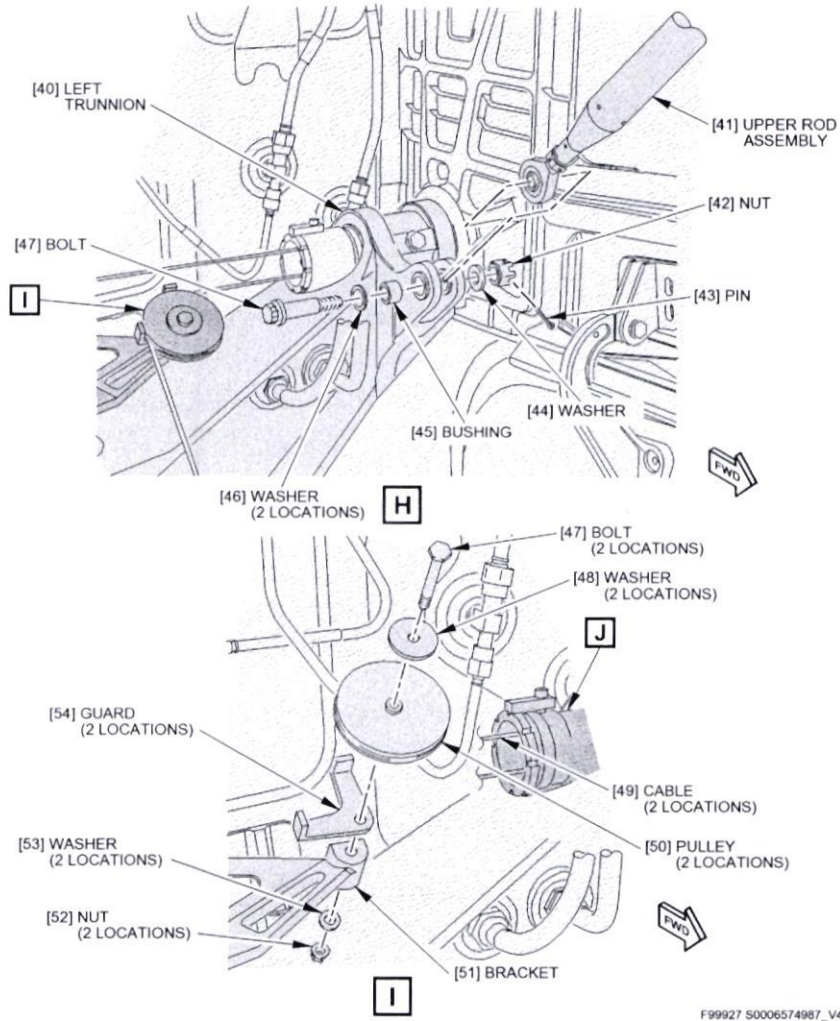
Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow:-

Work Area:-



F99927 50006574987_V4

S. L. M. 22/10/24

GAT 171 25/10/24

Figure 401. Nose Landing Gear Installation - Sheet 7
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

Restore Nose Landing Gear

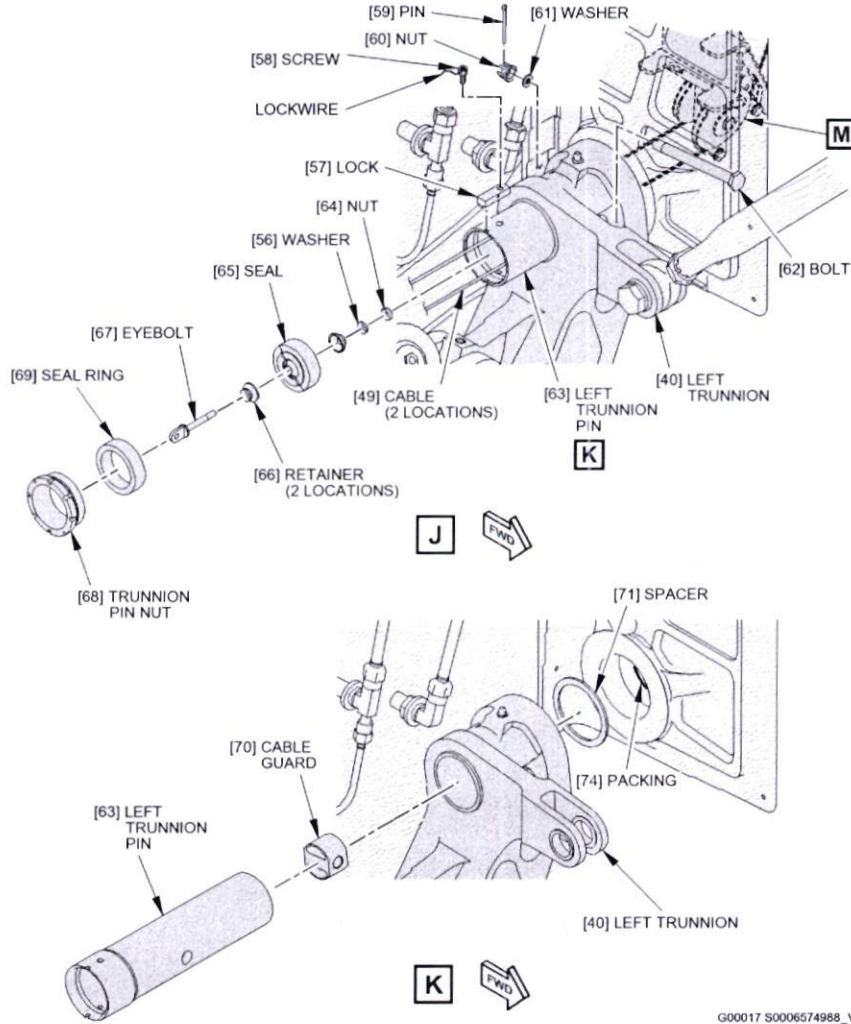
Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033



G00017 S0006574988_V8

Handwritten signatures and stamps:
 [Signature]
 25/11/2011
 GAT 171
 S.L. MR1952

Figure 401. Nose Landing Gear Installation - Sheet 8
 TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



737-600/700/800/900

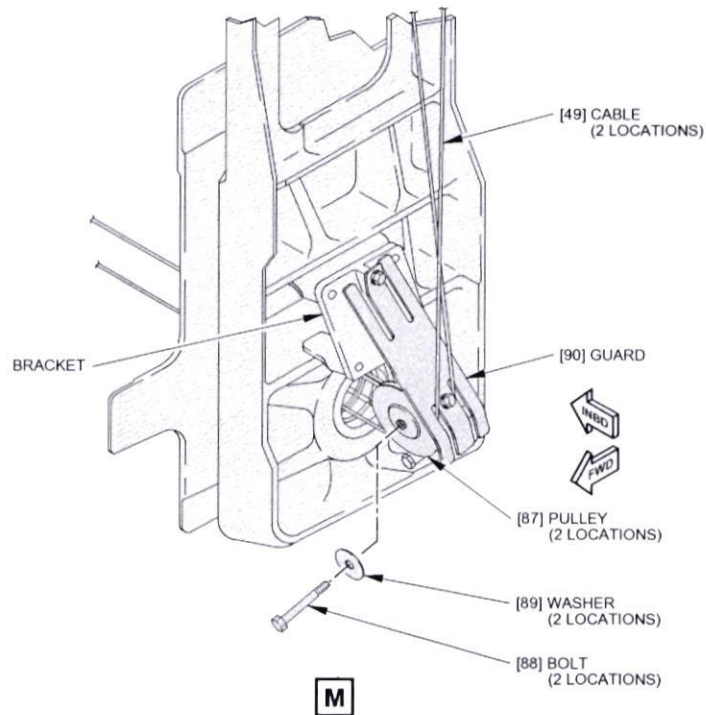
Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -



3032302 50000802915_V1

S.L
MR1952

GAT
171
25/14/2024

Figure 401. Nose Landing Gear Installation - Sheet 9
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



GCAA APPROVAL No : UAE.145.1232

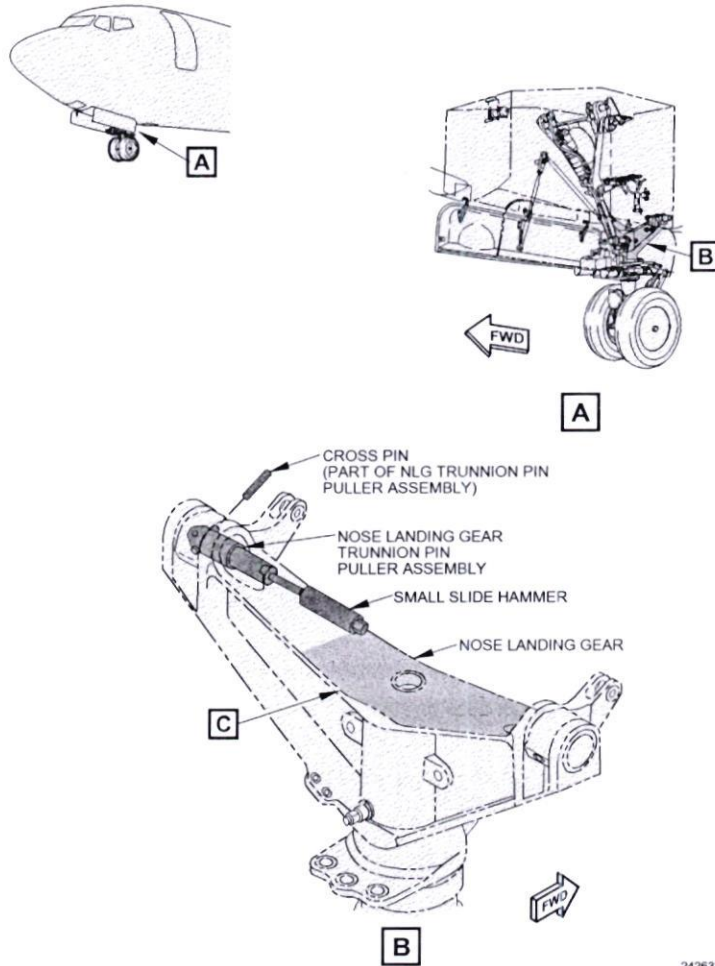
Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -



2426322 S0000560710_V1

Op *5/11/2024* *GAT 171*
S.L. *MK1952*

Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment - Sheet 1
TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

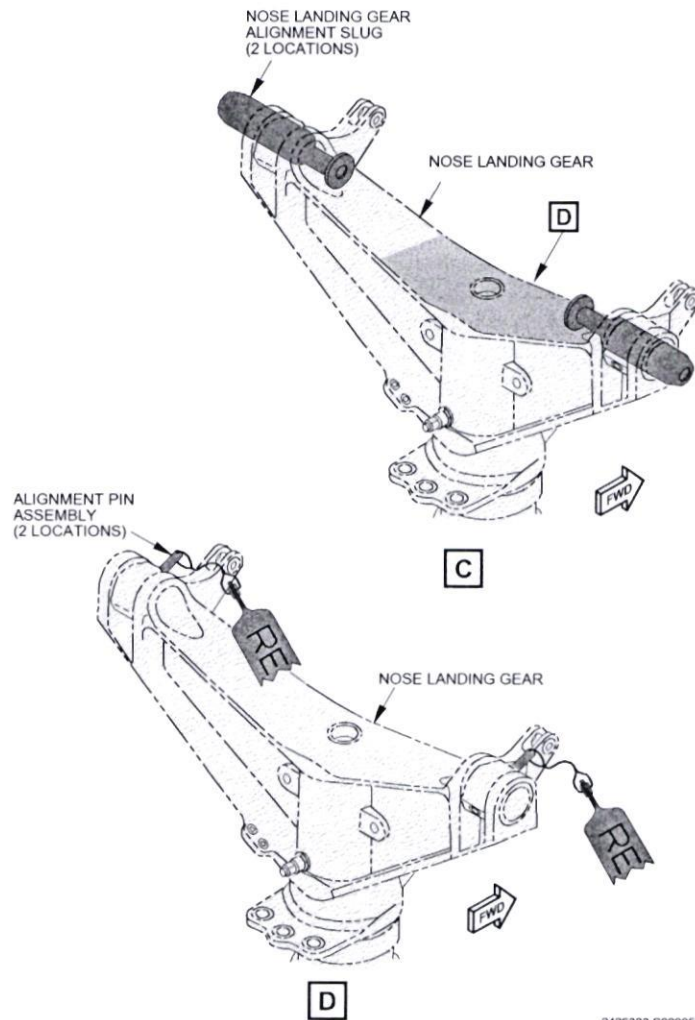
Rev # 41

Rev Date: Oct 17, 2024 PDT



* 3 2 - 0 9 0 - 0 0 - 0 1 *

GCAA APPROVAL No : UAE.145.1232



2426333 50000560711_V1

*S. Lang
12/19/52*

[Signature]
GAT
171

Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment - Sheet 2

TASK 32-21-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

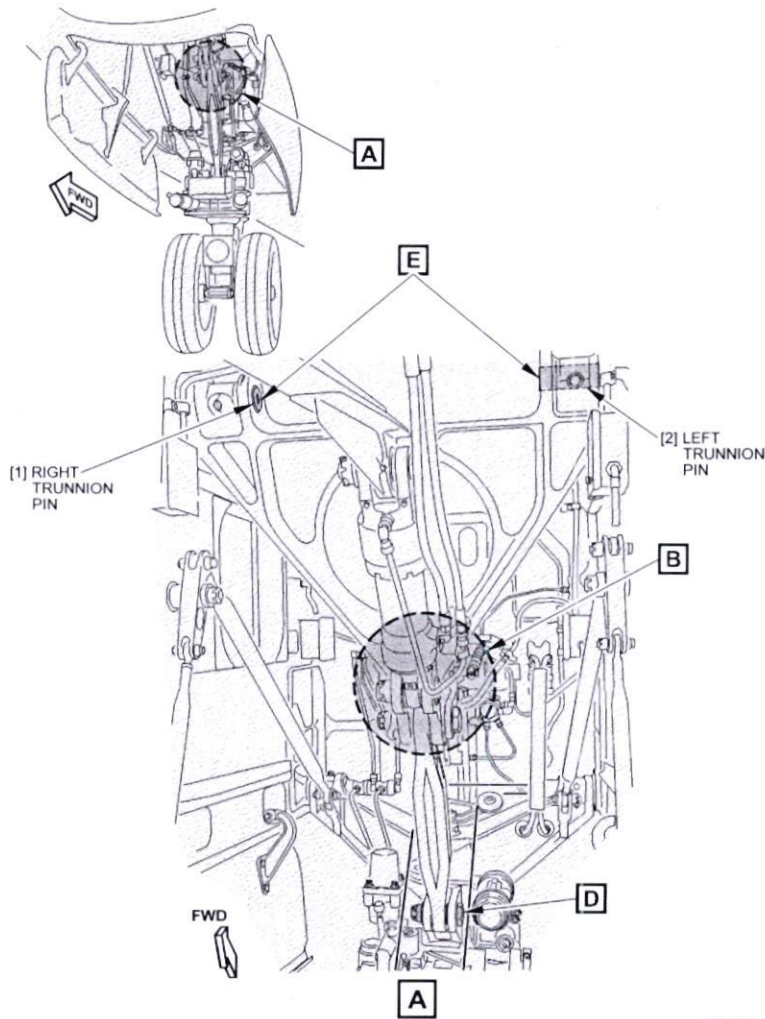
Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232



G27830 S0006575018_V2

Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 1
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



* 3 2 - 0 9 0 - 0 0 - 0 1 *

Restore Nose Landing Gear

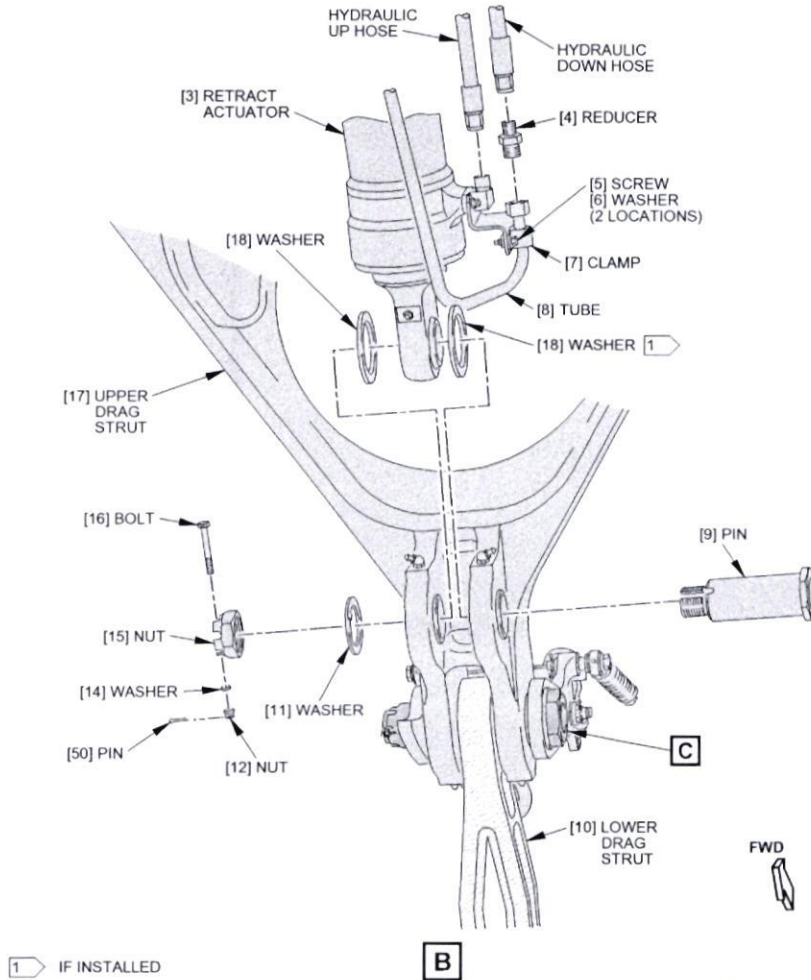
Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES



G27832 50006575019_V5

N/A

 24/11/2024
 GAT 171

Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 2
 TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

Restore Nose Landing Gear

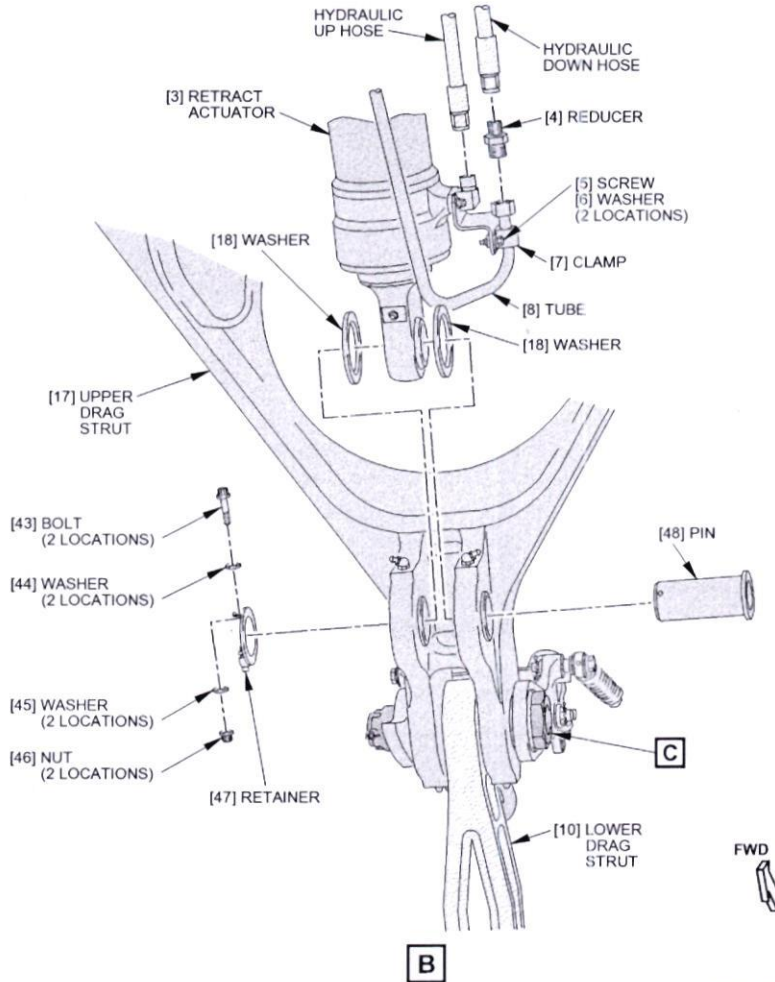
Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES



2944615 S0000721240_V2

Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 3
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

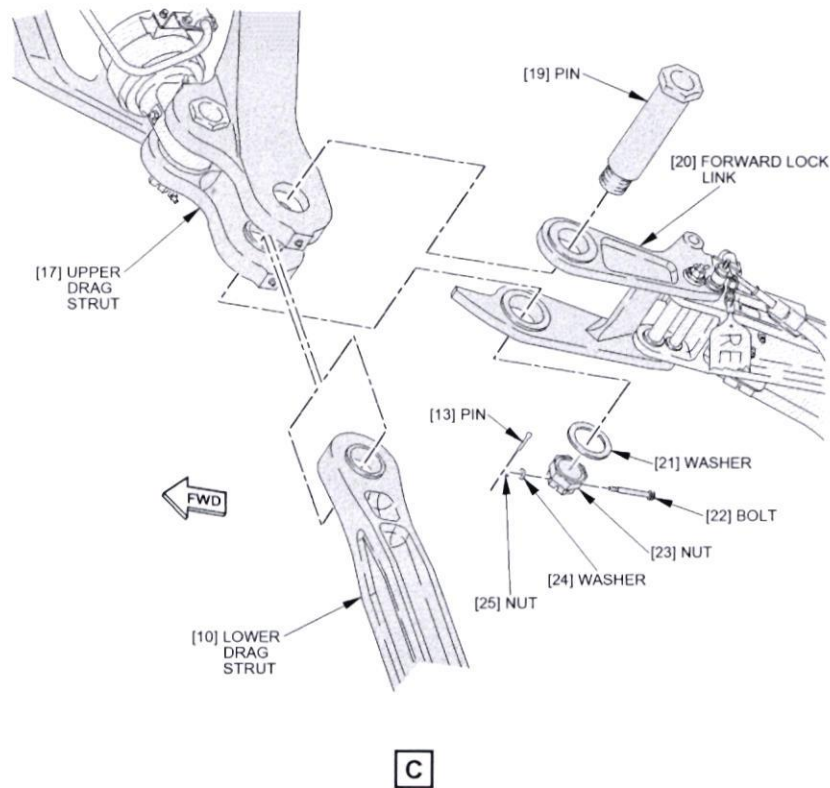
Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232



G27834 S0006575020_V3

S. Lutf
M121452

[Signature]
25/11/2024

GAT
171

Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 4
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

737-600/700/800/900

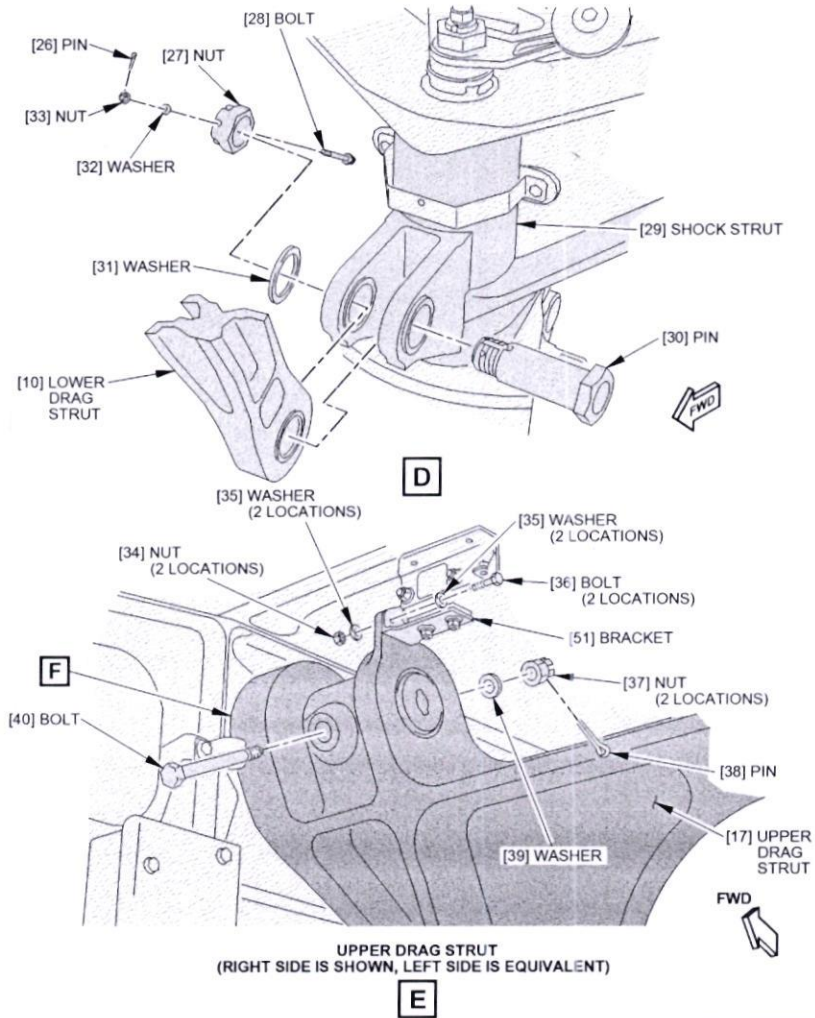
Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -



G27837 50006575021_V3

[Signature]
 25/12/2024
 GAT 171

s. long
 MK1952

[Signature]
 24/12/2024
 GAT 128

Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 5
 TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

737-600/700/800/900

Restore Nose Landing Gear

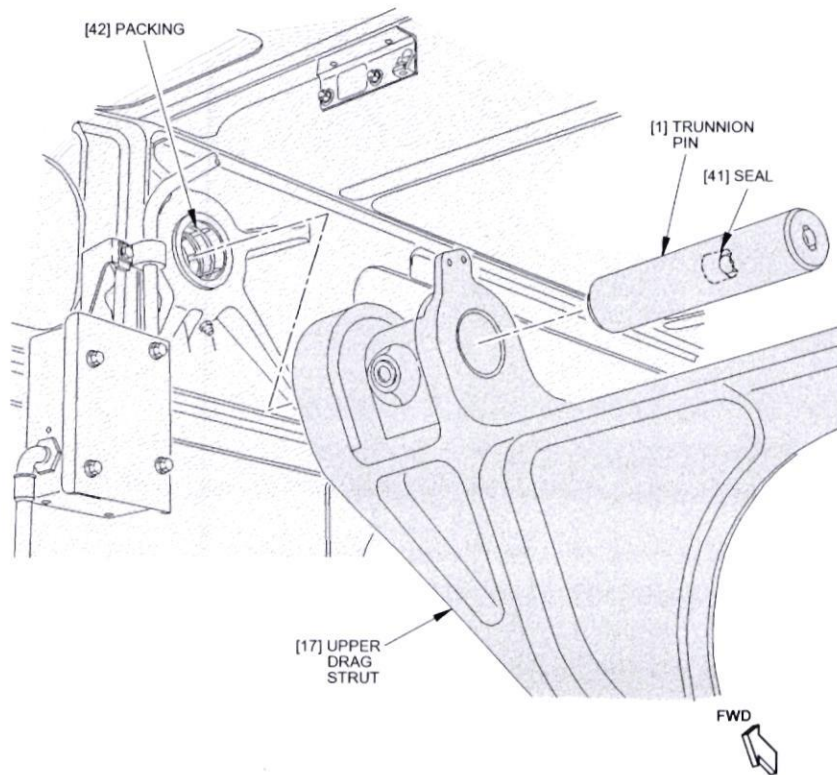
Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH PACKING PART NUMBER MS29513-031 OR MS29513-033



UPPER DRAG STRUT (RIGHT SIDE IS SHOWN, LEFT SIDE IS EQUIVALENT)

F

2973674 S0000752853_V2

GAT 171

Handwritten signature: S. Al... M21957

Handwritten signature: [Signature] 25/11/2024

Figure 401. Nose Landing Gear Drag Strut Installation - Sheet 6
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

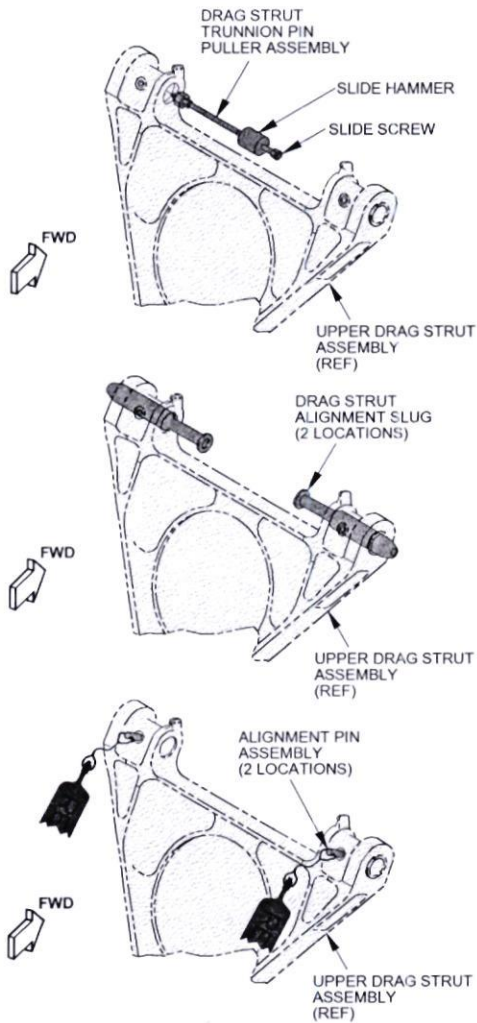
Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow:-

Work Area:-



2387744 S0000548478_V1

GAT
171

Handwritten signatures and initials:
 [Signature]
 S. King
 MR1957

Figure 402. Drag Strut Trunnion Pin Removal and Installation Equipment - Sheet 1
TASK 32-21-21-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

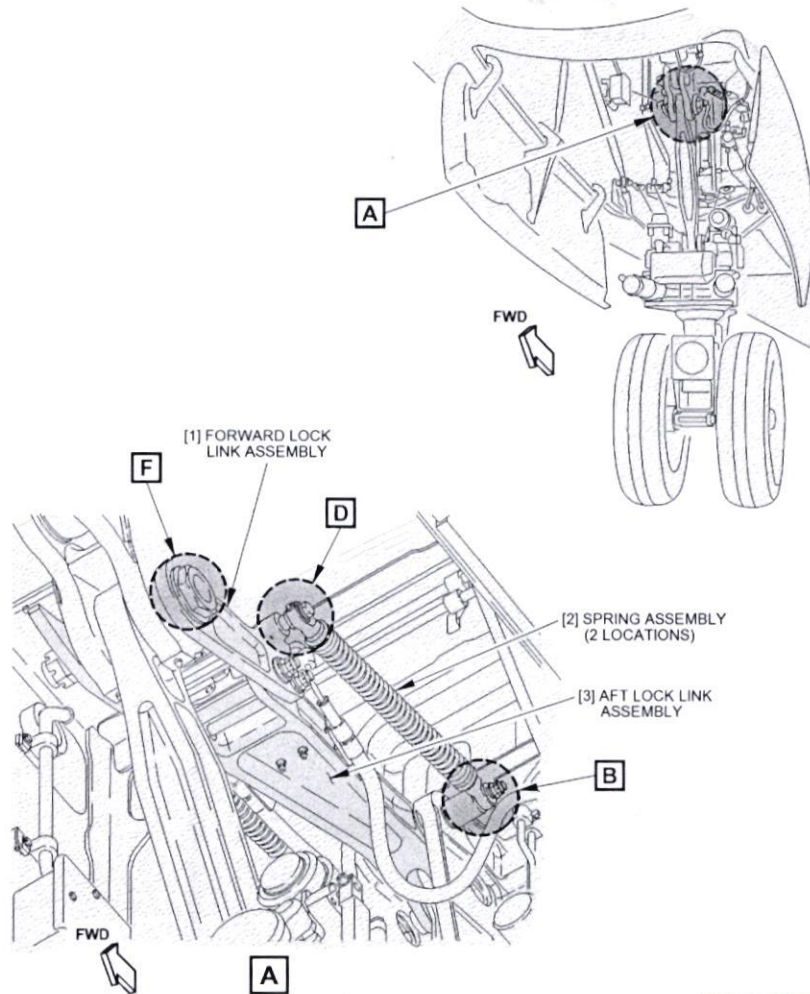
Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232



F95062 50006575286_V2

S. Lwt
M21957

[Signature]
 25/11/24

GAT
171

Figure 401. Nose Gear Lock Mechanism Installation - Sheet 1
TASK 32-33-51-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

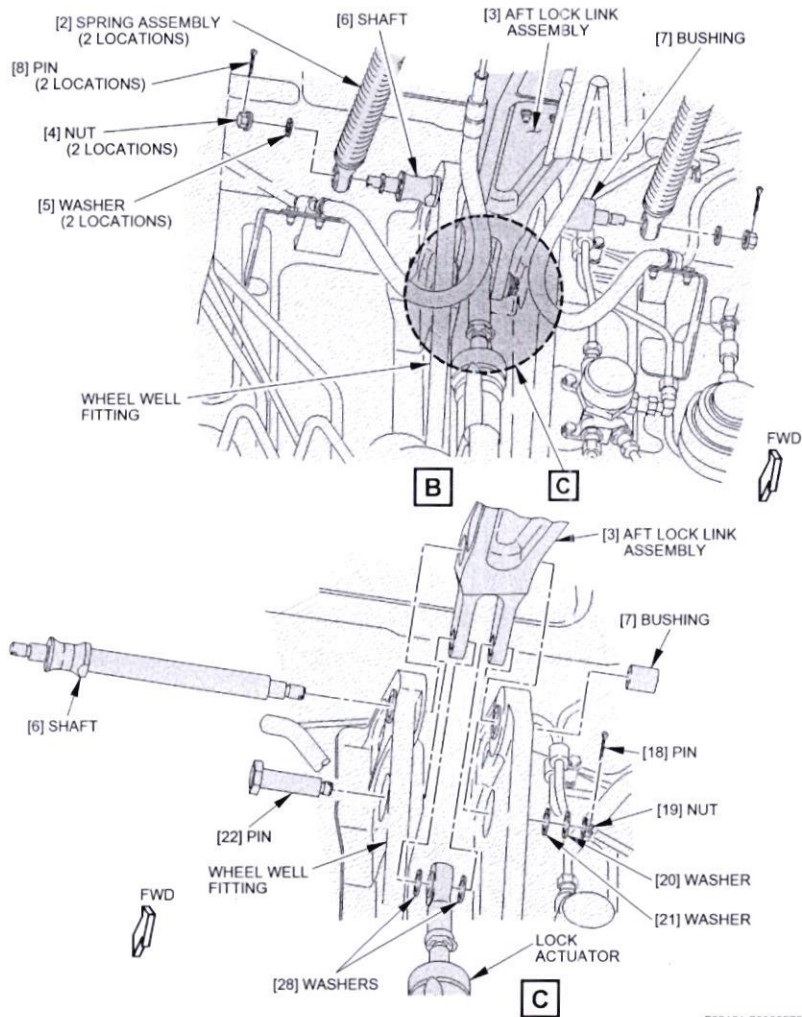
Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -



F95161 S0006575287_V5

[Signature]
 25/12/2024
 GAT
 171

S. Long
 MR1957

Figure 401. Nose Gear Lock Mechanism Installation - Sheet 2
 TASK 32-33-51-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



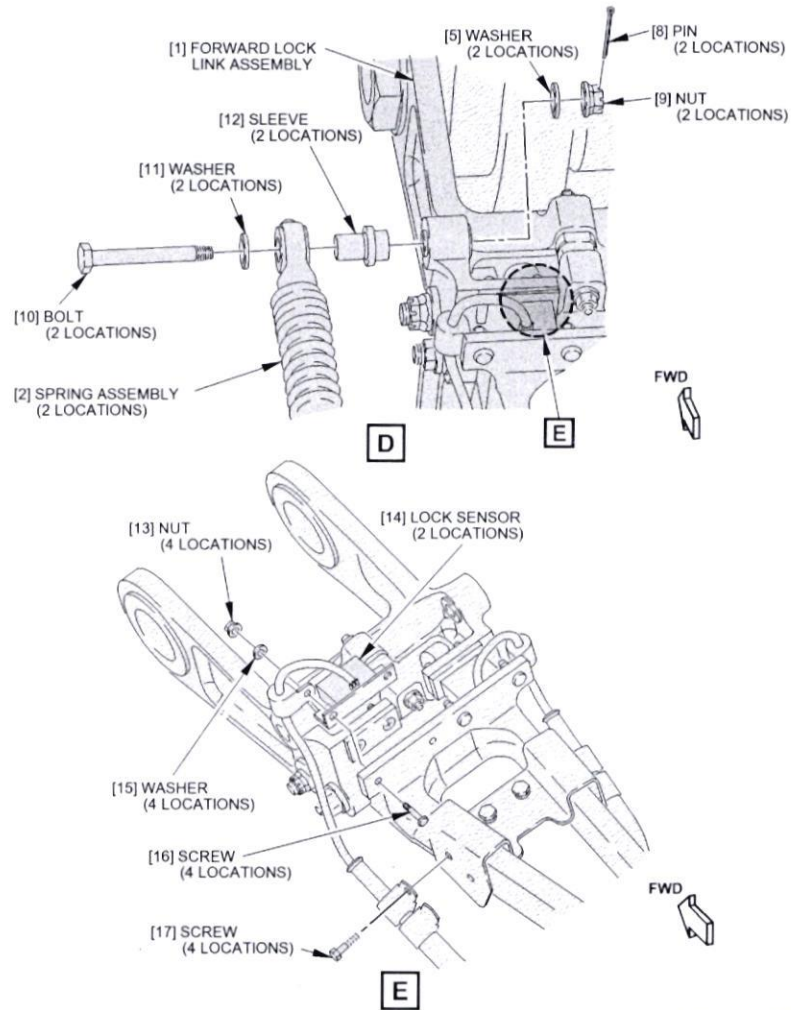
Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -



F96684 S0006575288_V2

Handwritten signature and date: 26/10/14

Handwritten signature: 25/11/14

GAT 171 (Circular stamp)

GAT 128 (Circular stamp)

Figure 401. Nose Gear Lock Mechanism Installation - Sheet 3
TASK 32-33-51-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

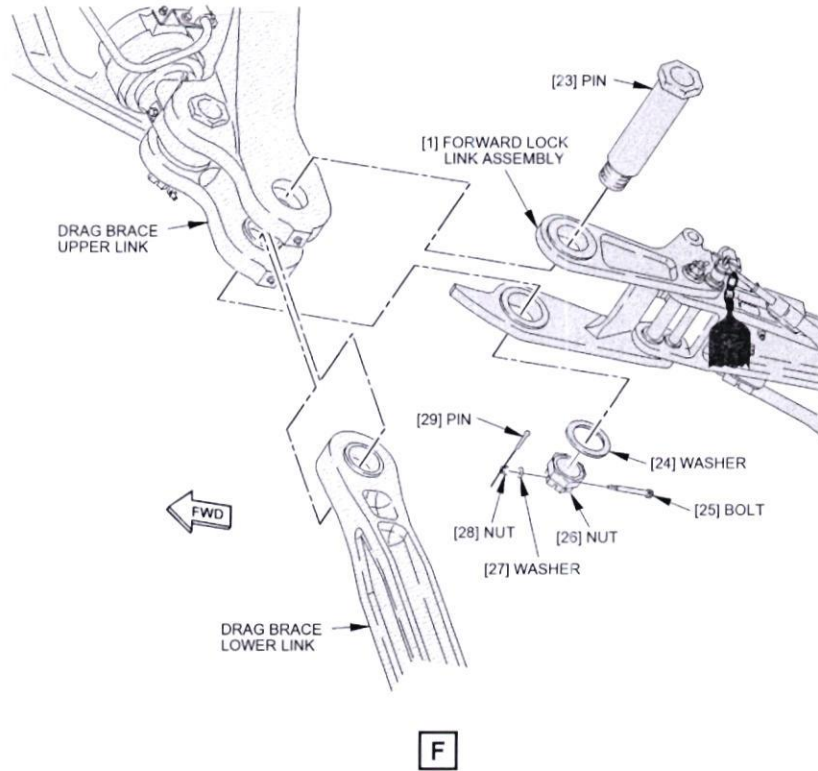
Restore Nose Landing Gear

Type: Routine Card

ATA: 32--

Flow: -

Work Area: -



F95770 S0006575289_V3

[Handwritten signature]
 25/14/2024
 GAT
 171

[Handwritten signature]
 MK1957

Figure 401. Nose Gear Lock Mechanism Installation - Sheet 4
 TASK 32-33-51-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

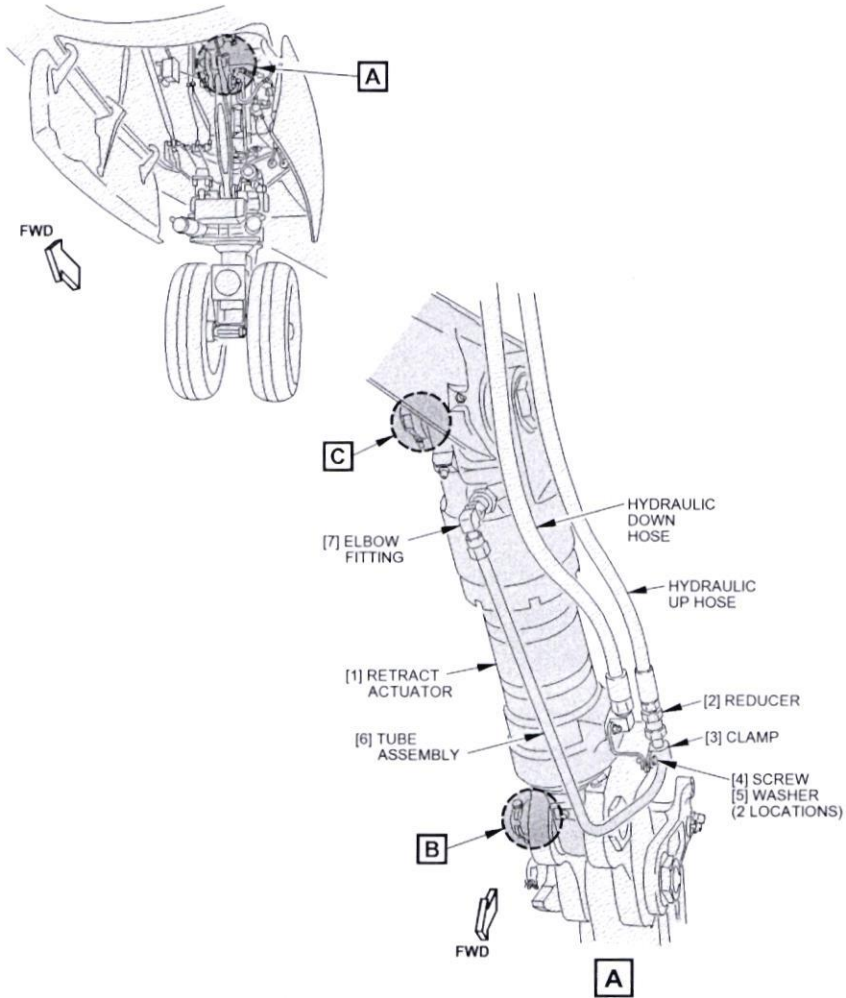
Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232



F83363 50006575262_V2

S. W. J. M. R. S. T.

[Signature]

GAT
171

25/11/2024

Figure 401. Nose Gear Retract Actuator Installation - Sheet 1
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



Restore Nose Landing Gear

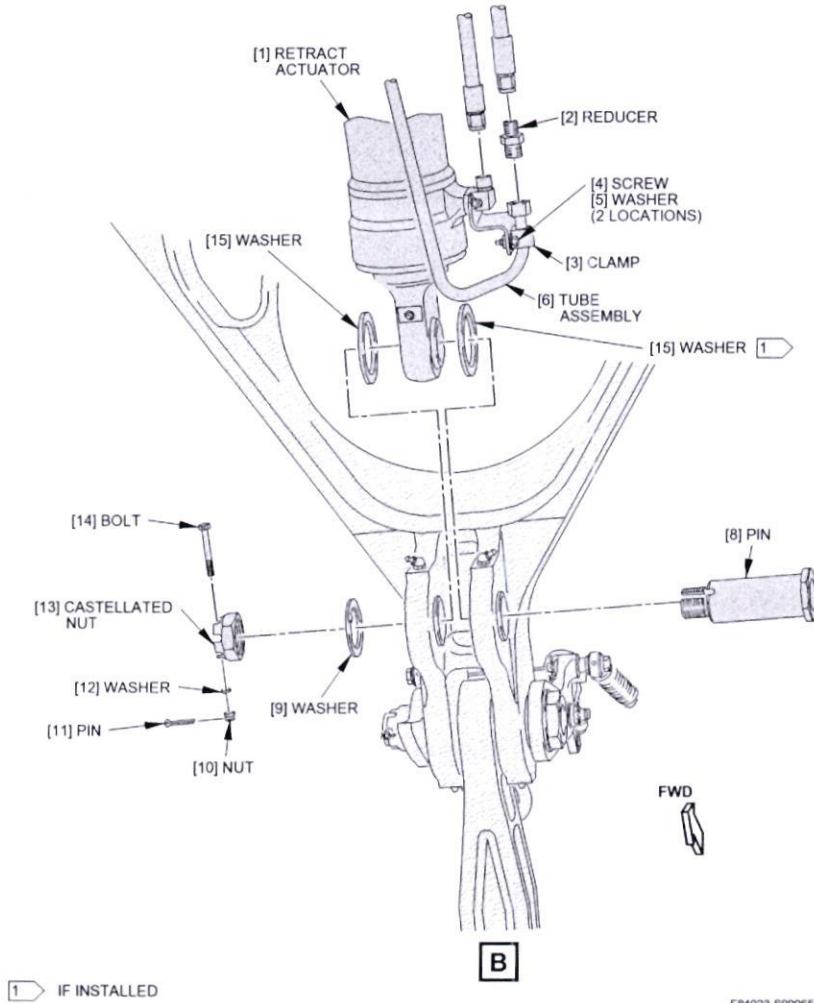
Type: Routine Card

ATA: 32--

Flow:-

Work Area:-

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES



F84023 90006575263_V3

Handwritten signature
 25/10/20
 GAT 171
Handwritten signature
 12/10/27

Figure 401. Nose Gear Retract Actuator Installation - Sheet 2
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



Restore Nose Landing Gear

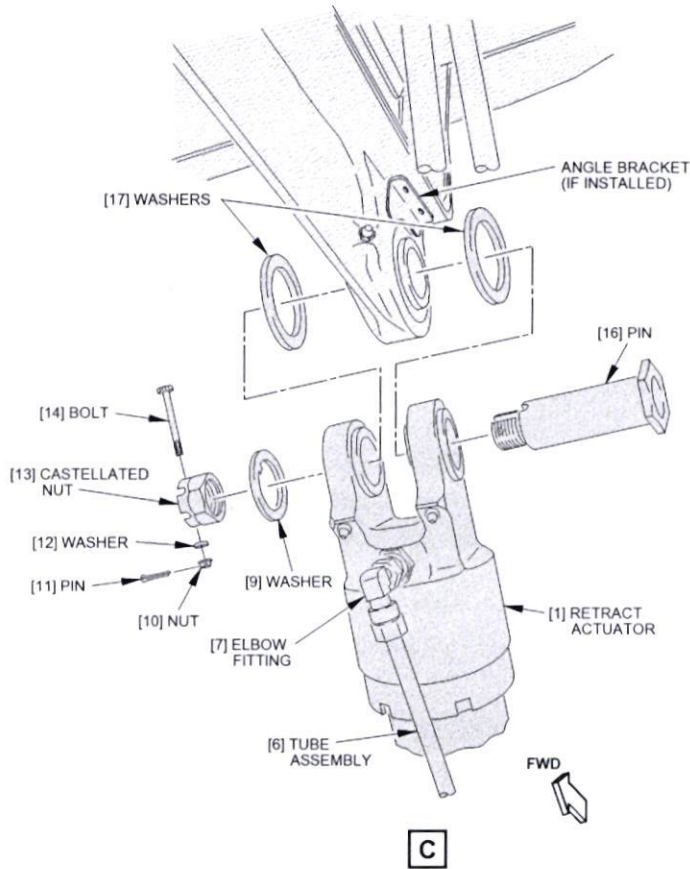
Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH OLD PIN ASSEMBLIES



F84027 50006575264_V3

N/A
24/12/2024
GAT
177

Figure 401. Nose Gear Retract Actuator Installation - Sheet 3
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Rev # 41



Rev Date: Oct 17, 2024 PDT

GCAA APPROVAL No : UAE.145.1232

Restore Nose Landing Gear

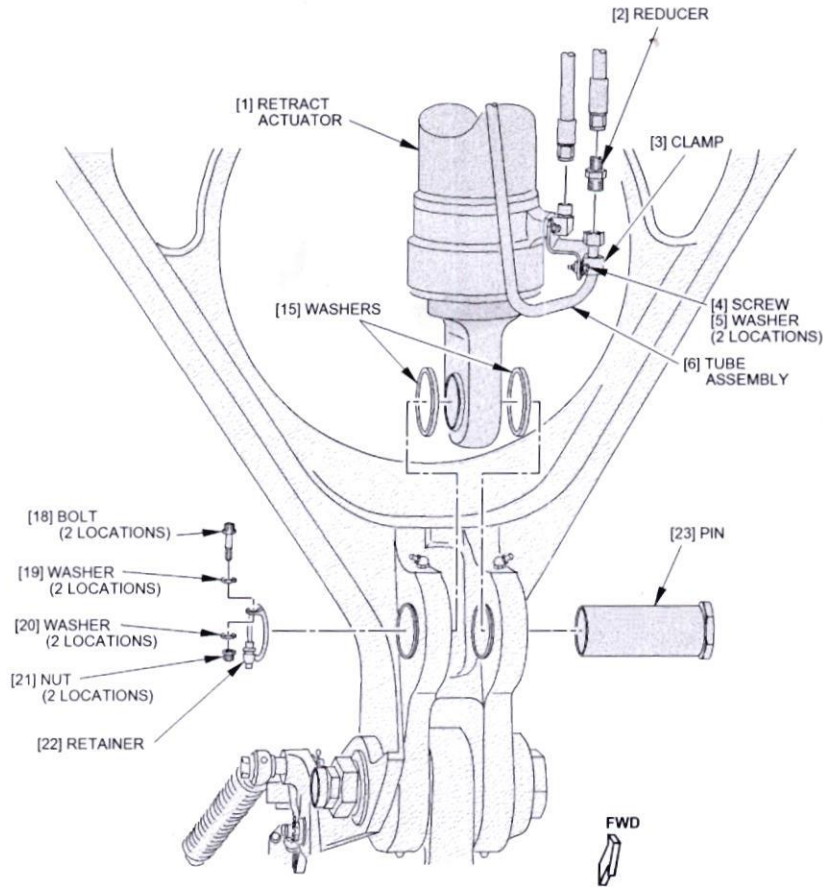
Type: Routine Card

ATA: 32--

Flow: -

Work Area: -

FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES



2090430 50000440997_V3

[Handwritten Signature]
 24/12/2024

GAT
171

Silent
 MR1957

Figure 401. Nose Gear Retract Actuator Installation - Sheet 4
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

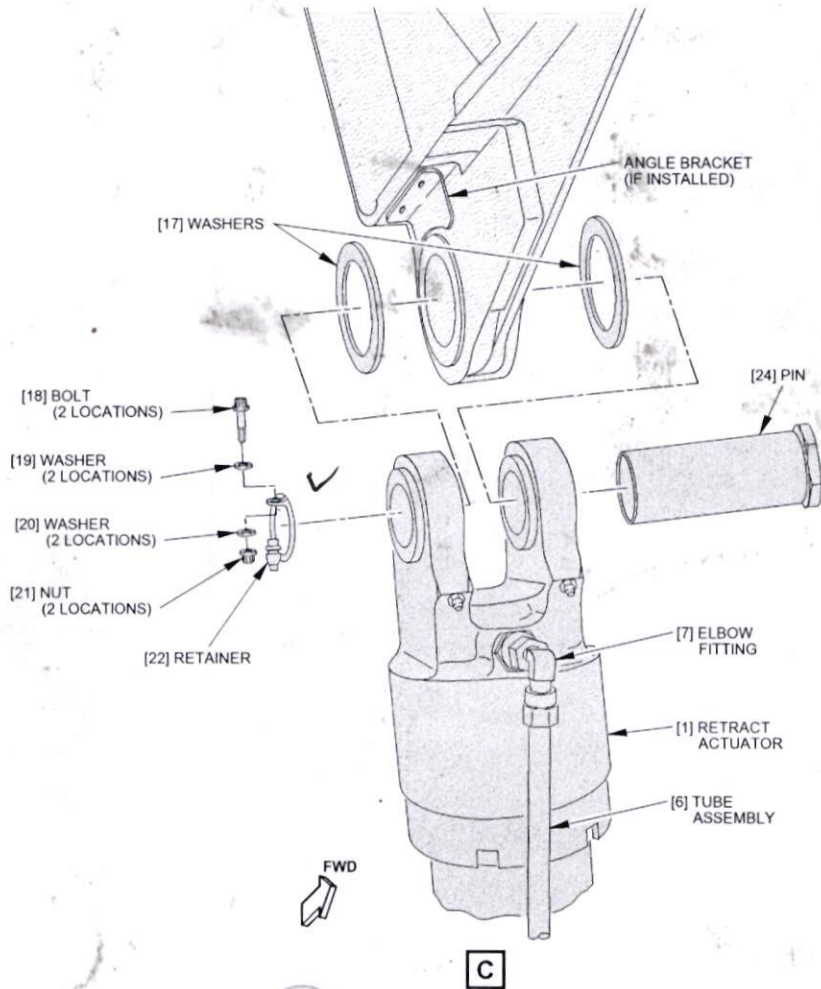
Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



FIG. EFFECTIVITY: JXB ALL; AIRPLANES WITH NEW PIN ASSEMBLIES



Handwritten notes:
 GAT 171
 S.W.T
 MR1952

2090508 50000440998_V3

Figure 401. Nose Gear Retract Actuator Installation - Sheet 5
TASK 32-33-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____
 Item: _____ Completed through item: _____ Sign: _____

Rev # 41

Rev Date: Oct 17, 2024 PDT



MRN24/009608/1224



UNSERVICEABLE TAG

Description: NOSE LANDING GEAR

Part No.: 162 A 1100-14 Serial No. (If applicable) GK1506895241

Removed from : A/C Regn/Eng Sl. No.: AB-FET Pos: NLG

Reason for Unserviceability : CUSTOMER WORK 2921939

Operator Name : FLY DUBAI

Signature: [Signature] Authorization Stamp: [GAT 171 Stamp] Date: 26/12/2011

This portion to be filled by Support / certifying staff of GAT or Authorised Personnel of Operator

TSN / TSO DONE: _____ Signature: _____ Staff Id : _____

CSN/CSO DONE: _____ Date : _____

- Regulations
- CAR 145
 - EASA Part 145
 - FAA Part 145
 - Others CSN
(Specify)

PLANNING COPY

UNSERVICEABLE

**Regulations**

- DGCA CAR 145
- EASA Part 145
- FAA Part 145
- Others GCAA (Specify)

STORE ACCEPTANCE TAG

PART 162A1100-14	MFR. SERIAL # GK15614Y5187	PART DESCRIPTION NOSE LANDING GEAR	PART TYPE Component	PART CONDITION OVERHAULED
LOT # / MFR. LOT #	QTY / UOM 1.00 EA	REF. DOC. #	RECEIPT NO. # CGR24/002607/1124	STOCK STATUS FLY DUBAI
SUPPLIER NAME FLYDUBAI	CERTIFICATE NO. RAP-24-10286	CERTIFICATE DATE 04/11/2024	EXPIRY/ CAL. DUE DATE NOT APPLICABLE	

WAREHOUSE / ZONE / BIN / RACK DETAILSH/S 03 / HANGAR 1**STORAGE REMARKS**

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY MR1591 /Sunil kumar Puli	INSPECTED DATE 11/11/2024
SIGNATURE 	STAMP



1. Approving Competent Authority / Country EASA		2. AUTHORISED RELEASE CERTIFICATE EASA FORM 1			3. Form Tracking Number RAP-24-10561	
4. Organisation Name and Address : REVIMA ASIA PACIFIC LTD. 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand				5. Work Order / Contract / Invoice FDB02/RG70000039/351960		
6. Item	7. Description	8. Part No.	9. Qty.	10. Serial No.	11. Status / Work	
1	PIN ASSY-TRUN RH	162A0302-1	1	E5598	OVERHAULED	
12. Remarks THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-07 REV: 36 DATED: 01st JUL,2022. CONTAINS LLP BARE CONFIGURATION P/N: 162A0302-2, SN: E5598. TSN: 39631:19 CSN: 14012 TSO/CSO: 0 AD APPLICATION: N/A. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.						
13a. Certifies that the items identified above were manufactured in conformity to : <input type="checkbox"/> approved design data and are in condition for safe operation. <input type="checkbox"/> non-approved design data specified in block 12.			14a. <input checked="" type="checkbox"/> Part-145.A.50 Release to Service <input type="checkbox"/> Other regulation specified in block 12 Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12, was accomplished in accordance with Part-145 and in respect to that work the items are considered ready for release to service.			
13b. Authorised Signature		13c. Approval / Authorisation Number		14b. Authorised Signature Pikul C.		14c. Certificate / Approval Ref. No EASA.145.0905
13d. Name		13e. Date (dd mmm yyyy)		14d. Name RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd mmm yyyy) 04 NOV 2024
USER/INSTALLER RESPONSABILITIES: This certificate does not automatically constitute authority to install the item(s). Where the user / installer performs work in accordance with regulations of an Airworthiness Authority different than the Airworthiness Authority specified in block 1, it is essential that the user / installer ensures that his / her Airworthiness Authority accepts items from the Airworthiness Authority specified in block 1. Statements in blocks 13a and 14a do not constitute installation certification. In all cases aircraft maintenance records must contain an installation certification issued in accordance with the national regulation by the user / installer before the aircraft may be flown.						



1. Approving Civil Aviation Authority/Country: FAA / United States		2.			AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG		3. Form Tracking Number: RAP-24-10667	
4. Organization Name and Address: REVIMA ASIA PACIFIC LTD. (Repair Station Certificate No. 43VY764D) 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand						5. Work Order/Contract/Invoice Number: FDB02/RG70000039/351960		
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:			
1	PIN ASSY-TRUN RH	162A0302-1	1	E5598	OVERHAULED			
12. Remarks: THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-07 REV: 36 DATED: 01st JUL,2022. CONTAINS LLP BARE CONFIGURATION P/N: 162A0302-2, SN: E5598. TSN: 39631:19 CSN: 14012 TSO/CSO : 0 AD APPLICATION: N/A. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.								
13a. Certifies the items identified above were manufactured in conformity to: <input type="checkbox"/> Approved design data and are in a condition for safe operation. <input type="checkbox"/> Non-approved design data specified in Block 12.					14a. <input checked="" type="checkbox"/> 14 CFR 43.9 Return to Service <input type="checkbox"/> Other regulation specified in Block 12 Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.			
13b. Authorized Signature:			13c. Approval/Authorization No.:		14b. Authorized Signature: <i>Pikul C.</i>		14c. Approval/Certificate No.: 43VY764D	
13d. Name (Typed or Printed):			13e. Date (dd/mmm/yyyy):		14d. Name (Typed or Printed): RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd/mmm/yyyy): 04 NOV 2024	
User/Installer Responsibilities								
<p>It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.</p> <p>Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.</p> <p>Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.</p>								



1. Approving Competent Authority / Country EASA		2. AUTHORISED RELEASE CERTIFICATE EASA FORM 1			3. Form Tracking Number RAP-24-10562	
4. Organisation Name and Address : REVIMA ASIA PACIFIC LTD. 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand				5. Work Order / Contract / Invoice FDB02/RG70000039/351967		
6. Item	7. Description	8. Part No.	9. Qty.	10. Serial No.	11. Status / Work	
1	PIN ASSY-TRUN LH	162A0301-1	1	E5511	OVERHAULED	
12. Remarks THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-07 REV: 36 DATED: 01st JUL,2022. CONTAINS LLP BARE CONFIGURATION P/N: 162A0301-2, SN: E5511. TSN: 39631:19 CSN: 14012 TSO/CSO: 0 AD APPLICATION: N/A. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.						
13a. Certifies that the items identified above were manufactured in conformity to : <input type="checkbox"/> approved design data and are in condition for safe operation. <input type="checkbox"/> non-approved design data specified in block 12.				14a. <input checked="" type="checkbox"/> Part-145.A.50 Release to Service <input type="checkbox"/> Other regulation specified in block 12 Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12, was accomplished in accordance with Part-145 and in respect to that work the items are considered ready for release to service.		
13b. Authorised Signature		13c. Approval / Authorisation Number		14b. Authorised Signature Pikul C.		14c. Certificate / Approval Ref. No EASA.145.0905
13d. Name		13e. Date (dd mmm yyyy)		14d. Name RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd mmm yyyy) 04 NOV 2024
USER/INSTALLER RESPONSABILITIES: This certificate does not automatically constitute authority to install the item(s). Where the user / installer performs work in accordance with regulations of an Airworthiness Authority different than the Airworthiness Authority specified in block 1, it is essential that the user / installer ensures that his / her Airworthiness Authority accepts items from the Airworthiness Authority specified in block 1. Statements in blocks 13a and 14a do not constitute installation certification. In all cases aircraft maintenance records must contain an installation certification issued in accordance with the national regulation by the user / installer before the aircraft may be flown.						



1. Approving Civil Aviation Authority/Country: FAA / United States		AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG			2. Form Tracking Number: RAP-24-10668	
4. Organization Name and Address: REVIMA ASIA PACIFIC LTD. (Repair Station Certificate No. 43VY764D) 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand					5. Work Order/Contract/Invoice Number: FDB02/RG70000039/351967	
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:	
1	PIN ASSY-TRUN LH	162A0301-1	1	E5511	OVERHAULED	
12. Remarks: THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-07 REV: 36 DATED: 01st JUL,2022. CONTAINS LLP BARE CONFIGURATION P/N: 162A0301-2, SN: E5511. TSN: 39631:19 CSN: 14012 TSO/CSO : 0 AD APPLICATION: N/A. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.						
13a. Certifies the items identified above were manufactured in conformity to: <input type="checkbox"/> Approved design data and are in a condition for safe operation. <input type="checkbox"/> Non-approved design data specified in Block 12.			14a. <input checked="" type="checkbox"/> 14 CFR 43.9 Return to Service <input type="checkbox"/> Other regulation specified in Block 12 Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.			
13b. Authorized Signature:		13c. Approval/Authorization No.:	14b. Authorized Signature: <i>Pikul C.</i>		14c. Approval/Certificate No.: 43VY764D	
13d. Name (Typed or Printed):		13e. Date (dd/mmm/yyyy):	14d. Name (Typed or Printed): RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd/mmm/yyyy): 04 NOV 2024	
User/Installer Responsibilities						
It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article. Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1. Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.						



1. Approving Competent Authority / Country EASA		2. AUTHORISED RELEASE CERTIFICATE EASA FORM 1			3. Form Tracking Number RAP-24-10563	
4. Organisation Name and Address : REVIMA ASIA PACIFIC LTD. 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand				5. Work Order / Contract / Invoice FDB02/RG70000039/351953		
6. Item	7. Description	8. Part No.	9. Qty.	10. Serial No.	11. Status / Work	
1	PIN-DRAG STRUT ASSY	162A2302-1	1	E5024	OVERHAULED	
12. Remarks THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-07 REV: 36 DATED: 01st JUL,2022. CONTAINS LLP BARE CONFIGURATION P/N: N/A, SN: E5024. TSN: 39631:19 CSN: 14012 TSO/CSO: 0 AD APPLICATION: N/A. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.						
13a. Certifies that the items identified above were manufactured in conformity to : <input type="checkbox"/> approved design data and are in condition for safe operation. <input type="checkbox"/> non-approved design data specified in block 12.			14a. <input checked="" type="checkbox"/> Part-145.A.50 Release to Service <input type="checkbox"/> Other regulation specified in block 12 Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12, was accomplished in accordance with Part-145 and in respect to that work the items are considered ready for release to service.			
13b. Authorised Signature		13c. Approval / Authorisation Number		14b. Authorised Signature Pikul C.		14c. Certificate / Approval Ref. No EASA.145.0905
13d. Name		13e. Date (dd mmm yyyy)		14d. Name RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd mmm yyyy) 04 NOV 2024
USER/INSTALLER RESPONSIBILITIES: This certificate does not automatically constitute authority to install the item(s). Where the user / installer performs work in accordance with regulations of an Airworthiness Authority different than the Airworthiness Authority specified in block 1, it is essential that the user / installer ensures that his / her Airworthiness Authority accepts items from the Airworthiness Authority specified in block 1. Statements in blocks 13a and 14a do not constitute installation certification. In all cases aircraft maintenance records must contain an installation certification issued in accordance with the national regulation by the user / installer before the aircraft may be flown.						



1. Approving Civil Aviation Authority/Country: FAA / United States		2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG			3. Form Tracking Number: RAP-24-10669	
4. Organization Name and Address: REVIMA ASIA PACIFIC LTD. (Repair Station Certificate No. 43VY764D) 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand					5. Work Order/Contract/Invoice Number: FDB02/RG70000039/351953	
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:	
1	PIN-DRAG STRUT ASSY	162A2302-1	1	E5024	OVERHAULED	
12. Remarks: THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-07 REV: 36 DATED: 01st JUL,2022. CONTAINS LLP BARE CONFIGURATION P/N: N/A, SN: E5024. TSN: 39631:19 CSN: 14012 TSO/CSO : 0 AD APPLICATION: N/A. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.						
13a. Certifies the items identified above were manufactured in conformity to: <input type="checkbox"/> Approved design data and are in a condition for safe operation. <input type="checkbox"/> Non-approved design data specified in Block 12.			14a. <input checked="" type="checkbox"/> 14 CFR 43.9 Return to Service <input type="checkbox"/> Other regulation specified in Block 12 Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.			
13b. Authorized Signature:		13c. Approval/Authorization No.:	14b. Authorized Signature: Pikul C.		14c. Approval/Certificate No.: 43VY764D	
13d. Name (Typed or Printed):		13e. Date (dd/mmm/yyyy):	14d. Name (Typed or Printed): RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd/mmm/yyyy): 04 NOV 2024	
User/Installer Responsibilities						
<p>It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.</p> <p>Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.</p> <p>Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.</p>						



1. Approving Competent Authority / Country EASA		2. AUTHORISED RELEASE CERTIFICATE EASA FORM 1			3. Form Tracking Number RAP-24-10564	
4. Organisation Name and Address : REVIMA ASIA PACIFIC LTD. 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand				5. Work Order / Contract / Invoice FDB02/RG70000039/351974		
6. Item	7. Description	8. Part No.	9. Qty.	10. Serial No.	11. Status / Work	
1	CYLINDER ASSY-OUTER	162A1110-5	1	ZKM2113	OVERHAULED	
12. Remarks THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-12 REV: 71 DATED: 01st JUL,2024. CONTAINS LLP BARE CONFIGURATION P/N: 162A1110-6, SN: ZKM2113. TSN: 39631:19 CSN: 14012 TSO/CSO: 0 AD APPLICATION: AD 2017-04-11. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.						
13a. Certifies that the items identified above were manufactured in conformity to : <input type="checkbox"/> approved design data and are in condition for safe operation. <input type="checkbox"/> non-approved design data specified in block 12.				14a. <input checked="" type="checkbox"/> Part-145.A.50 Release to Service <input type="checkbox"/> Other regulation specified in block 12 Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12, was accomplished in accordance with Part-145 and in respect to that work the items are considered ready for release to service.		
13b. Authorised Signature		13c. Approval / Authorisation Number		14b. Authorised Signature Pikul C.		14c. Certificate / Approval Ref. No EASA.145.0905
13d. Name		13e. Date (dd mmm yyyy)		14d. Name RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd mmm yyyy) 04 NOV 2024
USER/INSTALLER RESPONSABILITIES: This certificate does not automatically constitute authority to install the item(s). Where the user / installer performs work in accordance with regulations of an Airworthiness Authority different than the Airworthiness Authority specified in block 1, it is essential that the user / installer ensures that his / her Airworthiness Authority accepts items from the Airworthiness Authority specified in block 1. Statements in blocks 13a and 14a do not constitute installation certification. In all cases aircraft maintenance records must contain an installation certification issued in accordance with the national regulation by the user / installer before the aircraft may be flown.						



1. Approving Civil Aviation Authority/Country: FAA / United States		2.			AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG		3. Form Tracking Number: RAP-24-10670	
4. Organization Name and Address: REVIMA ASIA PACIFIC LTD. (Repair Station Certificate No. 43VY764D) 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand						5. Work Order/Contract/Invoice Number: FDB02/RG70000039/351974		
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:			
1	CYLINDER ASSY-OUTER	162A1110-5	1	ZKM2113	OVERHAULED			
12. Remarks: THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-12 REV: 71 DATED: 01st JUL,2024. CONTAINS LLP BARE CONFIGURATION P/N: 162A1110-6, SN: ZKM2113. TSN: 39631:19 CSN: 14012 TSO/CSO : 0 AD APPLICATION: AD 2017-04-11. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.								
13a. Certifies the items identified above were manufactured in conformity to: <input type="checkbox"/> Approved design data and are in a condition for safe operation. <input type="checkbox"/> Non-approved design data specified in Block 12.					14a. <input checked="" type="checkbox"/> 14 CFR 43.9 Return to Service <input type="checkbox"/> Other regulation specified in Block 12 Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.			
13b. Authorized Signature:			13c. Approval/Authorization No.:		14b. Authorized Signature: <i>Pikul C.</i>		14c. Approval/Certificate No.: 43VY764D	
13d. Name (Typed or Printed):			13e. Date (dd/mmm/yyyy):		14d. Name (Typed or Printed): RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd/mmm/yyyy): 04 NOV 2024	
User/Installer Responsibilities								
<p>It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.</p> <p>Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.</p> <p>Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.</p>								



1. Approving Competent Authority / Country EASA		2. AUTHORISED RELEASE CERTIFICATE EASA FORM 1			3. Form Tracking Number RAP-24-10565	
4. Organisation Name and Address : REVIMA ASIA PACIFIC LTD. 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand				5. Work Order / Contract / Invoice FDB02/RG70000039/351994		
6. Item	7. Description	8. Part No.	9. Qty.	10. Serial No.	11. Status / Work	
1	CYLINDER ASSY-INNER	162A1120-7	1	ZKM1837	OVERHAULED	
12. Remarks THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-12 REV: 71 DATED: 01st JUL,2024. CONTAINS LLP BARE CONFIGURATION P/N: 162A1120-8, SN: ZKM1837. TSN: 39631:19 CSN: 14012 TSO/CSO: 0 AD APPLICATION: N/A. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.						
13a. Certifies that the items identified above were manufactured in conformity to : <input type="checkbox"/> approved design data and are in condition for safe operation. <input type="checkbox"/> non-approved design data specified in block 12.			14a. <input checked="" type="checkbox"/> Part-145.A.50 Release to Service <input type="checkbox"/> Other regulation specified in block 12 Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12, was accomplished in accordance with Part-145 and in respect to that work the items are considered ready for release to service.			
13b. Authorised Signature		13c. Approval / Authorisation Number		14b. Authorised Signature Pikul C.		14c. Certificate / Approval Ref. No EASA.145.0905
13d. Name		13e. Date (dd mmm yyyy)		14d. Name RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd mmm yyyy) 04 NOV 2024
USER/INSTALLER RESPONSABILITIES: This certificate does not automatically constitute authority to install the item(s). Where the user / installer performs work in accordance with regulations of an Airworthiness Authority different than the Airworthiness Authority specified in block 1, it is essential that the user / installer ensures that his / her Airworthiness Authority accepts items from the Airworthiness Authority specified in block 1. Statements in blocks 13a and 14a do not constitute installation certification. In all cases aircraft maintenance records must contain an installation certification issued in accordance with the national regulation by the user / installer before the aircraft may be flown.						



1. Approving Civil Aviation Authority/Country: FAA / United States		2.			AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG		3. Form Tracking Number: RAP-24-10671	
4. Organization Name and Address: REVIMA ASIA PACIFIC LTD. (Repair Station Certificate No. 43VY764D) 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand						5. Work Order/Contract/Invoice Number: FDB02/RG70000039/351994		
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:			
1	CYLINDER ASSY-INNER	162A1120-7	1	ZKM1837	OVERHAULED			
12. Remarks: THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-12 REV: 71 DATED: 01st JUL,2024. CONTAINS LLP BARE CONFIGURATION P/N: 162A1120-8, SN: ZKM1837. TSN: 39631:19 CSN: 14012 TSO/CSO : 0 AD APPLICATION: N/A. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.								
13a. Certifies the items identified above were manufactured in conformity to: <input type="checkbox"/> Approved design data and are in a condition for safe operation. <input type="checkbox"/> Non-approved design data specified in Block 12.					14a. <input checked="" type="checkbox"/> 14 CFR 43.9 Return to Service <input type="checkbox"/> Other regulation specified in Block 12 Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.			
13b. Authorized Signature:		13c. Approval/Authorization No.:		14b. Authorized Signature: <i>Pikul C.</i>		14c. Approval/Certificate No.: 43VY764D		
13d. Name (Typed or Printed):		13e. Date (dd/mmm/yyyy):		14d. Name (Typed or Printed): RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd/mmm/yyyy): 04 NOV 2024		
User/Installer Responsibilities								
It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article. Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1. Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.								



1. Approving Competent Authority / Country EASA		2. AUTHORISED RELEASE CERTIFICATE EASA FORM 1			3. Form Tracking Number RAP-24-10566	
4. Organisation Name and Address : REVIMA ASIA PACIFIC LTD. 700/141 Moo 5, T.Klongtamru, A.Muang Chonburi, Chonburi 20000 Thailand					5. Work Order / Contract / Invoice FDB02/RG70000039/351993	
6. Item	7. Description	8. Part No.	9. Qty.	10. Serial No.	11. Status / Work	
1	LINK ASSY-TORSION	162A1315-1	1	AP9392	OVERHAULED	
12. Remarks THIS UNIT HAS BEEN OVERHAULED I.A.W. CMM: 32-21-12 REV: 71 DATED: 01st JUL,2024. CONTAINS LLP BARE CONFIGURATION P/N: 162A1315-2, SN: AP9392. TSN: 39631:19 CSN: 14012 TSO/CSO: 0 AD APPLICATION: N/A. PREVIOUS OPERATOR: FLYDUBAI, MSN: 40270, A/C: A6-FES. CUSTOMER ORDER: R2070524.						
13a. Certifies that the items identified above were manufactured in conformity to : <input type="checkbox"/> approved design data and are in condition for safe operation. <input type="checkbox"/> non-approved design data specified in block 12.				14a. <input checked="" type="checkbox"/> Part-145.A.50 Release to Service <input type="checkbox"/> Other regulation specified in block 12 Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12, was accomplished in accordance with Part-145 and in respect to that work the items are considered ready for release to service.		
13b. Authorised Signature		13c. Approval / Authorisation Number		14b. Authorised Signature Pikul C.		14c. Certificate / Approval Ref. No EASA.145.0905
13d. Name		13e. Date (dd mmm yyyy)		14d. Name RTS RAP 2124 Pikul CHINNAWONG		14e. Date (dd mmm yyyy) 04 NOV 2024
USER/INSTALLER RESPONSABILITIES: This certificate does not automatically constitute authority to install the item(s). Where the user / installer performs work in accordance with regulations of an Airworthiness Authority different than the Airworthiness Authority specified in block 1, it is essential that the user / installer ensures that his / her Airworthiness Authority accepts items from the Airworthiness Authority specified in block 1. Statements in blocks 13a and 14a do not constitute installation certification. In all cases aircraft maintenance records must contain an installation certification issued in accordance with the national regulation by the user / installer before the aircraft may be flown.						

