(WO-41)

flydubai	Task Card	Rotable Controlled Taskcard	Page 1 / 3	Tally Seq No: 41	TCREQLI.97771
		Work Package: FEW/H-25		A.M.M. REV:	

Task Card no:	Task Card Title:	A/C Area:
32-040-02 / ITEM-1	**RESTORE THE RIGHT MAIN LANDING GEAR ASSEMBLY.**	N/A

Part Number:	Serial Number:	Description:	Pos:
161A1100-54	MAL05346S	MLG COMPONENT INSTL	RH MLG

Operator Revision: 01/01.Dec.2011 Type Codes: RST
Special Codes:

MSG Codes: 8 Internal Check:

Step classificatior CMR: ☐ ETOPS: ☐ CDCCL: ☐ AD: ☐ ALI: ☐

A/C Registration: FEW
A/C Type: 73NG
Related Task Card:

Workorder No: 7911047
Work Package: FEW/H-25
Project No: BASE_CHECKS

ITEM-1 -

Zones: 744

ACCESS AND OTHER REQUIREMENTS

Panel	Description
651ET	651ET
651DB	651DB
651BB	651BB
651DT	651DT
651BT	651BT
651EB	651EB
651FB	651FB

DOCUMENT REFERENCES

Doc Typ	Reference
AMM	32-11-00-000
AMM	32-11-00-400
AMM	32-11-61-000
AMM	32-11-61-400
AMM	32-11-71-000
AMM	32-11-71-420
AMM	32-11-83-000
AMM	32-11-83-400
AMM	32-11-89-000
AMM	32-11-89-420
AMM	32-32-11-000
AMM	32-32-11-400
AMM	32-32-31-000
AMM	32-32-31-400
AMM	57-16-01-000
AMM	57-16-01-400
AMM	57-16-02-000
AMM	57-16-02-400

MATERIAL REQUIREMENTS

Partnumber	Description	Qty	SN/Batch no
No parts assigned to this Task Item			

TOOLING REQUIREMENTS

Partnumber	Description	S/N used (tools)
No tools assigned to this Task Item		

Task Card no:	Task Card Title:	A/C Area:
32-040-02 / ITEM-1	**RESTORE THE RIGHT MAIN LANDING GEAR ASSEMBLY.**	N/A

Part Number:	Serial Number:	Description:	Pos:
161A1100-54	MAL05346S	MLG COMPONENT INSTL	RH MLG

PERF. TASKCARD 32-040-02, Rev. 01 Step-1 / Job Description	Mech	Insp
Restore the right main landing gear assembly. INTERVAL NOTE: Whichever comes first. Per SL 32-165 Restoration should include: Component Installation P/N 161A1100 (CMM 32-11-12) Side Strut Assembly P/N 161A2100 (CMM 32-11-13) Reaction Link P/N 161A4100 (CMM 32-11-13) Forward Trunnion Bearing P/N 161A1185 (CMM 57-15-01) Downlock P/N 161A2120 (CMM 32-11-13) Walking Beam Installation P/N 161A7000 (CMM 32-32-27) which includes the Retraction Actuator P/N 273A2101* (CMM 32-32-37) Uplock Assembly P/N 161A6100 (CMM 32-32-34) Forward Trunnion Bearing Housing P/N 115A5240 (CMM 57-15-01) Aft Trunnion Bearing P/N 161A1500 (CMM 57-15-01) Shimmy Damper P/N 273A3610 (CMM 32-30-62)** All installation components should be restored (CMM 32-11-09) All system components and attachments should be restored (CMM 32-11-16) NOTE: TASK COMPONENT RELATED	MR07472	29.3.25 GAT 492 22.3.25 GAT 492 27.3.25 GAT 492

TASKCARD FINDING(Tick applicable Box): [✓]YES []NO
(Please enter Taskcard Reference also on Workorder)
FINDING WORKORDER NO . HMV24/000323/0225/17
HMV24/000323/0225/129
HMV24/000323/0225/120

If Job Stopped or interrupted :
Task Card Performed Up to Page : 12 Step: 1B NOTE 13.3.25 GAT 492
Remarks:
~~76~~ 14 H (13) 22.3.25 GAT 492
55 8 I (WARNING) 22.3.25 GAT 492 GAT 492

If Job Stopped or interrupted :
Task Card Performed Up to Page : 120 Step: 18 I(1) 22.3.25 GAT 492
Remarks:
56 15 G(8) 22.3.25
92 16G(a) ~~Satish Baby~~ 22/3/25 GAT 128

Manhours used to perform taskcard :

Station	Date	Accomplished
HMD	27.03.25	GAT 492

114 H 26) ~~Satish Baby~~ 22/3/25 GAT 128
~~HH~~52 F(24) ~~Satish Baby~~ 24/3/25 GAT 128
92 16G(6) ~~Satish Baby~~ 24/3/25 GAT 128
118 18H(44) ~~Satish Baby~~ 24/3/25 GAT 128

flydubai	Task Card	Rotable Controlled Taskcard	Page 3 / 3	Tally Seq No: 41	TCREQLI.97771
		Work Package: FEW/H-25		A.M.M. REV:	

Task Card no:	Task Card Title:	A/C Area:
32-040-02 / ITEM-1	**RESTORE THE RIGHT MAIN LANDING GEAR ASSEMBLY.**	**N/A**

Part Number:	Serial Number:	Description:	Pos:
161A1100-54	**MAL05346S**	**MLG COMPONENT INSTL**	**RH MLG**

CAT C Verification

GAT C-042

	P.NO	DESCRIPTION	S.NO	EXPIRY
1.	C15292	BONDING METER	2707	08/09/2025
2.	184-3135	FEELER GAUGE	2208225215	21/08/2025
3.	QD2R200	TORQUE WRENCH	021960028	23.12.25
4.	QD1R50	TORQUE WRENCH	09188 OK 25	19.06.24
5.	QD 3 R1600	TQ WRENCH	0612059121	21.05.2025
6.	20465	ANALOG DTI	BBLY92	01.5.25
7.	500-197-30	DIGITAL CALIPER	2409208199	01.04.25
8.	184-3135	FEELER GAGE	RG 013	26.2.26
9.	C32028-109	MLG HOIST	EH2306	20.5.2025
10.	C32030-10	STRAP RET	N/A	30.04.2025
11.	P90200-14	INFLATION TOOL	A4002	25.07.25
12.	TE. 6A	TQ METER	30067	28.04.25
13.	DG 200	PUSH PULL GAGE	503672	08.1.26
14.	J29006-1	BLEEDER HOSE	67953-2	- N/A
15.	QD2FR75B	TQ WRENCH	011912004193	03.2.26.

DOCUMENT CHECKED BY:
PLANNER MR No: MR1499
SIGN (INITIALS): SH
31/3/25

27.3.25

GAT 492

CR-1 CR-2

flydubai

737-600/700/800/900

Right Main Landing Gear Restoration

Type:Routine Card ATA:32-- Flow:- Work Area:-

EVENT DETAILS

Tail #	Work Order	Base	Start	End
-	-	-	-	-

REQUIREMENT 32-040-02

PERIODICITY

Version	Threshold	Interval	Limit	Threshold Date	Limit Date
1.1	21000 FC	21000 FC			
1.2	10 YR	10 YR			

Whichever comes first

SOURCES

Type	Reference	Title	Version	Version Date	Originator	Mandatory
MPD	32-040-02				BOEING	true
MRB	D626A001-MRBR		5	Mar2003	BOEING	true

INTERVAL NOTES: Whichever comes first.

AIRPLANE NOTES:

ENGINE NOTES:

ACCESS NOTES:

SPECIAL NOTES:

ENGINEERING COMMENTS:

ANY FINDINGS
YES ✓ NO
IRC No. : HMV 24/000323/0225/17
HMV24/000323/0225/127
HMV24/000323/0225/120
13.3.25 GAT 492

ZONES: 133, 134, 211, 212, 551, 651, 710, 713, 730, 734, 740, 744
ACCESS: 551BB, 551BT, 551CB, 551DB, 651BB, 651BT, 651CB, 651DB, 651DT, 651EB, 651ET, 651FB
SKILLS:
CERTIFICATIONS:
CONDITIONS:

PARTS

Operator Part No	Specification	Qty	Unit of Measure	Type	Description	Manufacturer
A00247	BMS5-95	1		Consumable	Sealant - Pressure And Environmental - Chromate Type	81205
A50009	BMS 5-142, TYPE II, CLASS B-1	1		Consumable	Sealant - Low Density, Non-Chromate Type. (Formerly Chromate - Synthetic Rubber)	81205

13.3.25 GAT 492 MR2143

GCAA APPROVAL No : UAE.145.1232

Type: Routine Card **ATA:**32-- **Flow:**- **Work Area:**-

A50009	BMS5-142 TYPE II CLASS B-1 OR B-2	1	Consumable	Sealant - Low Density, Non-Chromate Type. (Formerly Chromate - Synthetic Rubber)	81205
C00259	BMS10-11 TYPE I	1	Consumable	Coating - Chemical And Solvent Resistant Finish, Corrosion Inhibiting Primer	81205
D00006	-	1	Consumable	Compound - Antiseize Pure Nickel Special - Never-Seez NSBT-8N	81205
D00013	MIL-PRF-23827 (NATO G-354) (S	1	Consumable	Grease - Aircraft And Instrument Grease	81205
D00013	MIL-PRF-23827 (NATO G-354) (SUPERSEDES MIL-G-23827)	1	Consumable	Grease - Aircraft And Instrument Grease	81205
D00153	BMS3-11 TYPE IV	1	Consumable	Fluid - Hydraulic Fluid, Fire Resistant	81205
D00633	BMS3-33	1	Consumable	Grease - Aircraft General Purpose	81205
G00018	A-A-59503 TYPE I GRADE B, MIL	1	Consumable	Nitrogen - Gaseous, Pressurizing, 99.5 Percent Pure	81205
G00018	A-A-59503 TYPE I GRADE B, MIL-PRF-27401 TYPE I GRADE A	1	Consumable	Nitrogen - Gaseous, Pressurizing, 99.5 Percent Pure	81205
G01048	NASM20995	1	Consumable	Lockwire - MS20995C32, Corrosion Resistant Steel - 0.032 Inch (0.8121 mm) Diameter	81205
G01912	NASM20995	1	Consumable	Lockwire - MS20995NC32, Monel - 0.032 Inch (0.8121 mm) Diameter	81205
G02166	NASM20995	1	Consumable	Lockwire - MS20995NC20, Monel - 0.020 Inch (0.508 mm) Diameter	81205
G02314	BB-A-1034 SOURCE I GRADE A	1	Consumable	Air - Compressed, Breathing	81205

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

13.8.25 GAT 492 MR 2/13

G50136	BMS3-38	1	Consumable	Compound - Corrosion Inhibiting, Non-drying	81205
G50171	-	1	Consumable	Compound - Corrosion Inhibiting Compound, Interior Application - D5026NS or ZC-026	81205
G50237	BMS3-38	1	Consumable	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	81205
G50316	-	1	Consumable	Cloth - Clean, Dry, Lint-free, White, Cotton	81205

TOOLS

Operator Part No	Mfr Part No	Qty	Description	Manufacturer
COM-15434	-	1	OMNI ARM Aircraft Maintenance Fixture JXB ALL; 737-800 (Part #: NB-800, Supplier: 1T0V3) (Opt Part #: NBH, Supplier: 1T0V3)	81205
SPL-11076	-	1	Torque Wrench Adaptor (Part #: C32013-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-11946	-	1	Assembly - Wrench Adapter (C32029-31 Part of Kit C32029-38 or C32029-45) JXB ALL; 737-800 (Part #: C32029-45, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205)	81205
SPL-12460	-	1	Equipment - Removal/Installation, MLG and NLG springs (Part #: J32037-106, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-12498	-	1	MLG downlock release level tool (Part #: C32052-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1359	-	1	Assembly - Wrench Adapter (C32029-3, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1360	-	1	Assembly - Wrench Adapter (C32029-4, Part of Kit C32029-30)	81205

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

13.8.25 GAT 492 MR2143

			(Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	
SPL-1361	-	1	Assembly - Wrench Adapter (C32029-5, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1362	-	1	Assembly - Wrench Adapter (C32029-6, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1363	-	1	Assembly - Wrench Adapter (C32029-7, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1364	-	1	Assembly - Slug Driver (C32029-8, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1365	-	1	Protector - Thread (C32029-9, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1366	-	1	Protector - Thread (C32029-10, Part of Kit C32029-30)	81205

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MR2143 18.3.25 GAT 492

			(Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	
SPL-1367	-	1	Driver - Slug (C32029-11, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1368	-	1	Protector - Thread (C32029-12, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1369	-	1	Protector - Thread (C32029-13, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1370	-	1	Protector - Thread (C32029-14, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1371	-	1	Protector - Thread (C32029-15, Part of Kit C32029-30) (Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1372	-	1	Protector - Thread (C32029-16, Part of Kit C32029-30)	81205

WS 13.3.25 GAT 492 MP2443

			(Part #: C32029-30, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32029-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	
SPL-14021	-	1	Assembly - Wrench Adapter (C32029-48 Part of Kit C32029-45) (Part #: C32029-45, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER)	81205
SPL-14022	-	1	Assembly - Wrench Adapter (C32029-49 Part of Kit C32029-45) (Part #: C32029-45, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER)	81205
SPL-14023	-	1	Assembly - Wrench Adapter (C32029-50 Part of Kit C32029-45) (Part #: C32029-45, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER)	81205
SPL-14024	-	1	Equipment - Thread Protector (C32029-51 Part of Kit C32029-45) (Part #: C32029-45, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER)	81205
SPL-1521	-	1	Tool - Strut Inflation, Landing Gear (Part #: F70200-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1744	-	1	Set - Lock, Inboard Ground Spoiler Actuator (Contains 4 Lock Assemblies) (Part #: C27046-13, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C27046-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1862	-	1	Equipment - Removal/Installation, MLG Aft Trunnion Pin (Part #: C32031-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1868	-	1	Puller - Fuse Pin, Main Landing Gear Forward Trunnion Support (Part #: C32015-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -BBJ)	81205

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

GCAA APPROVAL No : UAE.145.1232

18.3.25 GAT 492

MR2113

SPL-1869	-	1	Fixture - Transportation, Main Landing Gear, Removal/Installation (Part #: C32034-227, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1870	-	1	Hoist - Main Landing Gear Retract Actuator (Part #: C32028-56, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Part #: C32028-58, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Part #: C32028-59, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32028-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32028-3, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-1871	-	1	Strap - Retention, NLG/MLG Inner Cylinder (Part #: C32030-10, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
SPL-4360	-	1	Wrench - Trunnion Spherical Bearing, MLG (Part #: C32023-12, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ) (Opt Part #: C32023-1, Supplier: 81205, A/P Effectivity: 737-600, -700, -700C, -700ER, -700QC, -800, -900, -900ER, -BBJ)	81205
STD-14686	-	1	Fittings - Hydraulic, Caps/Plugs (AMM 20-10-20-2)	81205
STD-4873	-	1	Rope - 0.5 inch (12.7 mm) Diameter	81205
STD-858	-	1	Tag - DO NOT OPERATE	81205

REFERENCES

Reference Number	Type	Title	Version Number	Version Date
07-11-01-580-815	AMM:task	Lift the Airplane with the Jacks	-	-
07-11-01-580-816	AMM:task	Lower the Airplane Off the Jacks	-	-
07-11-03-580-801	AMM:task	Lift the Main Landing Gear Axles with the Axle Jacks	-	-
10-11-05-201	AMM:pgblk	CHOCK INSTALLATION	-	-
10-11-05-500-801	AMM:task	Chock Installation in Winds or Wind Gusts to a Maximum of 35 Knots	-	-
10-11-05-500-802	AMM:task	Chock Installation in Winds of More than 35 Knots	-	-

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Type:Routine Card	ATA:32--	Flow:-	Work Area:-	
12-12-00-610-801	AMM:task	Hydraulic Reservoir Servicing	-	-
12-15-31-610-802	AMM:task	Main Landing Gear Shock Strut Servicing	-	-
12-15-31-610-805	AMM:task	Main Landing Gear Strut Servicing, Airplane on Jacks	-	-
12-21-11-301	AMM:pgblk	MAIN LANDING GEAR - SERVICING	-	-
12-21-11-640-801	AMM:task	Main Landing Gear Upper End Components Servicing	-	-
12-21-11-640-802	AMM:task	Main Landing Gear Lower End Components Servicing	-	-
20-10-44-400-801	AMM:task	Lockwire, Cotter Pins, and Lockrings - Installation	-	-
20-10-52-000-801	AMM:task	Flexible Hose Removal	-	-
20-10-52-400-801	AMM:task	Flexible Hose Installation	-	-
20-60-03	SWPM	SWPM 20-60-03	-	-
27-51-00-040-801	AMM:task	Trailing Edge Flap System Deactivation	-	-
27-51-00-440-801	AMM:task	Trailing Edge Flap System Reactivation	-	-
27-51-00-860-803	AMM:task	Extend the Trailing Edge Flaps	-	-
27-51-00-860-804	AMM:task	Retract the Trailing Edge Flaps	-	-
27-62-00-800-801	AMM:task	Speed Brake Hydraulic Systems A and B Pressurization	-	-
27-62-00-840-801	AMM:task	Put the Speed Brake Hydraulic systems A and B Back to the Condition Before the Pressurization	-	-
27-62-51-000-801	AMM:task	Ground Spoiler Interlock Valve Cable Removal	-	-
27-62-51-400-801	AMM:task	Ground Spoiler Interlock Valve Cable Installation	-	-
29-09-00-860-801	AMM:task	Hydraulic Reservoirs Pressurization	-	-
29-09-00-860-802	AMM:task	Hydraulic Reservoirs Depressurization	-	-
29-11-00-860-801	AMM:task	Hydraulic System A or B Pressurization	-	-
29-11-00-860-805	AMM:task	Hydraulic System A or B Power Removal	-	-
32-00-01-080-801	AMM:task	Landing Gear Downlock Pins Removal	-	-
32-00-01-480-801	AMM:task	Landing Gear Downlock Pins Installation	-	-
32-11-00-000-801	AMM:task	Main Landing Gear Removal	-	-
32-11-00-400-801	AMM:task	Main Landing Gear Installation	-	-
32-11-61-000-802	AMM:task	Main Landing Gear Lower Side Strut Removal	-	-

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Type:Routine Card	ATA:32--	Flow:-	Work Area:-	
32-11-61-400-802	AMM:task	Main Landing Gear Lower Side Strut Installation	-	-
32-11-81-870-801	AMM:task	Hydraulic Shimmy Damper - Bleeding	-	-
32-11-83-000-801	AMM:task	Main Landing Gear Forward Trunnion Bearing Assembly Removal	-	-
32-11-83-400-801	AMM:task	Main Landing Gear Forward Trunnion Bearing Assembly Installation	-	-
32-13-11-000-803	AMM:task	Main Landing Gear Door Assembly Removal	-	-
32-13-11-420-802	AMM:task	Main Landing Gear Center Door Installation	-	-
32-13-11-420-803	AMM:task	Main Landing Gear Door Assembly Installation	-	-
32-13-11-820-801	AMM:task	Main Landing Gear Shock Strut Door Adjustment	-	-
32-13-21-000-801	AMM:task	Main Landing Gear Wing Door Adjustment	-	-
32-13-21-000-802	AMM:task	Main Landing Gear Wing Door Removal	-	-
32-13-21-420-801	AMM:task	Main Landing Gear Wing Door Installation	-	-
32-32-00-710-801	AMM:task	Main Landing Gear Operational Test	-	-
32-32-00-710-802	AMM:task	Main Landing Gear Test - Component Replacement	-	-
32-32-51-400-801	AMM:task	Main Gear Downlock Actuator Installation	-	-
32-32-91-000-801	AMM:task	Main Gear Downlock Springs Removal	-	-
32-32-91-400-801	AMM:task	Main Gear Downlock Spring Installation	-	-
32-34-00-730-801	AMM:task	Main Gear Manual Extension System Test - Airplane on Jacks	-	-
32-34-21-000-801	AMM:task	Removal of the Manual Extension Mechanism for the Main Gear	-	-
32-34-21-400-801	AMM:task	Installation of the Manual Extension Mechanism for the Main Gear	-	-
32-41-00-870-802	AMM:task	Normal (System B) Hydraulic Brake System - Bleeding	-	-
32-41-41-000-801	AMM:task	Main Landing Gear Brake Removal	-	-
32-41-41-400-801	AMM:task	Main Landing Gear Brake Installation	-	-
32-42-00-710-802	AMM:task	Transducer Operational Test	-	-
32-42-00-720-803	AMM:task	Antiskid Valve Operational Test	-	-

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

13.3.25

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MP2143

GCAA APPROVAL No : UAE.145.1232

Type:Routine Card	ATA:32--	Flow:-	Work Area:-	
32-45-11-000-801	AMM:task	Main Landing Gear Wheel and Tire Assembly Removal	-	-
32-45-11-400-801	AMM:task	Main Landing Gear Wheel and Tire Assembly Installation	-	-
32-61-21-020-801	AMM:task	Main Landing Gear Uplock Sensor Removal	-	-
32-61-21-400-801	AMM:task	Main Landing Gear Uplock Sensor Installation	-	-
32-61-21-820-801	AMM:task	Main Landing Gear Uplock Sensor Clearance Adjustment	-	-
32-61-31-220-801	AMM:task	Main Landing Gear Down-and-Locked Sensor Clearance Measurement	-	-
57-15-00-200-802	AMM:task	Main Landing Gear Beam Hanger Link Free Play Check	-	-
57-16-01-000-801	AMM:task	MLG Forward Trunnion Housing Assembly - Removal	-	-
57-16-01-400-801	AMM:task	MLG Forward Trunnion Housing Assembly - Installation	-	-
57-16-02-000-801	AMM:task	Remove the MLG Aft Trunnion Bearing Assembly	-	-
57-16-02-400-801	AMM:task	Install the MLG Aft Trunnion Bearing Assembly	-	-

NOTES: Restore the right main landing gear assembly.

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

GCAA APPROVAL No : UAE.145.1232

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MP2143

JXB ALL

TASK 32-11-00-000-801

1. **Main Landing Gear Removal** (Figure 401) (Figure 402) (Figure 403) (Figure 404) (Figure 405)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title	MECH	INSP
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)		
10-11-05 P/B 201	CHOCK INSTALLATION		
20-10-52-000-801	Flexible Hose Removal (P/B 401)		
27-51-00-040-801	Trailing Edge Flap System Deactivation (P/B 201)		
27-51-00-860-803	Extend the Trailing Edge Flaps (P/B 201)		
27-62-00-800-801	Speed Brake Hydraulic Systems A and B Pressurization (P/B 201)		
27-62-51-000-801	Ground Spoiler Interlock Valve Cable Removal (P/B 401)		
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)		
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)		
32-11-61-000-802	Main Landing Gear Lower Side Strut Removal (P/B 401)		
32-13-11-000-803	Main Landing Gear Door Assembly Removal (P/B 401)		
32-13-21-000-802	Main Landing Gear Wing Door Removal (P/B 401)		
32-41-41-000-801	Main Landing Gear Brake Removal (P/B 401)		
32-45-11-000-801	Main Landing Gear Wheel and Tire Assembly Removal (P/B 401)		
57-15-00-200-802	Main Landing Gear Beam Hanger Link - Free Play Check (P/B 601)		
57-16-01-000-801	MLG Forward Trunnion Housing Assembly - Removal (P/B 401)		
57-16-01-400-801	MLG Forward Trunnion Housing Assembly - Installation (P/B 401)		
57-16-02-000-801	MLG Aft Trunnion Bearing Assembly Removal (P/B 401)		
57-16-02-400-801	MLG Aft Trunnion Bearing Assembly Installation (P/B 401)		

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description	MECH	INSP
COM-15434	OMNI ARM Aircraft Maintenance Fixture **737-800** (Part #: NB-800, Supplier: 1T0V3) (Opt Part #: NBH, Supplier: 1T0V3)		
SPL-1362	Wrench Adapter, C32029-6 or C32029-96 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-1363	Wrench Adapter, C32029-7 or C32029-98 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-1365	Thread Protector, C32029-9 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-1366	Thread Protector, C32029-10 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-1367	Slug Driver, C32029-11 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205)		

PARTIAL SIGN OFF STATUS:

Item: 1 Completed through item: B (NOTE) Sign:

Item: Completed through item: Sign:

		MECH	INSP
	(Opt Part #: C32029-45, Supplier: 81205)		
SPL-1371	Thread Protector, C32029-15 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-1372	Thread Protector, C32029-16 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-1744	Lock Set - Inboard Ground Spoiler Actuator (Contains 4 Lock Assemblies) **737-800** (Part #: C27046-13, Supplier: 81205) (Opt Part #: C27046-1, Supplier: 81205)		
SPL-1862	Equipment - Removal/Installation, MLG Aft Trunnion Pin **737-800** (Part #: C32031-22, Supplier: 81205) (Opt Part #: C32031-1, Supplier: 81205)		
SPL-1869	Fixture - Transportation, Main Landing Gear, Removal/ Installation **737-800** (Part #: C32034-343, Supplier: 81205) (Opt Part #: C32034-227, Supplier: 81205) (Opt Part #: C32034-273, Supplier: 81205) (Opt Part #: C32034-339, Supplier: 81205)		
SPL-1871	Strap - Retention, NLG/MLG Inner Cylinder **737-800** (Part #: C32030 -31, Supplier: 81205) (Opt Part #: C32030-10, Supplier: 81205)		

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

		MECH	INSP
SPL-14021	Wrench Adapter, C32029-48 or C32029-90 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-14022	Wrench Adapter, C32029-49 or C32029-88 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-14023	Wrench Adapter, C32029-50 or C32029-89 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-14024	Thread Protector , C32029-51 (included in C32029 Kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		

JXB ALL

C. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
734	Left Main Landing Gear
744	Right Main Landing Gear

D. Access Panels

Number	Name/Location
551BB	Lower Inboard Fixed Trailing Edge, Gear Adjustment Door
551BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

		MECH	INSP
551CB	Lower Inboard Fixed Trailing Edge, Gear Access Panel		
551DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel		
651BB	Lower Inboard Fixed Trailing Edge, Gear Door Adjustment		
651BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel		
651CB	Lower Inboard Fixed Trailing Edge, Gear Access Panel		
651DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel		

E. Prepare for the Removal

NOTE: Omni Arm Tool maintenance fixture, COM-15434, can be used as an alternate to Dolly Assembly fixture, SPL-1869. This procedure provides instructions to remove and install the main landing gear with the Dolly Assembly. Refer to manufacturers instructions for use of the Omni Arm Tool.

SUBTASK 32-11-00-480-001

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT. THIS CAN CAUSE INJURIES TO PERSONS, AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed on all the landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-11-00-480-025

(2) Make sure that the chocks are installed around the tires (PAGEBLOCK 10-11-05/201).

SUBTASK 32-11-00-020-001

WARNING: FULLY DEFLATE THE SHOCK STRUT BEFORE YOU REMOVE THE VALVE BODY. IF YOU REMOVE THE VALVE BODY WHEN THERE IS PRESSURE, THE PRESSURE CAN BLOW THE VALVE BODY OUT. IF YOU DO NOT OBEY, INJURY TO PERSONS CAN OCCUR.

CAUTION: DO NOT DEFLATE OR SERVICE THE SHOCK STRUTS WHILE THE TAIL STAND IS INSTALLED. IF YOU DEFLATE OR SERVICE THE SHOCK STRUTS, THE LOAD ON THE TAIL STAND CAN BE TOO LARGE. THIS CAN CAUSE DAMAGE TO EQUIPMENT.

(3) Deflate the shock strut [1] for the two main landing gear fully.

NOTE: This task is necessary to keep the airplane level when you do the subsequent steps.

(a) Remove the cap [4] for the air valve [2].

(b) Loosen the swivel nut [3] for a maximum of two turns.

NOTE: Fluid in the shock strut [1] will have bubbles when you release the pressure. Deflate the shock strut slowly to prevent the leakage of the fluid through the air valve [2].

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(c) Loosen the swivel nut [3] fully when all of the pressure in the shock strut [1] is released.

NOTE: The shock strut [1] is fully deflated when the dimension from the lower surface of the outer cylinder to the upper surface of the axle clevis on the inner cylinder is 0.9 in. (2.3 cm).

SUBTASK 32-11-00-480-002

WARNING: DO AN INSPECTION OF THE STRAP FOR CORRECT INSTALLATION AND DAMAGE BEFORE YOU USE IT. INCORRECT INSTALLATION OR DAMAGE CAN CAUSE THE STRAP TO BREAK. THIS CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(4) Install the retention strap, SPL-1871, with the marker facing up, to hold the shock strut [1] in its compressed position.

NOTE: Two straps are required to be used to hold the inner cylinder secure.

SUBTASK 32-11-00-480-022

(5) Lift the airplane with the jacks until the tires are off of the ground, do this task: Lift the Airplane with the Jacks, TASK 07-11-01-580-815.

SUBTASK 32-11-00-860-029

(6) Extend the trailing edge flaps fully, do this task: Extend the Trailing Edge Flaps, TASK 27-51-00-860-803.

NOTE: This step will provide access to the aft trunnion pin [19].

SUBTASK 32-11-00-860-030

(7) Do this task: Trailing Edge Flap System Deactivation, TASK 27-51-00-040-801.

SUBTASK 32-11-00-860-031

WARNING: MAKE SURE THAT PERSONS AND EQUIPMENT ARE CLEAR OF ALL CONTROL SURFACES BEFORE YOU SUPPLY HYDRAULIC POWER. AILERONS, RUDDERS, ELEVATORS, FLAPS, SLATS, SPOILERS, LANDING GEAR, AND THRUST REVERSERS CAN MOVE QUICKLY WHEN YOU SUPPLY HYDRAULIC POWER. THIS CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(8) Do this task: Speed Brake Hydraulic Systems A and B Pressurization, TASK 27-62-00-800-801.

SUBTASK 32-11-00-860-018

(9) Put the speed brake lever in the UP position to lift the ground spoilers.

SUBTASK 32-11-00-480-021

(10) Install the lock set, SPL-1744 , for the applicable ground spoiler.

SUBTASK 32-11-00-860-017

(11) Put the SPOILER A and B switches, on the P5-3 flight control panel, to the OFF position to remove the hydraulic power from the flight spoilers.

SUBTASK 32-11-00-860-004

(12) Do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-11-00-860-005

(13) Push the brake pedals a minimum of twelve times to remove the pressure from the brake lines.

SUBTASK 32-11-00-860-038

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(14) Open this circuit breaker and install safety tag:
CAPT Electrical System Panel, P18-3

Row	Col	Number	Name
E	4	(C00700)	HEATERS DRAIN MAST AIR

SUBTASK 32-11-00-860-003

(15) Open these circuit breakers and install safety tags:
F/O Electrical System Panel, P6-3

Row	Col	Number	Name
A	16	(C01345)	LANDING GEAR AUTOBRAKE BITE CONT 2
A	18	(C00583)	LANDING GEAR AUTOBRAKE BITE CONT 1
B	16	(C01346)	LANDING GEAR PARKING BRAKE
B	17	(C00129)	LANDING GEAR LATCH & PRESS WARN
C	15	(C01355)	LANDING GEAR AIR/GND SYS 2
C	16	(C01356)	LANDING GEAR AIR/GND SYS 1
D	1	(C01399)	PSEU PRI
D	2	(C01400)	PSEU ALTN
D	15	(C01401)	LANDING GEAR AIR/GND RELAY
D	16	(C01432)	LANDING GEAR ALTN EXTEND SOL
E	16	(C00196)	LANDING GEAR ANTISKID INBD
E	18	(C00195)	LANDING GEAR ANTISKID OUTBD

SUBTASK 32-11-00-200-002

(16) Do this task: Main Landing Gear Beam Hanger Link - Free Play Check, TASK 57-15-00-200-802.

NOTE: Free play check is necessary, when the landing gear is replaced according to the scheduled restoration task (MPD Item Number 32-040-01 or 32-040-02). If the landing gear is not being replaced as a scheduled restoration task (MPD Item Number 32-040-01 or 32-040-02), then the free play check is optional.

SUBTASK 32-11-00-010-003

(17) Remove the applicable access panel:
(a) For the left landing gear, open these access panels:

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Number	Name/Location
551BB	Lower Inboard Fixed Trailing Edge, Gear Adjustment Door
551CB	Lower Inboard Fixed Trailing Edge, Gear Access Panel

(b) For the right landing gear, open these access panels:

Number	Name/Location
651BB	Lower Inboard Fixed Trailing Edge, Gear Door Adjustment
651CB	Lower Inboard Fixed Trailing Edge, Gear Access Panel

(c) Open these access panels:

Number	Name/Location
551DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel
651DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel

SUBTASK 32-11-00-020-029

(18) Do this task: Main Landing Gear Wheel and Tire Assembly Removal, TASK 32-45-11-000-801.

SUBTASK 32-11-00-020-030

(19) Do this task: Main Landing Gear Brake Removal, TASK 32-41-41-000-801.

SUBTASK 32-11-00-020-002

(20) Do this task: Main Landing Gear Door Assembly Removal, TASK 32-13-11-000-803.

SUBTASK 32-11-00-020-003

(21) Do this task: Main Landing Gear Wing Door Removal, TASK 32-13-21-000-802.

SUBTASK 32-11-00-020-004

(22) Do this task: Main Landing Gear Lower Side Strut Removal, TASK 32-11-61-000-802.

SUBTASK 32-11-00-020-019

(23) Disconnect the reaction link assembly [30] from the outer cylinder trunnion [33].
(a) Remove the cotter pin [41], nut [40], and washer [39] from the crossbolt [31].
(b) Remove the crossbolt [31] from the reaction link pin [32].
(c) Remove the end cap [42] from the reaction link pin [32].
(d) Use the slug driver, C32029-11, SPL-1367, to remove the reaction link pin [32] that holds the reaction link assembly [30] to the trunnion [33].

SUBTASK 32-11-00-020-020

(24) Remove the nuts [37], washer [35], washer [36], and bolts [34] to disconnect the turnbuckle [38] from the bracket.

SUBTASK 32-11-00-020-021

MECH	INSP

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

	MECH	INSP

(25) Disconnect the retract actuator [43] from the main gear trunnion.
(a) Remove the cotter pin [56], nut [47], and washer [46] from the safety bolt [44].
(b) Remove the bolt [44] from the actuator attach pin [50].
(c) Use the wrench adapter, C32029-7 or C32029-98, SPL-1363, to hold the head of the actuator attach pin [50].
(d) Remove the nut [45] and washer [48] from the actuator attach pin [50].
(e) Remove the wrench adapter, C32029-7 or C32029-98, SPL-1363, from the head of the actuator attach pin [50].
(f) Install the thread protector, C32029-16, SPL-1372, on the actuator attach pin [50].
(g) Remove the actuator attach pin [50] to disconnect the retract actuator [43] from the main gear trunnion.
(h) Remove the thread protector, C32029-16, SPL-1372, from the actuator attach pin [50].
(i) Remove the spacers [49].

SUBTASK 32-11-00-020-022

(26) Disconnect the walking beam [51] from the main gear trunnion.
(a) Remove the cotter pin [57], nut [47], and washer [46] from the safety bolt [54].
(b) Remove the safety bolt [54] from the walking beam attach pin [52].
(c) Use the wrench adapter, C32029-6 or C32029-96, SPL-1362, to hold the head of the walking beam attach pin [52].
(d) Remove the nut [55] and washer [53].
(e) Remove the wrench adapter, C32029-6 or C32029-96, SPL-1362, from the head of the walking beam attach pin [52].
(f) Install the thread protector, C32029-15, SPL-1371, on the walking beam attach pin [52].
(g) Remove the walking beam attach pin [52] to disconnect the walking beam [51] from the main gear trunnion.
(h) Remove the thread protector, C32029-15, SPL-1371, from the walking beam attach pin [52].

SUBTASK 32-11-00-480-008

(27) Use rope to hold the disconnected ends of the walking beam [51] and retract actuator [43], such that the ends will be out of the way when you remove the main landing gear [93].

SUBTASK 32-11-00-010-004

(28) Remove the applicable access panel:
For the left wing, open this access panel:

Number	Name/Location
551BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel

For the right wing, open this access panel:

Number	Name/Location
651BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel

N/A

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

SUBTASK 32-11-00-020-017

(29) Remove the hose guide [18].

SUBTASK 32-11-00-020-006

(30) If you will remove the right gear, disconnect the interlock valve cable of the ground spoiler, do this task: Ground Spoiler Interlock Valve Cable Removal, TASK 27-62-51-000-801.

F. Main Landing Gear Removal

SUBTASK 32-11-00-020-007

(1) Disconnect the brake hose assembly [6] and hydraulic return shimmy damper line [114] (TASK 20-10-52-000-801).
- (a) Put a tag to identify the brake hose assembly [6] and hydraulic return shimmy damper line [114] to prevent the crossing of the hoses during the installation.
- (b) Disconnect the brake hose assembly [6] and hydraulic return shimmy damper line [114] from the elbow [5].
- (c) Put plugs in the brake hose assembly [6] and hydraulic return shimmy damper line [114].
- (d) Put caps on the elbow [5].
- (e) Move the brake hose assembly [6] and hydraulic return shimmy damper line [114] away from the work area.

SUBTASK 32-11-00-020-008

(2) Disconnect the electrical connector [11].
- (a) Remove the lockwire from the electrical connector [11].
- (b) Remove the screws [7], washers [8], clamps [9], and nuts [10] that hold the electrical connector [11] to the structure.
- (c) Disconnect the electrical connector [11] from the electrical connector.
- (d) Put cap on the electrical connector [11].
- (e) Pull the electrical connector [11] out of the hose guide [12].
- (f) Wind the electrical connector [11] into loop.

SUBTASK 32-11-00-020-009

(3) Disconnect the electrical connector [13].
- (a) Remove the lockwire from the electrical connector [13].
- (b) Remove the screws [14], washers [15], clamps [16], and nuts [17] that hold the electrical connector [13] to the hose guide [18].
- (c) Disconnect the electrical connector [13] from the electrical connector.
- (d) Put cap on the electrical connector [13].
- (e) Wind the electrical connector [13] into loop.

SUBTASK 32-11-00-020-025

(4) Remove the aft trunnion cross bolt [23].
- (a) Remove the pin [20], nut [21], and washer [22] from the aft trunnion cross bolt [23].
  - 1) Use the wrench adapter, C32029-48 or C32029-90, SPL-14021, to hold the head of the aft trunnion cross bolt [23].
- (b) Install the thread protector, C32029-10, SPL-1366, on the aft trunnion cross bolt [23].
- (c) Remove the aft trunnion cross bolt [23] to disconnect the aft trunnion pin [19] from the aft trunnion.

PARTIAL SIGN OFF STATUS:

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(d) Remove the thread protector, C32029-10, SPL-1366, from the aft trunnion cross bolt [23].

SUBTASK 32-11-00-020-024

(5) Remove the forward trunnion cross bolt [28].

(a) Remove the pin [24], nut [25], and washer [26] from the forward trunnion cross bolt [28].

1) Use the wrench adapter, C32029-49 or C32029-88, SPL-14022, or wrench adapter, C32029-50 or C32029-89, SPL-14023, to hold the head of the forward trunnion cross bolt [28].

(b) Install the thread protector, C32029-9, SPL-1365, or thread protector, C32029-51, SPL-14024, on the forward trunnion cross bolt [28].

(c) Remove the forward trunnion cross bolt [28] to disconnect the forward trunnion from the forward trunnion bearing pin [29].

(d) Remove the thread protector, C32029-9, SPL-1365, or thread protector, C32029-51, SPL-14024, from the forward trunnion cross bolt [28].

SUBTASK 32-11-00-480-009

CAUTION: MAKE SURE THAT THE FORWARD TRUNNION BEARING PIN DOES NOT MOVE ON THE PROTECTOR ASSEMBLY DURING THE REMOVAL OR INSTALLATION OF THE MAIN LANDING GEAR. IF THIS CONDITION OCCURS, IT CAN CAUSE DAMAGE TO THE PIN.

(6) Install the protector assembly [92], (C32034-228/-235).

NOTE: The protector assembly is a part of this tool set fixture, SPL-1869.

NOTE: The protector assembly will prevent damage to the forward trunnion bearing pin during removal or installation of the main landing gear.

(a) Put the protector assembly [92], (C32034-228/-235) on the forward trunnion bearing pin [29].

(b) Secure in place with tape.

SUBTASK 32-11-00-480-010

(7) Attach the support assembly [96], (C32034-212) to the main landing gear [93].

NOTE: A minimum of two persons are necessary to do this task.

(a) Make sure that the side strut attach assembly [94], (C32034-14) is not attached to the support assembly [96], (C32034-212) and jackball pin is removed from the hole in the jackball fitting.

NOTE: The jackball fitting is located on the support assembly [96], (C32034-212).

(b) Put the side strut attach assembly [94], (C32034-14) into the lower side strut attach point on the main landing gear [93].

1) Make sure that the stenciled plate of the assembly is in vertical and in the up position.

2) Install the washer, and nut finger tight on the threaded end of the side strut attach assembly [94], (C32034-14).

(c) Put the support assembly [96], (C32034-212) in its position on the aft side of the main landing gear [93].

1) Make sure that the jackball fitting on the support assembly is fully engaged on the jacking ball of the main landing gear.

(d) If it is necessary, align the holes in the support assembly [96], (C32034-212) with the jacking ball hole for the installation of the jackball pin.

PARTIAL SIGN OFF STATUS:

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(e) Put the jackball pin through the jackball fitting and jacking ball of the main landing gear [93].
(f) Install the lynch pin in the jackball pin.
(g) Loosen the nut that holds the side strut attach assembly [94], (C32034-14) such that the stenciled plate can be attached to the studs on the side strut support assembly [97], (C32034-12).
1) If it is necessary, adjust the adjustable square support tube of the side strut support assembly [97], (C32034-12).
(h) Tighten the two hand knobs to hold the stenciled plate to the studs on the side strut support assembly [97], (C32034-12).
(i) Tighten the nut that holds the side strut attach assembly [94], (C32034-14) to the lower side strut attach point.

SUBTASK 32-11-00-840-001

(8) Lift the airplane with the jacks until the lower surface of the housing assembly, for the forward trunnion bearing pin [29], is 97 ±2 in. (246 ±5 cm) above the ground (TASK 07-11-01-580-815) (View B-B, Figure 401).
(a) Make sure to keep the airplane level.

SUBTASK 32-11-00-480-011

(9) Attach the dolly assembly [98], (C32034-213) to the support assembly [96], (C32034-212).
NOTE: A minimum of two persons are necessary to do this task.
(a) If installed, remove the transportation support assembly and aft tire support assembly from the dolly assembly [98], (C32034-213).
(b) From the aft side, carefully move the dolly assembly [98], (C32034-213) in the forward direction under the wing.
(c) Push the dolly assembly [98], (C32034-213) aft such that the boom arm [91], (C32034-214) points forward of the main landing gear [93] and legs are parallel to the fuselage and in the center with the main landing gear [93].
(d) Lift or lower the boom arm [91], (C32034-214) until you can connect the lift link [95], (C32034-34) to the support assembly [96], (C32034-212).
(e) Install the pins to hold the lift link [95], (C32034-34) to the support assembly [96], (C32034-212).
(f) Connect the winch assembly [99], (C32034-207) to the support assembly [96], (C32034-212) with the snapper pin.

SUBTASK 32-11-00-020-011

WARNING: MAKE SURE THAT THE HYDRAULIC PRESSURE IS BELOW 5000 PSIG (34,474 KPA). IF YOU CONTINUE WHEN THE HYDRAULIC PRESSURE IS 5000 (34,474 KPA) OR HIGHER, A BINDING CONDITION WILL OCCUR. A BINDING CONDITION CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(10) Prepare the dolly assembly [98], (C32034-213) to disconnect the aft trunnion.
(a) Operate the hydraulic pump on the dolly assembly [98], (C32034-213) until approximately 1500 psi (10,342 kPa) is shown on the gage.
1) Make sure that the dolly assembly [98], (C32034-213) stays in center with the main landing gear [93].
(b) Make sure that the swivel locks for the single caster wheels are released.
(c) Use the pry lever bar to remove the load off of the caster wheels one at a time.

PARTIAL SIGN OFF STATUS:

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(d) Use the lever bar assemblies to turn the single caster wheels parallel to the legs of the dolly assembly [98], (C32034-213).
1) Lock the swivel locks for the single wheel casters.

(e) Operate the hydraulic pump on the dolly assembly [98], (C32034-213) to decrease the weight of the main landing gear from the aft trunnion pin [19].

NOTE: The amount of pressure required to perform this step will depend on how many components are installed on the gear at the time of removal, and it will vary approximately between 2500 psi (17,237 kPa) to 4500 psi (31,026 kPa).

(f) Put chocks on all the caster wheels.

SUBTASK 32-11-00-480-012

(11) Use the equipment, SPL-1862, to push the aft trunnion pin [19] into the aft trunnion.
(a) Attach the push adapter [75] to the shaft [71].
(b) Put the push adapter [75] through the pin access hole in the main landing gear beam.
(c) Push the slide hammer [72] until the aft trunnion pin [19] is disconnected from the aft trunnion bearing and it is inside the aft trunnion.

SUBTASK 32-11-00-480-013

(12) Use tie wraps to hold the aft trunnion pin [19] inside the aft trunnion.

NOTE: This will prevent the aft trunnion pin from falling out when you do the subsequent steps.

SUBTASK 32-11-00-950-003

(13) Put protective tape on the outer edge of the aft trunnion to protect its surface finish from scratching.

SUBTASK 32-11-00-420-021

WARNING: MAKE SURE THAT THE HYDRAULIC PRESSURE IS BELOW 5000 PSIG (34,474 KPA). IF YOU CONTINUE WHEN THE HYDRAULIC PRESSURE IS 5000 (34,474 KPA) OR HIGHER, A BINDING CONDITION WILL OCCUR. A BINDING CONDITION CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(14) Lower the aft trunnion.

NOTE: The initial main landing gear [93] removal path is done with the forward trunnion bearing pin [29] attached to the forward trunnion. This removal path follows a combination of movements from lowering and moving the aft trunnion inboard while the main landing gear pivots about the forward trunnion bearing pin.

(a) Make sure to lock the swivel locks for the caster wheels when the main landing gear [93] pivots about the forward trunnion bearing pin [29] as follows:
1) Use the pry lever bar to remove the load off of the caster wheels one at a time.
2) Lock the single wheel casters perpendicular to the forward trunnion pivot axis approximately 38 degrees, and 90 degrees for the double wheel caster.

(b) Put the fork end of the lever bar assemblies on the rectangular tubing above the single wheel casters.

(c) Hold the lever bar assemblies to slowly push the aft end of the dolly assembly [98], (C32034-213) inboard.

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

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1) Make sure that the swivel locks for all the caster wheels are released.

(d) Do the steps below to lower the aft trunnion inboard while at the same time the aft end of the main landing gear [93] moves inboard:

NOTE: When lifting or lowering the aft trunnion, put the single caster wheels parallel to the dolly assembly legs and lock the swivel locks.

1) Use the winch assembly [99], (C32034-207) to pull the bottom of the main landing gear [93] forward.
2) Use the manual valve on the hydraulic pump to lower the boom arm [91], (C32034-214).
3) Continue to lower and to move the aft trunnion inboard until it clears the following components:
   a) The lower flange of the main landing gear beam
   b) The seals for the lower trailing edge panel
   c) The seal for the support beam
   d) The aft trunnion just below the lower trailing edge panel.

SUBTASK 32-11-00-020-013

WARNING: MAKE SURE THAT THE HYDRAULIC PRESSURE IS BELOW 5000 PSIG (34,474 KPA). IF YOU CONTINUE WHEN THE HYDRAULIC PRESSURE IS 5000 (34,474 KPA) OR HIGHER, A BINDING CONDITION WILL OCCUR. A BINDING CONDITION CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(15) Disconnect the forward trunnion.

(a) Make sure that the swivel locks for the caster wheels are released.
(b) Use the lever bar assemblies to push the dolly assembly [98], (C32034-213) outboard such that the axis of the forward trunnion bearing pin [29] is perpendicular to the housing assembly and rear spar.
(c) Use the pry lever bar to remove the load off of the caster wheels one at a time.
(d) If it is necessary, use the lever bar assemblies and put the single caster wheels parallel to the legs of the dolly assembly [98], (C32034-213).
   1) Lock the swivel locks for the single caster wheels.
(e) Attach the outer cylinder strap assembly to the hoist ring on the end of the boom arm [91], (C32034–214) and around the outer cylinder at the aft trunnion.
(f) Remove the tension from the strap.
(g) Put the chocks approximately 2 in. (5 cm) aft of the single wheel casters, and in contact with the double wheel caster to limit the movement of the dolly assembly [98], (C32034-213).
(h) Make sure that the chock is in contact with the double wheel caster to prevent the movement of the dolly assembly [98], (C32034-213).
(i) Put the fork end of the lever bar assemblies on the rectangular tubing above the single wheel casters.
(j) Hold the lever bar assemblies to slowly push on the dolly assembly [98], (C32034–213) such that the forward trunnion will come off of the forward trunnion bearing pin [29].
   1) Make sure that the swivel locks for all the caster wheels are released.

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

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(k) Continue to push the dolly assembly [98], (C32034-213) while lowering the aft end of the main landing gear [93] with the use of the winch assembly [99], (C32034-207) and boom arm [91], (C32034-214).

(l) Continue this movement until the aft trunnion will clear the lower trailing edge panel seals, the forward trunnion is disconnected from the forward trunnion bearing pin [29].

JXB 004, 005, 007-009, 014 PRE SB 737-32-1448

NOTE: The retaining ring, the support ring assembly, and forward trunnion seal are still on the airplane until the forward trunnion is disconnected.

JXB 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448

NOTE: The retaining ring and seal are still on the airplane until the forward trunnion is disconnected.

JXB ALL

(m) Remove the outer cylinder strap assembly from the hoist ring and outer cylinder.

(n) Use the boom arm [91], (C32034-214) and winch assembly [99], (C32034-207) to lower the main landing gear [93], such that the dolly assembly [98], (C32034-213), with the main landing gear [93] can be removed below the wing.

SUBTASK 32-11-00-580-002

(16) Lower the main landing gear [93] on the tire support assemblies, such that the support assembly [96], (C32034-212) rests in the cradle of the transportation support assembly.

SUBTASK 32-11-00-480-014

(17) Use the retention strap, SPL-1871, to hold the main landing gear [93].

SUBTASK 32-11-00-480-023

(18) Tighten the adjustable square support tube of the side strut support assembly [97], (C32034-12) with the two adjustable handles.

SUBTASK 32-11-00-020-026

CAUTION: DO NOT TOW THE MAIN LANDING-GEAR SUPPORT EQUIPMENT MORE THAN 5 MPH (8 KM/H). THE MAIN LANDING-GEAR SUPPORT EQUIPMENT IS FOR SHOP OPERATION ONLY. IF YOU DO NOT OBEY, DAMAGE TO THE EQUIPMENT CAN OCCUR.

CAUTION: INSTALL THE RETENTION STRAP OR EQUIVALENT ON THE SHOCK STRUT WHEN YOU REMOVE THE MAIN LANDING-GEAR FIXTURE. THE SHOCK STRUT CAN EXTEND AND DAMAGE TO THE EQUIPMENT CAN OCCUR.

(19) Remove the dolly assembly [98], (C32034-213), with main landing gear [93] from the work area.

SUBTASK 32-11-00-210-006

(20) Do a general visual inspection of the main landing gear [93] at the forward trunnion bearing pin [29] and aft trunnion pin [19] locations and the visible surfaces of the

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—	20.3.24 GAT 492
D MDD108	20.3.25 GAT 492

forward trunnion bearing pin [29] and aft trunnion pin [19] for signs of corrosion and chrome plating damage.

(a) If any sign of corrosion or chrome plating damage is found for the forward trunnion bearing pin [29], do these tasks: NIL FINDING — N/A

- MLG Forward Trunnion Housing Assembly - Removal, TASK 57-16-01-000-801
- MLG Forward Trunnion Housing Assembly - Installation, TASK 57-16-01-400-801.

(b) If any sign of corrosion or chrome plating damage is found for the aft trunnion pin [19], do these tasks: NO CORROSION OBSERVED.

- MLG Aft Trunnion Bearing Assembly Removal, TASK 57-16-02-000-801
- MLG Aft Trunnion Bearing Assembly Installation, TASK 57-16-02-400-801.

Figure 401. Main Landing Gear Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

Sheet 4 - Effectivity: JXB 004, 005, 007-009, 014 PRE SB 737-32-1448

Sheet 5 - Effectivity: JXB 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448

Sheet 6 - Effectivity: JXB ALL

Sheet 7 - Effectivity: JXB ALL

Sheet 8 - Effectivity: JXB ALL

Figure 402. Main Landing Gear Trunnion Pin Puller Equipment

Sheet 1 - Effectivity: JXB ALL

Figure 403. Main Landing Gear Support Equipment

Sheet 1 - Effectivity: JXB ALL

Figure 404. Main Landing Gear Clearance Requirements

Sheet 1 - Effectivity: JXB ALL

Figure 405. Main Landing Gear Shock Strut Servicing Chart

Sheet 1 - Effectivity: JXB ALL

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TASK 32-11-61-000-803

2. Main Landing Gear Side Strut Removal (Figure 403)

NOTE: This procedure is a scheduled maintenance task.

A. General

(1) This task supplies instructions to remove the side strut.

B. References

Reference	Title
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-32-91-000-801	Main Gear Downlock Springs Removal (P/B 401)

C. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1359	Wrench Adapter, C32029-3 or C32029-93 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1368	Thread Protector, C32029-12 or C32029-101 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1369	Thread Protector , C32029-13 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)

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SPL-1370 Thread Protector, C32029-14 (included in C32029 kit)

737-800

(Part #: C32029-87, Supplier: 81205)

(Opt Part #: C32029-38, Supplier: 81205)

(Opt Part #: C32029-45, Supplier: 81205)

JXB ALL

D. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
713	Nose Landing Gear
734	Left Main Landing Gear
744	Right Main Landing Gear

E. Prepare for the Removal

SUBTASK 32-11-61-480-006

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR COULD RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed on all the landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-11-61-580-008

(2) Do this task: Lift the Airplane with the Jacks, TASK 07-11-01-580-815.

SUBTASK 32-11-61-860-003

(3) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-11-61-020-010

(4) Do this task: Main Gear Downlock Springs Removal, TASK 32-32-91-000-801.

F. Main Landing Gear Side Strut Removal

SUBTASK 32-11-61-020-011

(1) Disconnect the upper side strut [3] from the downlock strut [1].

(a) Remove the pin [6] from the downlock link pin [2].

(b) Remove the nut [5] and the washer [4] from the downlock link pin [2].

(c) Remove the downlock link pin [2] to disconnect the upper side strut [3] from the downlock strut [1].

NOTE: Use the thread protector, C32029-13, SPL-1369, for the downlock link pin [2] when you do this step.

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PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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SUBTASK 32-11-61-020-012

(2) Disconnect the lower side strut [8] from the pushrod [41].
- (a) Remove the cotter pin [55], nut [42], and washer [43] from the bolt [46].
  1) Discard the cotter pin [55].
- (b) Remove the bolt [46], washer [44], and bushing [45] to disconnect the lower side strut [8] from the pushrod [41].
- (c) Isolate the pushrod [41] from the lower side strut [8].

SUBTASK 32-11-61-480-007

(3) Use a rope to hold the pushrod [41] and the inner door in its position.

SUBTASK 32-11-61-020-013

(4) Disconnect the lower side strut [8] from the shock strut [51].
- (a) Remove the pin [49], nut [50], and washer [48] from the cross bolt [47].
- (b) Remove the cross bolt [47] from the side strut pin [52].
- (c) Remove the side strut nut [54] and the side strut washer [53] from the side strut pin [52].
- (d) Remove the side strut pin [52] to disconnect the lower side strut [8] from the shock strut [51].

  NOTE: Use the thread protector, C32029-14, SPL-1370, for the side strut pin when you do this step.

SUBTASK 32-11-61-020-014

(5) Disconnect the upper side strut [3] from the reaction link assembly [15].
- (a) Remove the pin [19], nut [18], and washer [17] from the cross bolt [16].
- (b) Remove the cross bolt [16] from the side strut pin [22].
- (c) Remove the side strut nut [20] and the side strut washer [21] from the side strut pin [22].

  NOTE: Use wrench adapter, C32029-3 or C32029-93, SPL-1359 , to hold the head of the side strut pin [22] when you loosen the side strut nut [20].
- (d) Remove the side strut pin [22] that holds the upper side strut [3] to the reaction link assembly [15].

  NOTE: Use the thread protector, C32029-12 or C32029-101, SPL-1368 , for the side strut pin when you do this step.

SUBTASK 32-11-61-020-015

(6) Remove the assembly of the upper side strut [3] and the lower side strut [8] from the airplane.

NOTE: The side strut assembly weighs approximately 90.11 lb (40.87 kg).

Figure 403. Main Landing Gear Side Strut Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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JXB ALL

TASK 32-32-31-000-801

3. **Main Gear Uplock Assembly Removal** (Figure 401) (Figure 402)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
10-11-05-500-801	Chock Installation in Winds or Wind Gusts to a Maximum of 35 Knots (P/B 201)
10-11-05-500-802	Chock Installation in Winds More than 35 Knots - Handling (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-34-21-000-801	Removal of the Manual Extension Mechanism for the Main Gear (P/B 401)
32-61-21-020-801	Main Landing Gear Uplock Sensor Removal (P/B 401)

B. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors
730	Subzone - Left Main Landing Gear and Landing Gear Doors
740	Subzone - Right Main Landing Gear and Landing Gear Doors

C. Prepare for the Removal

SUBTASK 32-32-31-480-001

WARNING: MAKE SURE THAT YOU INSTALL THE DOWNLOCK PINS ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) Make sure that the downlock pins are installed in the nose and main landing gear (TASK 32-00-01-480-801).

SUBTASK 32-32-31-860-001

(2) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-32-31-580-001

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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(3) Install chocks around the tires of the main landing gear (TASK 10-11-05-500-801 or TASK 10-11-05-500-802).

SUBTASK 32-32-31-020-001

(4) Remove the uplock sensors from the uplock assembly [9] (TASK 32-61-21-020-801).

D. Main Gear Uplock Assembly Removal

SUBTASK 32-32-31-020-009

(1) Put tags on the two hydraulic lines to show their connection locations.

SUBTASK 32-32-31-020-002

(2) Disconnect the two hydraulic lines from the uplock actuator.

SUBTASK 32-32-31-480-002

(3) Install plugs in the two hydraulic lines.

SUBTASK 32-32-31-020-020

(4) Disconnect the control rod of the manual extension mechanism from the uplock assembly [9] (TASK 32-34-21-000-801).

SUBTASK 32-32-31-020-003

(5) If it is necessary to remove the hanger assembly [19], remove the pins [13], nuts [12], uplock washers [11], and uplock hanger pins [10] to disconnect the hanger assembly [19].

SUBTASK 32-32-31-020-004

(6) Remove the pin [8], nut [7], uplock washer [6], and uplock attach pin [14] to disconnect the uplock assembly [9] from the structure.

SUBTASK 32-32-31-020-005

(7) Remove uplock assembly [9].

SUBTASK 32-32-31-020-006

(8) Remove the pin [5], nut [4], washer [3], bolt [1] and bushing [2] to disconnect the hanger assembly [19] from the structure.

SUBTASK 32-32-31-020-007

(9) Remove the hanger assembly [19].

SUBTASK 32-32-31-020-008

(10) If the uplock actuator on the replacement uplock assembly does not have hydraulic fittings installed, do these steps:
(a) Remove the union [18] and the restrictor [16] from the uplock actuator.
(b) Remove and discard the packing [15], and packing [17].
(c) Install plugs in the ports on the uplock actuator.

Figure 401. Main Landing Gear Uplock Assembly Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

Sheet 4 - Effectivity: JXB ALL

Sheet 5 - Effectivity: JXB ALL

Figure 402. Main Landing Gear Uplock Assembly Adjustment

Sheet 1 - Effectivity: JXB ALL

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

JXB ALL

TASK 32-11-89-000-801

4. **Main Landing Gear Downlock Strut Removal** (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)
07-11-03-580-801	Lift the Main Landing Gear Axles with the Axle Jacks (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-080-801	Landing Gear Downlock Pins Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-32-91-000-801	Main Gear Downlock Springs Removal (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1361	Wrench Adapter, C32029-39 or C32029-95 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1369	Thread Protector , C32029-13 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1370	Thread Protector, C32029-14 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205)

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

	MECH	INSP

(Opt Part #: C32029-45, Supplier: 81205)

SPL-12498 MLG downlock release level tool

737-800

(Part #: C32052-1, Supplier: 81205)

STD-14686 Fittings - Hydraulic, Caps/Plugs (ref. AMM 20-10-XX)

C. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
713	Nose Landing Gear
734	Left Main Landing Gear
744	Right Main Landing Gear

D. Prepare for the Removal

SUBTASK 32-11-89-480-001

WARNING: MAKE SURE THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR COULD RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed in the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-11-89-860-001

(2) Lift the airplane on jacks until the main landing gear wheels are off of the ground (TASK 07-11-01-580-815).

SUBTASK 32-11-89-860-002

(3) Remove the power from hydraulic system A and B, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-11-89-020-001

(4) Do this task: Main Gear Downlock Springs Removal, TASK 32-32-91-000-801.

E. Main Landing Gear Downlock Strut Removal

SUBTASK 32-11-89-020-002

(1) Do these steps to disconnect the hydraulic lines [2] from the downlock actuator [4]:
- (a) Put tags on the hydraulic lines [2] to prevent the crossing of the hydraulic lines [2] during the installation.
- (b) Disconnect the hydraulic lines [2] from the downlock actuator [4].
- (c) Install the STD-14686 cap/plug, on the hydraulic lines [2].
- (d) Remove and discard the packings [3].
- (e) Install the STD-14686 cap/plug, in the ports of the downlock actuator [4].

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

	MECH	INSP

SUBTASK 32-11-89-020-008

(2) Use a downlock release tool, SPL-12498, to manually release the Main Landing Gear (MLG) downlock mechanism.

SUBTASK 32-11-89-020-003

(3) Do these steps to disconnect the rod end of the downlock actuator [4] from the downlock strut [5]:
- (a) Remove the spacer [7] from the rod end pin [8].
- (b) From the forward side, remove the rod end pin [8] that holds the rod end of the downlock actuator [4] to the downlock strut [5].
- (c) Remove the spacers [9] and washers [10].
- (d) Slowly lower the rod end of the downlock actuator [4] to the upper end of the upper side strut.
- (e) Use a tape to attach the rod end of the downlock actuator [4] to the upper end of the upper side strut.

NOTE: This will prevent the downlock actuator from moving when the subsequent steps are done.

SUBTASK 32-11-89-020-004

(4) Do these steps to disconnect the electrical harnesses [11] from the downlock strut [5]:
- (a) Remove the nuts [20], washers [19], and bolts [18] to disconnect the brackets [17] from the downlock strut [5].
  - 1) Remove the brackets [17] with the sensors attached.
- (b) Isolate the brackets [17] from the downlock strut [5].

CONNECTORS D46012P & D46018P DISCONNECTED.

SUBTASK 32-11-89-080-001

(5) Remove the downlock pin from the MLG in order to remove the downlock strut [5] (TASK 32-00-01-080-801).

SUBTASK 32-11-89-020-005

(6) Do these steps to disconnect the downlock strut [5] from the reaction link assembly [6]:
- (a) Remove the pin [22] from the downlock link pin [12].
- (b) Remove the nut [21] and washer [23] from the downlock link pin [12].
  - 1) Use a wrench adapter, C32029-39 or C32029-95, SPL-1361 , to hold the head of the downlock link pin [12] when the nut [21] is loosen.
- (c) Remove the downlock link pin [12] that holds the downlock strut [5] to the reaction link assembly [6].
  - 1) Use a thread protector, C32029-14, SPL-1370.

SUBTASK 32-11-89-020-006

(7) Do these steps to disconnect the downlock strut [5] from the upper side strut:
- (a) Remove the pin [16] from the downlock link pin [13].
- (b) Remove the nut [15] and washer [14] from the downlock link pin [13].
- (c) Remove the downlock link pin [13].
  - 1) Use a thread protector, C32029-13, SPL-1369.

SUBTASK 32-11-89-020-007

(8) Remove the downlock strut [5] from the airplane.

SUBTASK 32-11-89-580-001

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(9) If the removal of the downlock strut [5] is difficult do this task to decrease the weight on the gear: Lift the Main Landing Gear Axles with the Axle Jacks, TASK 07-11-03-580-801.

Figure 401. Main Landing Downlock Strut Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

GAT 492

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH | INSP

JXB ALL

TASK 32-11-71-000-801

5. **Main Landing Gear Reaction Link Assembly Removal** (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)
29-09-00-860-802	Hydraulic Reservoirs Depressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-32-91-000-801	Main Gear Downlock Springs Removal (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1359	Wrench Adapter, C32029-3 or C32029-93 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1360	Wrench Adapter, C32029-4 or C32029-94 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1364	Slug Driver Assy, C32029-8 (include in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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SPL-1368	Thread Protector, C32029-12 or C32029-101 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-1369	Thread Protector , C32029-13 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-11946	Wrench Adapter, C32029-31 or C32029-99 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)	B	20.3.24 GAT 492
STD-14686	Fittings - Hydraulic, Caps/Plugs (ref. AMM 20-10-XX)		

C. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
713	Nose Landing Gear
734	Left Main Landing Gear
744	Right Main Landing Gear

D. Prepare for the Removal

SUBTASK 32-11-71-480-001

WARNING: MAKE SURE THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed on all the landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

		B	20.3.24 GAT 492
		B MDD008	20.3.24 GAT 492

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

	MECH	INSP

SUBTASK 32-11-71-580-002

(2) Do this task: Lift the Airplane with the Jacks, TASK 07-11-01-580-815.

SUBTASK 32-11-71-860-005

(3) Remove the power from the hydraulic systems A and B, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-11-71-860-001

(4) Remove the pressure from the hydraulic reservoirs, do this task: Hydraulic Reservoirs Depressurization, TASK 29-09-00-860-802.

E. Main Landing Gear Reaction Link Assembly Removal

SUBTASK 32-11-71-020-001

(1) Do this task: Main Gear Downlock Springs Removal, TASK 32-32-91-000-801.

SUBTASK 32-11-71-020-002

(2) Do these steps to disconnect the hydraulic line [37] and hydraulic line [37A] from the hydraulic tubes [2]:

(a) Disconnect the hydraulic line [37] and hydraulic line [37A] from the hydraulic tubes [2] attached to the bracket assembly [38].

(b) Install the STD-14686 caps/plugs, on the hydraulic line [37] and hydraulic line [37A].

(c) Put tags on the hydraulic line [37] and hydraulic line [37A] to prevent the crossing of the lines during the installation.

(d) Disconnect the hydraulic tubes [2] from the downlock actuator [4].

(e) Install the STD-14686 caps/plugs, on the hydraulic tubes [2].

(f) Install the STD-14686 caps/plugs, on the unions in the ports of the downlock actuator [4].

SUBTASK 32-11-71-020-003

(3) Do these steps to disconnect the reaction link assembly [1] from the downlock actuator [4]:

(a) Remove the cotter pin [11], nut [10], and the washer [9] from the rod end pin [6].

(b) Remove the rod end pin [6] that attaches the head end of the downlock actuator [4] to the reaction link assembly [1].

(c) Remove the washers [7] and washers [8].

(d) Slowly lower the head end of the downlock actuator [4] to the upper end of the upper side strut [21].

(e) Use a tape to attach the head end of the downlock actuator [4] to the upper end of the upper side strut [21].

NOTE: This task will prevent the downlock actuator from moving when you do the subsequent steps.

SUBTASK 32-11-71-020-004

(4) Do these steps to disconnect the reaction link assembly [1] from the upper lock link [5]:

(a) Remove the cotter pin [16] from the downlock link pin [13].

(b) Remove the nut [15] and the washer [14] from the downlock link pin [13].

1) Use the wrench adapter wrench adapter, C32029-4 or C32029-94, SPL-1360 , to hold the head of the downlock link pin [13] when you loosen the nut [15].

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(c) Remove the downlock link pin [13] that attaches the reaction link assembly [1] to the upper lock link [5].
1) Use the thread protector, C32029-13, SPL-1369, when you do this step.

SUBTASK 32-11-71-020-005

(5) Do these steps to disconnect the reaction link assembly [1] from the upper side strut [21]:
(a) Remove the cotter pin [23], nut [22] and washer [24] from crossbolt [17].
(b) Remove the crossbolt [17] from the side strut pin [20].
(c) Remove the side side strut nut [18] and the side strut washer [19] from the side strut pin [20].
1) Use the wrench adapter wrench adapter, C32029-3 or C32029-93, SPL-1359 , to hold the head of the side strut pin [20] when you loosen the side strut nut [18].
(d) Attach a strap to the upper end of the upper side strut [21].
1) If you will not remove the hanger link assembly [35], attach a strap to the upper end of the upper side strut [21] and loop the strap over the hanger link assembly [35]. Route the strap such that it will not interfere with removal of the reaction link assembly [1].
2) If you will remove the hanger link assembly [35], attach a strap to the upper end of the upper side strut [21] and loop the strap over the main landing gear beam support hanger, located aft of the hanger link assembly [35]. Route the strap such that it will not interfere with removal of the reaction link assembly [1].

WARNING: YOU MUST HAVE TWO PERSONNEL TO LIFT THIS COMPONENT. THE COMPONENT IS HEAVY. IF YOU DO NOT OBEY, INJURY TO PERSONNEL CAN OCCUR.

(e) Remove the side strut pin [20] that attach the reaction link assembly [1] to the upper side strut [21].

NOTE: The reaction link assembly [1] weighs approximately 63.2 lb (28.7 kg).

1) Use the thread protector, C32029-12 or C32029-101, SPL-1368 , when you do this step.

SUBTASK 32-11-71-020-006

(6) Do these steps to disconnect the reaction link assembly [1] from the hanger link assembly [35]:
(a) Remove the pin assembly [31] from the reaction link assembly [1].
(b) Remove the cotter pin [28], nut [29] and washer [30] from the crossbolt [25].
(c) Remove the crossbolt [25] from the end cap [27].
(d) Remove the washer [26] from the crossbolt [25].
(e) Isolate the end cap [27], bracket assembly [38], hydraulic tube [2], electrical harness [12], and the spacer [32] and spacer [36] as an assembly from the reaction link assembly [1].
(f) If it is necessary to disconnect the electrical harness [12], do this step:
1) Disconnect the electrical harness [12] by removing the bolts for the bracket which holds the sensor to the upper lock link [5].
(g) Remove the hanger link pin [34] that attaches the reaction link assembly [1] to the hanger link assembly [35].

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

1) Use the slug driver slug driver assy, C32029-8, SPL-1364 , when you do this step.

SUBTASK 32-11-71-020-013

(7) If it is necessary, do the following steps to remove the hanger link assembly [35] from the supporting structure:
(a) Remove the cotter pin [53], nut [52] and washer [51] from the crossbolt [50].
(b) Remove the cross bolt [50] from the hanger link assembly pin [46].
(c) Remove the nut [49] and washer [48] from the hanger link assembly pin [46] with wrench adapter, C32029-31 or C32029-99, SPL-11946 .
(d) Remove the hanger link assembly pin [46] that attaches the hanger link assembly to the supporting structure.
(e) Remove the hanger link assembly [35].

SUBTASK 32-11-71-020-007

(8) Do these steps to disconnect the reaction link assembly [1] from the trunnion [41]:
(a) Remove the cotter pin [44], nut [43] and washer [42] from the crossbolt [39].
(b) Remove the crossbolt [39] from the trunnion pin [40].
(c) Remove the end cap [45] from the trunnion pin [40].
(d) Remove the trunnion pin [40] that attaches the reaction link assembly [1] to the trunnion [41].
1) Use the slug driver slug driver assy, C32029-8, SPL-1364 , when you do this task.

SUBTASK 32-11-71-020-008

WARNING: YOU MUST HAVE TWO PERSONNEL TO LIFT THIS COMPONENT. THE COMPONENT IS HEAVY. IF YOU DO NOT OBEY, INJURY TO PERSONNEL CAN OCCUR.

(9) Remove the reaction link assembly [1] from the airplane.

NOTE: The reaction link assembly weighs approximately 63.2 lb (28.7 kg).

Figure 401. Main Landing Gear Reaction Link Assembly Installation

Sheet 1 - Effectivity: JXB ALL
Sheet 2 - Effectivity: JXB ALL
Sheet 3 - Effectivity: JXB ALL
Sheet 4 - Effectivity: JXB ALL
Sheet 5 - Effectivity: JXB ALL
Sheet 6 - Effectivity: JXB ALL

MECH | INSP

JXB ALL

TASK 32-32-11-000-801

6. Removal Of The Actuator Assembly For The Main Gear (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
10-11-05 P/B 201	CHOCK INSTALLATION
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1870	Hoist - Main Landing Gear Retract Actuator **737-800** (Part #: C32028-109, Supplier: 81205) (Part #: C32028-110, Supplier: 81205) (Part #: C32028-111, Supplier: 81205) (Opt Part #: C32028-1, Supplier: 81205) (Opt Part #: C32028-3, Supplier: 81205) (Opt Part #: C32028-56, Supplier: 81205) (Opt Part #: C32028-58, Supplier: 81205) (Opt Part #: C32028-59, Supplier: 81205)

JXB ALL

C. Location Zones

Zone	Area
211	Flight Compartment - Left
212	Flight Compartment - Right
551	Left Wing - Rear Spar To Landing Gear Support Beam
651	Right Wing - Rear Spar to Landing Gear Support Beam
730	Subzone - Left Main Landing Gear and Landing Gear Doors

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

		MECH	INSP
740	Subzone - Right Main Landing Gear and Landing Gear Doors		

D. Access Panels

Number	Name/Location
551BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel
551DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel
651BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel
651DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel

E. Prepare for the Removal

SUBTASK 32-32-11-480-001

WARNING: MAKE SURE THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) Make sure the downlock pins are installed in the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-32-11-860-001

(2) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-32-11-580-001

(3) Install chocks around the tires of the main landing gear (CHOCK INSTALLATION, PAGEBLOCK 10-11-05/201).

SUBTASK 32-32-11-010-001

(4) Open the applicable access panels:

Number	Name/Location
551BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel
551DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel
651BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel
651DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel

F. Actuator Assembly Removal

SUBTASK 32-32-11-020-001

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(1) Disconnect the hydraulic line [27] UP and hydraulic line [28] DN from the actuator assembly [2].

SUBTASK 32-32-11-480-002

(2) Install plugs in the hydraulic hoses.

SUBTASK 32-32-11-480-003

(3) Install the hoist, SPL-1870.

SUBTASK 32-32-11-980-001

(4) Hold the head end of the actuator assembly [2] in position with the hoist, SPL-1870.

SUBTASK 32-32-11-020-002

(5) Remove these parts to disconnect the beam hanger [3] from the structure:
- (a) pin [5]
- (b) nut [6]
- (c) washer [7]
- (d) bolt [4].

SUBTASK 32-32-11-020-003

(6) Hold the beam assembly [1] and actuator assembly [2] and remove these parts to disconnect the beam assembly [1] from the main gear trunnion:
- (a) pin [12], nut [11], washer [10], and bolt [17]
- (b) nut [18] and washer [19]
- (c) walking beam attach pin [16].

SUBTASK 32-32-11-020-004

(7) Hold the beam assembly [1] and actuator assembly [2] and remove these parts to disconnect the actuator assembly [2] from the main gear trunnion:
- (a) pin [12], nut [11], washer [10], and bolt [8]
- (b) nut [9] and washer [13]
- (c) spacers [14]
- (d) actuator attach pin [15].

SUBTASK 32-32-11-020-005

(8) Remove the beam assembly [1] and actuator assembly [2] with the hoist, SPL-1870.

SUBTASK 32-32-11-080-001

(9) Remove the hoist, SPL-1870 from the beam assembly [1] and actuator assembly [2].

SUBTASK 32-32-11-020-006

(10) Do these steps to remove the actuator assembly [2] from the beam assembly [1] and the beam hanger [3]:
- (a) Remove the pin [12], nut [11], washer [10], and bolt [17].
- (b) Remove the nut [21] and washer [22].
- (c) Remove the pin [20].
- (d) Remove the actuator assembly [2].

SUBTASK 32-32-11-020-007

(11) Do these steps if the replacement actuator assembly [2] does not have a restrictor [23] and union [26] installed:
- (a) Remove the restrictor [23] and the union [26] from the actuator assembly [2].
- (b) Remove and discard the packing [24], and packing [25].
- (c) Install plugs in the ports on the actuator assembly [2].

Figure 401. Main Gear Actuator Assembly Installation

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

Sheet 4 - Effectivity: JXB ALL

Sheet 5 - Effectivity: JXB ALL

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TASK 32-11-83-000-801

7. **Main Landing Gear Forward Trunnion Bearing Assembly Removal** (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-11-00-000-801	Main Landing Gear Removal (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1868	Puller - Fuse Pin, Main Landing Gear Forward Trunnion Support **737-800** (Part #: C32015-34, Supplier: 81205) (Opt Part #: C32015-1, Supplier: 81205)

JXB ALL

C. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
551	Left Wing - Rear Spar To Landing Gear Support Beam
651	Right Wing - Rear Spar to Landing Gear Support Beam
734	Left Main Landing Gear
744	Right Main Landing Gear

D. Prepare for the Removal

SUBTASK 32-11-83-480-001

WARNING: MAKE SURE THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

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(1) If the downlock pins are not installed on all the landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-11-83-020-001

(2) Do this task: Main Landing Gear Removal, TASK 32-11-00-000-801.

SUBTASK 32-11-83-020-008

(3) Remove the fairlead [13] and fairlead [24] from the bracket to access the forward trunnion bearing assembly [1].

NOTE: Optional to suspend the fairlead [13] and fairlead [24] on the cables.

(a) Remove the bolts [14], washers [15] and nuts [16] to remove the fairlead [13] from the bracket.

(b) Remove the bolts [25], washers [26] and nuts [27] to remove the fairlead [24] from the bracket.

SUBTASK 32-11-83-020-009

(4) Remove the clamp block assemblies [17] to access the forward trunnion bearing assembly [1].

(a) Remove the screws [18], washers [19] and spacers [20] from the clamp block assemblies [17] and nutplates [23].

(b) Remove the channels [21] and clamp blocks [22] from the tubing.

E. Main Landing Gear Forward Trunnion Bearing Assembly Removal

SUBTASK 32-11-83-020-007

(1) Remove the clamp [10] from the hydraulic flow regulator, do this step:

(a) Remove the screws [11] and washers [12].

SUBTASK 32-11-83-020-002

(2) Remove the forward trunnion bearing assembly [1], do these steps:

(a) Remove the nuts [2], washers [5] and the fuse pin caps [3] on each end of the rods [4].

(b) Remove the rods [4] from the fuse pins [6].

(c) Remove the nuts [9] and the special washers [7] from the fuse pins [6].

(d) Use puller, SPL-1868, to remove the fuse pins [6] that hold the housing assembly [8] to the forward trunnion support.

(e) Remove the forward trunnion bearing assembly [1] from the airplane.

Figure 401. Main Landing Gear Forward Trunnion Bearing Assembly Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

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JXB ALL

TASK 32-32-31-400-801

8. **Main Gear Uplock Assembly Installation** (Figure 401) (Figure 402)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)
12-12-00-610-801	Hydraulic Reservoir Servicing (P/B 301)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-080-801	Landing Gear Downlock Pins Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-13-21-000-801	Main Landing Gear Wing Door Adjustment (P/B 501)
32-34-21-400-801	Installation of the Manual Extension Mechanism for the Main Gear (P/B 401)
32-61-21-400-801	Main Landing Gear Uplock Sensor Installation (P/B 401)
32-61-21-820-801	Main Landing Gear Uplock Sensor Clearance Adjustment (P/B 501)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-12460	Equipment - Removal/Installation, MLG and NLG springs **737-800** (Part #: J32037-106, Supplier: 81205)
STD-858	Tag - DO NOT OPERATE
STD-4873	Rope - 0.5 inch (12.7 mm) Diameter

C. Consumable Materials

Reference	Description	Specification
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827

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		(NATO G-354) (Supersedes MIL-G-23827)		
D00153	Fluid - Hydraulic Fluid, Fire Resistant (Interchangeable And Intermixable With BMS 3-11 Type V)	BMS3-11 Type IV		
D00633	Grease - Aircraft General Purpose	BMS3-33	D	22-3-75 GAT 492
G01048	Lockwire - MS20995C32, Corrosion Resistant Steel - 0.032 Inch (0.8128 mm) Diameter	NASM20995		
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38		
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38		
G50316	Cloth - Clean, Dry, Lint-free, White, Cotton			

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity	MECH	INSP
5	Pin	32-32-31-02-055	JXB ALL		
		32-32-31-02A-025	JXB ALL		
8	Pin	32-32-31-02-030	JXB ALL		
		32-32-31-02A-005	JXB ALL		
9	Uplock assembly	32-32-31-02-105	JXB ALL		
		32-32-31-02A-050	JXB 004, 005		
		32-32-31-02A-055	JXB 007-009, 014, 026-031, 033, 036, 037, 040-051, 053-999	D mDock	22-3-75 GAT 492
13	Pin	32-32-31-02-145	JXB ALL		
		32-32-31-02A-115	JXB ALL		
15	Packing	32-32-41-01-035	JXB ALL		
17	Packing	32-32-41-01-030	JXB ALL		
27	Pin	32-13-21-01-202	JXB 004, 005, 007-009, 014, 026-031, 033		

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				MECH	INSP
		32-13-21-04-210	JXB 036, 037, 040-051, 053-999		
35	Pin	32-32-31-02-210	JXB ALL		
		32-32-31-02A-150	JXB ALL		

E. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
710	Subzone - Landing Gear: Nose Landing Gear and Landing Gear Doors
730	Subzone - Left Main Landing Gear and Landing Gear Doors
740	Subzone - Right Main Landing Gear and Landing Gear Doors

F. Main Gear Uplock Assembly Installation

SUBTASK 32-32-31-420-001

(1) Do these steps if the replacement uplock actuator on the uplock assembly [9] does not have the union [18] or restrictor [16] installed:
   (a) Lubricate the new packing [15], new packing [17], union [18], and restrictor [16] with D00153 hydraulic fluid .
   (b) Remove the plugs from the ports on the uplock actuator.
   (c) Install the packing [15], packing [17], union [18], and restrictor [16] in the uplock actuator.

SUBTASK 32-32-31-600-001

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(2) Apply a thin layer of G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (optional), on the parts that follow:
   (a) The threads and thread reliefs of the uplock attach pin [14] and nut [7].
   (b) The faces of the uplock washer [6].
   (c) The pin [8].

SUBTASK 32-32-31-640-001

(3) Lubricate the shank of the uplock attach pin [14] with D00633 grease .

SUBTASK 32-32-31-420-002

(4) Put the uplock assembly [9] in position on the structure.

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SUBTASK 32-32-31-420-003

(5) Install the uplock attach pin [14], uplock washer [6], and nut [7] to attach the uplock assembly [9] to the structure.

SUBTASK 32-32-31-420-004

(6) Tighten the nut [7] to 50.0 in-lb (5.6 N·m) - 100.0 in-lb (11.3 N·m) above the run-on torque. 80 in lb

(a) Loosen to the nearest castellation, if it is necessary, and install the new pin [8].

CR-2

INDEPENDENT INSPECTION 22.3.25 05:05 GAT 412

SUBTASK 32-32-31-600-002

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(7) Apply a thin layer of G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (optional), to the parts that follow:

(a) The threads and thread reliefs of the uplock hanger pins [10] and nuts [12].

(b) The faces of the uplock washers [11].

(c) The pins [13].

SUBTASK 32-32-31-640-002

(8) Lubricate the shank of the uplock hanger pin [10] with D00633 grease .

SUBTASK 32-32-31-420-005

(9) Install uplock hanger pin [10], uplock washers [11], and nuts [12] to connect the hanger assembly [19] to the uplock assembly [9].

SUBTASK 32-32-31-420-006

(10) Tighten the nuts [12] to 20.0 in-lb (2.3 N·m) - 24.0 in-lb (2.7 N·m) above the run-on torque. (24 in lb)

(a) Loosen to the nearest castellation, if it is necessary, and install the new pins [13].

CR-2

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SUBTASK 32-32-31-820-001

CR-2

(11) Loosen the jamnut [20] on the rod end of the hanger assembly [19].

(a) Adjust the rod end until the distance from the center of the bolt [1] to the top of the threaded end of the hanger assembly [19] is 2.17 in. (55.1 mm). 2.17 IN

SUBTASK 32-32-31-600-003

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WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(12) Apply a thin layer of G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (optional), on the parts that follow:

(a) The threads and thread reliefs of the bolt [1] and nut [4].

(b) The faces of the washer [3].

(c) The pin [5].

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SUBTASK 32-32-31-640-003

(13) Lubricate the shanks of the bolt [1] with a light film of D00633 grease .
(a) Wipe off excess grease.

SUBTASK 32-32-31-420-007

(14) Install the bolt [1], bushing [2], washer [3], nut [4], and pin [5] to attach the hanger assembly [19] to the structure.

NOTE: The installation of the hanger assembly rod end does not require the cotter pin to engage to the self locking nut castellations.

SUBTASK 32-32-31-420-013

(15) Check and adjust if needed, the 2.17 in. (55.12 mm) dimension from the center of bolt [1], to the top of the threaded end of the hanger under the jamnut [20].

SUBTASK 32-32-31-420-008

(16) Tighten the jamnut [20] on the rod end of the hanger assembly [19] to 200 in-lb (23 N·m) - 250 in-lb (28 N·m). (220 in lb)

SUBTASK 32-32-31-420-009

(17) Install G01048 MS20995C32 lockwire , on the jamnut [20].

SUBTASK 32-32-31-420-030

(18) Connect the control rod of the manual extension mechanism to the uplock assembly [9] (TASK 32-34-21-400-801).

SUBTASK 32-32-31-420-012

(19) Do these steps to connect the two hydraulic lines to the uplock actuator:
(a) Remove the plugs from the two hydraulic lines.
(b) Connect the two hydraulic lines to the uplock actuator.
(c) Tighten the B-nuts on the hydraulic lines to the value of 140 ±7 in-lb (16 ±1 N·m).
(d) Remove the tags from the two hydraulic lines.

SUBTASK 32-32-31-420-010

(20) Install uplock sensors on the uplock assembly [9], do this task: Main Landing Gear Uplock Sensor Installation, TASK 32-61-21-400-801.

SUBTASK 32-32-31-420-011

(21) Do the adjustment for the uplock sensors on the uplock assembly [9], do this task: Main Landing Gear Uplock Sensor Clearance Adjustment, TASK 32-61-21-820-801.

G. Main Landing Gear Uplock Spring and Spool Replacement with Spring Removal Tool

SUBTASK 32-32-31-020-015

(1) Remove the uplock spring [32] and spool [37].

NOTE: This procedure uses two spring removal tools. If only one tool is used, it is necessary to remove the spring on the nut side of the pin first before removing the spring on the head side of the pin.

(a) Install the spring removal tools from spring removal and installation tool, SPL-12460, on the spools [37].
(b) Extend the spools [37] and uplock springs [32] a small amount to remove spring pressure from the pins [41].
(c) Remove the pin [35], nut [33], and washer [36] from the pin [41].

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(d) Remove the pin [41] to disconnect the uplock springs [32] and spools [37] from the uplock assembly.
1) Make sure that you catch the spacers [39], spacers [40], washers [36], washers [38], and sleeves [34] that come loose when the pins [41] are removed.
(e) Remove the tension from the uplock springs [32] and remove the spools [37] and uplock springs [32] from the spring removal tools.
(f) Remove each spool [37] from uplock spring [32].

SUBTASK 32-32-31-420-017

(2) Install the uplock spring [32] and spool [37].

NOTE: This procedure uses two spring removal tools. If only one tool is used, it is necessary to install the spring on the head side of the pin first before installing the spring on the nut side of the pin.

(a) Install one end of the uplock spring [32] over one spool [37] and install the other end of the uplock spring [32] over the other spool [37].
(b) Put the uplock springs [32] and spools [37] in the spring removal tools from spring removal and installation tool, SPL-12460.
(c) Lubricate the chrome surfaces of the pins [41] with D00013 grease , or D00633 grease .
(d) Put one washer [36] and one sleeve [34] on each pin [41].
(e) Put the pins [41] through the spools [37] on the uplock springs [32].
(f) Install one of the washer [38] and one spacer [39] or spacer [40] on each pin [41].
(g) Use a spring removal tools to align the spools [37] in the ends of the uplock springs [32] with the holes in the uplock hook assembly [42] and link assembly [43].
(h) Push the pins [41] through the uplock hook assembly [42] and link assembly [43] such that you can install the second spacer [39] or spacer [40] and washers [38].
(i) Push the pins [41] completely through the spools [37] on the other spring.
(j) Install the sleeves [34], washers [36], and nuts [33] on the pins [41].
(k) Remove the spring removal tools.
(l) Tighten the nuts [33] to 100 in-lb (11 N·m). 100 in lb
(m) Tighten the nuts [33] to the nearest castellation and install the pins [35].

SUBTASK 32-32-31-420-022

(3) If only the uplock spring [32] and/or spool [37] is replaced, it is not necessary to perform a landing gear retraction test if the spring and/or spool [37] are replaced per this procedure.

H. Main Landing Gear Uplock Spring and Spool Replacement without Spring Removal Tool

SUBTASK 32-32-31-020-016

(1) Remove the broken or fractured uplock spring.

SUBTASK 32-32-31-020-019

(2) Remove spring(s) if they are not broken.
(a) Do not use metallic objects which could nick or damage the spring surface.
(b) Wrap a STD-4873 0.5 inch (12.7 mm) diameter rope, or G50316 cotton cloth , around one end of the uplock spring [32].
(c) Pull the spring end over the spool assembly.

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NOTE: Approximately 30 lbf (133 N) of force per 1.0 in. (25.4 mm) of spring extension is necessary.

(d) Slowly release tension after spring end is out of the spool.

(e) Remove spring from other end of spool.

SUBTASK 32-32-31-020-017

(3) Install the uplock spring [32].

(a) Do not use metallic objects which could nick or damage the spring surface.

(b) With one end of the uplock spring [32] on one spools [37], put a STD-4873 0.5 inch (12.7 mm) diameter rope, or G50316 cotton cloth , around the other end of the uplock spring [32].

(c) Pull the free spring end over the other spool assembly.

NOTE: Approximately 30 lbf (133 N) of force per 1.0 in. (25.4 mm) of spring extension is necessary.

(d) Remove the STD-4873 0.5 inch (12.7 mm) diameter rope, or G50316 cotton cloth , from the end of the uplock spring [32].

(e) It is not necessary to perform a landing gear retraction test if the uplock spring [32] is replaced per the above procedure.

SUBTASK 32-32-31-020-018

(4) Remove the spools [37].

(a) Remove the pin [35], nut [33], and washers [36] from the pin [41].

(b) Remove the pins [41].

1) Make sure that you catch the spacers [39], spacers [40], washers [36], washers [38], sleeves [34], and spools [37] that come loose when the pins [41] are removed.

SUBTASK 32-32-31-420-023

(5) Replace the uplock springs [32] and spools [37]. REASSEMBLED IN ASSEMBLED CONDITION

(a) Lubricate the chrome surfaces of the pin [41] with D00013 grease , or D00633 grease .

(b) Put one of the washers [36], one sleeve [34], and one spool [37] on each pin [41].

(c) Install one of the washer [38] and one spacer [39] or spacer [40] on each pin [41].

(d) Push the pins [41] completely through the uplock hook assembly [42] and link assembly [43] such that you can install the second spacer [39] or spacer [40] and washer [38].

(e) Install the spools [37], sleeves [34], washers [36], and nuts [33] on the pins [41].

(f) Tighten the nuts [33] to 100 in-lb (11 N·m). ~~100 in-lb~~. WS 22.3.25

(g) Tighten the nuts [33] to the nearest castellation and install the pins [35].

CAUTION: DO NOT USE METAL TOOLS. METAL TOOLS CAN CAUSE DAMAGE TO THE PART YOU WILL DO WORK ON OR ADJACENT PARTS.

(h) With one end of the uplock spring [32] on one spool [37], put a STD-4873 0.5 inch (12.7 mm) diameter rope, or G50316 cotton cloth , around the other end of the uplock spring [32].

(i) Pull the free spring end over the other spool [37] (approximately) 30 lbf (133 N) per 1.0 in. (25.4 mm) of spring extension is required).

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(j) Remove the STD-4873 0.5 inch (12.7 mm) diameter rope, or G50316 cotton cloth , from the end of the uplock spring [32].

SUBTASK 32-32-31-420-029

(6) If only the uplock spring [32] or spool [37] is replaced, it is not necessary to perform a landing gear retraction test if the spring is replaced per this procedure.

I. Installation Test for the Main Gear Uplock Assembly

SUBTASK 32-32-31-580-002

WARNING: KEEP PERSONNEL AND EQUIPMENT AWAY FROM THE APPLICABLE LANDING GEAR. THE LANDING GEAR EXTENDS AND RETRACTS QUICKLY. THIS CAN CAUSE INJURIES TO PERSONNEL, AND DAMAGE TO EQUIPMENT.

(1) Do this task: Lift the Airplane with the Jacks, TASK 07-11-01-580-815.

SUBTASK 32-32-31-020-014

(2) Disconnect the main gear door rod actuator [29] from the main gear doors [28], by removing the pin [27], nut [26], washer [25], washer [23], spacer [24], and bolt [22].

NOTE: The main gear doors will remain hanging during the installation test.

SUBTASK 32-32-31-080-001

(3) Remove the downlock pin only from the main gear you are working on and leave the pin in the other main gear and the nose gear (TASK 32-00-01-080-801).

SUBTASK 32-32-31-860-002

(4) Pressurize the hydraulic system A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-32-31-870-001

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED IN ALL OF THE LANDING GEAR BEFORE YOU MOVE THE CONTROL LEVER. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONNEL AND DAMAGE TO EQUIPMENT.

(5) Extend and retract the main landing gear you are working on several times.

NOTE: It will bleed the air from the uplock actuator and the lines. Examine for the correct gear operation.

SUBTASK 32-32-31-040-001

(6) Retract the main landing gear you are working on:

NOTE: The main gear doors will remain hanging during the installation test.

(a) Move the landing gear lever to the OFF position.

NOTE: This will depressurize the landing gear system while personnel are in the wheel well.

(b) Install a STD-858 DO NOT OPERATE tag, on the landing gear handle.

SUBTASK 32-32-31-200-001

WARNING: MAKE SURE THAT THE LANDING GEAR LEVER IS IN THE OFF POSITION BEFORE YOU GO INTO THE WHEEL WELL AREA. ACCIDENTAL MOVEMENT OF THE LANDING GEAR COULD CAUSE INJURY TO PERSONNEL.

(7) Make sure that the gap is 0.375 +0.036 / -0.000 in. (9.52 +0.92 / -0.00 mm) (Figure 402).

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(a) Use a 0.375 in. (9.525 mm) diameter pin, or equivalent, inserted between the uplock roller and the back of the uplock assembly hook.
(b) If adjustment is required, extend the landing gear.

WARNING: MAKE SURE THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

1) Make sure that the downlock pins are installed in the main landing gear (TASK 32-00-01-480-801).
2) Adjust the length of the hanger assembly [19], plus or minus one half turn, until the gap requirement is met.
   a) Loosen the jamnut [20].
   b) Adjust the hanger assembly [19] length one half turn longer or shorter.
   c) Tighten the jamnut [20] to 250 in-lb (28 N·m) - 200 in-lb (23 N·m).
   d) Remove the downlock pin from the gear being worked on only and repeat this step until the gap requirement is met.
   e) Install G01048 MS20995C32 lockwire , between the jamnut [20] and hanger assembly [19].

SUBTASK 32-32-31-800-001

(8) Extend the landing gear:
   (a) Remove the STD-858 DO NOT OPERATE tag, from the landing gear handle.
   (b) Move the landing gear lever to the DOWN position.
   (c) Install the downlock pins in the main landing gear (TASK 32-00-01-480-801).

SUBTASK 32-32-31-420-014

(9) Reconnect the main gear door rod actuator [29] to the main gear doors [28], by installing the spacer [24], washer [23], washer [25], bolt [22], nut [26], and pin [27].

SUBTASK 32-32-31-420-015

(10) Make sure that the main gear door is adjusted (TASK 32-13-21-000-801).

SUBTASK 32-32-31-210-001

(11) Examine the hydraulic connections for leaks.

SUBTASK 32-32-31-860-003

(12) Remove power to hydraulic system A (TASK 29-11-00-860-805).

SUBTASK 32-32-31-580-003

(13) Do this task: Lower the Airplane Off the Jacks, TASK 07-11-01-580-816.

SUBTASK 32-32-31-840-001

(14) Examine the hydraulic reservoirs for the correct servicing (TASK 12-12-00-610-801).
   (a) Do a servicing if it is necessary.

INDEPENDENT INSPECTION

INDEPENDENT INSPECTION

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

	MECH	INSP

JXB ALL

TASK 57-16-01-000-801

9. MLG Forward Trunnion Housing Assembly - Removal (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
32-11-83-000-801	Main Landing Gear Forward Trunnion Bearing Assembly Removal (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-4360	Wrench - Trunnion Spherical Bearing, MLG **737-800** (Part #: C32023-12, Supplier: 81205) (Opt Part #: C32023-1, Supplier: 81205)

JXB ALL

C. Location Zones

Zone	Area
551	Left Wing - Rear Spar To Landing Gear Support Beam
651	Right Wing - Rear Spar to Landing Gear Support Beam

D. MLG Forward Trunnion Housing Assembly Removal

SUBTASK 57-16-01-010-001

(1) Remove the housing assembly from the airplane, do this task: Main Landing Gear Forward Trunnion Bearing Assembly Removal, TASK 32-11-83-000-801.

JXB 004, 005, 007-009, 014 PRE SB 737-32-1448

SUBTASK 57-16-01-020-001

(2) Do the steps that follow to disassemble the housing assembly [5] (Figure 401):

(a) Remove the lockwire [2] from the retainer nut [1].

(b) Remove the retainer nut [1], pin assembly [4], and race assembly [3].

1) Use wrench, SPL-4360, or equivalent, to remove the retainer nut [1].

(c) Remove the retaining ring [8], support ring assembly [7], and seal [6].

JXB 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448

SUBTASK 57-16-01-020-005

(3) Do the steps that follow to disassemble the housing assembly [5] (Figure 401):

(a) Remove the lockwire [2] from the retainer nut [1].

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

(b) Remove the retainer nut [1], pin assembly [4], and race assembly [3].
1) Use wrench, SPL-4360, or equivalent, to remove the retainer nut [1].
(c) Remove the retaining ring [8] and seal [6].

Figure 401. Forward Trunnion Housing Assembly Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB 004, 005, 007-009, 014 PRE SB 737-32-1448

Sheet 3 - Effectivity: JXB 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448

Sheet 4 - Effectivity: JXB ALL

Sheet 5 - Effectivity: JXB 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448

MECH	INSP

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH | INSP

JXB ALL

TASK 57-16-01-400-801

10. **MLG Forward Trunnion Housing Assembly - Installation** (Figure 401) (Figure 401 (Sheet 4))

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
32-11-83-400-801	Main Landing Gear Forward Trunnion Bearing Assembly Installation (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-4360	Wrench - Trunnion Spherical Bearing, MLG **737-800** (Part #: C32023-12, Supplier: 81205) (Opt Part #: C32023-1, Supplier: 81205)

JXB ALL

C. Consumable Materials

Reference	Description	Specification
D00633	Grease - Aircraft General Purpose	BMS3-33
G01912	Lockwire - MS20995NC32, Monel - 0.032 Inch (0.8128 mm) Diameter	NASM20995

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
5	Housing assembly	32-11-00-04-095	JXB ALL
8	Retaining ring	32-11-00-02-045	JXB 004, 005, 007-009, 014
		32-11-00-02-048	JXB 026-031, 033, 036, 037, 040-051, 053-999

E. Location Zones

Zone	Area
551	Left Wing - Rear Spar To Landing Gear Support Beam

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

651 Right Wing - Rear Spar to Landing Gear Support Beam

F. MLG Forward Trunnion Housing Assembly Installation

SUBTASK 57-16-01-020-002

(1) Install the pin assembly [4] and race assembly [3] into the housing assembly [5].

(a) Apply a coating of D00633 grease , to these parts before assembly.

(b) Make sure that the splines of the race assembly [3] are in full contact with the housing assembly [5] splines before you install the retainer nut [1].

SUBTASK 57-16-01-640-001

(2) Install the retainer nut [1].

(a) Apply D00633 grease , to the retainer nut threads.

(b) Tighten the retainer nut [1] to 50 ft-lb (68 N·m) - 75 ft-lb (102 N·m).

1) Use the wrench, SPL-4360, or equivalent, to tighten the retainer nut [1].

CAUTION: MAKE SURE THAT THE CLEARANCE BETWEEN THE RACE ASSEMBLY AND THE HOUSING ASSEMBLY IS LESS THAN 0.0050 INCH (0.1270 MM). IF THE CLEARANCE IS MORE THAN 0.0050 INCH (0.1270 MM) DURING THE INSTALLATION OF THE FORWARD TRUNNION BEARINGS, DAMAGE TO THE AIRPLANE CAN OCCUR.

(c) Make sure that the clearance between the aft face of the race assembly [3] and the housing assembly [5] is 0.0050 in. (0.1270 mm) or less (Figure 401 (Sheet 4)).

NOTE: It is not required to continue with the installation until a correct clearance is measured after tightening the nut retainer.

1) If a clearance greater than 0.0050 in. (0.1270 mm) is found, disassemble the retainer nut [1], pin assembly [4], and race assembly [3] to look for unwanted material between the mated parts.

SUBTASK 57-16-01-020-003

(3) Install the lockwire [2] (G01912 MS20995NC32 lockwire), at 2 locations, onto the retainer nut [1], and housing assembly [5].

(a) Install G01912 MS20995NC32 lockwire , with the double twist method.

JXB 004, 005, 007-009, 014 PRE SB 737-32-1448

SUBTASK 57-16-01-020-004

(4) Install the seal [6], support ring assembly [7], and the retaining ring [8].

NOTE: You can use the removed seal, support ring assembly, and retaining ring if in a serviceable condition.

(a) Apply a coating of D00633 grease , to these parts.

JXB 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448

SUBTASK 57-16-01-420-001

CAUTION: DO NOT USE THE EXISTING PARTS (P/N S38624 SEAL AND P/N 161A1194-1 RETAINER) TO REPLACE THE NEW OR CHANGED PARTS (P/N 5649-0615-369 SEAL AND RETAINER P/N 161A1194-3) ON FWD MLG TRUNNION HOUSING ASSY. IF YOU DO NOT OBEY THESE INSTRUCTIONS, DAMAGE TO THE MAIN LANDING GEAR CAN OCCUR.

INDEPENDENT INSPECTION

INDEPENDENT INSPECTION

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

(5) Install the seal [6] and the retaining ring [8] (Figure 401).

NOTE: You can use the removed seal and retaining ring if in a serviceable condition.

(a) Apply a coating of D00633 grease , to these parts.

(b) Make sure that you install the seal [6] in its correct position.

JXB ALL

SUBTASK 57-16-01-410-001

(6) Put D00633 grease , into the grease fitting on the housing assembly [5] until the grease can be seen at the aft edge of the bearing race.

SUBTASK 57-16-01-410-002

(7) Do this task: Main Landing Gear Forward Trunnion Bearing Assembly Installation, TASK 32-11-83-400-801.

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	GAT 492

PARTIAL SIGN OFF STATUS:

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JXB ALL

TASK 57-16-02-000-801

11. **MLG Aft Trunnion Bearing Assembly Removal** (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-11-00-000-801	Main Landing Gear Removal (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-11076	Torque Wrench Adaptor **737-800** (Part #: C32013-15, Supplier: 81205) (Opt Part #: C32013-1, Supplier: 81205)

JXB ALL

C. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
734	Left Main Landing Gear
744	Right Main Landing Gear

D. Prepare for the Removal

SUBTASK 57-16-02-010-005

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED IN ALL OF THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed in the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 57-16-02-010-002

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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(2) To get access to the aft trunnion bearing assembly, do these steps:
(a) Do this task: Lift the Airplane with the Jacks, TASK 07-11-01-580-815.
(b) Do this task: Main Landing Gear Removal, TASK 32-11-00-000-801.
1) Remove the aft trunnion pin [6] from the aft trunnion bearing assembly.

NOTE: This pin was attached to the aft trunnion with tie wraps within the procedure done before.

E. MLG Aft Trunnion Bearing Assembly Removal

SUBTASK 57-16-02-020-001

(1) Remove the lock tab [4] from the lock plate.
(a) Remove the screws [2], washers [3], and lockwire.

SUBTASK 57-16-02-020-002

(2) Use torque wrench adaptor, SPL-11076 , to remove the retainer nut [1].

SUBTASK 57-16-02-020-003

(3) Remove the split ball assembly [7] and outer race assembly [5].

Figure 401. MLG Aft Trunnion Bearing Assembly Installation

Sheet 1 - Effectivity: JXB ALL

Sheet 2 - Effectivity: JXB ALL

Sheet 3 - Effectivity: JXB ALL

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

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JXB ALL
TASK 57-16-02-400-801

12. **MLG Aft Trunnion Bearing Assembly Installation** (Figure 401)
NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-11-00-400-801	Main Landing Gear Installation (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-11076	Torque Wrench Adaptor **737-800** (Part #: C32013-15, Supplier: 81205) (Opt Part #: C32013-1, Supplier: 81205)

JXB ALL

C. Consumable Materials

Reference	Description	Specification
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827 (NATO G-354) (Supersedes MIL-G-23827)
D00633	Grease - Aircraft General Purpose	BMS3-33

D. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
734	Left Main Landing Gear

PARTIAL SIGN OFF STATUS:

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744 Right Main Landing Gear

E. MLG Aft Trunnion Bearing Assembly Installation

SUBTASK 57-16-02-020-004

(1) Install the split ball assembly [7] and outer race assembly [5].

NOTE: Installation of the race assembly grease fitting should be on the upper inboard side of the trunnion pin when looking forward, approximately 2 o'clock orientation for the left hand side and 11 o'clock orientation for the right hand side.

(a) Apply a thin coat of D00633 grease , or D00013 grease , to all parts before you install them.

NOTE: D00633 grease , is the preferred grease because it offers protection against corrosion and oxidation.

(b) Make sure that the wider splines of the outer race assembly [5] and lock plate are aligned (View C, Figure 401).

SUBTASK 57-16-02-020-005

(2) Install the retainer nut [1].

(a) Tighten the retainer nut [1] to 10 ft-lb (14 N·m) - 15 ft-lb (20 N·m). 15 ft-lbs

1) Use the torque wrench adaptor, SPL-11076, to tighten the retainer nut [1].

2) After the initial torque, allow excess grease on the outer race assembly [5] and split ball assembly [7] to extrude from the bearing housing.

a) Recheck the torque to ensure it is still within the torque limits.

CAUTION: MAKE SURE THAT THE END OF THE OUTER RACE ASSEMBLY IS FLUSH WITH THE TOP OF THE RETAINER NUT WHILE YOU TIGHTEN THE RETAINER NUT. IF YOU DO NOT OBEY, DAMAGE CAN OCCUR TO THE INCORRECT POSITION OF THE GREASE FITTING TO THE LANDING GEAR.

(b) Make sure that the end of the outer race assembly [5] is flush with the top of the retainer nut [1] while you tighten the retainer nut [1] (View C, Figure 401).

SUBTASK 57-16-02-020-006

(3) Install the lock tab [4] to the lock plate.

(a) Back the retainer nut [1] off until tab can engage slot in nut.

1) To do this, these steps can help:

a) The lock tab [4] can be flipped.

b) The lock tab [4] can be moved to adjacent threaded holes in the lock plate.

(b) Install the screws [2], washers [3], and lockwire.

1) Tighten the screws [2] to 50 in-lb (6 N·m) - 80 in-lb (9 N·m). 80 in-lbs

2) Install the lockwire by the double-twist procedure.

SUBTASK 57-16-02-020-007

(4) Lubricate the aft trunnion bearing assembly at the lubrication fitting using D00633 grease .

(a) Move the spherical bearing until positioned as shown in View A-A, Figure 401.

NOTE: Grease shall appear at inner diameter of bearing.

SUBTASK 57-16-02-020-008

(5) Install the aft trunnion pin [6] into the aft trunnion bearing assembly.

F. Put the Airplane Back to Its Usual Condition

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INDEPENDENT INSPECTION 21-3-25 08:45 GAT 412

INDEPENDENT INSPECTION 21-3-25 08:53 GAT 412

INDEPENDENT INSPECTION 21-3-25 09:05 GAT 412

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SUBTASK 57-16-02-010-003

(1) Do this task: Main Landing Gear Installation, TASK 32-11-00-400-801.

SUBTASK 57-16-02-010-006

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED IN ALL OF THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(2) If the downlock pins are not installed in the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

PARTIAL SIGN OFF STATUS:

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TASK 32-11-83-400-801

13. **Main Landing Gear Forward Trunnion Bearing Assembly Installation** (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
12-21-11-640-801	Main Landing Gear Upper End Components Servicing (P/B 301)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-11-00-400-801	Main Landing Gear Installation (P/B 401)

B. Consumable Materials

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS5-95
A50009	Sealant - Low Density, Non-Chromate Type. (Formerly Chromate - Synthetic Rubber)	BMS5-142 Type II Class B-1 or B-2
C00259	Coating - Chemical And Solvent Resistant Finish, Corrosion Inhibiting Primer	BMS10-11 Type I
D00006	Compound - Antiseize Pure Nickel Special - Never-Seez NSBT	
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38

C. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
7	Special washer	32-11-00-04-040	JXB ALL
8	Housing assembly	32-11-00-04-095	JXB ALL

D. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
551	Left Wing - Rear Spar To Landing Gear Support Beam

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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651	Right Wing - Rear Spar to Landing Gear Support Beam		
734	Left Main Landing Gear		
744	Right Main Landing Gear		

E. Prepare for the Installation

SUBTASK 32-11-83-480-002

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT. THIS CAN CAUSE INJURIES TO PERSONS, AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed on all the landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

F. Main Landing Gear Forward Trunnion Bearing Assembly Installation

SUBTASK 32-11-83-420-001

(1) Install the forward trunnion bearing assembly [1]:

(a) Apply a thin layer of G50136 corrosion inhibiting material , to the fuse pins [6] as follows:

1) The outer diameters.
2) The threads.
3) The head mating surfaces.

(b) Apply two layers of C00259 primer , to the special washers [7].

(c) Put the housing assembly [8] in its position for the connection to the forward trunnion support.

NOTE: The zerk fitting on the housing assembly [8] will point up and aft.

(d) Align the holes in the housing assembly [8] to the holes in the forward trunnion support.

(e) Install the special washers [7] as follows:

1) Clean any G50136 corrosion inhibiting material , from surfaces where A00247 sealant , or A50009 sealant , is applied.

NOTE: Cross contamination is not permitted.

CAUTION: DO NOT APPLY SEALANT TO WASHER SURFACES THAT TOUCH WITH THE PIN HEADS OR NUTS. THIS CAN CAUSE DAMAGE TO THE EQUIPMENT.

2) Apply sealant as follows:

a) If plain special washers [7] are installed, apply A00247 sealant , or A50009 sealant , to one side of the special washer [7] that will be mating to the forward trunnion support.

N/A

b) If countersunk special washers [7] are installed, apply A00247 sealant , or A50009 sealant , to the non-countersunk side.

<1> For each countersunk special washer [7], turn the non-countersunk side against the forward trunnion support.

3) Install the special washers [7] on the forward trunnion support.

4) Visually align the special washers [7] with the forward trunnion support bore.

NOTE: Washers are to stay aligned with the fuse pins during installation of the fuse pins.

PARTIAL SIGN OFF STATUS:

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(f) Apply a thin layer of G50136 corrosion inhibiting material , to the mating surfaces of the special washers [7] to the fuse pin bolt heads or nuts [9].
(g) Put the fuse pins [6] through the special washers [7], forward trunnion support, and housing assembly [8].
(h) Use a clean cloth to clean unwanted G50136 corrosion inhibiting material , from the threads of the fuse pins [6].
(i) Apply D00006 Never-Seez NSBT compound , to the inner surfaces of nuts [9] and to the threads of the fuse pins [6].
(j) Install the nuts [9] on the fuse pins [6].
(k) Check that the run-up torque on the nuts [9] is 117 in-lb (13.2 N·m) or greater.
1) If the run-up torque is less than 117 in-lb (13.2 N·m), replace the nuts [9].
(l) Tighten the nuts [9] to 1300 in-lb (146.9 N·m) - 1350 in-lb (152.5 N·m). 1350 in lbs
1) Make sure that the special washers [7] stay aligned with the support bore when you tighten each nut [9].
2) Do a visual inspection that the special washers [7] are centered.
3) Re-tighten the nuts [9] to 1300 in-lb (146.9 N·m) - 1350 in-lb (152.5 N·m). 1350 in lbs
NOTE: Re-tighten the nuts before the squeeze-out life of the sealant.
NOTE: Wait a minimum 10 minutes for sealant with a squeeze out life less than or equal to 48 hours.
Wait a minimum 60 minutes for sealant with a squeeze out life greater than 48 hours.
(m) Use a clean cloth to clean unwanted G50136 corrosion inhibiting material , at the outer edges of the special washers [7].
(n) Clean any G50136 corrosion inhibiting material , from surfaces where A00247 sealant , or A50009 sealant , is applied.
NOTE: Cross contamination is not permitted.
(o) Apply a fillet seal to the edge of the special washers [7] with A00247 sealant , or A50009 sealant .
(p) Put the rods [4] into the fuse pins [6].
(q) Install the fuse pin caps [3], washers [5], and nuts [2] on the ends of the rods [4].
(r) Tighten the nuts [2] to 50 in-lb (5.6 N·m) - 80 in-lb (9.0 N·m). (80 in lbs)

SUBTASK 32-11-83-420-003
(2) Install the clamp [10], screws [11], and washers [12] on the hydraulic flow regulator.

SUBTASK 32-11-83-420-004
(3) Install the clamp block assemblies [17].
(a) Install the clamp blocks [22] and channels [21] to the tubing.
(b) Install the spacers [20], washers [19] and screws [18] to the clamp block assemblies [17] and nutplates [23] to secure to the structure.

SUBTASK 32-11-83-420-005
(4) Install the fairlead [13] and fairlead [24] to the brackets..
(a) Install the fairlead [13] to the bracket with bolts [14], washers [15] and nuts [16].
(b) Tighten the bolts [14] 20 in-lb (2.3 N·m) - 25 in-lb (2.8 N·m).
(c) Install the fairlead [24] to the bracket with the bolts [25], washers [26] and nuts [27].
(d) Tighten the bolts [25] 20 in-lb (2.3 N·m) - 25 in-lb (2.8 N·m).

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INDEPENDENT INSPECTION 21.3.25 08:05 GAT 412
INDEPENDENT INSPECTION 21.3.25 08:20 GAT 412
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Type: Routine Card **ATA:** 32-- **Flow:** - **Work Area:** -

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(e) Visually inspect the cables at the fairlead [13] and fairlead [24] upon reinstallation to the brackets for any damage. SUBTASK 32-11-83-420-002 (5) Do this task: Main Landing Gear Installation, TASK 32-11-00-400-801. G. Put the Airplane Back to Its Usual Condition SUBTASK 32-11-83-610-001 (1) Lubricate the housing assembly [8], do this task: Main Landing Gear Upper End Components Servicing, TASK 12-21-11-640-801.	M12eS16 21/3/25	21.3.25 GAT 492

PARTIAL SIGN OFF STATUS:

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TASK 32-32-11-400-801

14. **Installation Of The Actuator Assembly For The Main Gear** (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)
12-21-11 P/B 301	MAIN LANDING GEAR - SERVICING
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-080-801	Landing Gear Downlock Pins Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-32-00-710-801	Main Landing Gear Operational Test (P/B 501)
32-32-00-710-802	Main Landing Gear Test - Component Replacement (P/B 501)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1870	Hoist - Main Landing Gear Retract Actuator **737-800** (Part #: C32028-109, Supplier: 81205) (Part #: C32028-110, Supplier: 81205) (Part #: C32028-111, Supplier: 81205) (Opt Part #: C32028-1, Supplier: 81205) (Opt Part #: C32028-3, Supplier: 81205) (Opt Part #: C32028-56, Supplier: 81205) (Opt Part #: C32028-58, Supplier: 81205) (Opt Part #: C32028-59, Supplier: 81205)

JXB ALL

C. Consumable Materials

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

Type: Routine Card | **ATA:** 32-- | **Flow:** - | **Work Area:** -

MECH | INSP

Reference	Description	Specification
D00153	Fluid - Hydraulic Fluid, Fire Resistant (Interchangeable And Intermixable With BMS 3-11 Type V)	BMS3-11 Type IV
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
1	Beam assembly	32-32-11-01-085	JXB ALL
		32-32-93-01-095	JXB ALL
2	Actuator assembly	32-32-11-01-135	JXB ALL
5	Pin	32-32-93-01-010	JXB ALL
12	Pin	32-32-11-01-010	JXB ALL
		32-32-11-01-095	JXB ALL
		32-32-93-01-030	JXB ALL
24	Packing	32-32-11-01-170	JXB ALL
25	Packing	32-32-11-01-165	JXB ALL
26	Union	32-32-11-01-155	JXB ALL
		32-32-11-01-160	JXB ALL

E. Location Zones

Zone	Area
211	Flight Compartment - Left
212	Flight Compartment - Right
551	Left Wing - Rear Spar To Landing Gear Support Beam
651	Right Wing - Rear Spar to Landing Gear Support Beam
730	Subzone - Left Main Landing Gear and Landing Gear Doors
740	Subzone - Right Main Landing Gear and Landing Gear Doors

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PARTIAL SIGN OFF STATUS:

Item: ________ Completed through item: ________ Sign: ________

Item: ________ Completed through item: ________ Sign: ________

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F. Access Panels

Number	Name/Location
551BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel
551DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel
651BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel
651DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel

G. Prepare for the Installation

SUBTASK 32-32-11-100-001

(1) Clean the surfaces of these items that will attach the beam hanger [3] to the structure:
(a) attachment fitting on the beam hanger [3]
(b) bolt [4], washer [7], and nut [6].

SUBTASK 32-32-11-100-002

(2) Clean these items that will attach the actuator assembly [2] to the main gear trunnion:
(a) attachment fitting on the main gear trunnion
(b) actuator attach pin [15]
(c) spacers [14], washer [13], and nut [9]
(d) pin [12], nut [11], washer [10], and bolt [8].

SUBTASK 32-32-11-100-003

(3) Clean the surfaces of these items that will attach the beam assembly [1] and actuator assembly [2] to the main gear trunnion:
(a) attachment fitting on the main gear trunnion
(b) walking beam attach pin [16]
(c) washer [19] and nut [18]
(d) pin [12], nut [11], washer [10], and bolt [17].

SUBTASK 32-32-11-600-004

(4) Apply D00633 grease , to the chrome-plated surfaces of the bolt [4], pin [20], actuator attach pin [15], and the walking beam attach pin [16].

SUBTASK 32-32-11-600-001

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(5) Apply a thin layer of G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to the thread reliefs and threads of the bolt [4], nut [6], and to the face of the washer [7].

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

	MECH	INSP

SUBTASK 32-32-11-600-003

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(6) Apply a thin layer of G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these parts:
- (a) the faces of the washer [13] and washer [19]
- (b) the threads and thread reliefs on the actuator attach pin [15] and walking beam attach pin [16]
- (c) the threads and thread reliefs of the bolt [8], bolt [17], faces of the washers [10], threads and thread reliefs of the nuts [11], and pin [12].

SUBTASK 32-32-11-420-001

(7) Do these steps if the replacement actuator assembly [2] does not have the restrictor [23] and union [26] installed:
- (a) Lubricate these items with D00153 hydraulic fluid :
  - 1) packing [24]
  - 2) packing [25]
  - 3) restrictor [23]
  - 4) union [26].
- (b) Remove the plugs from the ports on the actuator assembly [2].
- (c) Install the packing [24] and packing [25] on the actuator assembly [2].
- (d) Tighten the restrictor [23] to 500 ±25 in-lb (56 ±3 N·m) on the actuator assembly [2].
- (e) Tighten the union [26] to 270 ±14 in-lb (31 ±2 N·m) on the actuator assembly [2].

N/A

H. Actuator Assembly Installation

SUBTASK 32-32-11-420-002

(1) Do these steps to connect the actuator assembly [2] to the beam assembly [1]:
- (a) Install the actuator assembly [2] between the beam hanger [3] and the beam assembly [1] with the hydraulic ports down.

CAUTION: MAKE SURE THAT THE HEAD OF THE PIN IS POINTED AFT. THE INCORRECT ORIENTATION OF THE PIN CAN CAUSE DAMAGE TO THE MAIN-LANDING-GEAR (MLG) SUPPORT BEAM.

- (b) Install the pin [20].
- (c) Install the washer [22] and nut [21]. 45 ft-lb.
- (d) Tighten the nut [21] to 35 ft-lb (47.45 N·m) - 50 ft-lb (67.79 N·m).
- (e) If it is necessary, loosen the nut [21] to align the nearest castellation.
- (f) Install the bolt [17], washer [10], and nut [11].
- (g) Tighten the nut [11] to 50 in-lb (5.65 N·m) - 75 in-lb (8.47 N·m). 70 in-lb
- (h) If it is necessary, loosen the nut [11] to align the nearest castellation.
- (i) Install the pin [12].

SUBTASK 32-32-11-480-004

(2) Connect the beam assembly [1] and actuator assembly [2] to the hoist, SPL-1870.

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INDEPENDENT INSPECTION GAT 412 22-3-25 03:05

INDEPENDENT INSPECTION GAT 412 22-3-25 03:15

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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737-600/700/800/900

Right Main Landing Gear Restoration

Type: Routine Card **ATA:**32-- **Flow:-** **Work Area:-**

	MECH	INSP

SUBTASK 32-32-11-420-003

(3) Put the beam hanger [3] in its position in the bracket on the structure.

SUBTASK 32-32-11-420-004

(4) Install these parts to connect the beam hanger [3] to the bracket on the wing structure:

(a) bolt [4], washer [7], and nut [6]

1) Tighten the nut [6] to 250 in-lb (28.25 N·m) - 300 in-lb (33.90 N·m). 280 in-lb

2) If it is necessary, loosen the nut [6] to align the nearest castellation.

(b) Install the pin [5].

SUBTASK 32-32-11-420-005

(5) Install these parts to connect the actuator assembly [2] to the main gear trunnion:

(a) actuator attach pin [15], spacers [14], washer [13], and nut [9]

1) Remove the clamps for the electrical conduit on the inboard side of the nut [9] and move the conduit to get clearance to tighten the nut [9].

2) Tighten the nut [9] to 50 ft-lb (67.79 N·m) - 75 ft-lb (101.69 N·m). 70 ft-lb

3) If it is necessary, loosen the nut [9] to align the nearest castellation.

(b) bolt [8], washer [10], and nut [11]

1) Tighten the nut [11] to 45 in-lb (5.08 N·m) - 55 in-lb (6.21 N·m).

2) If it is necessary, loosen the nut [11] to align the nearest castellation.

(c) pin [12].

SUBTASK 32-32-11-410-004

(6) Put the electrical conduit back in its position and install the clamps.

SUBTASK 32-32-11-420-006

(7) Hold the beam assembly [1], actuator assembly [2], and install these parts to connect the beam assembly [1] to the main gear trunnion:

(a) walking beam attach pin [16] with the pin head forward

(b) washer [19] and nut [18] (70 ft-lb)

1) Tighten the nut [18] to 50 ft-lb (67.79 N·m) - 75 ft-lb (101.69 N·m).

2) If it is necessary, loosen the nut [18] to align the nearest castellation.

(c) bolt [17], washer [10], and nut [11] 50 in-lb.

1) Tighten the nut [11] to 45 in-lb (5.08 N·m) - 55 in-lb (6.21 N·m).

2) If it is necessary, loosen the nut [11] to align the nearest castellation.

(d) pin [12].

SUBTASK 32-32-11-080-002

(8) Remove the hoist, SPL-1870.

SUBTASK 32-32-11-080-003

(9) Remove the plugs from hydraulic line [27] and hydraulic line [28].

SUBTASK 32-32-11-420-007

(10) Connect the hydraulic line [27] UP to the actuator assembly [2].

SUBTASK 32-32-11-420-008

(11) Tighten the B-nut on the hydraulic line [27] UP to a value of 475 in-lb (53.67 N·m) - 525 in-lb (59.32 N·m). 500 in-lb

SUBTASK 32-32-11-420-009

(12) Connect the hydraulic line [28] DN to the actuator assembly [2].

INDEPENDENT INSPECTION

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

GCAA APPROVAL No : UAE.145.1232

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SUBTASK 32-32-11-420-010

(13) Tighten the B-nut on the hydraulic line [28] DN to a value of 256 in-lb (28.92 N·m) - 284 in-lb (32.09 N·m). (280 in-lb)

I. Main Gear Actuator Assembly - Bleeding (Airplane on Jacks)

SUBTASK 32-32-11-580-004

WARNING: MAKE SURE ALL PERSONS AND EQUIPMENT ARE CLEAR OF THE FLIGHT CONTROLS SURFACES, THE THRUST REVERSERS, AND THE LANDING GEAR. THESE COMPONENTS CAN MOVE SUDDENLY WHEN YOU SUPPLY HYDRAULIC POWER. THIS CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) Lift the airplane for landing gear retraction, do this task: Lift the Airplane with the Jacks, TASK 07-11-01-580-815.

SUBTASK 32-32-11-640-001

(2) Lubricate the attach points (MAIN LANDING GEAR - SERVICING, PAGEBLOCK 12-21-11/301).

SUBTASK 32-32-11-080-005

(3) Remove the downlock pins from the main landing gear, do this task: Landing Gear Downlock Pins Removal, TASK 32-00-01-080-801.

SUBTASK 32-32-11-860-004

(4) For hydraulic system A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-32-11-700-002

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED IN THE NOSE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE NOSE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(5) Do a retraction test of the main landing gear, do this task: Main Landing Gear Operational Test, TASK 32-32-00-710-801 or Main Landing Gear Test - Component Replacement, TASK 32-32-00-710-802.

SUBTASK 32-32-11-480-008

(6) Extend the main landing gear and install the downlock pins, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-32-11-210-003

(7) Examine the hydraulic connections for leaks.

SUBTASK 32-32-11-210-004

(8) Make sure that there is no interference of the hoses and actuator assembly [2] during gear operation.

SUBTASK 32-32-11-860-005

(9) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-32-11-580-005

(10) Do this task: Lower the Airplane Off the Jacks, TASK 07-11-01-580-816.

J. Main Gear Actuator Assembly - Bleeding (Airplane on Ground Without Jacks) - Alternate Method

PARTIAL SIGN OFF STATUS:

Item: 14 Completed through item: H (13) Sign: 22.3.25

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

SUBTASK 32-32-11-210-005

(1) Make sure that the hydraulic lines are installed correctly and they are not twisted.

SUBTASK 32-32-11-860-006

WARNING: OBEY THE PROCEDURE FOR THE INSTALLATION OF THE DOWNLOCK PINS. IF YOU MOVE THE CONTROL LEVER FOR THE LANDING GEAR TO THE UP POSITION, THE LANDING GEAR CAN RETRACT. THIS CAN CAUSE INJURIES TO PERSONNEL, AND DAMAGE TO EQUIPMENT.

(2) If the downlock pins are not installed in the nose or main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-32-11-860-012

(3) For hydraulic system A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-32-11-860-008

(4) Push the override button, and move the control lever for the landing gear from the DN position to the UP position and to the DN position three times.

NOTE: Hold the lever in each position for three seconds before you move the control lever to the subsequent position.

SUBTASK 32-32-11-860-009

(5) Move the control lever to the DN position.

SUBTASK 32-32-11-860-010

(6) When you move the control lever from the DN position to the OFF position, make sure that the condition that follows occurs:
(a) The rod end of the actuator tries to extend.

SUBTASK 32-32-11-860-011

(7) Move the control lever to the DN position.

SUBTASK 32-32-11-200-001

(8) Make sure that there are no leaks at the hydraulic connections.

SUBTASK 32-32-11-860-007

(9) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-32-11-600-005

(10) Lubricate the attach points (MAIN LANDING GEAR - SERVICING, PAGEBLOCK 12-21-11/301).

K. Put the Airplane Back to Its Usual Condition

SUBTASK 32-32-11-410-003

(1) Close the applicable access panel:

Number	Name/Location
551BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel
551DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

		MECH	INSP
651BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel		
651DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel	MR0742	

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH | INSP

JXB ALL

TASK 32-11-71-420-801

15. Main Landing Gear Reaction Link Assembly Installation (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)
29-09-00-860-801	Hydraulic Reservoirs Pressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-32-00-710-801	Main Landing Gear Operational Test (P/B 501)
32-32-00-710-802	Main Landing Gear Test - Component Replacement (P/B 501)
32-32-51-400-801	Main Gear Downlock Actuator Installation (P/B 401)
32-32-91-400-801	Main Gear Downlock Spring Installation (P/B 401)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1359	Wrench Adapter, C32029-3 or C32029-93 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1360	Wrench Adapter, C32029-4 or C32029-94 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1368	Thread Protector, C32029-12 or C32029-101 (included in C32029 kit)

MR 0742

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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	737-800 (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-1369	Thread Protector , C32029-13 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-11946	Wrench Adapter, C32029-31 or C32029-99 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)	MR 0742	22-8-24 GAT 492

JXB ALL

C. Consumable Materials

Reference	Description	Specification
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827 (NATO G-354) (Supersedes MIL-G-23827)
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
11	Pin	32-32-51-01-003	JXB ALL
44	Pin	32-11-61-03-010	JXB ALL

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

E. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
713	Nose Landing Gear
734	Left Main Landing Gear
744	Right Main Landing Gear

F. Prepare for the Installation

SUBTASK 32-11-71-480-004

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED IN ALL OF THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed in the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

G. Main Landing Gear Reaction Link Assembly Installation

SUBTASK 32-11-71-420-001

(1) Do these steps to connect the reaction link assembly [1] to the trunnion [41]:

(a) Lubricate the chrome plated surfaces of the trunnion pin [40] and end cap [45] with D00633 grease .

WARNING: YOU MUST HAVE TWO PERSONNEL TO LIFT THIS COMPONENT. THE COMPONENT IS HEAVY. IF YOU DO NOT OBEY, INJURY TO PERSONNEL CAN OCCUR.

(b) Position the outboard end of the reaction link assembly [1] for attachment to the trunnion [41].

NOTE: The reaction link assembly weighs approximately 63.2 lb (28.7 kg).

(c) Put the trunnion pin [40] through the reaction link assembly [1] and trunnion [41].

(d) Install the end cap [45] on the trunnion pin [40].

(e) Put the crossbolt [39] through the trunnion pin [40] and end cap [45].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: MAKE SURE THAT YOU REMOVE UNWANTED CORROSION INHIBITING COMPOUND FROM SURFACES THAT YOU WILL LUBRICATE. IF YOU APPLY CORROSION INHIBITING COMPOUND

MECH	INSP

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT COULD OCCUR.

(f) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound , (preferred) or G50136 corrosion inhibiting material , (alternate) to these items:
1) Do not apply corrosion preventive compound to the chrome plated surfaces.
2) The threads and thread reliefs of the crossbolt [39].
3) The threads of the nut [43].
4) The faces of the washer [42].

(g) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

(h) Install the washer [42] and nut [43] on the crossbolt [39].

(i) If it is necessary, loosen the nut [43] to the nearest castellation that aligns with the hole for the cotter pin [44].

(j) Install the cotter pin [44] in the crossbolt [39].

SUBTASK 32-11-71-020-014

(2) If it is previously removed, do the following steps to install the hanger link assembly [35] into the supporting structure:

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: MAKE SURE THAT YOU REMOVE UNWANTED CORROSION INHIBITING COMPOUND FROM SURFACES THAT YOU WILL LUBRICATE. IF YOU APPLY CORROSION INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT COULD OCCUR.

(a) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred) or G50136 corrosion inhibiting material , (alternate) to these items:
1) Do not apply corrosion preventive compound to the chrome plated surfaces.
2) The threads and thread reliefs of the hanger link assembly pin [46] and bolt [50].
3) The threads of the nut [49] and nut [52].
4) The faces of the washer [48] and washer [51].
5) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

(b) Prior to installation of the hanger link assembly pin [46], lubricate chrome plated surfaces with D00633 grease .

(c) Install the hanger link assembly pin [46] that supports the hanger link assembly [35] to the supporting structure, through the hanger link assembly [35].

(d) Install the washer [48] and nut [49] on the hanger link assembly pin [46].

(e) Use thewrench adapter, C32029-31 or C32029-99, SPL-11946 , to hold the hanger link assembly pin [46] to tighten the nut [49].

(f) Tighten the nut [49] to 95 ft-lb (128.8 N·m) - 115 ft-lb (155.9 N·m) above run-on torque. 100 ft lb.

(g) If it is necessary, loosen the nut [49] to align the nearest castellation.

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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(h) Install the bolt [50] in the hanger link assembly pin [46].
(i) Install the cotter pin [53], nut [52], and washer [51] to the bolt [50].
(j) Tighten the nut [52] to 20 in-lb (2.3 N·m) - 24 in-lb (2.7 N·m) above run-on torque. 24 in-lb

SUBTASK 32-11-71-420-002

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(3) Do these steps to connect the reaction link assembly [1] to the hanger link assembly [35]:
(a) Lubricate the chrome plated surfaces of the hanger link pin [34] and end cap [27] with D00633 grease .

WARNING: YOU MUST HAVE TWO PERSONNEL TO LIFT THIS COMPONENT. THE COMPONENT IS HEAVY. IF YOU DO NOT OBEY, INJURY TO PERSONNEL CAN OCCUR.

(b) Put the inboard end of the reaction link assembly [1] in its position for attachment to the hanger link assembly [35].
(c) Put the hanger link pin [34] through the reaction link assembly [1] and hanger link assembly [35].
(d) Attach the assembly of the end cap [27], bracket assembly [38], hydraulic tube [2], electrical harnesses [12], spacers [32], and spacer [36] on each end of the hanger link pin [34].
(e) If it is necessary to connect the electrical harnesses [12], do this step:
1) Install the bolts for the bracket that holds the sensor to the upper lock link [5] to connect the electrical harnesses [12].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: MAKE SURE THAT YOU REMOVE UNWANTED CORROSION INHIBITING COMPOUND FROM SURFACES THAT YOU WILL LUBRICATE. IF YOU APPLY CORROSION INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT COULD OCCUR.

(f) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound , (preferred) or G50136 corrosion inhibiting material , (alternate) to the faces of the washer [26].
(g) Put the washer [26] on the crossbolt [25].
(h) Put the crossbolt [25] through the end caps [27].

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: MAKE SURE THAT YOU REMOVE UNWANTED CORROSION INHIBITING COMPOUND FROM SURFACES THAT YOU WILL

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LUBRICATE. IF YOU APPLY CORROSION INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT COULD OCCUR.

(i) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound , (preferred) or G50136 corrosion inhibiting material , (alternate) to these items:

1) Do not apply corrosion preventive compound to the chrome plated surfaces.
2) The threads and thread reliefs of the crossbolt [25].
3) The threads of the nut [29].
4) The faces of the washer [30].

(j) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

(k) Install the washer [30] and nut [29] on the crossbolt [25].

(l) Tighten the nut [29] to 30 in-lb (3.4 N·m) - 50 in-lb (5.6 N·m). (50 in lb)

(m) If it is necessary, loosen the nut [29] to the nearest castellation that aligns with the hole for the cotter pin [28].

(n) Install the cotter pin [28] in the crossbolt [25].

(o) Put the pin assembly [31] through the bracket assembly [38] and reaction link assembly [1].

SUBTASK 32-11-71-420-003

(4) Do these steps to connect the reaction link assembly [1] to the upper side strut [21]:

(a) Lubricate the chrome plated surfaces of the side strut pin [20] with D00633 grease .

WARNING: YOU MUST HAVE TWO PERSONNEL TO LIFT THIS COMPONENT. THE COMPONENT IS HEAVY. IF YOU DO NOT OBEY, INJURY TO PERSONNEL CAN OCCUR.

(b) Remove the strap that supports the upper side strut [21] to the hanger link assembly [35].

(c) Position the upper side strut [21] for attachment to the reaction link assembly [1].

(d) Put the side strut pin [20] through the upper side strut [21] and reaction link assembly [1].

1) Use the thread protector, C32029-12 or C32029-101, SPL-1368, when you do this step.

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(e) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound , (preferred) or G50136 corrosion inhibiting material , (alternate) to these items:

1) Do not apply corrosion preventive compound to the chrome plated surfaces.

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2) The threads and thread reliefs of the side strut pin [20] and crossbolt [17].
3) The threads of the side strut nut [18] and nut [22].
4) The faces of the side strut washer [19] and washer [24].

(f) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

(g) Install the side strut washer [19] and side strut nut [18] on the side strut pin [20].

(h) Tighten the side strut nut [18] to 600 in-lb (67.8 N·m) - 900 in-lb (101.7 N·m).

1) Use the wrench adapter, C32029-3 or C32029-93, SPL-1359, to hold the head of the side strut pin [20] when you tighten the side strut nut [18].

(i) If it is necessary, loosen the side strut nut [18] to the nearest castellation that aligns with the hole for the crossbolt [17].

(j) Install the crossbolt [17] in the side strut pin [20].

(k) Install the washer [24] and nut [22] on the crossbolt [17].

(l) Tighten the nut [22] to 30 in-lb (3.4 N·m) - 50 in-lb (5.6 N·m). (50 in lb)

(m) If it is necessary, loosen the nut [22] to the nearest castellation that aligns with the hole for the cotter pin [23].

(n) Install the cotter pin [23] in the crossbolt [17].

SUBTASK 32-11-71-420-004

(5) Do these steps to connect the reaction link assembly [1] to the upper lock link [5]:

(a) Lubricate the chrome plated surfaces of the downlock link pin [13] with D00633 grease .

(b) Position the upper lock link [5] for attachment to the reaction link assembly [1].

(c) Put the downlock link pin [13] through the upper lock link [5] and reaction link assembly [1].

1) Use the thread protector, C32029-13, SPL-1369, when you do this task.

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(d) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred) or G50136 corrosion inhibiting material , (alternate) to these items:

1) Do not apply corrosion preventive compound to the chrome plated surfaces.
2) The threads and thread reliefs of the downlock link pin [13].
3) The threads of the nut [15].
4) The faces of the washer [14].

(e) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

(f) Install the washer [14] and nut [15] on the downlock link pin [13].

(g) Tighten the nut [15] to 360 in-lb (40.7 N·m) - 600 in-lb (67.8 N·m). (500 in lb)

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INDEPENDENT INSPECTION 22-3-25 02:22 GAT 412
INDEPENDENT INSPECTION 22-3-25 02:27 GAT 412

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1) Use the wrench adapter wrench adapter, C32029-4 or C32029-94, SPL-1360, to hold the head of the downlock link pin [13] when you tighten the nut [15].

(h) If it is necessary, loosen the nut [15] to the nearest castellation that aligns with the hole for the cotter pin [16].

(i) Install the cotter pin [16] in the downlock link pin [13].

SUBTASK 32-11-71-640-001

(6) Do these steps to connect the downlock actuator [4] to the reaction link assembly [1]:

(a) Lubricate the inside surface of the lug, head end of the downlock actuator [4], and rod end pin [6] with D00633 grease .

1) IfD00633 grease , is not available, use D00013 grease .

(b) Position the head end of the downlock actuator [4] for attachment to the reaction link assembly [1].

1) If it is necessary, retract or extend the downlock actuator [4] to align the head end to the reaction link assembly [1].

(c) Put the rod end pin [6] through the reaction link assembly [1], washers [7], washers [8], and head end of the downlock actuator [4].

(d) Install the washer [9] and nut [10] on the rod end pin [6].

(e) Tighten the nut [10] to 15 in-lb (1.7 N·m) - 25 in-lb (2.8 N·m). (20 in-lb)

(f) If it is necessary, loosen the nut [10] to the nearest castellation that aligns with the hole for the cotter pin [11].

(g) Install the cotter pin [11] in the rod end pin [6].

SUBTASK 32-11-71-420-005

(7) Do these steps to connect the hydraulic tubes [2] to the downlock actuator [4]:

(a) Remove the caps from the unions in the ports of the downlock actuator [4].

(b) Remove the caps from the hydraulic tubes [2] and connect them to the downlock actuator [4].

(c) Tighten the hydraulic tubes [2] to 140 ±7 in-lb (15.8 ±0.8 N·m).

(d) Loosen the hydraulic tubes [2] and re-tighten them to 133 in-lb (15.0 N·m) - 147 in-lb (16.6 N·m).

SUBTASK 32-11-71-420-006

(8) Do these steps to connect the hydraulic line [37] and hydraulic line [37A] to the hydraulic tubes [2] that are attached to the bracket assembly [38]:

(a) Remove the caps from the hydraulic line [37] and hydraulic line [37A].

(b) Connect the hydraulic line [37] and hydraulic line [37A] to the hydraulic tubes [2].

(c) Tighten the hydraulic line [37] to 140 ±7 in-lb (15.8 ±0.8 N·m).

(d) Tighten the hydraulic line [37A] to 270 ±14 in-lb (30.5 ±1.6 N·m).

H. Main Landing Gear Reaction Link Assembly Installation Test

SUBTASK 32-11-71-420-007

(1) Do this task: Main Gear Downlock Spring Installation, TASK 32-32-91-400-801.

SUBTASK 32-11-71-710-001

(2) Do this task: Main Gear Downlock Actuator Installation, TASK 32-32-51-400-801.

SUBTASK 32-11-71-860-006

(3) Do this task: Hydraulic Reservoirs Pressurization, TASK 29-09-00-860-801.

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SUBTASK 32-11-71-860-004

(4) Do this task: Main Landing Gear Operational Test, TASK 32-32-00-710-801.

NOTE: If a hydraulic cart is not available, it is optional to perform the Main Landing Gear Test - Component Replacement, TASK 32-32-00-710-802 instead of the operational test.

SUBTASK 32-11-71-210-001

(5) Do a check of the hydraulic connections for leakage.

I. Put the Airplane Back to Its Usual Condition

SUBTASK 32-11-71-860-002

(1) For hydraulic system A, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

SUBTASK 32-11-71-580-003

(2) Do this task: Lower the Airplane Off the Jacks, TASK 07-11-01-580-816.

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TASK 32-11-89-420-801

16. Main Landing Gear Downlock Strut Installation (Figure 401)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)
12-21-11-640-801	Main Landing Gear Upper End Components Servicing (P/ B 301)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
29-11-00-860-805	Hydraulic System A or B Power Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-32-51-400-801	Main Gear Downlock Actuator Installation (P/B 401)
32-32-91-400-801	Main Gear Downlock Spring Installation (P/B 401)
32-61-31-220-801	Main Landing Gear Down-and-Locked Sensor Clearance Measurement (P/B 501)

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1361	Wrench Adapter, C32029-39 or C32029-95 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1369	Thread Protector , C32029-13 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1370	Thread Protector, C32029-14 (included in C32029 kit)

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737-800

(Part #: C32029-87, Supplier: 81205)

(Opt Part #: C32029-38, Supplier: 81205)

(Opt Part #: C32029-45, Supplier: 81205)

JXB ALL

C. Consumable Materials

Reference	Description	Specification
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827 (NATO G-354) (Supersedes MIL-G-23827)
D00153	Fluid - Hydraulic Fluid, Fire Resistant (Interchangeable And Intermixable With BMS 3-11 Type V)	BMS3-11 Type IV
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
3	Packing	32-32-51-01-024	JXB ALL
5	Downlock strut	32-11-61-01-187	JXB 004, 005
		32-11-61-01-195	JXB 004, 005
		32-11-61-01-200	JXB 004, 005
		32-11-61-01-555	JXB 007-009, 014, 026-031, 033, 036, 037, 040-051, 053-999
16	Pin	32-11-61-01-175	JXB ALL
22	Pin	32-11-71-01-010	JXB ALL
		32-11-71-01-085	JXB ALL

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E. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
713	Nose Landing Gear
734	Left Main Landing Gear
744	Right Main Landing Gear

F. Prepare for the Installation

SUBTASK 32-11-89-480-002

WARNING: MAKE SURE THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR CAN RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed in the nose and main landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

G. Main Landing Gear Downlock Strut Installation

SUBTASK 32-11-89-420-001

(1) Do these steps to connect the downlock strut [5] to the upper side strut:

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: MAKE SURE THAT YOU REMOVE UNWANTED CORROSION INHIBITING COMPOUND FROM SURFACES THAT YOU WILL LUBRICATE. IF YOU APPLY CORROSION INHIBITING COMPOUND TO JOINTS THAT TURN, FAILURE OF THE LANDING GEAR TO EXTEND OR RETRACT COULD OCCUR.

(a) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:

NOTE: Do not apply corrosion preventive compound to the chrome plated surfaces.

1) The threads and thread reliefs of the downlock link pin [13]
2) The threads of the nut [15]
3) The faces of the washer [14].

(b) Remove any excess corrosion preventive G50237 Cor-Ban 27L Compound .

(c) Lubricate the chrome plated surfaces of the downlock link pin [13] with D00633 grease .

(d) Put the downlock strut [5] in its position for attachment to the upper side strut.

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(e) Put the downlock link pin [13] through the upper side strut and the downlock strut [5].
NOTE: Use thread protector, C32029-13, SPL-1369, when you do this task.
(f) Install the washer [14] and nut [15] on the downlock link pin [13].
(g) Tighten the nut [15] to 360 in-lb (40.7 N·m) – 600 in-lb (67.8 N·m). 500 in lb
(h) Install the pin [16] in the downlock link pin [13].
1) If it is necessary, loosen the nut [15] to the nearest castellation that aligns with the holes for the pin [16].

SUBTASK 32-11-89-420-002

(2) Do these steps to connect the downlock strut [5] to the reaction link assembly [6]:

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(a) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound (preferred), or G50136 corrosion inhibiting material (alternate), to these items:
NOTE: Do not apply corrosion preventive compound to the chrome plated surfaces.
1) The threads and thread reliefs of the downlock link pin [12]
2) The threads of the nut [21]
3) The faces of the washer [23].
(b) Remove any excess corrosion preventive G50237 Cor-Ban 27L Compound .
(c) Lubricate the chrome plated surfaces of the downlock link pin [12] with D00633 grease .
(d) Put the downlock strut [5] in its position for attachment to the reaction link assembly [6].
(e) Put the downlock link pin [12] through the downlock strut [5] and reaction link assembly [6].
NOTE: Use thread protector, C32029-14, SPL-1370, when you do this task.
(f) Install the washer [23] and nut [21] on the downlock link pin [12].
(g) Tighten the nut [21] to 360 in-lb (40.7 N·m) – 600 in-lb (67.8 N·m). 500 in lb
NOTE: Use wrench adapter, C32029-39 or C32029-95, SPL-1361, to hold the head of the downlock link pin when you tighten the nut.
(h) Install the pin [22] in the downlock link pin [12].
1) If it is necessary, loosen the nut [21] to the nearest castellation that aligns with the holes for the pin [22].

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SUBTASK 32-11-89-640-001

(3) Do these steps to connect the rod end of the downlock actuator [4] to the downlock strut [5]:

(a) Lubricate the inside surface of the rod end of the downlock actuator [4] and rod end pin [8] with D00633 grease .

NOTE: Use D00013 grease , if D00633 grease , is not available.

(b) Put the rod end of the downlock actuator [4] in its position for attachment to the downlock strut [5].

1) If it is necessary, retract or extend the downlock actuator [4] to align the rod end of the downlock actuator [4] to the lug on the downlock strut [5].

(c) From the forward side, put the rod end pin [8] through the downlock strut [5], spacers [9], washers [10], and rod end of the downlock actuator [4].

(d) Install the spacer [7] on the rod end pin [8].

SUBTASK 32-11-89-420-003

(4) Do these steps to connect the hydraulic lines [2] to the downlock actuator [4]:

(a) Remove the plugs from the ports on the downlock actuator [4].

(b) Lubricate the new packings [3] with D00153 hydraulic fluid , and install the new packings [3] in the ports of the downlock actuator [4].

(c) Remove the caps from the unions and connect the hydraulic lines [2] to the downlock actuator [4].

(d) Tighten B-nuts on the hydraulic lines [2] to 140 in-lb (16 N·m).

(e) Loosen the hydraulic lines [2] and re-tighten them to 133 in-lb (15 N·m) - 147 in-lb (17 N·m). 144 in lb

(f) Remove the tags from the hydraulic lines [2].

SUBTASK 32-11-89-420-005

(5) Do this step to connect the electrical harness [11] to the downlock strut [5]: D46012P & D46015P CONNECTED

(a) Install the nuts [20], washers [19], and bolts [18] to connect the brackets [17] to the downlock strut [5].

SUBTASK 32-11-89-820-001

(6) Do this task: Main Landing Gear Down-and-Locked Sensor Clearance Measurement, TASK 32-61-31-220-801.

H. Put the Airplane Back to Its Usual Condition

SUBTASK 32-11-89-420-004

(1) Do this task: Main Gear Downlock Spring Installation, TASK 32-32-91-400-801.

SUBTASK 32-11-89-610-001

(2) Lubricate the downlock strut [5], do this task: Main Landing Gear Upper End Components Servicing, TASK 12-21-11-640-801.

SUBTASK 32-11-89-860-003

(3) For the hydraulic systems A and B, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-11-89-710-001

(4) Test the downlock actuator [4], do this task: Main Gear Downlock Actuator Installation, TASK 32-32-51-400-801.

SUBTASK 32-11-89-790-001

(5) Examine the connections of the hydraulic lines for leakage.

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Item: 16G(6) Completed through item: 16G(6) Sign:

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SUBTASK 32-11-89-080-002

(6) Do this task: Lower the Airplane Off the Jacks, TASK 07-11-01-580-816.

SUBTASK 32-11-89-780-001

(7) Remove the power from hydraulic systems A and B, do this task: Hydraulic System A or B Power Removal, TASK 29-11-00-860-805.

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TASK 32-11-61-400-803

17. **Main Landing Gear Side Strut Installation** (Figure 403)

NOTE: This procedure is a scheduled maintenance task.

A. General

(1) This task supplies instructions to install the side strut.

B. References

Reference	Title
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)
12-21-11-640-801	Main Landing Gear Upper End Components Servicing (P/B 301)
20-10-44-400-801	Lockwire, Cotter Pins, and Lockrings - Installation (P/B 401)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-13-11-820-801	Main Landing Gear Shock Strut Door Adjustment (P/B 501)
32-32-00-710-801	Main Landing Gear Operational Test (P/B 501)
32-32-00-710-802	Main Landing Gear Test - Component Replacement (P/B 501)
32-32-91-400-801	Main Gear Downlock Spring Installation (P/B 401)

C. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description
SPL-1359	Wrench Adapter, C32029-3 or C32029-93 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-38, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)
SPL-1361	Wrench Adapter, C32029-39 or C32029-95 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205)

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(Opt Part #: C32029-38, Supplier: 81205)

(Opt Part #: C32029-45, Supplier: 81205)

SPL-1368 Thread Protector, C32029-12 or C32029-101 (included in C32029 kit)

737-800

(Part #: C32029-87, Supplier: 81205)

(Opt Part #: C32029-38, Supplier: 81205)

(Opt Part #: C32029-45, Supplier: 81205)

SPL-1369 Thread Protector , C32029-13 (included in C32029 kit)

737-800

(Part #: C32029-87, Supplier: 81205)

(Opt Part #: C32029-38, Supplier: 81205)

(Opt Part #: C32029-45, Supplier: 81205)

SPL-1370 Thread Protector, C32029-14 (included in C32029 kit)

737-800

(Part #: C32029-87, Supplier: 81205)

(Opt Part #: C32029-38, Supplier: 81205)

(Opt Part #: C32029-45, Supplier: 81205)

JXB ALL

D. Consumable Materials

Reference	Description	Specification
D00633	Grease - Aircraft General Purpose	BMS3-33
G50136	Compound - Corrosion Inhibiting, Non-drying Paste	BMS3-38
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

E. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
3	Upper side strut	32-11-61-01-480	JXB ALL
6	Pin	32-11-61-01-175	JXB ALL
19	Pin	32-11-71-01-055	JXB ALL
49	Pin	32-11-61-03-040	JXB ALL

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

				MECH	INSP
55	Cotter pin	32-13-21-01-203	JXB 004, 005, 007-009, 014, 026-031, 033		
		32-13-21-01-204	JXB 004, 005, 007-009, 014, 026-031, 033		
		32-13-21-04-215	JXB 036, 037, 040-051, 053-999		
		32-13-21-04-220	JXB 036, 037, 040-051, 053-999		

F. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
713	Nose Landing Gear
734	Left Main Landing Gear
744	Right Main Landing Gear

G. Prepare for the Installation

SUBTASK 32-11-61-480-008

WARNING: MAKE SURE THAT THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR COULD RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(1) If the downlock pins are not installed on all the landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

H. Main Landing Gear Side Strut Installation

SUBTASK 32-11-61-420-008

(1) Connect the upper side strut [3] to the reaction link assembly [15].
  (a) Lubricate the chrome plated surfaces of the side strut pin [22] with the D00633 grease .
  (b) Remove all unwanted D00633 grease .
  (c) Put the upper side strut [3] in its position for connection to the reaction link assembly [15].

CAUTION: MAKE SURE THAT THE HEAD OF THE PIN POINTS IS IN THE FORWARD DIRECTION. IF THE HEAD OF THE PIN POINTS IS IN AFT DIRECTION, DAMAGE TO COMPONENTS AND THE AIRPLANE STRUCTURE WILL OCCUR.

  (d) Put the side strut pin [22] through the upper side strut [3] and reaction link assembly [15].

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

1) Make sure that the head of the side strut pin [22] points in the forward direction.
   a) Use a thread protector, C32029-12 or C32029-101, SPL-1368, for the side strut pin.

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

INDEPENDENT INSPECTION 21.3.25 09:17 GAT 412

(e) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound , or G50136 corrosion inhibiting material , to these items:
   1) Do not apply corrosion preventive compound to the chrome plated surfaces.
   2) The threads and thread reliefs of the side strut pin [22] and cross bolt [16].
   3) The threads of the side strut nut [20] and nut [18].
   4) The faces of the side strut washer [21] and washer [17].
(f) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .
(g) Install the side strut washer [21] and side strut nut [20] on the side strut pin [22].
(h) Tighten the side strut nut [20] to 600 in-lb (67.79 N·m) - 900 in-lb (101.69 N·m). 900 in-lb
   1) Use a wrench adapter, C32029-3 or C32029-93, SPL-1359, to hold the head of the side strut pin when you tighten the side strut nut.

INDEPENDENT INSPECTION 21.3.25 09:22 GAT 412 CR-2

(i) If it is necessary, loosen the side strut nut [20] to the nearest castellation to align the hole for the cross bolt [16].
(j) Install the cross bolt [16] in the side strut pin [22].
(k) Install the washer [17] and nut [18] on the cross bolt [16].
(l) Tighten the nut [18] to 30 in-lb (3.39 N·m) - 50 in-lb (5.65 N·m).
(m) If it is necessary, loosen the nut [18] to the nearest castellation to align the holes for the new pin [19].
(n) Install the new pin [19] in the cross bolt [16].

CR-2 CR-2

INDEPENDENT INSPECTION 21.3.25 09:27 GAT 412

SUBTASK 32-11-61-420-009

(2) Connect the lower side strut [8] to the shock strut [51].
   (a) Lubricate the chrome plated surfaces of the side strut pin [52] with D00633 grease .
   (b) Remove all unwanted D00633 grease .
   (c) Put the lower side strut [8] in its position for attachment to the shock strut [51].
   (d) Put the side strut pin [52] through the lower side strut [8] and clevis on the shock strut [51].
      1) Use a thread protector, C32029-14, SPL-1370, for the side strut pin.

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL

MR 0742

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: **DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.**

(e) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound , or G50136 corrosion inhibiting material , to these items:
1) Do not apply corrosion preventive compound to the chrome plated surfaces.
2) The threads and thread reliefs of the side strut pin [52] and cross bolt [47].
3) The threads of the side strut nut [54] and nut [50].
4) The faces of the side strut washer [53] and washer [48].

(f) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .

(g) Install the side strut washer [53] and side strut nut [54] on the side strut pin [52].

(h) Tighten the side strut nut [54] to 95 ft-lb (128.80 N·m) – 115 ft-lb (155.92 N·m) more than the run on torque. 110ft-lb
1) Use a wrench adapter, C32029-39 or C32029-95, SPL-1361, to hold the head of the side strut pin when you tighten the side strut nut.

INDEPENDENT INSPECTION 21.3.25 09:32 GAT 412 CR-2

(i) Install the cross bolt [47] in the side strut pin [52].

(j) If it is necessary, loosen the side strut nut [54] to the nearest castellation to align the hole for the cross bolt [47].

(k) Install the washer [48] and nut [50] on the cross bolt [47].

INDEPENDENT INSPECTION 21.3.25 09:37 GAT 412 CR-2

(l) Tighten the nut [50] to 20 in-lb (2.26 N·m) - 24 in-lb (2.71 N·m) more than the run on torque. 24in.lb

(m) If it is necessary, loosen the nut [50] to the nearest castellation to align the hole for the new pin [49].

CAUTION: **MAKE SURE THAT THE COTTER PIN IS INSTALLED CORRECT. A CHAFF CONDITION CAN OCCUR TO THE SHEATHING ON THE CABLE OF THE GROUND SPOILER INTERLOCK. IF YOU DO NOT OBEY, DAMAGE TO EQUIPMENT CAN OCCUR.**

CR-2

(n) Install the new pin [49] in the cross bolt [47].
1) Make sure that you install the new pin [49] in accordance with TASK 20-10-44-400-801, type B preferred installation method.

(o) Check the interior of side strut pin [52] and reapply corrosion preventive G50237 Cor-Ban 27L Compound , as required.

INDEPENDENT INSPECTION 21.3.25 09:42 GAT 412

SUBTASK 32-11-61-080-003

(3) Remove the rope that holds the pushrod [41] in its position.

SUBTASK 32-11-61-420-010

(4) Connect the lower side strut [8] to the pushrod [41].
(a) Lubricate the shank of the bolt [46] and bushing [45] with D00633 grease .
(b) Remove all unwanted D00633 grease .
(c) Align the hole in the pushrod [41] with the hole in the lower side strut [8].
(d) Put the bushing [45] through the lower side strut [8] and pushrod [41].

CR-2 MR 0712

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(e) Install the washer [44] on the bolt [46].
(f) Put the bolt [46] through the lower side strut [8] and pushrod [41].
(g) Install the washer [43] and nut [42] on the bolt [46].
1) Do not tighten the nut.

NOTE: Final torque to the nut will be done after final adjustments to the doors are made in the subsequent step.

SUBTASK 32-11-61-420-011

(5) Connect the upper side strut [3] to the downlock strut [1].
(a) Lubricate the chrome plated surfaces of the downlock link pin [2] with D00633 grease .
(b) Remove all unwanted D00633 grease .
(c) Put the downlock strut [1] in its position for connection to the upper side strut [3].
(d) Put the downlock link pin [2] through the upper side strut [3] and the downlock strut [1].
1) Use a thread protector, C32029-13, SPL-1369, for the downlock link pin.

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(e) Apply a thin layer of the corrosion preventive G50237 Cor-Ban 27L Compound , to these items:

NOTE: Do not apply corrosion preventive compound to the chrome plated surfaces.

NOTE: If G50237 Cor-Ban 27L Compound , is not available, you can use G50136 corrosion inhibiting material , as an approved alternate.

1) The threads and thread reliefs of the downlock link pin [2]
2) The threads of the nut [5]
3) The faces of the washer [4].

(f) Remove all unwanted corrosion preventive G50237 Cor-Ban 27L Compound .
(g) Install the washer [4] and nut [5] on the downlock link pin [2].
(h) Tighten the nut [5] to 360 in-lb (40.67 N·m) - 600 in-lb (67.79 N·m). 500 in-lb
(i) If it is necessary, loosen the nut [5] to the nearest castellation to align the hole for the new pin [6].
(j) Install the new pin [6] in the downlock link pin [2].

SUBTASK 32-11-61-820-002

(6) If the length of pushrod [41] is changed, do the step that follows:

NOTE: No adjustment to the doors is necessary if the pushrod was not changed.

N/A

INDEPENDENT INSPECTION 21.8.24 09:52 GAT 412

INDEPENDENT INSPECTION 21.8.24 09:57 GAT 412

CR-2 CR-2

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(a) Do this task: Main Landing Gear Shock Strut Door Adjustment, TASK 32-13-11-820-801.

SUBTASK 32-11-61-420-016

(7) Tighten the nut [42].

(a) Install the new cotter pin [55] in the hole of the bolt [46].

(b) If it is necessary, loosen the nut [42] to the nearest castellation to align the holes for the cotter pin [55].

I. Put the Airplane Back to Its Usual Condition

SUBTASK 32-11-61-420-012

(1) Do this task: Main Gear Downlock Spring Installation, TASK 32-32-91-400-801.

SUBTASK 32-11-61-640-003

(2) Lubricate the lube fittings on the side strut, do this task: Main Landing Gear Upper End Components Servicing, TASK 12-21-11-640-801.

SUBTASK 32-11-61-780-004

(3) For hydraulic systems A, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-11-61-710-002

(4) Do this task: Main Landing Gear Operational Test, TASK 32-32-00-710-801.

NOTE: It is optional to perform the Main Landing Gear - Component Replacement, TASK 32-32-00-710-802 instead of the operational test if a hydraulic cart is not available.

SUBTASK 32-11-61-580-009

(5) Do this task: Lower the Airplane Off the Jacks, TASK 07-11-01-580-816.

INDEPENDENT INSPECTION 21-3-25 10:10 GAT 412

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

JXB ALL

TASK 32-11-00-400-801

18. **Main Landing Gear Installation** (Figure 401) (Figure 402) (Figure 403) (Figure 404) (Figure 405) (32-13-11-990-801)

NOTE: This procedure is a scheduled maintenance task.

A. References

Reference	Title
07-11-01-580-815	Lift the Airplane with the Jacks (P/B 201)
07-11-01-580-816	Lower the Airplane Off the Jacks (P/B 201)
12-15-31-610-802	Main Landing Gear Shock Strut Servicing, Airplane on the Ground (P/B 301)
12-15-31-610-805	Main Landing Gear Strut Servicing, Airplane on Jacks (P/B 301)
12-21-11-640-801	Main Landing Gear Upper End Components Servicing (P/B 301)
12-21-11-640-802	Main Landing Gear Lower End Components Servicing (P/B 301)
20-10-44-400-801	Lockwire, Cotter Pins, and Lockrings - Installation (P/B 401)
20-10-52-400-801	Flexible Hose Installation (P/B 401)
27-51-00-040-801	Trailing Edge Flap System Deactivation (P/B 201)
27-51-00-440-801	Trailing Edge Flap System Reactivation (P/B 201)
27-51-00-860-804	Retract the Trailing Edge Flaps (P/B 201)
27-62-00-800-801	Speed Brake Hydraulic Systems A and B Pressurization (P/B 201)
27-62-00-840-801	Put the Speed Brake Hydraulic systems A and B Back to the Condition Before the Pressurization (P/B 201)
27-62-51-400-801	Ground Spoiler Interlock Valve Cable Installation (P/B 401)
29-11-00-860-801	Hydraulic System A or B Pressurization (P/B 201)
32-00-01-080-801	Landing Gear Downlock Pins Removal (P/B 201)
32-00-01-480-801	Landing Gear Downlock Pins Installation (P/B 201)
32-11-61-400-802	Main Landing Gear Lower Side Strut Installation (P/B 401)
32-11-81-870-801	Hydraulic Shimmy Damper - Bleeding (P/B 401)
32-11-83-000-801	Main Landing Gear Forward Trunnion Bearing Assembly Removal (P/B 401)

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

		MECH	INSP
32-11-83-400-801	Main Landing Gear Forward Trunnion Bearing Assembly Installation (P/B 401)		
32-13-11-420-802	Main Landing Gear Center Door Installation (P/B 401)		
32-13-11-420-803	Main Landing Gear Door Assembly Installation (P/B 401)		
32-13-11-820-801	Main Landing Gear Shock Strut Door Adjustment (P/B 501)		
32-13-11-990-801	Figure: Main Landing Gear Shock Strut Doors Installation (P/B 401)		
32-13-21-000-801	Main Landing Gear Wing Door Adjustment (P/B 501)		
32-13-21-420-801	Main Landing Gear Wing Door Installation (P/B 401)		
32-32-00-710-801	Main Landing Gear Operational Test (P/B 501)		
32-32-00-710-802	Main Landing Gear Test - Component Replacement (P/B 501)		
32-34-00-730-801	Main Gear Manual Extension System Test - Airplane on Jacks (P/B 501)		
32-41-00-870-802	Normal (System B) Hydraulic Brake System - Bleeding (P/B 201)		
32-41-41-400-801	Main Landing Gear Brake Installation (P/B 401)		
32-42-00-710-802	Transducer Operational Test (P/B 501)		
32-42-00-720-803	Antiskid Valve Operational Test (P/B 501)		
32-45-11-400-801	Main Landing Gear Wheel and Tire Assembly Installation (P/B 401)		
SWPM 20-60-03	Special Protection of Electrical Connectors		

B. Tools/Equipment

NOTE: When more than one tool part number is listed under the same "Reference" number, the tools shown are alternates to each other within the same airplane series. Tool part numbers that are replaced or non-procurable are preceded by "Opt:", which stands for Optional.

Reference	Description	MECH	INSP
COM-15434	OMNI ARM Aircraft Maintenance Fixture **737-800** (Part #: NB-800, Supplier: 1T0V3) (Opt Part #: NBH, Supplier: 1T0V3)		
SPL-1362	Wrench Adapter, C32029-6 or C32029-96 (included in C32029 kit)		

PARTIAL SIGN OFF STATUS: -

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

		MECH	INSP
	737-800		
	(Part #: C32029-87, Supplier: 81205)		
	(Opt Part #: C32029-38, Supplier: 81205)		
	(Opt Part #: C32029-45, Supplier: 81205)		
SPL-1363	Wrench Adapter, C32029-7 or C32029-98 (included in C32029 kit)		
	737-800		
	(Part #: C32029-87, Supplier: 81205)		
	(Opt Part #: C32029-38, Supplier: 81205)		
	(Opt Part #: C32029-45, Supplier: 81205)		
SPL-1365	Thread Protector, C32029-9 (included in C32029 kit)		
	737-800		
	(Part #: C32029-87, Supplier: 81205)		
	(Opt Part #: C32029-38, Supplier: 81205)		
	(Opt Part #: C32029-45, Supplier: 81205)		
SPL-1366	Thread Protector, C32029-10 (included in C32029 kit)	MR 0742	GAT 492
	737-800		
	(Part #: C32029-87, Supplier: 81205)		
	(Opt Part #: C32029-38, Supplier: 81205)		
	(Opt Part #: C32029-45, Supplier: 81205)		
SPL-1371	Thread Protector, C32029-15 (included in C32029 kit)		
	737-800		
	(Part #: C32029-87, Supplier: 81205)		
	(Opt Part #: C32029-38, Supplier: 81205)		
	(Opt Part #: C32029-45, Supplier: 81205)		
SPL-1372	Thread Protector, C32029-16 (included in C32029 kit)		
	737-800		
	(Part #: C32029-87, Supplier: 81205)		
	(Opt Part #: C32029-38, Supplier: 81205)		
	(Opt Part #: C32029-45, Supplier: 81205)		
SPL-1521	Tool - Strut Inflation, Landing Gear		
	737-800		

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

		MECH	INSP
	(Part #: F70200-35, Supplier: 81205) (Opt Part #: F70200-1, Supplier: 81205) (Opt Part #: F70200-14, Supplier: 81205) (Opt Part #: F70200-17, Supplier: 81205) (Opt Part #: F70200-18, Supplier: 81205)		
SPL-1744	Lock Set - Inboard Ground Spoiler Actuator (Contains 4 Lock Assemblies) **737-800** (Part #: C27046-13, Supplier: 81205) (Opt Part #: C27046-1, Supplier: 81205)		
SPL-1862	Equipment - Removal/Installation, MLG Aft Trunnion Pin **737-800** (Part #: C32031-22, Supplier: 81205) (Opt Part #: C32031-1, Supplier: 81205)		
SPL-1869	Fixture - Transportation, Main Landing Gear, Removal/ Installation **737-800** (Part #: C32034-343, Supplier: 81205) (Opt Part #: C32034-227, Supplier: 81205) (Opt Part #: C32034-273, Supplier: 81205) (Opt Part #: C32034-339, Supplier: 81205)	MR 0742	GAT 492
SPL-1871	Strap - Retention, NLG/MLG Inner Cylinder **737-800** (Part #: C32030 -31, Supplier: 81205) (Opt Part #: C32030-10, Supplier: 81205)		
SPL-14021	Wrench Adapter, C32029-48 or C32029-90 (included in C32029 kit) **737-800** (Part #: C32029-87, Supplier: 81205) (Opt Part #: C32029-45, Supplier: 81205)		
SPL-14022	Wrench Adapter, C32029-49 or C32029-88 (included in C32029 kit) **737-800**		

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(Part #: C32029-87, Supplier: 81205)

(Opt Part #: C32029-45, Supplier: 81205)

SPL-14023 Wrench Adapter, C32029-50 or C32029-89 (included in C32029 kit)

737-800

(Part #: C32029-87, Supplier: 81205)

(Opt Part #: C32029-45, Supplier: 81205)

SPL-14024 Thread Protector , C32029-51 (included in C32029 Kit)

737-800

(Part #: C32029-87, Supplier: 81205)

(Opt Part #: C32029-45, Supplier: 81205)

JXB ALL

C. Consumable Materials

Reference	Description	Specification
D00633	Grease - Aircraft General Purpose	BMS3-33
G00018	Nitrogen - Gaseous, Pressurizing, 99.5 Percent Pure	A-A-59503 Type I Grade B, MIL-PRF-27401 Type I Grade A
G02166	Lockwire - MS20995NC20, Monel - 0.020 Inch (0.508 mm) Diameter	NASM20995
G02314	Air - Compressed, Breathing	BB-A-1034 Source I Grade A
G50171	Compound - Corrosion Inhibiting Compound, Interior Application - D5026NS or ZC-026	
G50237	Compound - Corrosion Inhibiting, Non-drying - Cor-Ban 27L	BMS3-38

D. Expendables/Parts

AMM Item	Description	AIPC Reference	AIPC Effectivity
20	Pin	32-11-00-02-010	JXB ALL
24	Pin	32-11-00-02-070	JXB ALL

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

				MECH	INSP
41	Cotter pin	32-11-61-03-010	JXB ALL		
56	Cotter pin	32-32-11-01-010	JXB ALL		
57	Cotter pin	32-32-93-01-030	JXB ALL		

E. Location Zones

Zone	Area
133	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Left
134	Main Landing Gear Wheel Well, Body Station 663.75 to Body Station 727.00 - Right
734	Left Main Landing Gear
744	Right Main Landing Gear

F. Access Panels

Number	Name/Location
551BB	Lower Inboard Fixed Trailing Edge, Gear Adjustment Door
551BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel
551CB	Lower Inboard Fixed Trailing Edge, Gear Access Panel
551DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel
651BB	Lower Inboard Fixed Trailing Edge, Gear Door Adjustment
651BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel
651CB	Lower Inboard Fixed Trailing Edge, Gear Access Panel
651DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel

G. Prepare for the Installation

NOTE: Omni Arm Tool maintenance fixture, COM-15434, can be used as an alternate to Dolly Assembly fixture, SPL-1869. This procedure provides instructions to remove and install the main landing gear with the Dolly Assembly. Refer to manufacturers instructions for use of the Omni Arm Tool.

SUBTASK 32-11-00-480-015

CAUTION: MAKE SURE THAT THE FORWARD TRUNNION BEARING PIN DOES NOT MOVE ON THE PROTECTOR ASSEMBLY DURING THE REMOVAL OR INSTALLATION OF THE MAIN LANDING GEAR. IF THIS CONDITION OCCURS, IT CAN CAUSE DAMAGE TO THE PIN.

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PARTIAL SIGN OFF STATUS:

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(1) If the protector assembly [92] (C32034-228/-235) from fixture, SPL-1869, is not already installed, do the steps that follow:

NOTE: The protector assembly [92] will prevent damage to the forward trunnion bearing pin [29] during the installation of the main landing gear [93].

NOTE: For more protection of the transition radius of the forward trunnion bearing pin [29], you can apply tape around the pin before gear installation. The tape should be removed after installation.

JXB 004, 005, 007-009, 014 PRE SB 737-32-1448

NOTE: The retaining ring, the support ring assembly, and forward trunnion seal should be attached on the forward trunnion bearing pin before the forward trunnion is connected.

N/A

JXB 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448

NOTE: The retaining ring and seal should be attached on the forward trunnion bearing pin before the forward trunnion is connected.

JXB ALL

(a) Put the upper end of the protector assembly [92] (C32034-228/-235) in the large clearance between the forward trunnion bearing pin [29] and housing assembly.

(b) Attach the lower end of the protector assembly [92] (C32034-228/-235) over the bottom of the housing assembly, such that you hear a click when you attach it.

SUBTASK 32-11-00-200-005

(2) Make sure that the main landing gear center door has the fully captured clip assemblies for bolt retention installed (Figure 32-13-11-990-801, TASK 32-13-11-420-802).

SUBTASK 32-11-00-480-016

(3) Install the retention strap, SPL-1871, with the marker facing up, to hold the shock strut [1] in its compressed position.

NOTE: Two straps are required to hold the inner cylinder secure.

SUBTASK 32-11-00-640-001

(4) Make sure that the aft trunnion pin [19] can easily move.

(a) If you cannot easily move the aft trunnion pin [19], remove it from the aft trunnion and lubricate it with D00633 grease .

(b) Put the aft trunnion pin [19] into the aft trunnion.

SUBTASK 32-11-00-480-017

(5) Use tie wraps to hold the aft trunnion pin [19] inside the aft trunnion.

NOTE: This will prevent the aft trunnion pin from falling out of the outer cylinder trunnion when you do the subsequent steps.

SUBTASK 32-11-00-950-001

(6) Put protective tape on the outer edge of the aft trunnion to protect its surface finish from scratching.

SUBTASK 32-11-00-860-007

(7) Do this task: Trailing Edge Flap System Deactivation, TASK 27-51-00-040-801.

H. Main Landing Gear Installation

PARTIAL SIGN OFF STATUS:

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SUBTASK 32-11-00-480-018

(1) Attach the support assembly [96] and side strut attach assembly [94] to the main landing gear [93].

NOTE: A minimum of two persons are necessary to do this task.

(a) Make sure that the side strut attach assembly [94] is not attached to the support assembly [96].

(b) Make sure that the jackball pin is removed from the hole in the jackball fitting of the support assembly [96].

(c) Put the side strut attach assembly [94] into the lower side strut attach point on the main landing gear [93], and such that the stenciled plate is in the vertical position.

1) Install the washer on the threaded end of the side strut attach assembly [94].

2) Make sure that the stenciled plate is still in a vertical position and install the nut finger tight.

(d) Put the support assembly [96] in its position on the aft side of the main landing gear [93].

1) Make sure that the jackball fitting on the support assembly [96] is fully engaged on the jacking ball of the main landing gear.

(e) If it is necessary, align the holes for the installation of the jackball pin.

(f) Put the jackball pin through the jackball fitting of the support assembly [96] and jacking ball of the main landing gear [93].

(g) Install the lynch pin in the jackball pin.

(h) Loosen the nut that holds the side strut attach assembly [94], such that the stenciled plate can be attached to the studs on the side strut support assembly [97].

1) If it is necessary, adjust the adjustable square support tube of the side strut support assembly [97].

(i) Tighten the two hand knobs to hold the stenciled plate to the studs on the side strut support assembly [97].

(j) Tighten the nut that holds the side strut attach assembly [94] to the lower side strut attach point.

(k) Tighten the two adjustable handles to hold the adjustable square support tube of the side strut support assembly [97].

SUBTASK 32-11-00-480-019

(2) Attach the dolly assembly [98] to the support assembly [96].

NOTE: A minimum of two persons are necessary to do this task.

(a) Put the dolly assembly [98] in its position such that the boom arm [91] is on the forward side of the main landing gear [93] and legs are in center with the main landing gear [93].

(b) Use the manual valve on the hydraulic pump to lift or lower the boom arm [91] until you can connect the lift link [95] to the support assembly [96].

(c) Connect the lift link [95] to the support assembly [96].

(d) Install the pins to hold the lift link [95] to the support assembly [96].

(e) Connect the winch assembly [99] to the support assembly [96] with the snapper pin.

SUBTASK 32-11-00-480-020

PARTIAL SIGN OFF STATUS:

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(3) Make sure that the airplane is lifted with the jacks and lower surface of the housing, for the forward trunnion bearing pin [29], measures 97 ±2 in. (246 ±5 cm) above the ground (TASK 07-11-01-580-815) (View B-B, Figure 401).
(a) Make sure that to keep the airplane level.

SUBTASK 32-11-00-580-003

(4) Use the boom arm [91] to lift the main landing gear [93], such that the transportation support assembly and aft tire support assembly can be removed from the dolly assembly [98].

SUBTASK 32-11-00-980-001

(5) From the aft side, carefully move the dolly assembly [98] with main landing gear [93] forward below the wing and put it in its position below the wheel well area.
(a) Move the dolly assembly [98] such that the axis of the forward trunnion bearing pin [29] will be perpendicular to the housing assembly and rear spar.

SUBTASK 32-11-00-020-027

WARNING: MAKE SURE THAT THE HYDRAULIC PRESSURE IS BELOW 5000 PSIG (34,474 KPA). IF YOU CONTINUE WHEN THE HYDRAULIC PRESSURE IS 5000 (34,474 KPA) OR HIGHER, A BINDING CONDITION WILL OCCUR. A BINDING CONDITION CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(6) Do these steps to connect the forward trunnion:
(a) Lift the main landing gear [93] with the boom arm [91] and winch assembly [99], such that the forward trunnion and forward trunnion bearing pin [29] are aligned.
1) Remove the transportation support assembly and aft tire support assembly from the dolly assembly [98].
(b) Attach the outer cylinder strap assembly to the hoist ring on the end of the boom arm [91] and around the outer cylinder at the aft trunnion.
(c) Remove the tension from the strap.
(d) Use the pry lever bar to remove the load off of the caster wheels one at a time.
(e) Use the lever bar assemblies to turn all the caster wheels parallel to the dolly assembly [98] legs, such that the forward trunnion can be inserted straight onto the forward trunnion bearing pin [29].
1) Make sure that the swivel locks on the caster wheels are not locked.
(f) Make sure that the forward trunnion stays aligned with the forward trunnion bearing pin [29] and lock the swivel locks.
(g) Put chocks on all caster wheels but put the forward chock approximately 2 in. (5 cm) forward of the double wheel caster.
(h) Put the fork end of the lever bar assemblies on the rectangular tubing above the single wheel casters.
(i) Use the lever bar assemblies to slowly push the dolly assembly [98] forward.
1) Make sure that the aft trunnion will clear the lower trailing edge panel seals.
2) Make sure that the chocks are in contact with the single wheel casters to prevent the movement of the dolly assembly [98].
(j) Slowly lift the aft end of the main landing gear [93] with the winch assembly [99] and lifting the boom arm [91] at the same time while you continue to push the dolly assembly [98] forward.
1) Use the manual valve on the hydraulic pump to lift the boom arm [91].

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2) If it is necessary, tighten the outer cylinder strap to remove the tension, such that the forward trunnion will not come off of the forward trunnion bearing pin [29].

(k) Continue the movement of the dolly assembly [98] until the forward trunnion is fully inserted onto the forward trunnion bearing pin [29].

SUBTASK 32-11-00-210-004

(7) Visually examine the protective coating on the transition radius of the forward trunnion pin for chipping, cracks, or corrosion. NO DAMAGE ON THE TRANSITION RADIUS

(a) If the transition radius is damaged, replace the trunnion pin, do these tasks: N/A

- Main Landing Gear Forward Trunnion Bearing Assembly Removal, TASK 32-11-83-000-801
- Main Landing Gear Forward Trunnion Bearing Assembly Installation, TASK 32-11-83-400-801.

SUBTASK 32-11-00-620-004

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(8) Apply a thin layer of G50237 Cor-Ban 27L Compound , to these items:

(a) The shank, thread, and thread relief of the forward trunnion cross bolt [28]

(b) The thread of the nut [25]

(c) The spline and face of the washer [26]

(d) The new pin [24].

SUBTASK 32-11-00-160-005

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(9) Remove all unwanted G50237 Cor-Ban 27L Compound .

SUBTASK 32-11-00-420-029

(10) Do these steps to install the forward trunnion cross bolt [28]:

(a) Install the thread protector, C32029-9, SPL-1365, or thread protector, C32029-51, SPL-14024, on the forward trunnion cross bolt [28].

(b) Put the forward trunnion cross bolt [28] through the door actuator bracket [27], forward trunnion, and forward trunnion bearing pin [29].

(c) Remove the thread protector, C32029-9, SPL-1365, or thread protector, C32029-51, SPL-14024, from the forward trunnion cross bolt [28].

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INDEPENDENT INSPECTION 21.3.25 08:45 GAT 412

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(d) Install the washer [26] and nut [25] on the forward trunnion cross bolt [28].		
(e) Tighten the nut [25] to 21 ft-lb (28.47 N·m) - 25 ft-lb (33.90 N·m). (25 ft-lb)		
CR-2 1) Use the wrench adapter, C32029-49 or C32029-88, SPL-14022, or wrench adapter, C32029-50 or C32029-89, SPL-14023, to hold the head of the forward trunnion cross bolt [28].		WS 21.3.25 0830 GAT 492
(f) Install the pin [24] in the forward trunnion cross bolt [28].		
(g) If it is necessary, loosen the nut [25] to the nearest castellation to align the holes for the pin [24].		WS 21.3.25 08:40 GAT 492
CR-2 (h) Do a check of the nut [117], on the door actuator bracket [27], to make sure that it is not loose.		
1) If the nut [117] is loose, tighten it.		
SUBTASK 32-11-00-080-011		
(11) Remove the outer cylinder strap assembly from the hoist ring and outer cylinder.		
SUBTASK 32-11-00-420-022		
WARNING: MAKE SURE THAT THE HYDRAULIC PRESSURE IS BELOW 5000 PSIG (34,474 KPA). IF YOU CONTINUE WHEN THE HYDRAULIC PRESSURE IS 5000 (34,474 KPA) OR HIGHER, A BINDING CONDITION WILL OCCUR. A BINDING CONDITION CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.		WS 21.3.25 GAT 492
(12) Operate the dolly assembly [98] to put the aft trunnion in its position.		WS 21.3.25 GAT 492
NOTE: The main landing gear [93] installation path is done with the forward trunnion bearing pin [29] attached to the forward trunnion. This installation path is a combination of moving the aft trunnion inboard, lifting the aft trunnion, and moving the aft trunnion outboard as the main landing gear pivots about the forward trunnion bearing pin [29].		WS 21.3.25 GAT 492
(a) Make sure to lock the caster wheels when the main landing gear [93] pivots about the forward trunnion bearing pin [29] as follows:		
1) Use the pry lever bar to remove the load off of the caster wheels one at a time.		
2) Lock the single wheel casters perpendicular to the forward trunnion pivot axis approximately 38 degrees, and 90 degrees for the double wheel caster.		
(b) Put the fork end of the lever bar assemblies on the rectangular tubing above the single wheel casters.		
(c) Do a check to make sure that the locks for the caster wheels are released.		
(d) Hold the lever bar assemblies to slowly push the aft end of the dolly assembly [98] inboard, such that the aft trunnion will clear the main landing gear beam lower flange, lower trailing edge panel seals, and seal support beam when the aft trunnion is lifted.	MR 0742	WS 21.3.25 GAT 492
(e) If it is necessary, use the lever bar assemblies and pry lever bar to put the caster wheels parallel to the legs of the dolly assembly [98].		
(f) Lock the caster wheels.		
(g) Lift and move the aft trunnion outboard at the same time while you clear the main landing gear [93] from the main landing gear beam as follows:		
1) Release the tension on the winch assembly [99].		
2) Use the manual valve on the hydraulic pump to lift the boom arm [91].		
(h) Put the aft trunnion in its position directly below the aft trunnion bearing.		

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1) If the forward tire support assembly is on the way of the tires, remove it.
(i) Put chocks on all the casters.
(j) Continue to slowly lift the aft trunnion of the main landing gear [93] until the aft trunnion and aft trunnion bearing are aligned.
1) Make sure that the chock is in contact with the double wheel caster to prevent the forward movement of the dolly assembly [98].

SUBTASK 32-11-00-080-006

(13) Remove the two tie wraps that hold the aft trunnion pin [19].

SUBTASK 32-11-00-080-007

(14) Remove the protective tape from the outer edge of the aft trunnion.

SUBTASK 32-11-00-640-002

(15) Lubricate the chrome plated surfaces of the aft trunnion pin [19] with D00633 grease .

SUBTASK 32-11-00-160-001

(16) Remove all unwanted D00633 grease .

SUBTASK 32-11-00-620-001

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(17) Apply a thin layer of G50237 Cor-Ban 27L Compound , to these items:
(a) The shank, thread, and thread relief of the aft trunnion cross bolt [23]
(b) The thread of the nut [21]
(c) The spline and face of the washer [22]
(d) The new pin [20].

SUBTASK 32-11-00-160-002

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(18) Remove all unwanted G50237 Cor-Ban 27L Compound .

SUBTASK 32-11-00-420-007

WARNING: MAKE SURE THAT THE HYDRAULIC PRESSURE IS BELOW 5000 PSIG (34,474 KPA). IF YOU CONTINUE WHEN THE HYDRAULIC PRESSURE IS 5000 (34,474 KPA) OR HIGHER, A BINDING

MECH	INSP

CONDITION WILL OCCUR. A BINDING CONDITION CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(19) Connect the aft trunnion.
- (a) Put a flashlight, or other bright lights, at the top of the main landing gear [93] and shine them back past the aft trunnion.
- (b) Look through the aft trunnion pin access hole so you can see the aft trunnion pin [19]through the aft trunnion bearing.
- (c) Move the dolly assembly [98] from side to side or lift and lower it until you can see equal amounts of light all the way around the aft trunnion pin [19] through the aft trunnion bearing.

  NOTE: When you have the aft trunnion pin [19] and aft trunnion bearing correctly aligned, the aft trunnion pin will install very easily.
- (d) Use the aft trunnion equipment, SPL-1862, to pull the aft trunnion pin [19] from the aft trunnion.
  1) Put the pull adapter [74] through the pin access hole in the main landing gear beam assembly.
  2) Attach the pull adapter [74] and pull nut [73] to the aft trunnion pin [19].
  3) Pull the slide hammer [72] until the aft trunnion pin [19] is out of the aft trunnion bearing such that the aft trunnion will connect to the main landing gear beam assembly.
- (e) Move the aft trunnion bearing until it is in line with the aft trunnion pin [19].
- (f) If it is necessary, use the aft trunnion equipment, SPL-1862, to turn the aft trunnion pin [19] clockwise until the holes for the aft trunnion cross bolt [23] are aligned.
- (g) Install the thread protector, C32029-10, SPL-1366, on the aft trunnion cross bolt [23].
- (h) Put the aft trunnion cross bolt [23] through the aft trunnion and aft trunnion pin [19].
- (i) Remove the thread protector, C32029-10, SPL-1366, from the aft trunnion cross bolt [23].
- (j) Install the washer [22] and nut [21] on the aft trunnion cross bolt [23].
- (k) Tighten the nut [21] to 160 in-lb (18.08 N·m) - 190 in-lb (21.47 N·m). 180 in-lb

  NOTE: You can use long wrench with narrow body, ratchet wrench or crowfoot with 4 inches extension to tighten the nut.
  1) Use the wrench adapter, C32029-48 or C32029-90, SPL-14021, to hold the head of the aft trunnion cross bolt [23].
- (l) Install the pin [20] in the aft trunnion cross bolt [23].
- (m) If it is necessary, loosen the nut [21] to the nearest castellation to align the holes for the pin [20].

SUBTASK 32-11-00-080-004

(20) Remove the main landing gear fixture, SPL-1869, from the main landing gear [93].

SUBTASK 32-11-00-080-008

(21) Remove the protector assembly.

SUBTASK 32-11-00-210-005

(22) Visually examine the protective coating on the transition radius of the forward trunnion pin for chipping, cracks, or corrosion.
- (a) If the transition radius is damaged, replace the trunnion pin, do these tasks:

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INDEPENDENT INSPECTION 21.8.25 09:15 GAT 412

PARTIAL SIGN OFF STATUS:

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Type: Routine Card | **ATA:** 32-- | **Flow:** - | **Work Area:** -

	MECH	INSP
• Main Landing Gear Forward Trunnion Bearing Assembly Removal, TASK 32-11-83-000-801 • Main Landing Gear Forward Trunnion Bearing Assembly Installation, TASK 32-11-83-400-801. SUBTASK 32-11-00-080-009 (23) Remove retention strap, SPL-1871, from the main landing gear [93]. SUBTASK 32-11-00-212-001 (24) Before you connect the electrical connector, examine the connector for corrosion.		21-3-25 GAT 492
WARNING: DO THE STEPS BELOW IF THE AIRPLANE OPERATES WHERE DEICING FLUID THAT CONTAINS POTASSIUM FORMATE OR POTASSIUM ACETATE IS USED. ALSO, DO THE STEPS FOR ALL AIRPLANES THAT YOU FOUND CORROSION IN THE ELECTRICAL CONNECTORS IN THE MAIN WHEEL WELL. THE ELECTRICAL CONNECTORS ARE IN A SYSTEM THAT IS NECESSARY FOR SAFE FLIGHT. (a) If there was a corrosion, refer to SWPM 20-60-03 to correct the problem. (b) Apply the G50171 D5026NS or ZC-026 compound , to the connector (SWPM 20-60-03).		21-3-25 GAT 492
SUBTASK 32-11-00-420-008 (25) Do these steps to connect the electrical connector [13]: (a) Attach the two clamps [16] to the electrical connector [13]. (b) Install the two screws [14], two washers [15], and two nuts [17] to the two clamps [16] that hold the electrical connector [13] to the hose guide [18]. (c) Remove the cap from the electrical connector [13]. (d) Connect the electrical connector [13] to the electrical connector. (e) Install G02166 MS20995NC20 lockwire , on the electrical connector [13] (TASK 20-10-44-400-801). SUBTASK 32-11-00-420-009 (26) Do these steps to connect the electrical connector [11]: (a) Put the electrical connector [11] through the hose guide [12]. (b) Attach the two clamps [9] to the electrical connector [11]. (c) Install the two screws [7], two washers [8], and two nuts [10] to the two clamps [9] that hold the electrical connector [11] to the structure. (d) Remove the cap from the electrical connector [11]. (e) Connect the electrical connector [11] to the electrical connector. (f) Install G02166 MS20995NC20 lockwire , on the electrical connector [11] (TASK 20-10-44-400-801).	MR1701	22/3/25 GAT 128
SUBTASK 32-11-00-420-010 (27) Do these steps to connect the brake hose assembly [6] and hydraulic return shimmy damper line [114] (TASK 20-10-52-400-801): (a) Remove the caps from the brake hose assembly [6] and hydraulic return shimmy damper line [114]. (b) Remove the caps from the elbow [5]. (c) Connect the brake hose assembly [6] and hydraulic return shimmy damper line [114] to the elbow [5].	MR 0742	21-3-25 GAT 492

PARTIAL SIGN OFF STATUS:

Item: 18 H(25) Completed through item: H(26) Sign: 22/3/25 GAT 128

Item: ______ Completed through item: ______ Sign: ______

MR1701

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(d) Remove the tags from the brake hose assembly [6] and hydraulic return shimmy damper line [114].

SUBTASK 32-11-00-100-001

(28) Clean the surfaces of these items that will attach the retract actuator [43] to the main gear trunnion:
(a) Attachment fitting on the main gear trunnion
(b) actuator attach pin [50]
(c) spacers [49]
(d) washers [48]
(e) nut [45]
(f) cotter pin [56], nut [47], washer [46], and bolt [44].

SUBTASK 32-11-00-100-002

(29) Clean the surfaces of these items that will attach the walking beam [51] to the main gear trunnion:
(a) Attachment fitting on the main gear trunnion
(b) walking beam attach pin [52]
(c) washer [53]
(d) nut [55]
(e) cotter pin [57], nut [47], washer [46], and bolt [54].

SUBTASK 32-11-00-600-001

(30) Apply D00633 grease , to the chrome-plated surfaces of the reaction link pin [32], actuator attach pin [50], and walking beam attach pin [52].

SUBTASK 32-11-00-620-002

WARNING: **USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.**

CAUTION: **DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.**

(31) Apply a thin layer of G50237 Cor-Ban 27L Compound , to these items:

NOTE: **Do not apply corrosion preventive compound to the chrome plated surfaces.**

(a) The threads and thread reliefs on the actuator attach pin [50] and walking beam attach pin [52].
(b) The threads and thread reliefs of the bolt [44] and bolt [54].
(c) The threads and thread reliefs of the nut [45], nut [47], and nut [55].
(d) The faces of the washer [46], washer [48], and washer [53].
(e) The cotter pin [41], cotter pin [56], and cotter pin [57].

SUBTASK 32-11-00-160-003

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(32) Remove all unwanted G50237 Cor-Ban 27L Compound .

SUBTASK 32-11-00-080-010

(33) Remove the rope that holds the disconnected ends of the walking beam [51] and retract actuator [43].

SUBTASK 32-11-00-010-005

(34) Remove the clamps for the electrical conduit on the inboard side of the nut [45] and move the conduit to get clearance to tighten the nut [45].

SUBTASK 32-11-00-420-024

(35) Do these steps to connect the retract actuator [43] to the main gear trunnion:
- (a) Put the retract actuator [43] in its position for connection to the main gear trunnion.
- (b) Install the thread protector, C32029-16, SPL-1372, on the actuator attach pin [50].
- (c) Put the actuator attach pin [50] through the retract actuator [43], spacers [49] and main gear trunnion.
- (d) Remove the thread protector, C32029-16, SPL-1372, from the actuator attach pin [50].
- (e) Install the washer [48] and nut [45] on the actuator attach pin [50].
- (f) Use the wrench adapter, C32029-7 or C32029-98, SPL-1363, to hold the head of the actuator attach pin [50].
- (g) Tighten the nut [45] to 50 ft-lb (67.79 N·m) - 75 ft-lb (101.69 N·m). 70 ft-lb
- (h) If it is necessary, loosen the nut [45] to the nearest castellation to align the holes for the bolt [44].
- (i) Remove the wrench adapter, C32029-7 or C32029-98, SPL-1363, from the actuator attach pin [50].
- (j) Put the bolt [44] through the actuator attach pin [50].
- (k) Install the washer [46] and nut [47] on the actuator attach pin [50].
- (l) Tighten the nut [47] to 45 in-lb (5.08 N·m) - 55 in-lb (6.21 N·m). 50 in lb
- (m) If it is necessary, loosen the nut [47] to the nearest castellation to align the holes for the cotter pin [56].
- (n) Install the new cotter pin [56] in the bolt [44].

SUBTASK 32-11-00-410-003

(36) Put the electrical conduit back in its position and install the clamps.

SUBTASK 32-11-00-420-025

(37) Do these steps to connect the walking beam [51] to the main gear trunnion:
- (a) Put the walking beam [51] in its position for the connection to the main gear trunnion.
- (b) Install the thread protector, C32029-15, SPL-1371, on the walking beam attach pin [52].

INDEPENDENT INSPECTION 22-3-25 07:05 GAT 412 CR-2

INDEPENDENT INSPECTION 22-3-25 07:11 GAT 412 CR-2 CR-2

INDEPENDENT INSPECTION 22-3-25 07:14 GAT 412

PARTIAL SIGN OFF STATUS:

Item: 18 Completed through item: 4 (34) Sign: 21.3.25 GAT 492

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

(c) Put the walking beam attach pin [52] through the walking beam [51], and main gear trunnion.
(d) Remove the thread protector, C32029-15, SPL-1371, from the walking beam attach pin [52].
(e) Install the washer [53] and nut [55] on the walking beam attach pin [52].
(f) Install the wrench adapter, C32029-6 or C32029-96, SPL-1362, to hold the head of the walking beam attach pin [52].
(g) Tighten the nut [55] to 50 ft-lb (67.79 N·m) - 75 ft-lb (101.69 N·m). 70 ft-lb
(h) If it is necessary, loosen the nut [55] to the nearest castellation to align the holes for the bolt [54].
(i) Remove the wrench adapter, C32029-6 or C32029-96, SPL-1362, from the walking beam attach pin [52].
(j) Put the bolt [54] through the walking beam attach pin [52].
(k) Install the washer [46] and nut [47] on the walking beam attach pin [52].
(l) Tighten the nut [47] to 45 in-lb (5.08 N·m) - 55 in-lb (6.21 N·m). 50 in-lb
(m) If it is necessary, loosen the nut [47] to the nearest castellation to align the holes for the cotter pin [57].
(n) Install the new cotter pin [57] in the bolt [54].

SUBTASK 32-11-00-420-026

(38) Install the nuts [37], washers [35], washers [36], and bolts [34] to connect the turnbuckle [38] to the bracket.

SUBTASK 32-11-00-620-003

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES, IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

CAUTION: DO NOT APPLY CORROSION-INHIBITING COMPOUND ON GREASE JOINTS, OR SEALED BEARINGS. THESE COMPOUNDS REMOVE GREASE AND OTHER LUBRICANTS. THEY ARE PENETRATING COMPOUNDS. THEY WILL MOVE AROUND THE SEALS AND INTO THE BEARINGS. THIS WILL CAUSE DAMAGE TO THE BEARINGS, AND JOINTS.

(39) Apply a thin layer of the G50237 Cor-Ban 27L Compound , to these items:

NOTE: Do not apply corrosion preventive compound to the chrome plated surfaces.

(a) The threads and thread reliefs of the crossbolt [31]
(b) The threads of the nut [40]
(c) The faces of the washer [39].

SUBTASK 32-11-00-640-003

(40) Apply D00633 grease , to the chrome plated surfaces of the reaction link pin [32] and end cap [42].

SUBTASK 32-11-00-160-004

WARNING: USE NITRILE GLOVES FOR SKIN PROTECTION WHEN YOU USE COR-BAN 27L, G50237. IF IT GETS ON YOUR SKIN, IMMEDIATELY REMOVE IT WITH WATER. IF THIS MATERIAL GETS IN YOUR EYES,

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

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MECH	INSP

IMMEDIATELY FLUSH YOUR EYES WITH WATER. GET MEDICAL AID. THIS MATERIAL CONTAINS FLAMMABLE AGENTS WHICH CAN CAUSE INJURIES TO PERSONNEL.

(41) Remove all unwanted G50237 Cor-Ban 27L Compound , and D00633 grease .

SUBTASK 32-11-00-420-027

(42) Do these steps to connect the reaction link assembly [30] to the trunnion [33]:
- (a) Put the outboard end of the reaction link assembly [30] in its position for attachment to the trunnion [33].
- (b) Put the reaction link pin [32] through the reaction link assembly [30] and trunnion [33].
- (c) Install the end cap [42] on the reaction link pin [32].
- (d) Put the crossbolt [31] through the reaction link pin [32] and end cap [42].
- (e) Install the washer [39] and nut [40] on the crossbolt [31].
- (f) Tighten the nut [40] to 20 in-lb (2 N·m) - 24 in-lb (3 N·m). 23 in-lb
- (g) If it is necessary, loosen the nut [40] to the nearest castellation to align the holes for the cotter pin [41].
- (h) Install the new cotter pin [41] in the crossbolt [31].

INDEPENDENT INSPECTION 22-3-25 07:42 GAT 412 CR-2

SUBTASK 32-11-00-420-019

(43) Do this task: Main Landing Gear Lower Side Strut Installation, TASK 32-11-61-400-802.

CAUTION: MAKE SURE THAT THE HEAD OF THE BOLT POINTS IN THE AFT DIRECTION. IF THE HEAD OF THE BOLT POINTS FORWARD, DAMAGE TO COMPONENTS AND THE AIRPLANE STRUCTURE WILL OCCUR.

- (a) For the bolt that connects the lower side strut to the upper side strut, make sure that the bolt head points in the aft direction.

INDEPENDENT INSPECTION 22-3-25 07:47 GAT 412 CR-2

SUBTASK 32-11-00-200-001

WARNING: MAKE SURE THAT THE SYSTEM-TO-SYSTEM SEPARATIONS ARE CORRECT. SEPARATIONS THAT ARE NOT CORRECT COULD CAUSE UNWANTED CONDITIONS, WHICH COULD INCLUDE CHAFING, FIRE OR ELECTROMAGNETIC INTERFERENCE. THIS CAN CAUSE INJURIES TO PERSONNEL, OR CAN MAKE FLIGHT DANGEROUS, OR CAN CAUSE DAMAGE TO THE SYSTEMS.

(44) Make sure that there is a minimum distance of 0.13 in. (3.30 mm) between the wire bundles [111] and hangar link [112] (Figure 404). 1.5 in

SUBTASK 32-11-00-420-020

(45) Install the hose guide [18].

SUBTASK 32-11-00-410-002

(46) Install the applicable panel:
- (a) For the left wing, do this step:
  Close this access panel:

Number	Name/Location
551BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel

- (b) For the right wing, do this step:

PARTIAL SIGN OFF STATUS:

Item: 18H (44) Completed through item: 18H (44) Sign: 24/3/25 GAT 128

Item: _____ Completed through item: _____ Sign: _____

MR1701

MECH | INSP

Close this access panel:

Number	Name/Location
651BT	Upper Inboard Fixed Trailing Edge, MLG Actuator Access Panel

SUBTASK 32-11-00-420-031

(47) If you will install the right gear, connect the interlock valve cable of the ground spoiler (TASK 27-62-51-400-801).

SUBTASK 32-11-00-420-013

(48) Do this task: Main Landing Gear Wing Door Installation, TASK 32-13-21-420-801.

INDEPENDENT INSPECTION 22.3.24 08:20 GAT 412

SUBTASK 32-11-00-420-028

(49) Do this task: Main Landing Gear Door Assembly Installation, TASK 32-13-11-420-803.

SUBTASK 32-11-00-420-032

(50) Do this task: Main Landing Gear Brake Installation, TASK 32-41-41-400-801.

SUBTASK 32-11-00-420-033

(51) Do this task: Main Landing Gear Wheel and Tire Assembly Installation, TASK 32-45-11-400-801.

SUBTASK 32-11-00-410-001

(52) Install the applicable panel:

(a) For the left landing gear, do this step:

Close these access panels:

Number	Name/Location
551BB	Lower Inboard Fixed Trailing Edge, Gear Adjustment Door
551CB	Lower Inboard Fixed Trailing Edge, Gear Access Panel

(b) For the right landing gear, do this step:

Close these access panels:

Number	Name/Location
651BB	Lower Inboard Fixed Trailing Edge, Gear Door Adjustment
651CB	Lower Inboard Fixed Trailing Edge, Gear Access Panel

(c) Close these access panels:

Number	Name/Location
551DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel
651DB	Lower Inboard Fixed Trailing Edge, Lube Actuator & MLG Beam Outboard Attach Pin Access Panel

I. Put the Airplane Back to Its Usual Condition

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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MR 0742	22.3.25
	27.8.25
	27.8.25

SUBTASK 32-11-00-080-013

(1) Remove the ground spoiler actuator lock lock set, SPL-1744 .

SUBTASK 32-11-00-860-015

(2) Remove the safety tags and close these circuit breakers:
F/O Electrical System Panel, P6-3

Row	Col	Number	Name
A	16	(C01345)	LANDING GEAR AUTOBRAKE BITE CONT 2
A	18	(C00583)	LANDING GEAR AUTOBRAKE BITE CONT 1
B	16	(C01346)	LANDING GEAR PARKING BRAKE
B	17	(C00129)	LANDING GEAR LATCH & PRESS WARN
C	15	(C01355)	LANDING GEAR AIR/GND SYS 2
C	16	(C01356)	LANDING GEAR AIR/GND SYS 1
D	1	(C01399)	PSEU PRI
D	2	(C01400)	PSEU ALTN
D	15	(C01401)	LANDING GEAR AIR/GND RELAY
D	16	(C01432)	LANDING GEAR ALTN EXTEND SOL
E	16	(C00196)	LANDING GEAR ANTISKID INBD
E	18	(C00195)	LANDING GEAR ANTISKID OUTBD

SUBTASK 32-11-00-860-039

(3) Remove the safety tag and close this circuit breaker:
CAPT Electrical System Panel, P18-3

Row	Col	Number	Name
E	4	(C00700)	HEATERS DRAIN MAST AIR

SUBTASK 32-11-00-860-009

(4) Do this task: Trailing Edge Flap System Reactivation, TASK 27-51-00-440-801.

SUBTASK 32-11-00-860-010

(5) Do this task: Retract the Trailing Edge Flaps, TASK 27-51-00-860-804.

SUBTASK 32-11-00-860-020

WARNING: MAKE SURE THAT PERSONS AND EQUIPMENT ARE CLEAR OF ALL CONTROL SURFACES BEFORE YOU SUPPLY HYDRAULIC POWER. AILERONS, RUDDERS, ELEVATORS, FLAPS, SLATS, SPOILERS, LANDING GEAR, AND THRUST REVERSERS CAN MOVE QUICKLY

PARTIAL SIGN OFF STATUS:

Item: 18 Completed through item: 1 (1) Sign: 22.3.25

Item: _____ Completed through item: _____ Sign: _____

MECH	INSP

WHEN YOU SUPPLY HYDRAULIC POWER. THIS CAN CAUSE INJURY TO PERSONS AND DAMAGE TO EQUIPMENT.

(6) Do this task: Speed Brake Hydraulic Systems A and B Pressurization, TASK 27-62-00-800-801.

SUBTASK 32-11-00-860-021

(7) Move the speed brake lever to the DOWN position to lower the spoilers.

SUBTASK 32-11-00-860-032

(8) Do this task: Put the Speed Brake Hydraulic systems A and B Back to the Condition Before the Pressurization, TASK 27-62-00-840-801.

SUBTASK 32-11-00-820-001

(9) If you changed the length of one or more of the door rods, do the steps that follows:

NOTE: No adjustment to the doors is necessary if the door rods were not changed.

(a) Adjust the shock strut doors of the main landing gear (TASK 32-13-11-820-801).

(b) Adjust the outer door of the main landing gear (TASK 32-13-21-000-801).

SUBTASK 32-11-00-610-001

(10) Lubricate the main landing gear, do these tasks:

- Main Landing Gear Upper End Components Servicing, TASK 12-21-11-640-801
- Main Landing Gear Lower End Components Servicing, TASK 12-21-11-640-802.

SUBTASK 32-11-00-619-001

(11) To fully service the shock strut with the airplane on jacks, do these steps to service the shock strut with fluid:

NOTE: Hold the shock strut in a vertical position to do these steps.

(a) Make sure that the shock strut is fully deflated, refer to service placard for the fully deflated dimension (Figure 405).

(b) Fill the shock strut with fluid (TASK 12-15-31-610-805).

(c) Close the valves and install their caps.

SUBTASK 32-11-00-619-002

(12) To inflate the shock strut with the airplane on jacks, do these steps:

NOTE: To do these steps, you must have accomplished the steps to add fluid to the strut, at the start of this task.

(a) Install the tool, SPL-1521, on the gas valve.

(b) Inflate the shock strut with G00018 nitrogen , until you reach the fully extended pressure, refer to the servicing placard for the correct pressure (Figure 405).

NOTE: If nitrogen is not available, you can use G02314 air , as an alternative, to inflate the shock strut. N/A

(c) Remove the tool, SPL-1521, and install the cap on the gas valve.

SUBTASK 32-11-00-619-003

(13) If you did not service the strut with fluid, at the start of this task, do this task: Main Landing Gear Shock Strut Servicing, Airplane on the Ground, TASK 12-15-31-610-802. N/A

SUBTASK 32-11-00-080-002

(14) If the downlock pins are installed on all the landing gear, do this task: Landing Gear Downlock Pins Removal, TASK 32-00-01-080-801.

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

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✓	✓ 27.3.24
✓	✓ 27.3.24
✓	✓ 27.3.24
✓ MAINT	✓ 27.3.24

SUBTASK 32-11-00-710-002

(15) Do this task: Main Landing Gear Operational Test, TASK 32-32-00-710-801.

NOTE: It is optional to perform the main landing gear test - component replacement, instead of the operational test if a hydraulic cart is not available (TASK 32-32-00-710-802).

SUBTASK 32-11-00-710-003

(16) Do this task: Main Gear Manual Extension System Test - Airplane on Jacks, TASK 32-34-00-730-801.

SUBTASK 32-11-00-480-003

WARNING: MAKE SURE THE DOWNLOCK PINS ARE INSTALLED ON ALL THE LANDING GEAR. WITHOUT THE DOWNLOCK PINS, THE LANDING GEAR COULD RETRACT AND CAUSE INJURIES TO PERSONS AND DAMAGE TO EQUIPMENT.

(17) If the downlock pins are not installed on all the landing gear, do this task: Landing Gear Downlock Pins Installation, TASK 32-00-01-480-801.

SUBTASK 32-11-00-580-001

(18) Do this task: Lower the Airplane Off the Jacks, TASK 07-11-01-580-816.

SUBTASK 32-11-00-860-011

(19) For hydraulic systems A and B, do this task: Hydraulic System A or B Pressurization, TASK 29-11-00-860-801.

SUBTASK 32-11-00-870-001

(20) Bleed the brake hydraulic system, do this task: Normal (System B) Hydraulic Brake System - Bleeding, TASK 32-41-00-870-802.

SUBTASK 32-11-00-790-001

(21) Examine the connections of the hydraulic brake line for leakage.

SUBTASK 32-11-00-870-003

(22) Bleed the shimmy damper, do this task: Hydraulic Shimmy Damper - Bleeding, TASK 32-11-81-870-801.

SUBTASK 32-11-00-790-002

(23) Examine the connections of the shimmy damper for leakage.

SUBTASK 32-11-00-710-004

(24) Do this task: Antiskid Valve Operational Test, TASK 32-42-00-720-803.

SUBTASK 32-11-00-710-001

(25) Do a test of the operation of the antiskid transducer (TASK 32-42-00-710-802).

SUBTASK 32-11-00-610-002

(26) Check the pressure and extension of the shock strut [1].

(a) Measure the pressure and extension of the shock strut [1].

(b) Make sure that the pressure you measure is still correct for the extension of the shock strut [1] (Figure 405).

(c) If the pressure is not correct for the extension of the shock strut [1], then the shock strut [1] does not have the proper amount of fluid and you must fully service the shock strut [1] (TASK 12-15-31-610-802).

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

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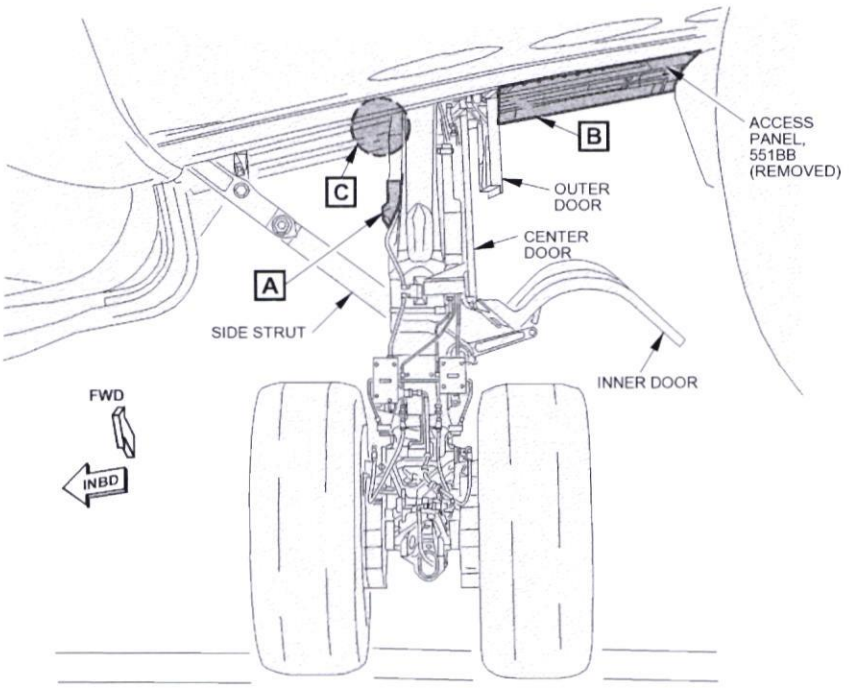
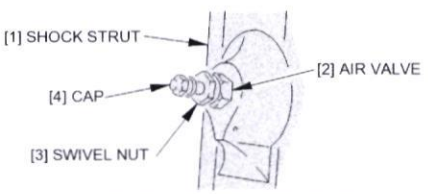


LEFT MAIN LANDING GEAR
(RIGHT MAIN LANDING GEAR IS EQUIVALENT)



SHOCK STRUT INFLATION VALVE
A

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Figure 401. Main Landing Gear Installation - Sheet 1
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

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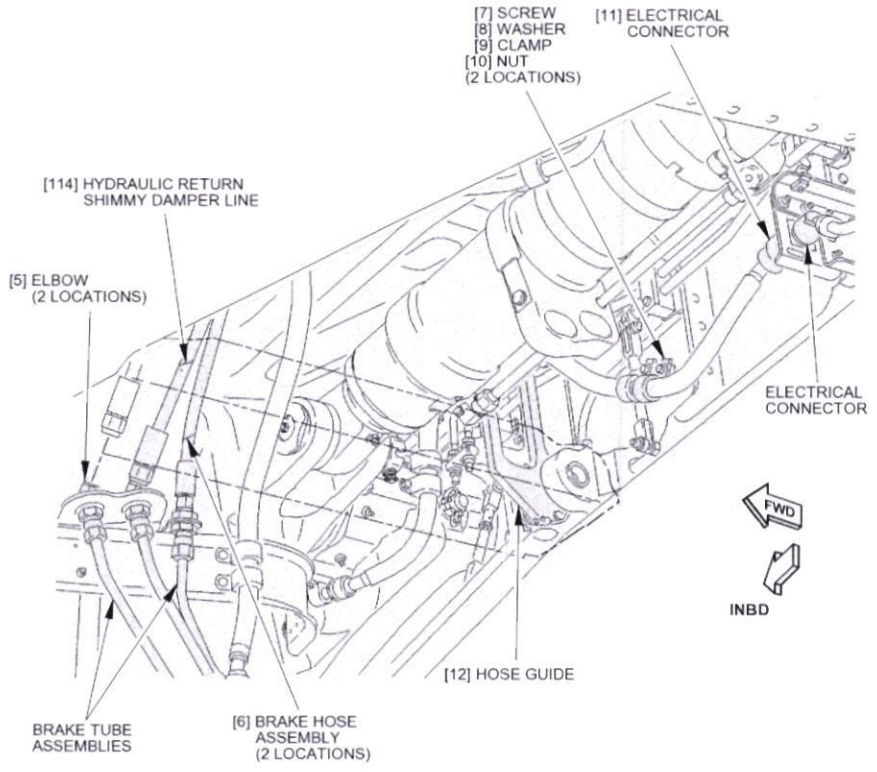

HYDRAULIC LINES AND ELECTRICAL CONNECTORS

B

G02341 S0006574756_V3

Figure 401. Main Landing Gear Installation - Sheet 2
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

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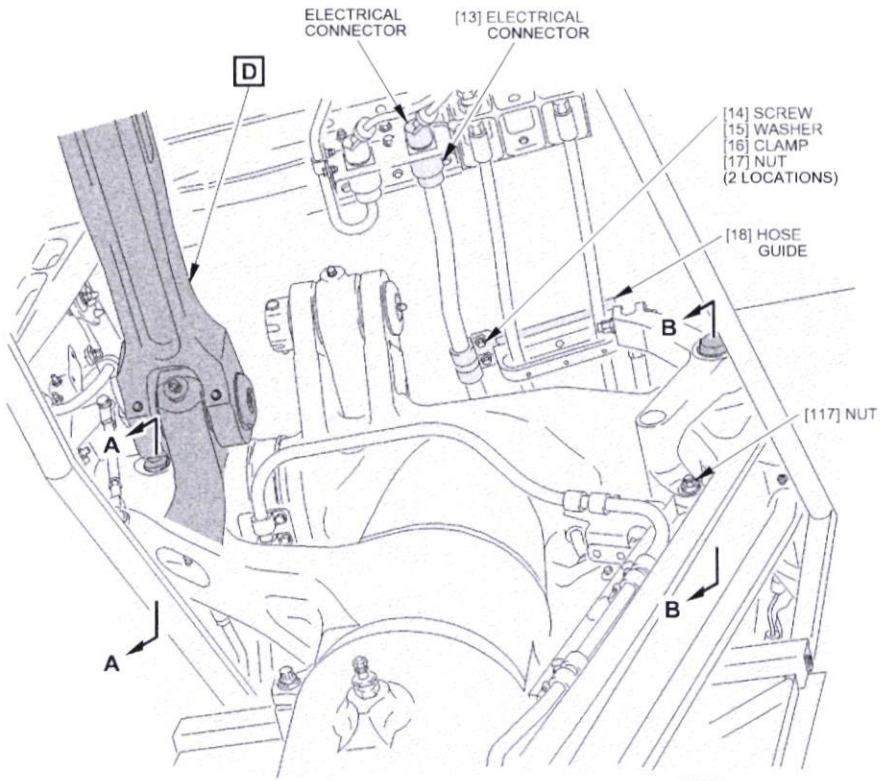

ELECTRICAL CONNECTOR AND TRUNNION SUPPORTS

C

G02347 S0006574757_V6

Figure 401. Main Landing Gear Installation - Sheet 3
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 004, 005, 007-009, 014 PRE SB 737-32-1448

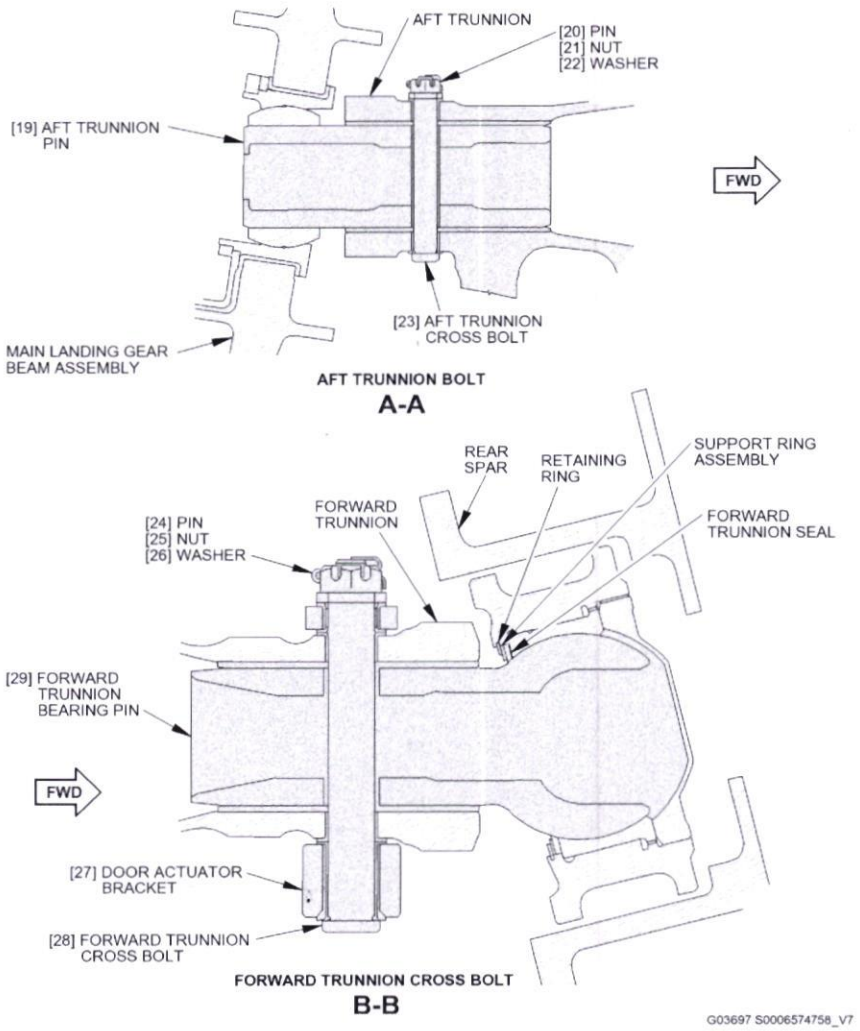

Figure 401. Main Landing Gear Installation - Sheet 4
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448

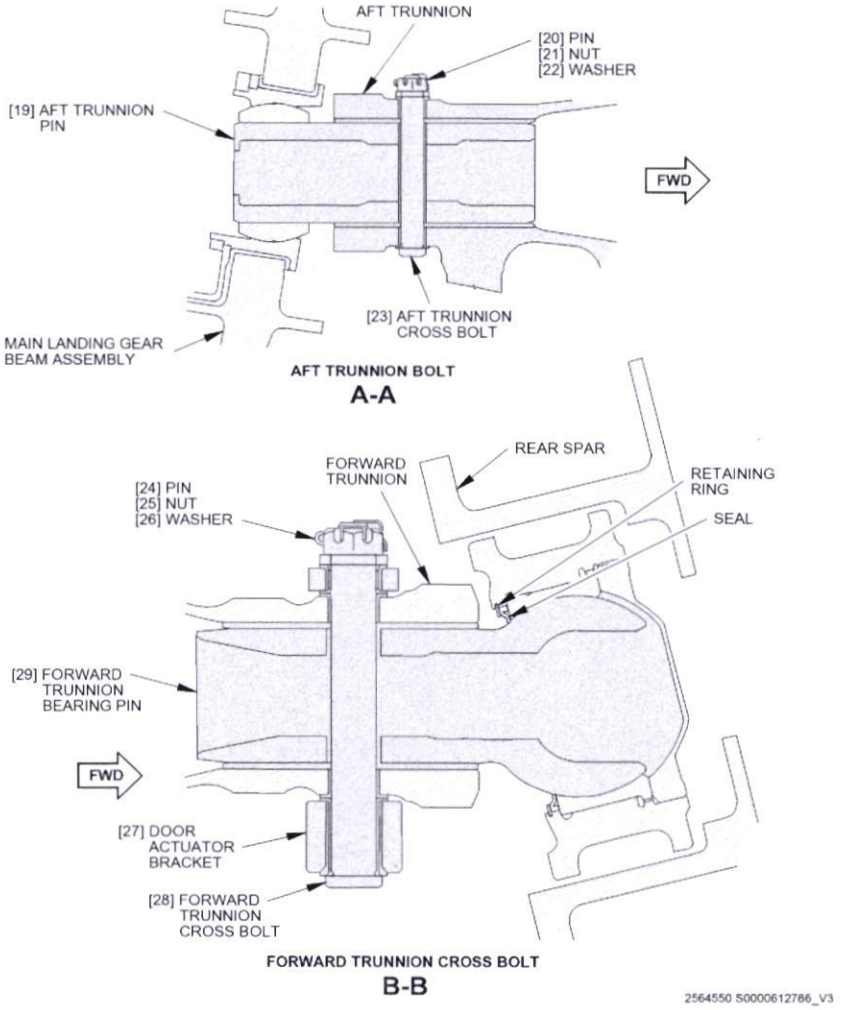

Figure 401. Main Landing Gear Installation - Sheet 5
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

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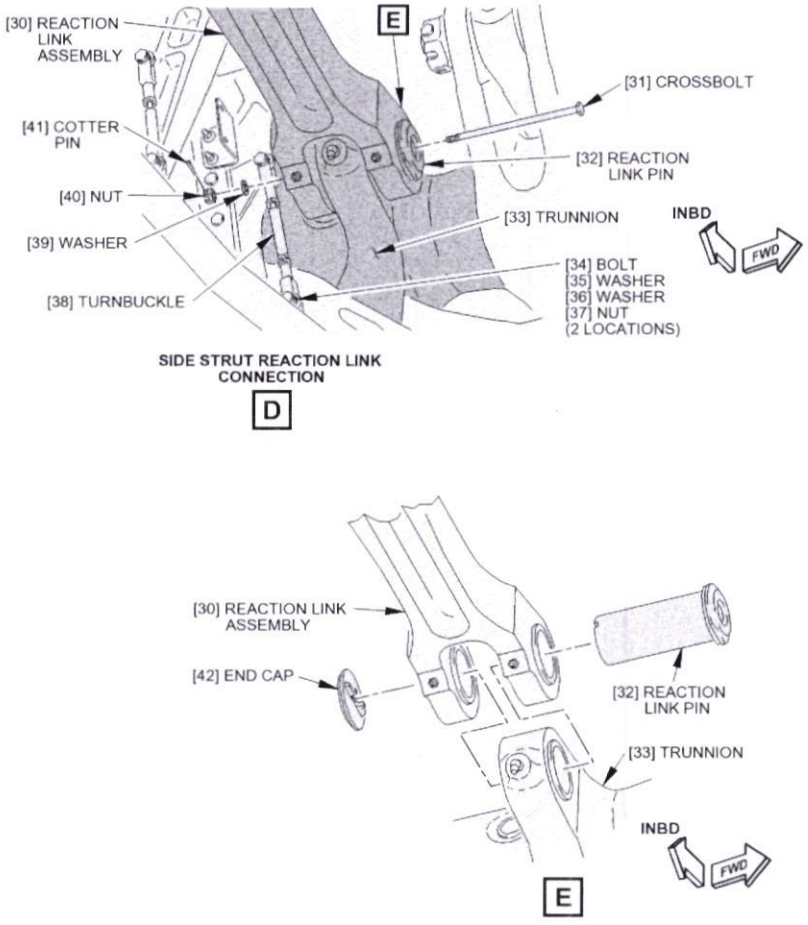

Figure 401. Main Landing Gear Installation - Sheet 6
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

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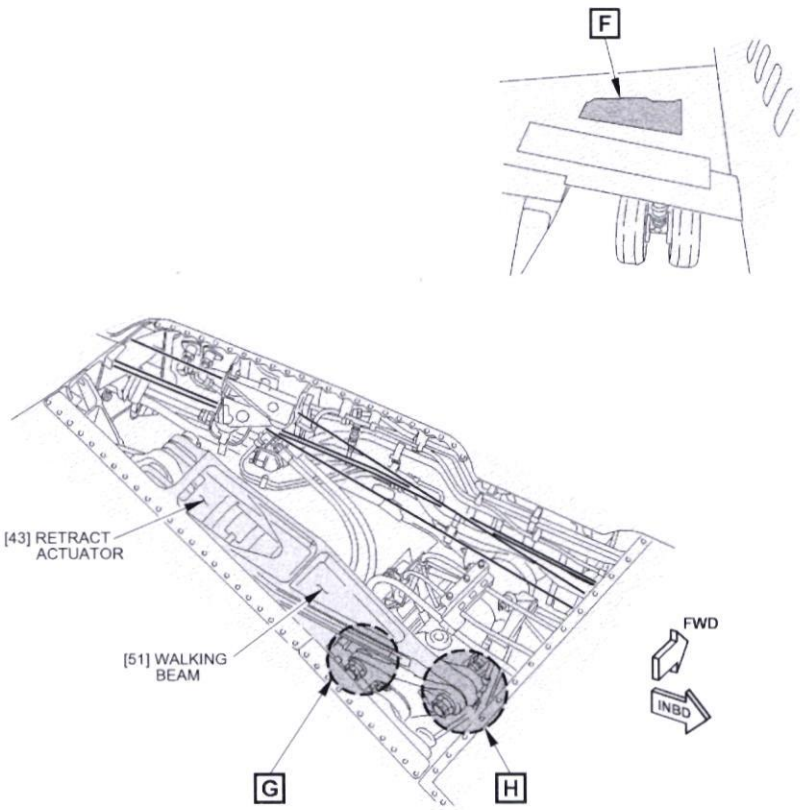

Figure 401. Main Landing Gear Installation - Sheet 7
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

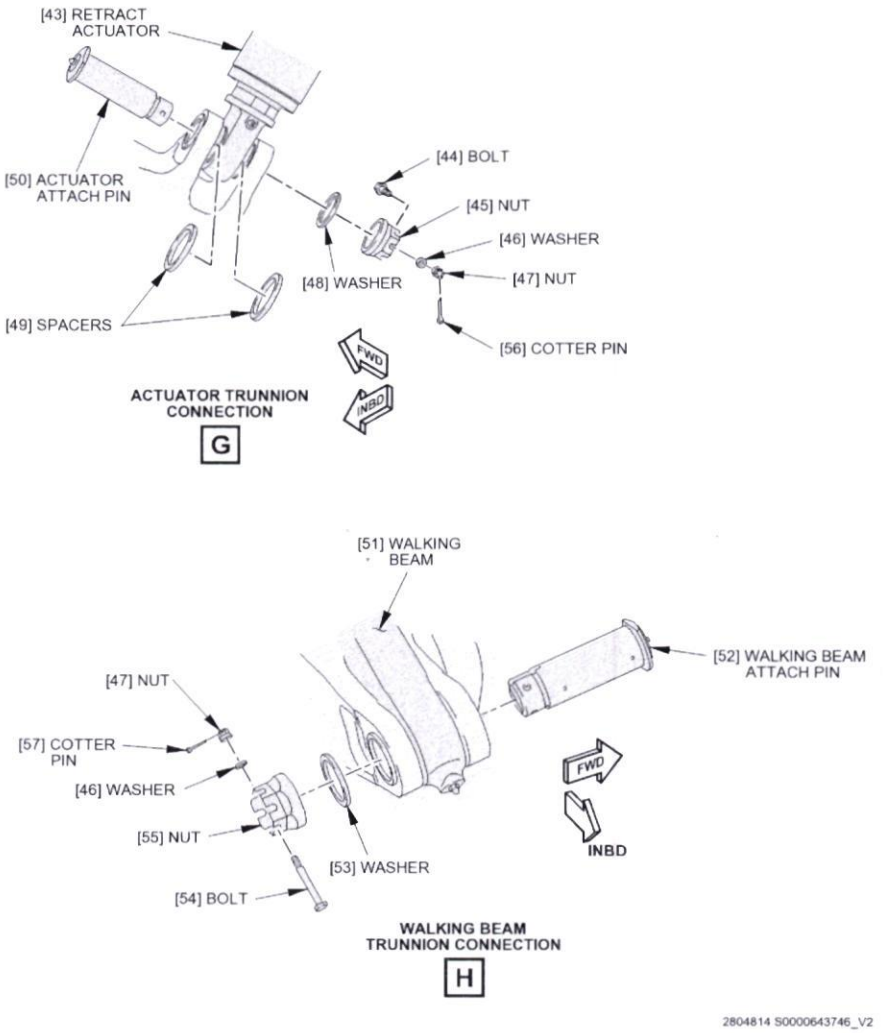

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Figure 401. Main Landing Gear Installation - Sheet 8
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

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Item: _____ Completed through item: _____ Sign: _____

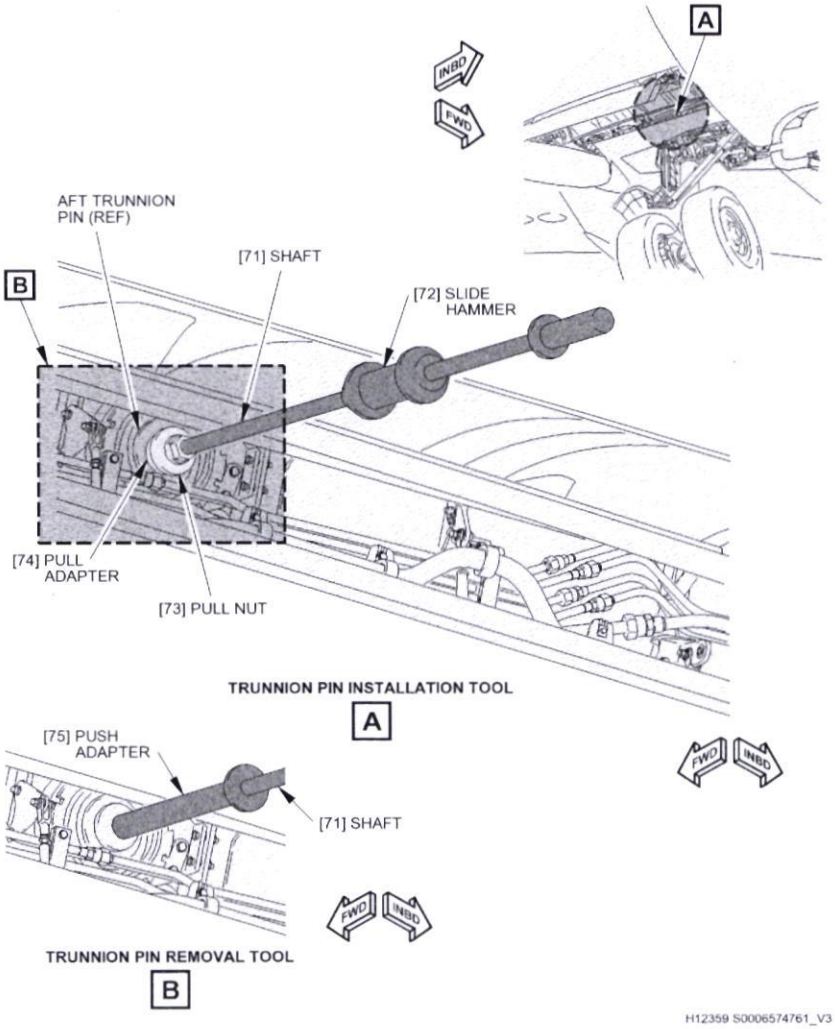

Figure 402. Main Landing Gear Trunnion Pin Puller Equipment - Sheet 1
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

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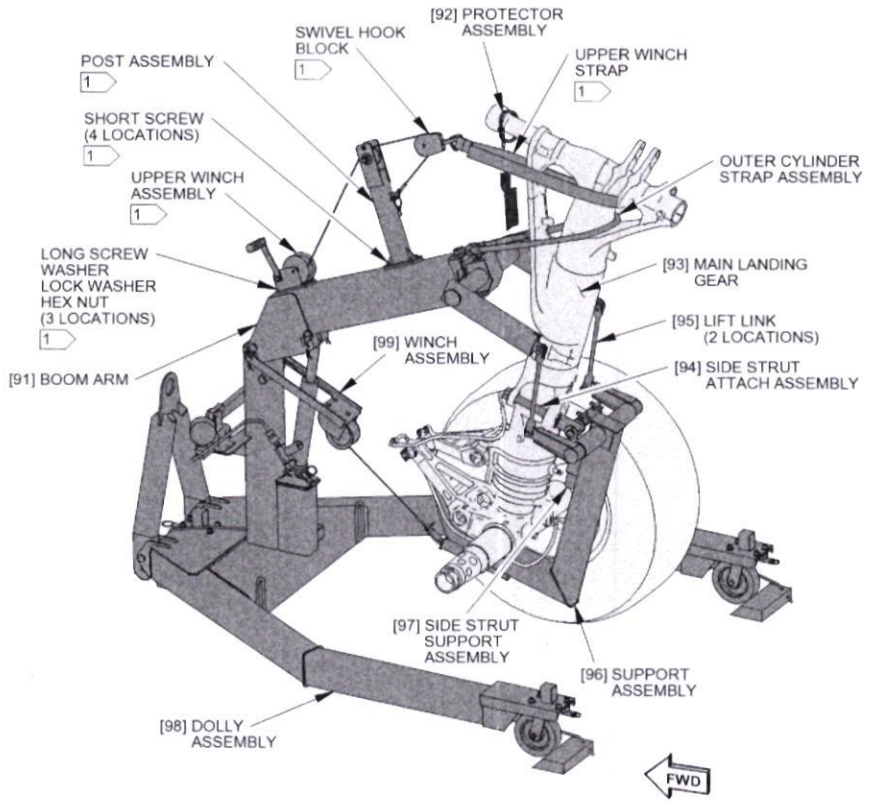
Item: _____ Completed through item: _____ Sign: _____



1 REMOVABLE PARTS TO SUPPORT MLG INSTALLATION. NOT NEEDED DURING MLG REMOVAL.

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Figure 403. Main Landing Gear Support Equipment - Sheet 1
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

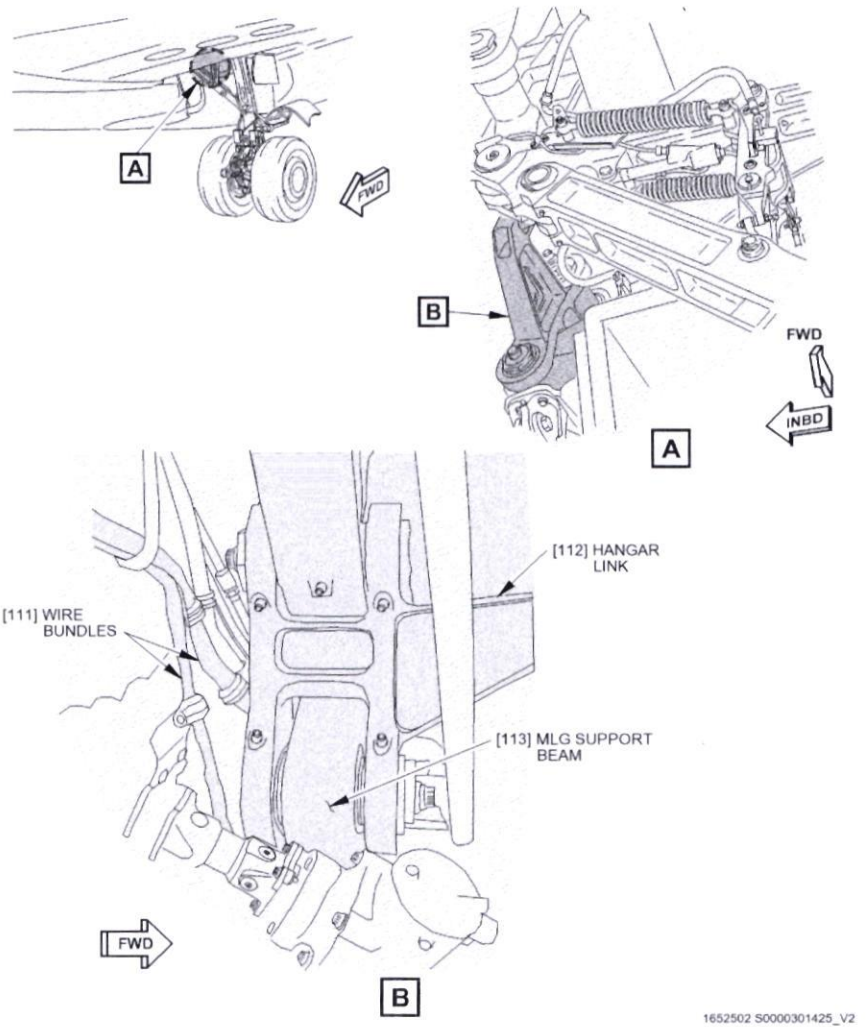
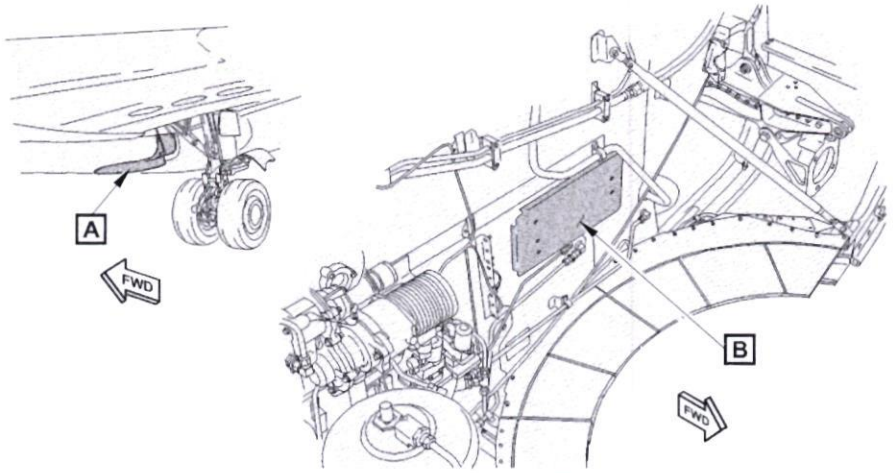


Figure 404. Main Landing Gear Clearance Requirements - Sheet 1
TASK 32-11-00-000-801

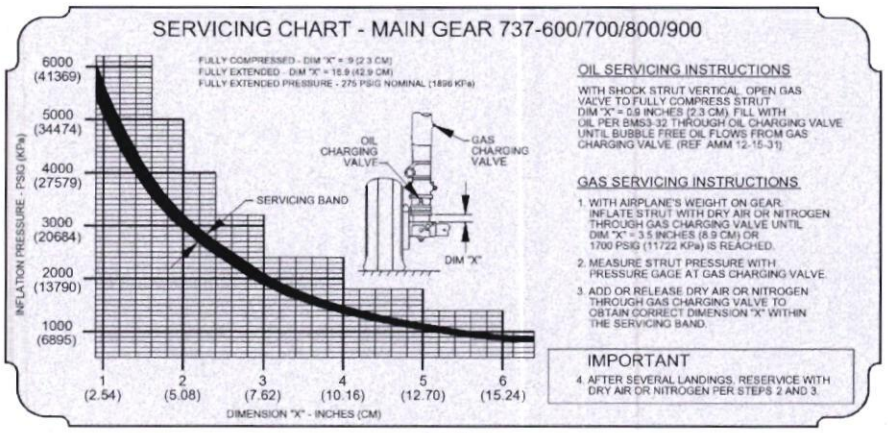
PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



A



B

F84899 S0006561283_V4

20.3.25

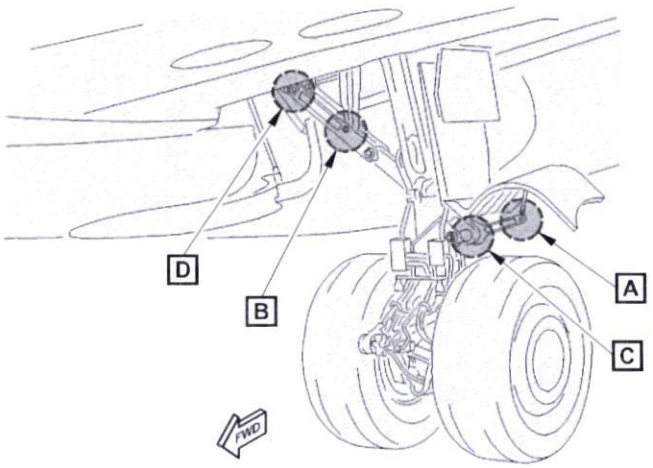
GAT 492

Figure 405. Main Landing Gear Shock Strut Servicing Chart - Sheet 1
TASK 32-11-00-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



LEFT MAIN LANDING GEAR
(RIGHT MAIN LANDING GEAR IS EQUIVALENT)

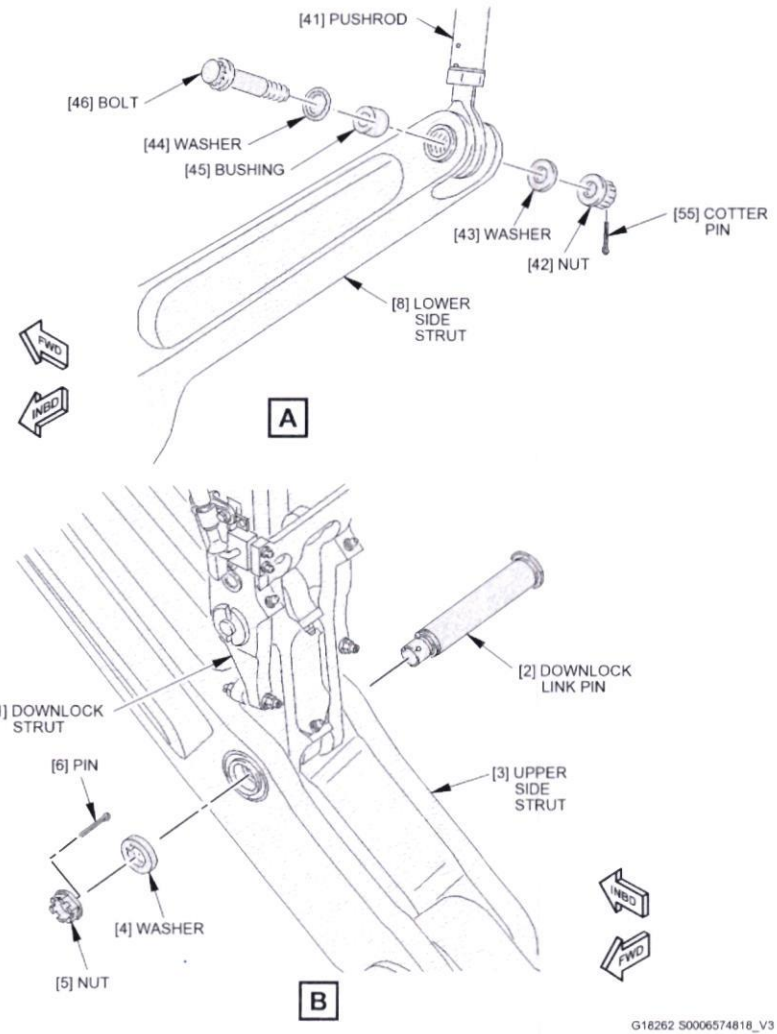
G18251 S0006574817_V2

Figure 403. Main Landing Gear Side Strut Installation - Sheet 1
TASK 32-11-61-000-803

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



20.3.24

GAT 492

Figure 403. Main Landing Gear Side Strut Installation - Sheet 2
TASK 32-11-61-000-803

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

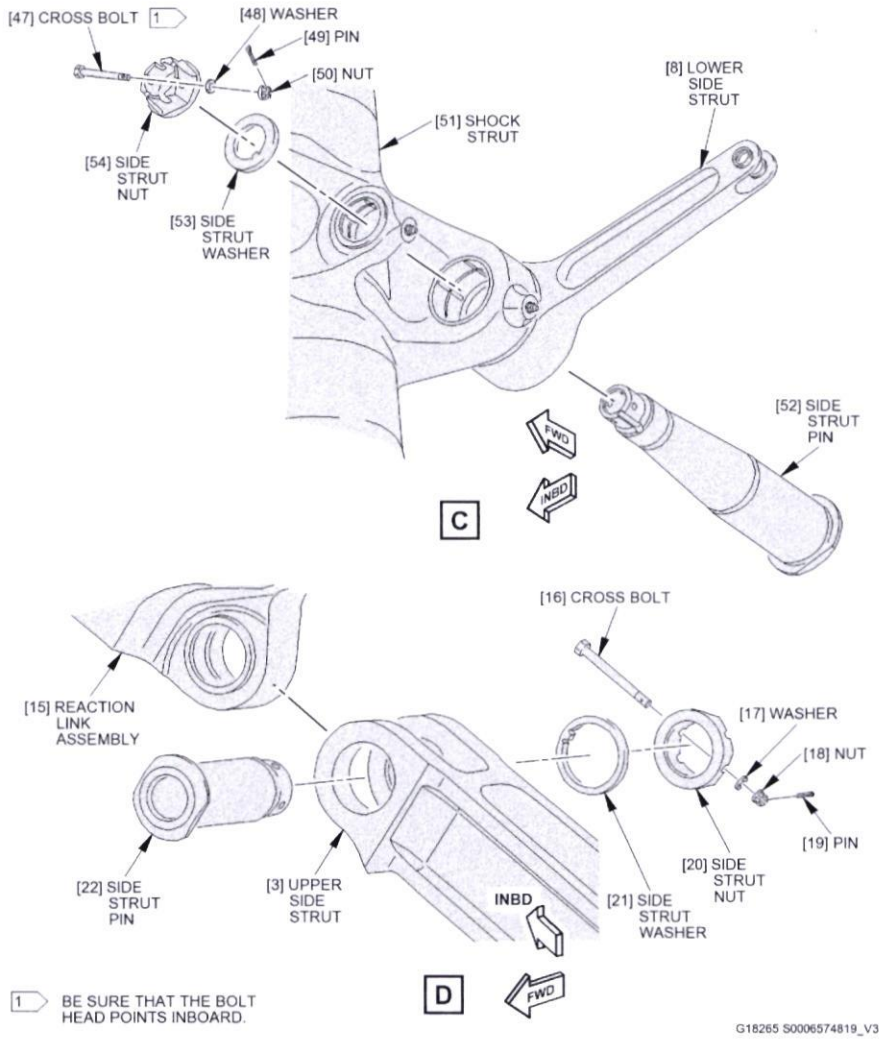


Figure 403, Main Landing Gear Side Strut Installation - Sheet 3
TASK 32-11-61-000-803

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

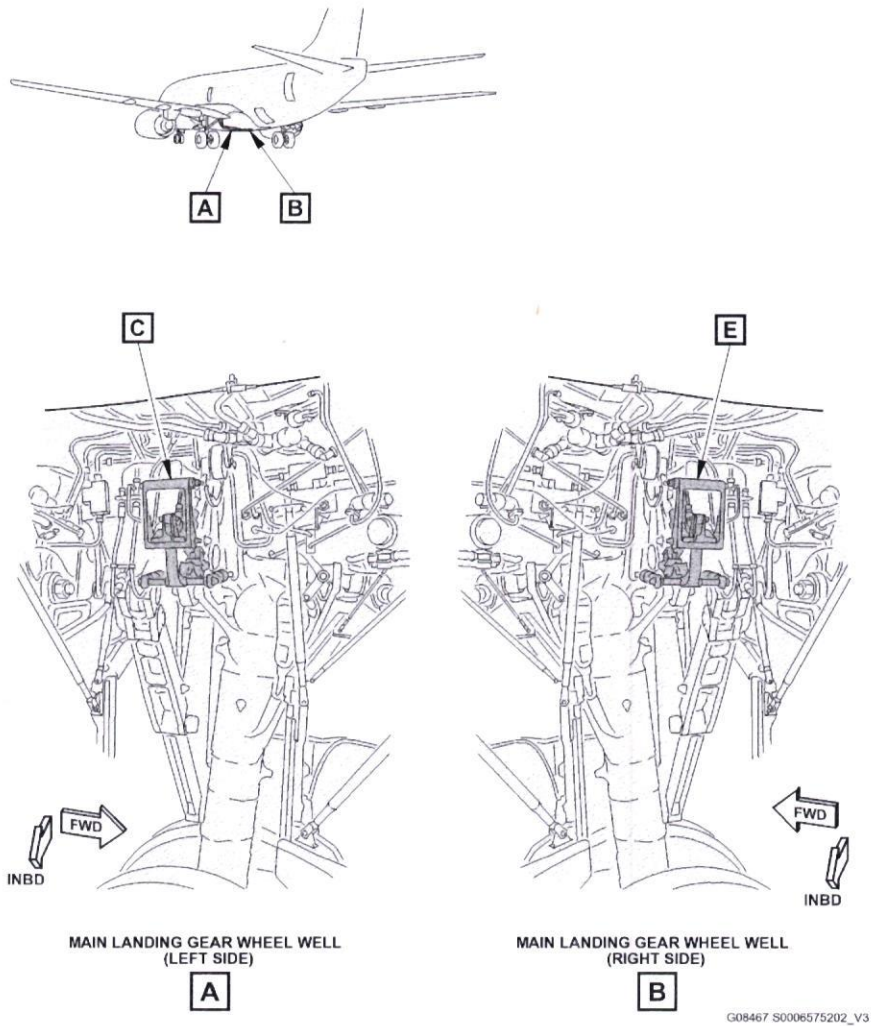


Figure 401. Main Landing Gear Uplock Assembly Installation - Sheet 1
TASK 32-32-31-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

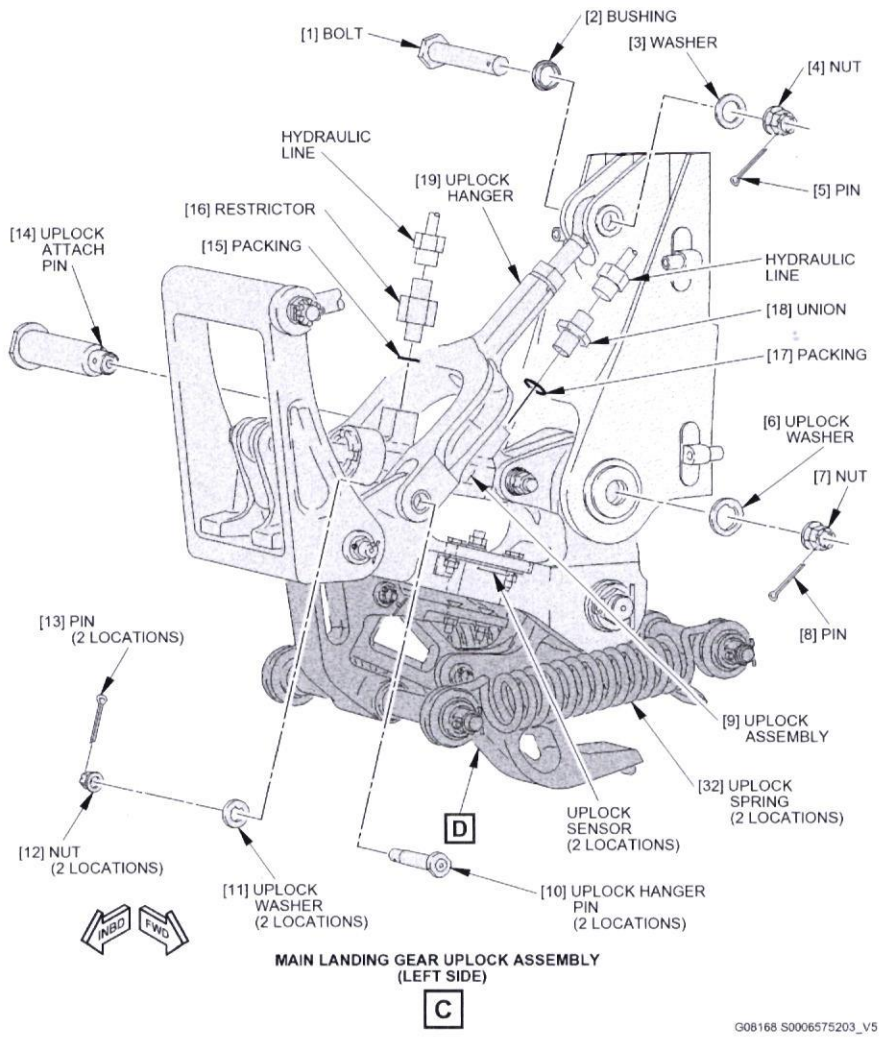
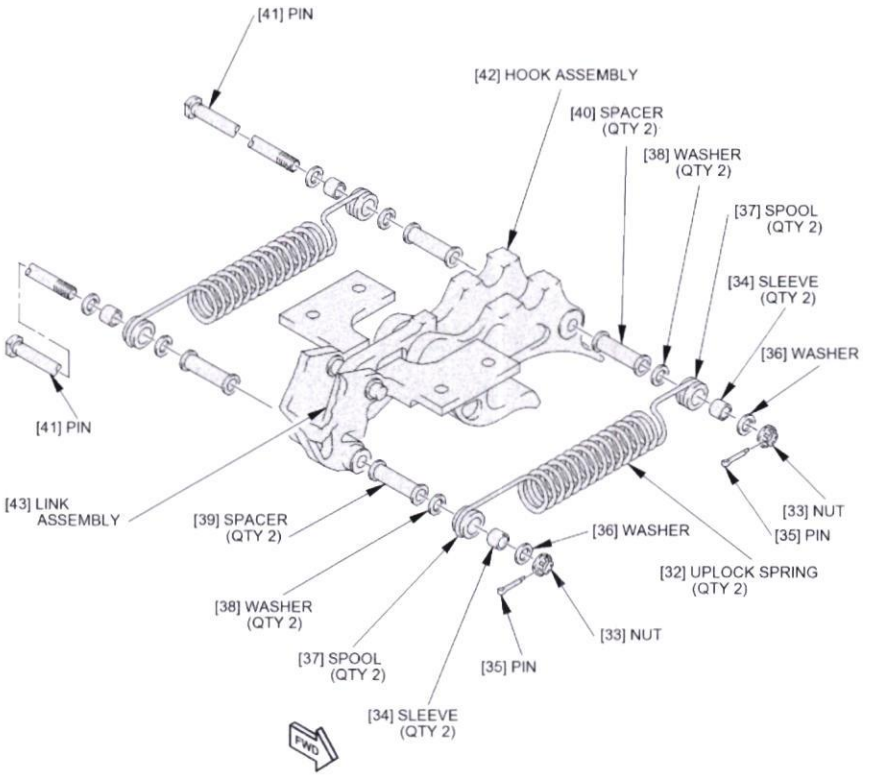


Figure 401. Main Landing Gear Uplock Assembly Installation - Sheet 2
TASK 32-32-31-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



20.8.25

GAT 492

Figure 401. Main Landing Gear Uplock Assembly Installation - Sheet 3
TASK 32-32-31-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

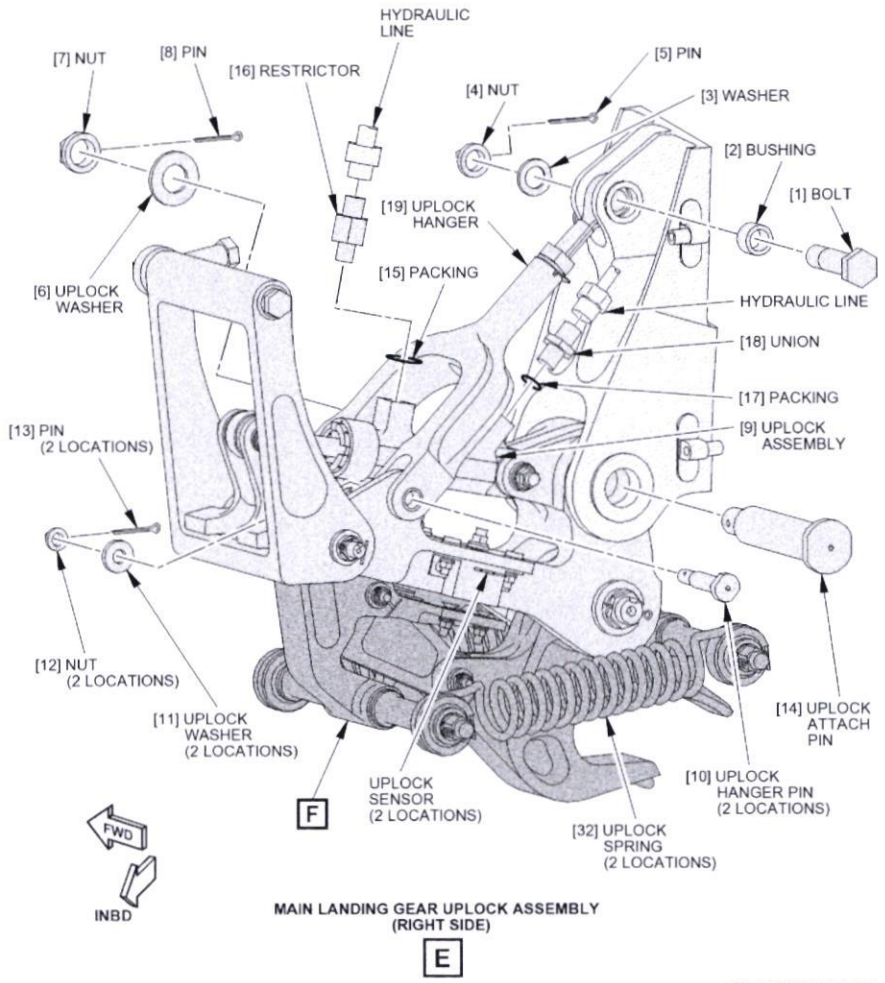
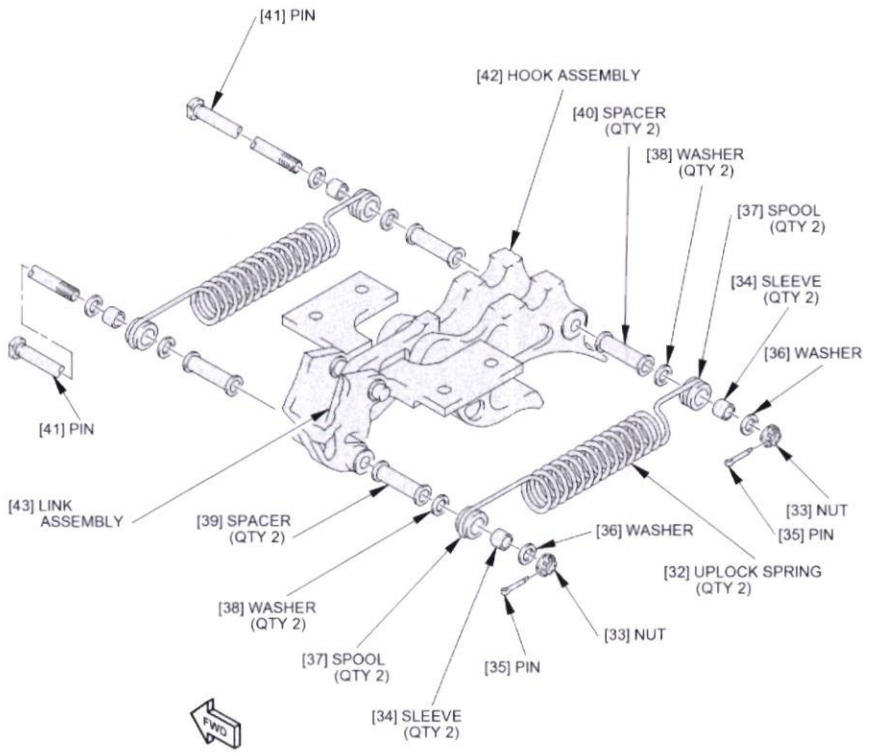


Figure 401. Main Landing Gear Uplock Assembly Installation - Sheet 4
TASK 32-32-31-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



20.3.25

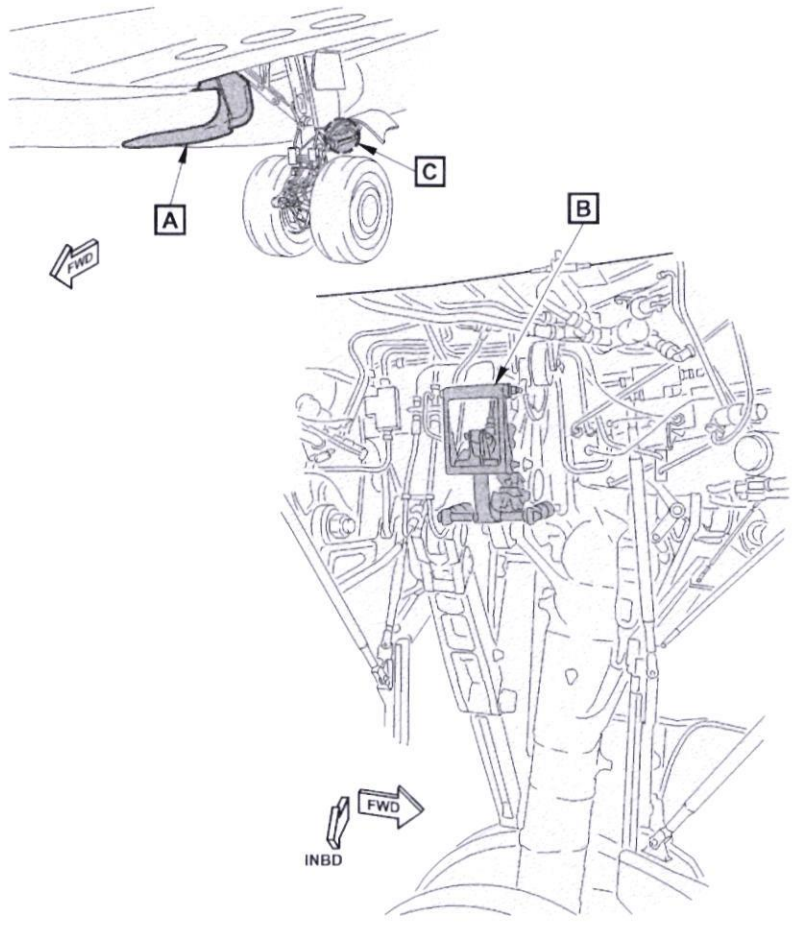
GAT 492

Figure 401. Main Landing Gear Uplock Assembly Installation - Sheet 5
TASK 32-32-31-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



MAIN LANDING GEAR WHEEL WELL
(LEFT SIDE IS SHOWN, RIGHT SIDE IS EQUIVALENT)
A

M50911 S0006575204_V2

Figure 402. Main Landing Gear Uplock Assembly Adjustment - Sheet 1
TASK 32-32-31-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

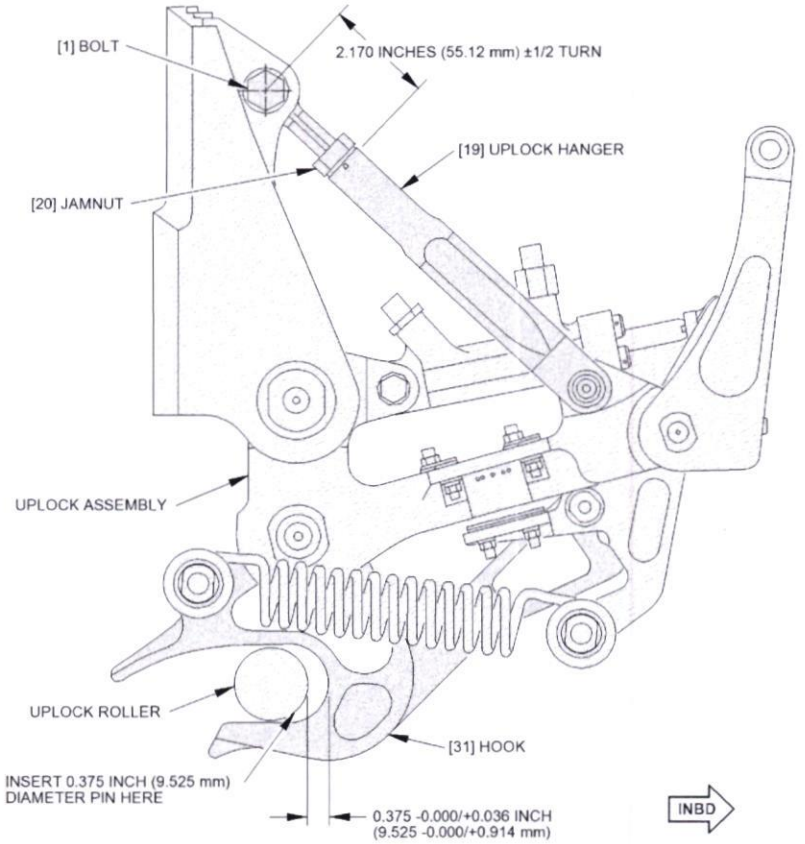


Figure 402. Main Landing Gear Uplock Assembly Adjustment - Sheet 2
TASK 32-32-31-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

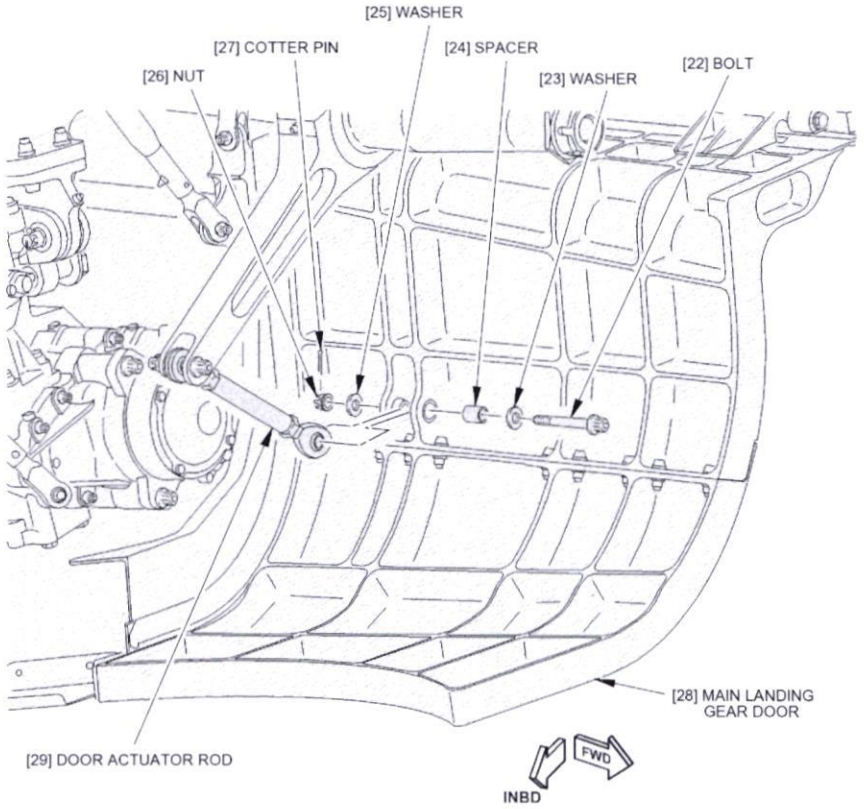


Figure 402. Main Landing Gear Uplock Assembly Adjustment - Sheet 3
TASK 32-32-31-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

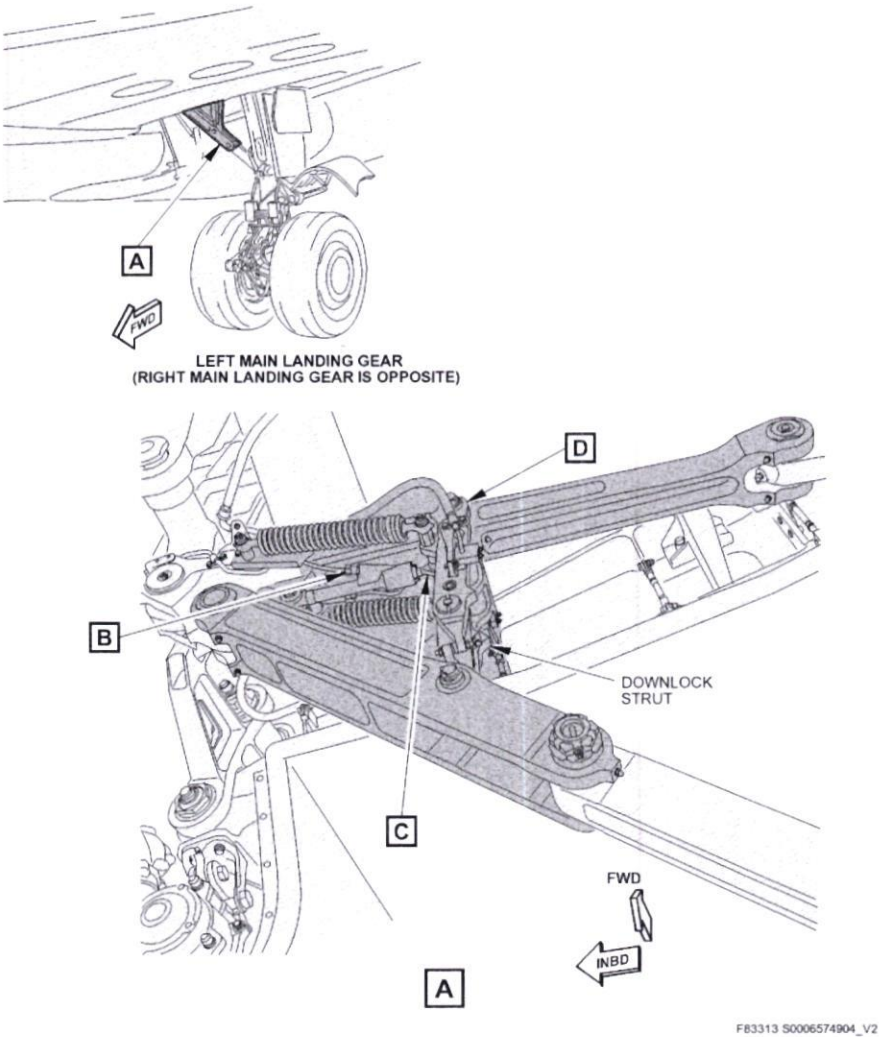
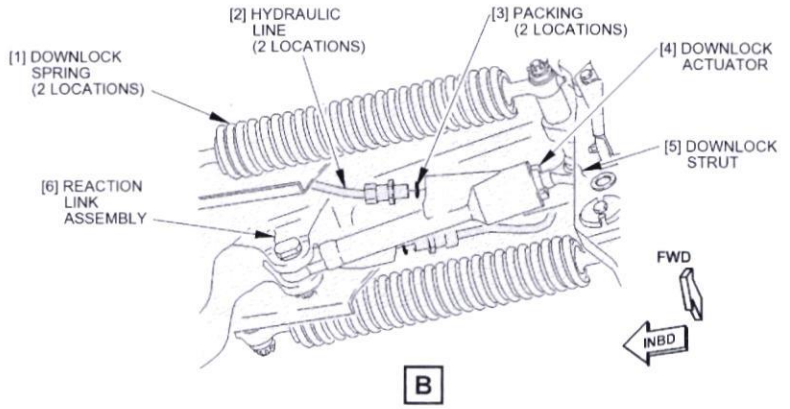


Figure 401. Main Landing Downlock Strut Installation - Sheet 1
TASK 32-11-89-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



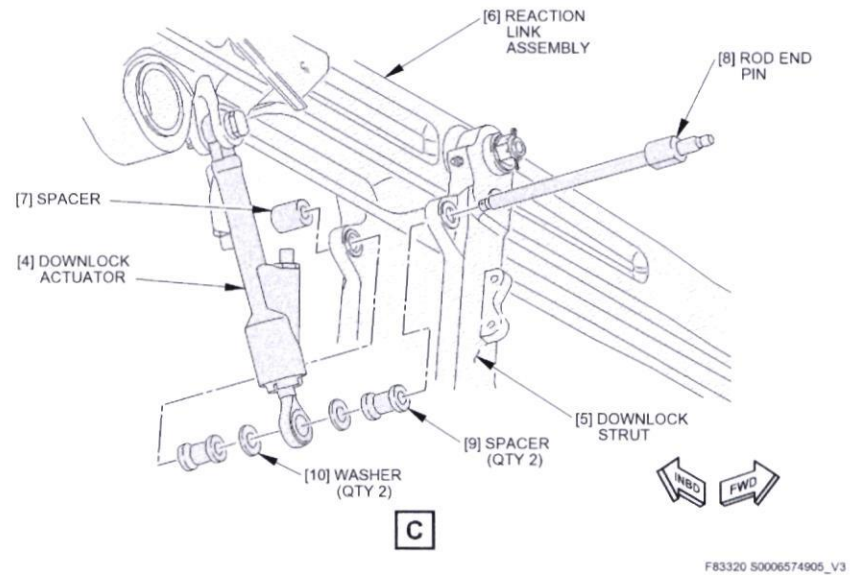


Figure 401. Main Landing Downlock Strut Installation - Sheet 2
TASK 32-11-89-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

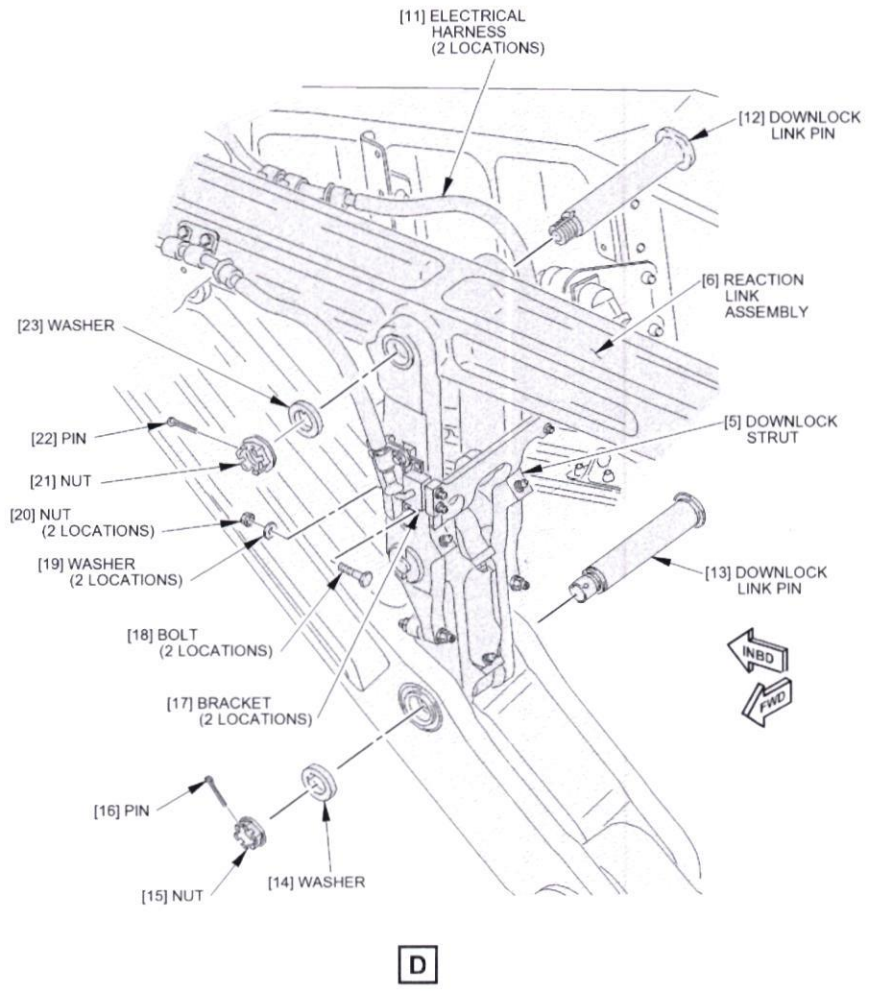
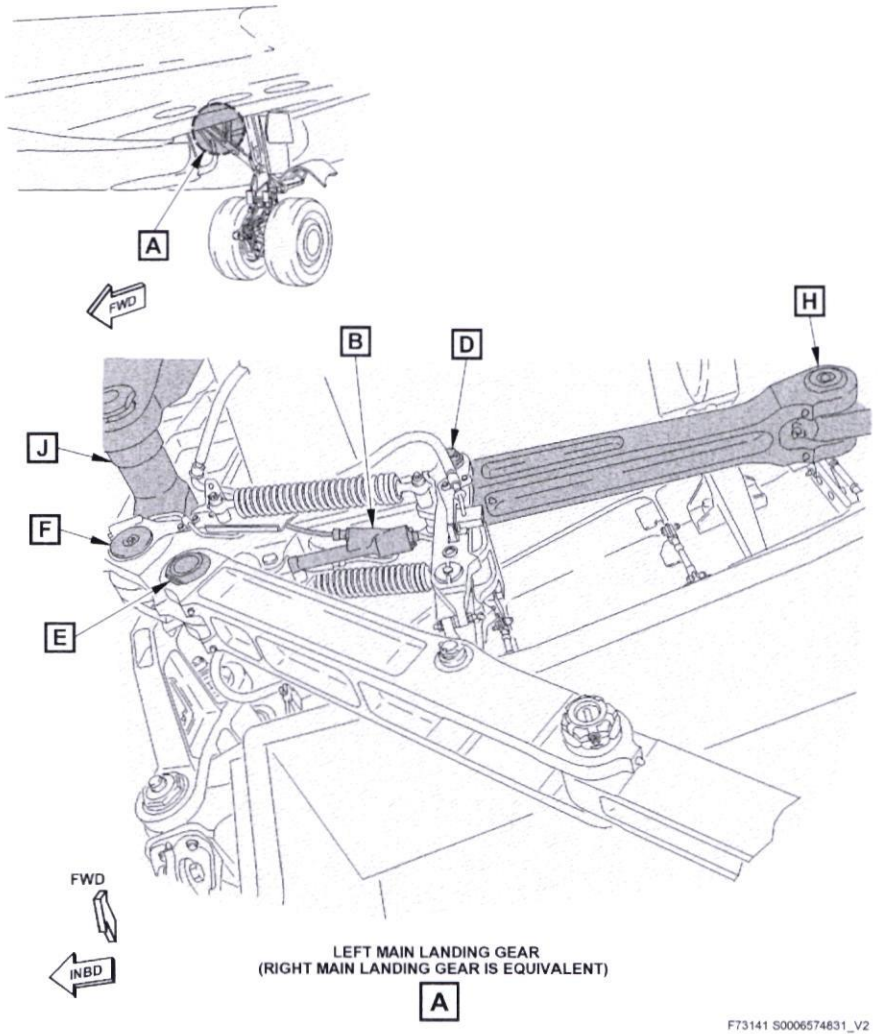


Figure 401. Main Landing Downlock Strut Installation - Sheet 3
TASK 32-11-89-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



LEFT MAIN LANDING GEAR
(RIGHT MAIN LANDING GEAR IS EQUIVALENT)

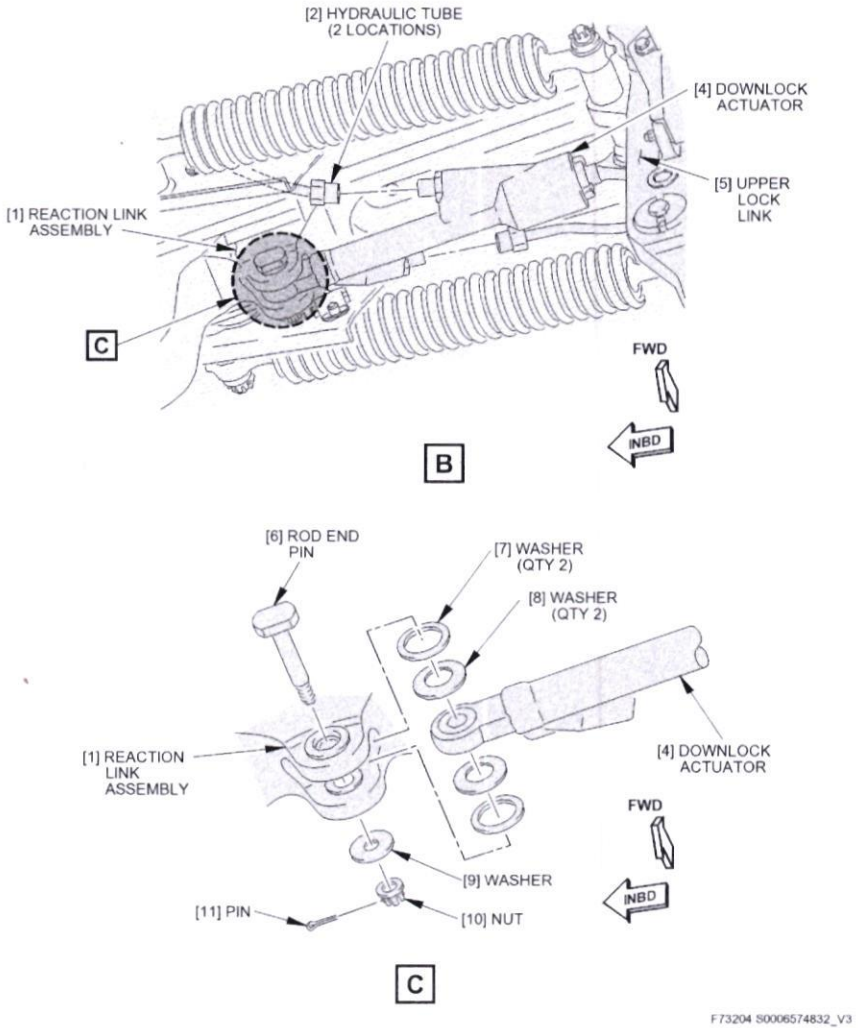
F73141 S0006574831_V2

Figure 401. Main Landing Gear Reaction Link Assembly Installation - Sheet 1
TASK 32-11-71-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



20-3-25

GAT 492

Figure 401. Main Landing Gear Reaction Link Assembly Installation - Sheet 2
TASK 32-11-71-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

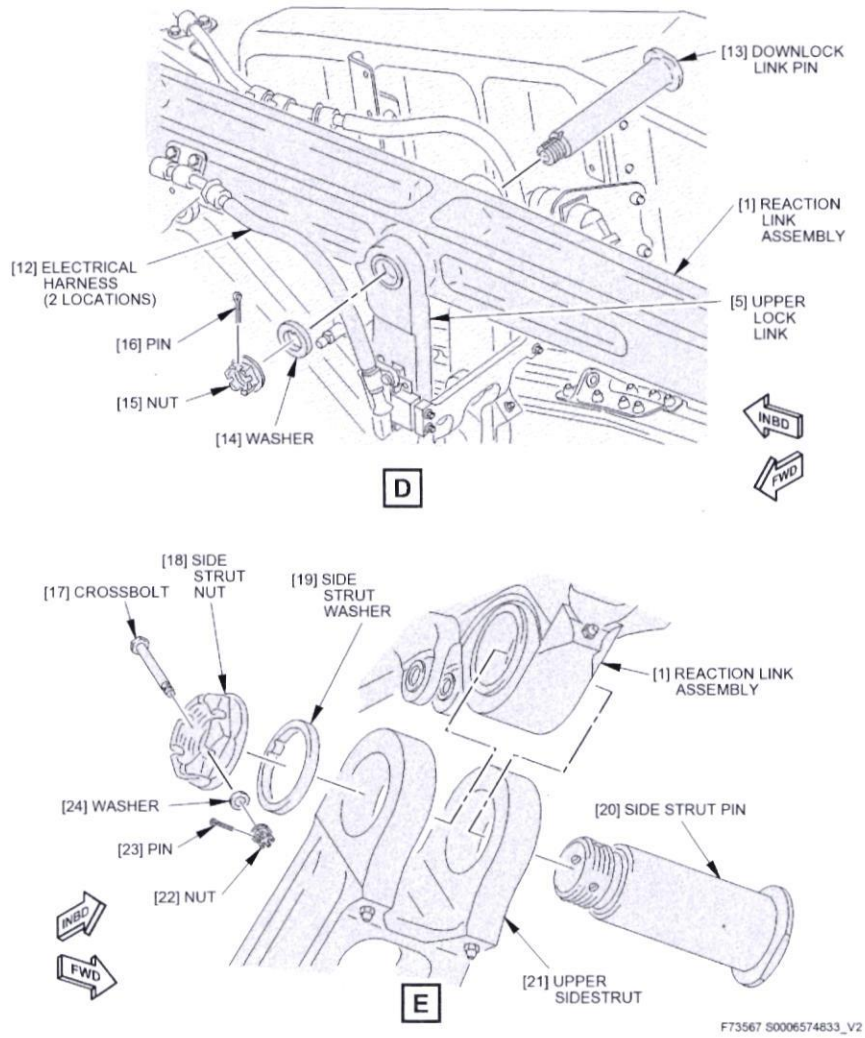


Figure 401. Main Landing Gear Reaction Link Assembly Installation - Sheet 3
TASK 32-11-71-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

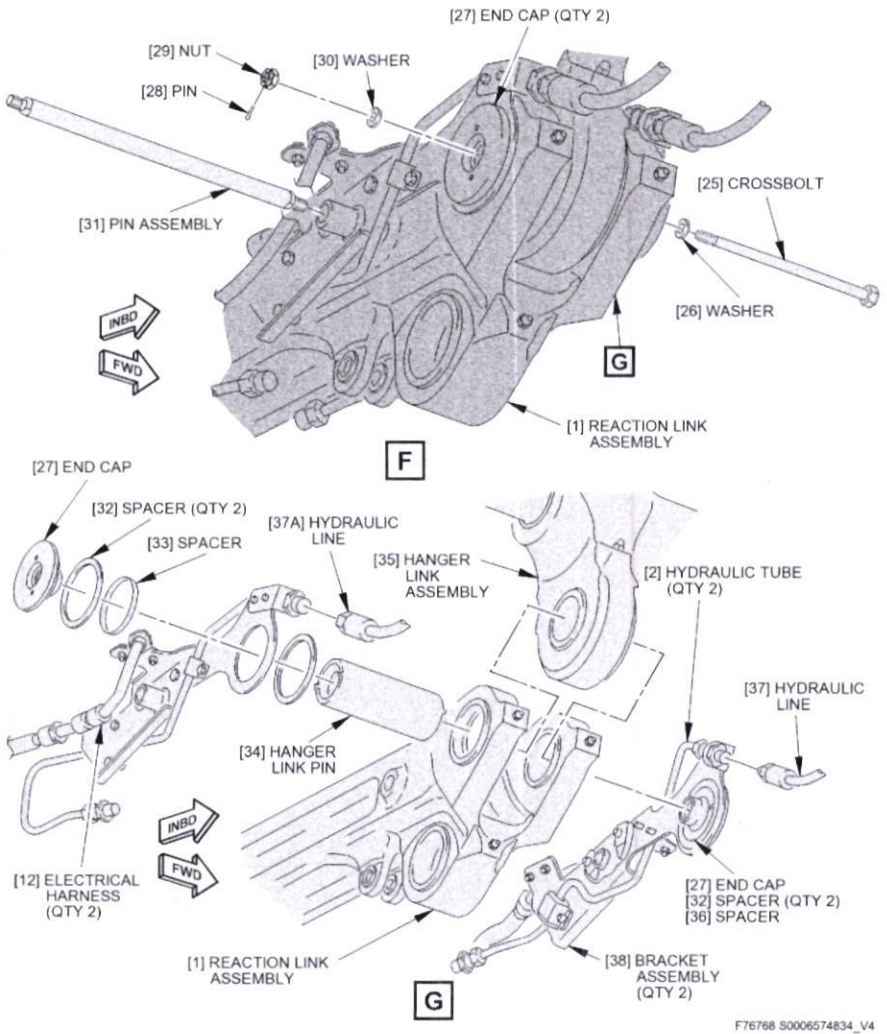


Figure 401. Main Landing Gear Reaction Link Assembly Installation - Sheet 4
TASK 32-11-71-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

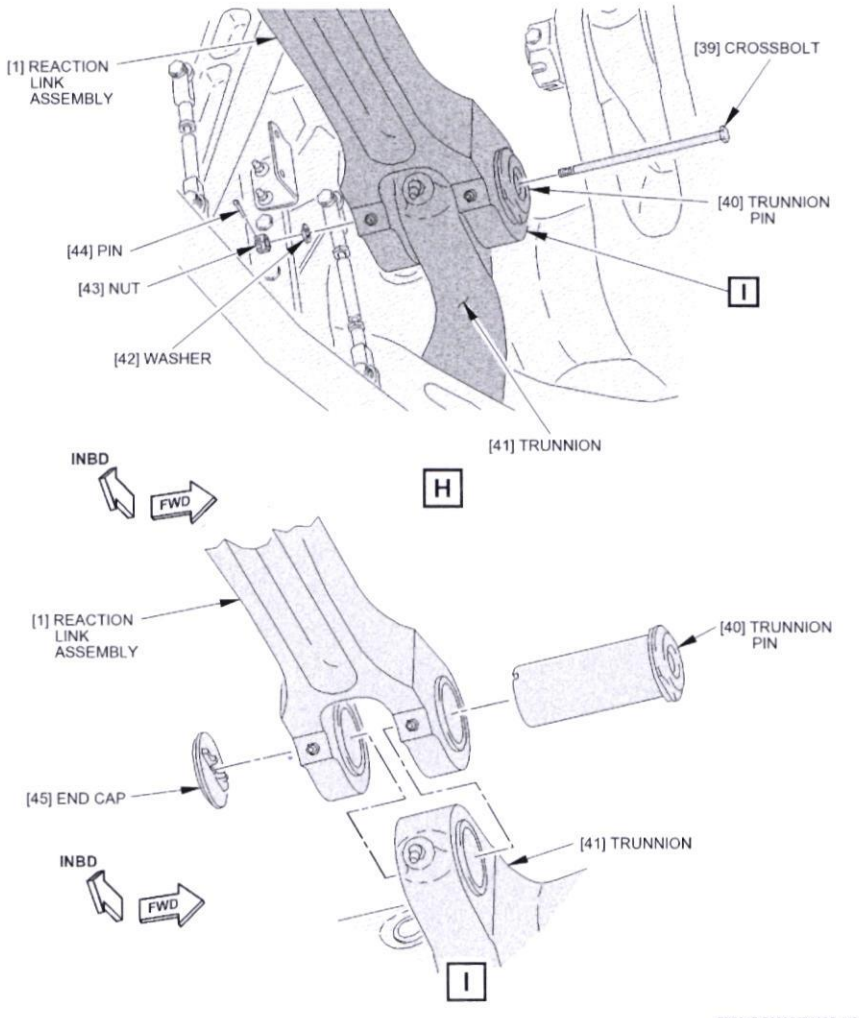


Figure 401. Main Landing Gear Reaction Link Assembly Installation - Sheet 5
TASK 32-11-71-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

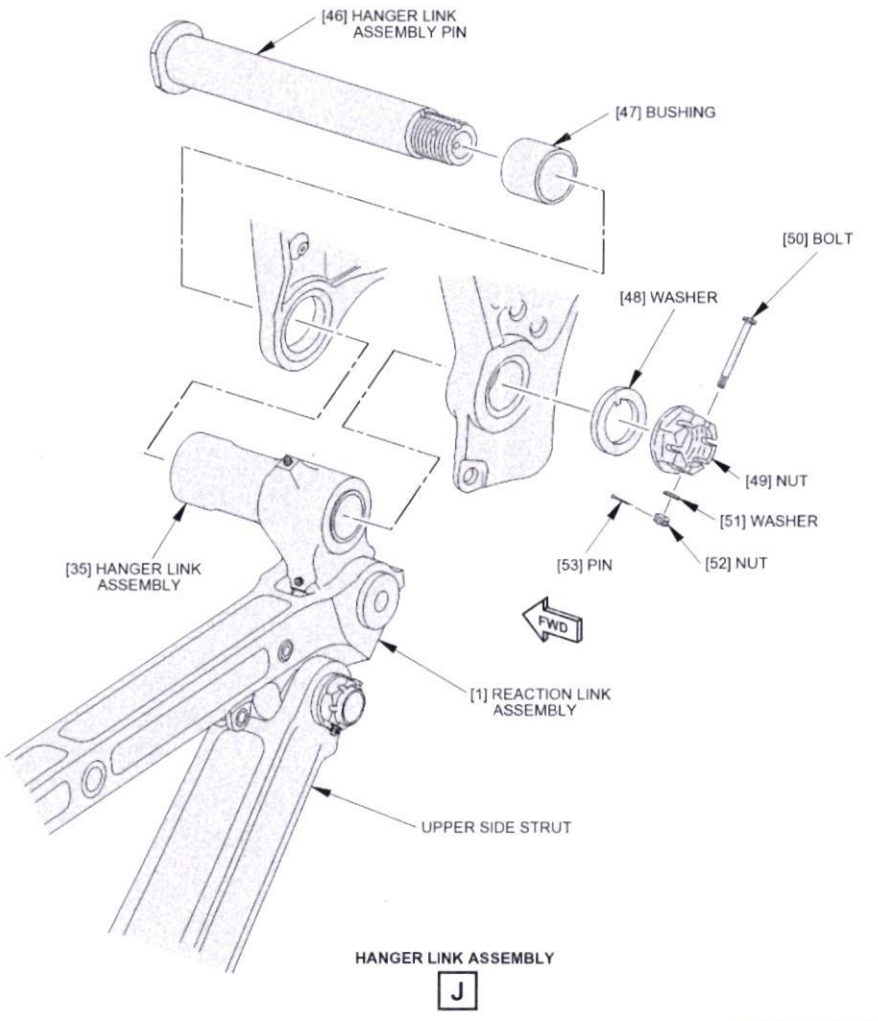


Figure 401. Main Landing Gear Reaction Link Assembly Installation - Sheet 6
TASK 32-11-71-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

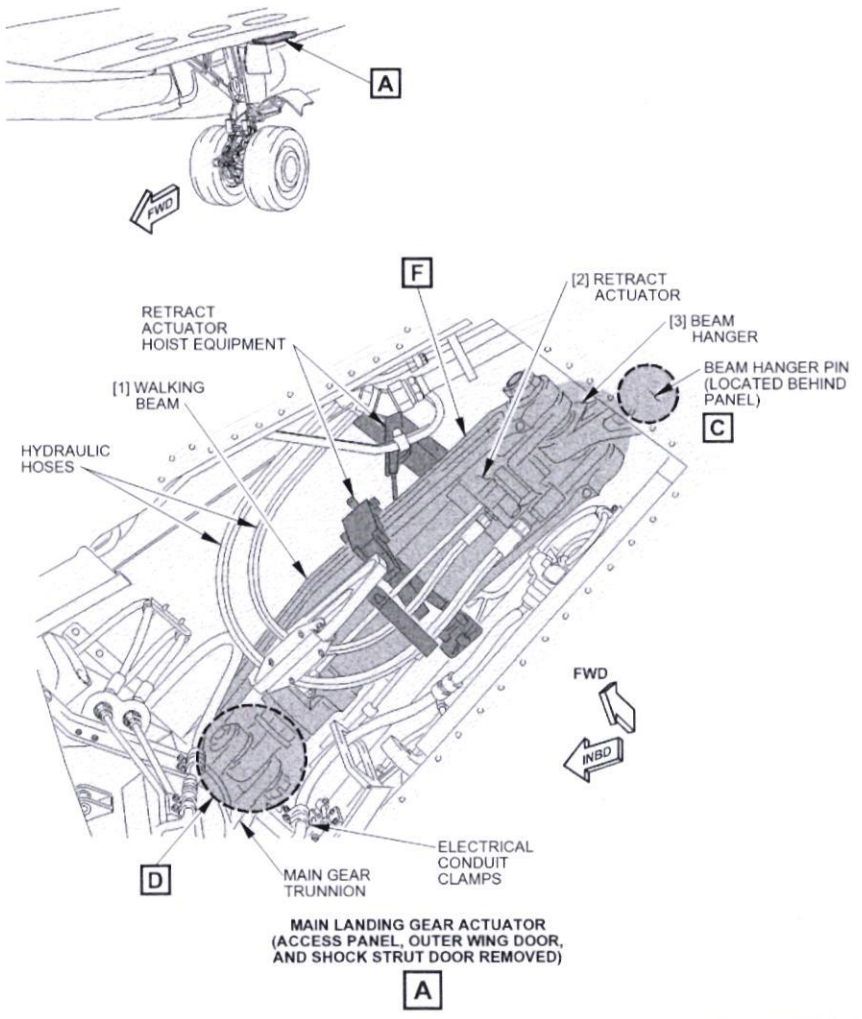
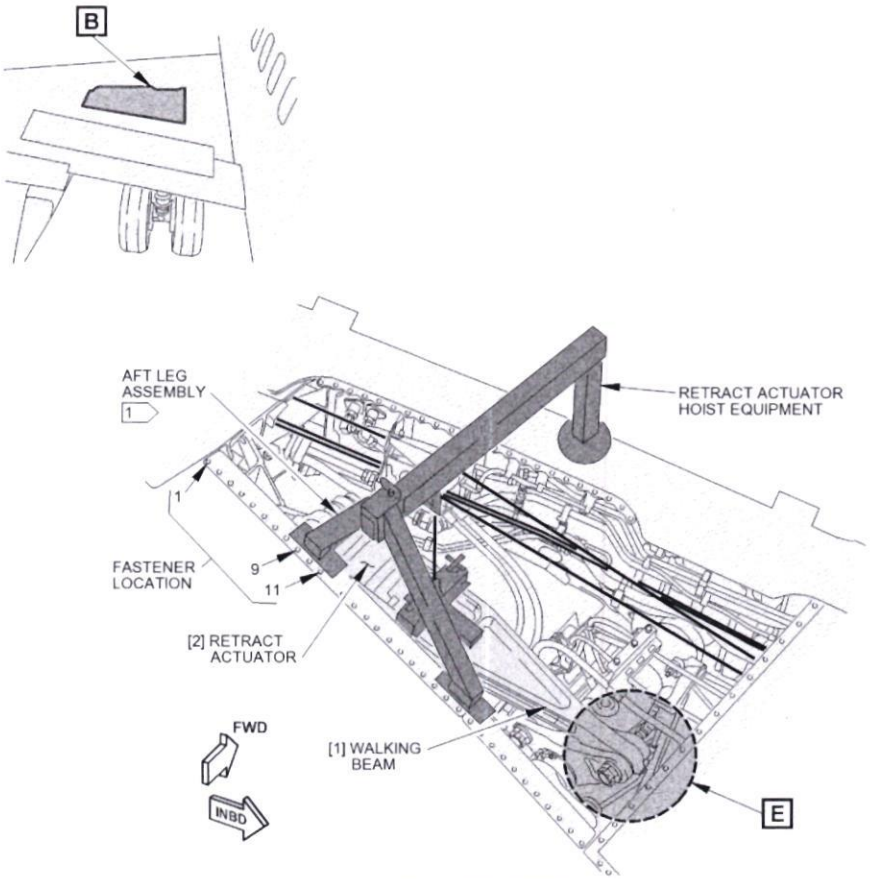


Figure 401. Main Gear Actuator Assembly Installation - Sheet 1
TASK 32-32-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



MAIN LANDING GEAR ACTUATOR
(ACCESS PANEL REMOVED AND RETRACT ACTUATOR HOIST TOOL INSTALLED)

1 USE FASTENER LOCATIONS NUMBERS 9 AND 11 FOR THE AFT OUTBOARD LEG OF THE HOIST EQUIPMENT.

B

G20722 S0006575181_V2

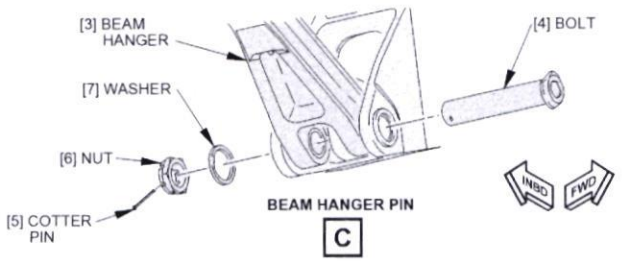
20.8.25 GAT 492

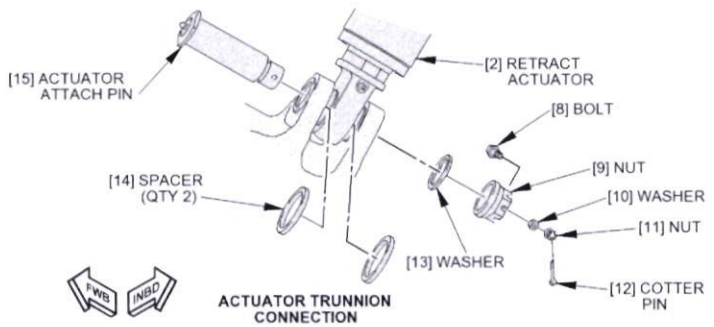
Figure 401. Main Gear Actuator Assembly Installation - Sheet 2
TASK 32-32-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____





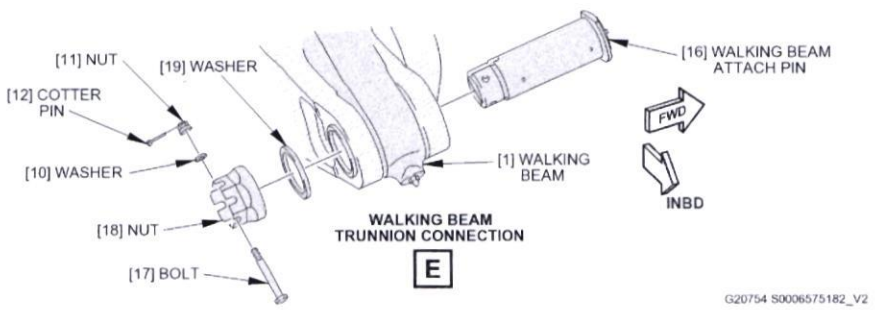
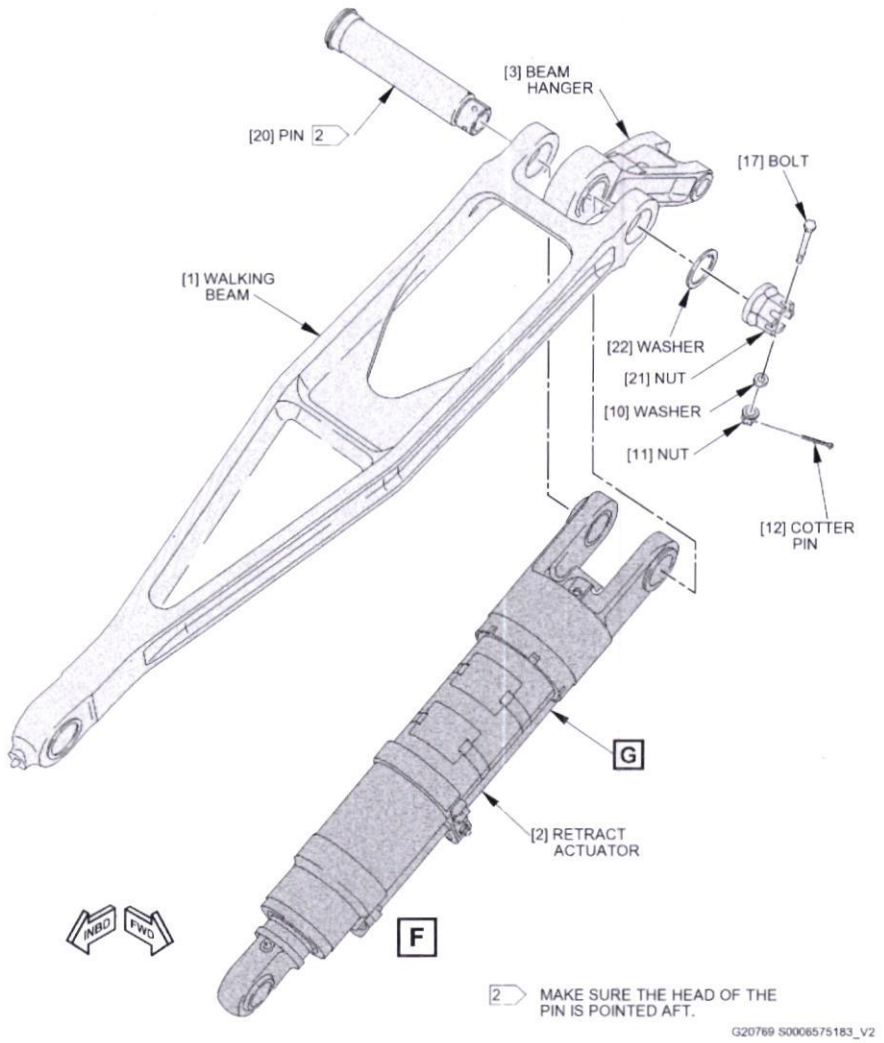


Figure 401. Main Gear Actuator Assembly Installation - Sheet 3
TASK 32-32-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



20-3-25

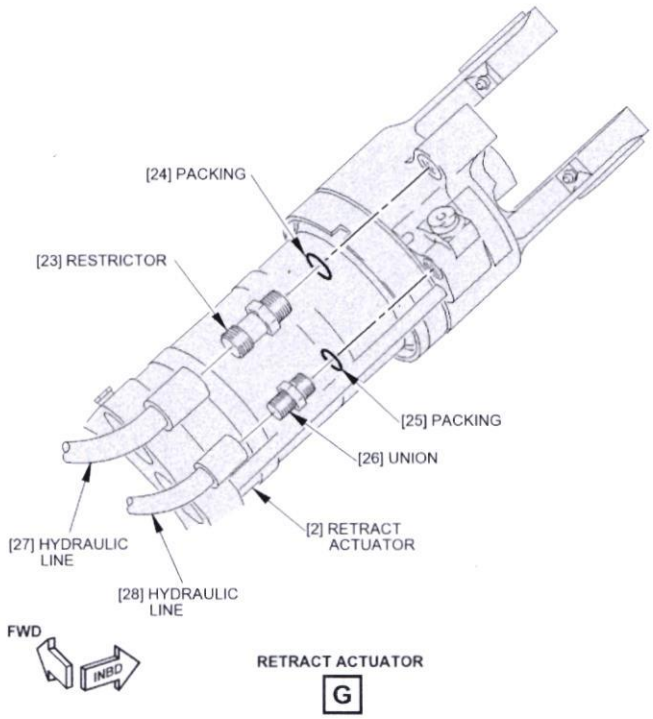
GAT 492

Figure 401. Main Gear Actuator Assembly Installation - Sheet 4
TASK 32-32-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



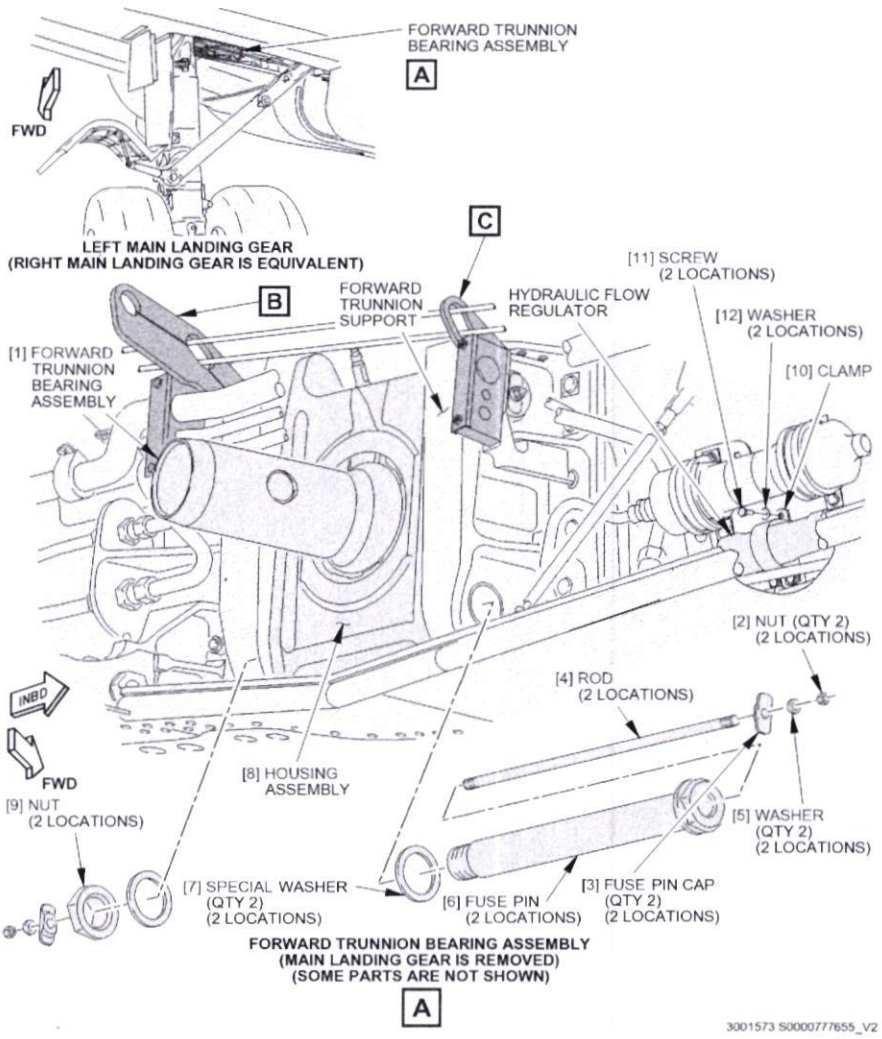
G21285 S0006575184_V3

Figure 401. Main Gear Actuator Assembly Installation - Sheet 5
TASK 32-32-11-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



20.3.24 GAT 492

Figure 401. Main Landing Gear Forward Trunnion Bearing Assembly Installation - Sheet 1
TASK 32-11-83-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

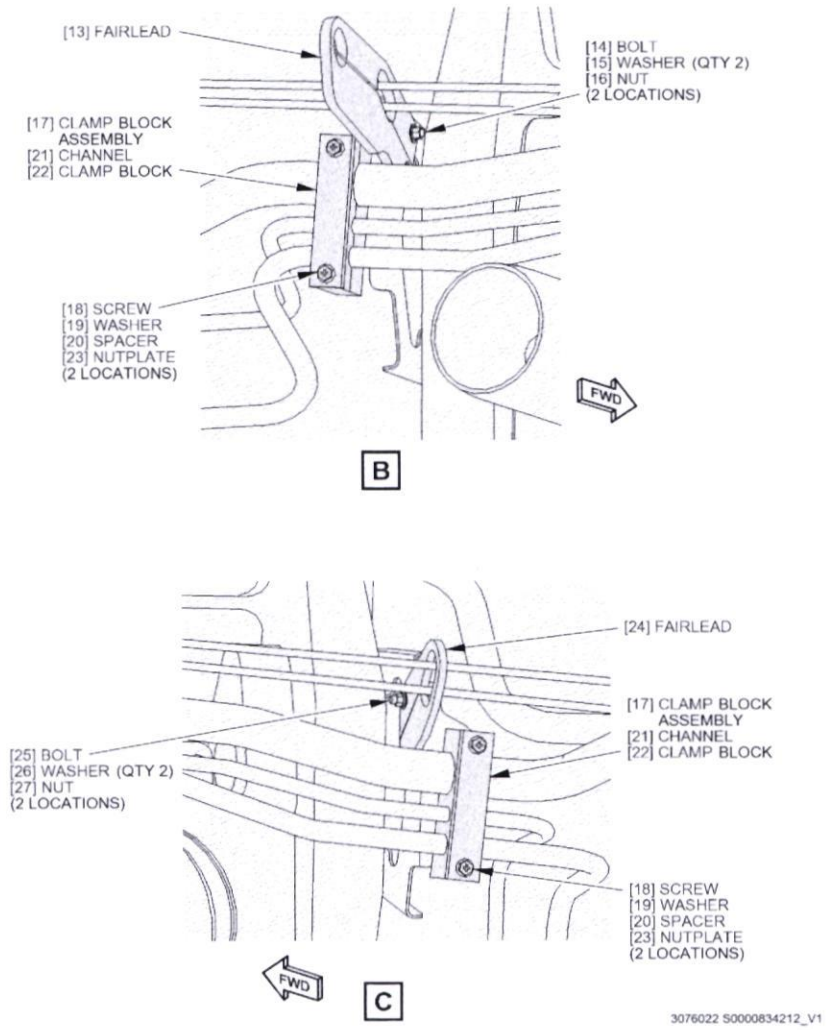


Figure 401. Main Landing Gear Forward Trunnion Bearing Assembly Installation - Sheet 2
TASK 32-11-83-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

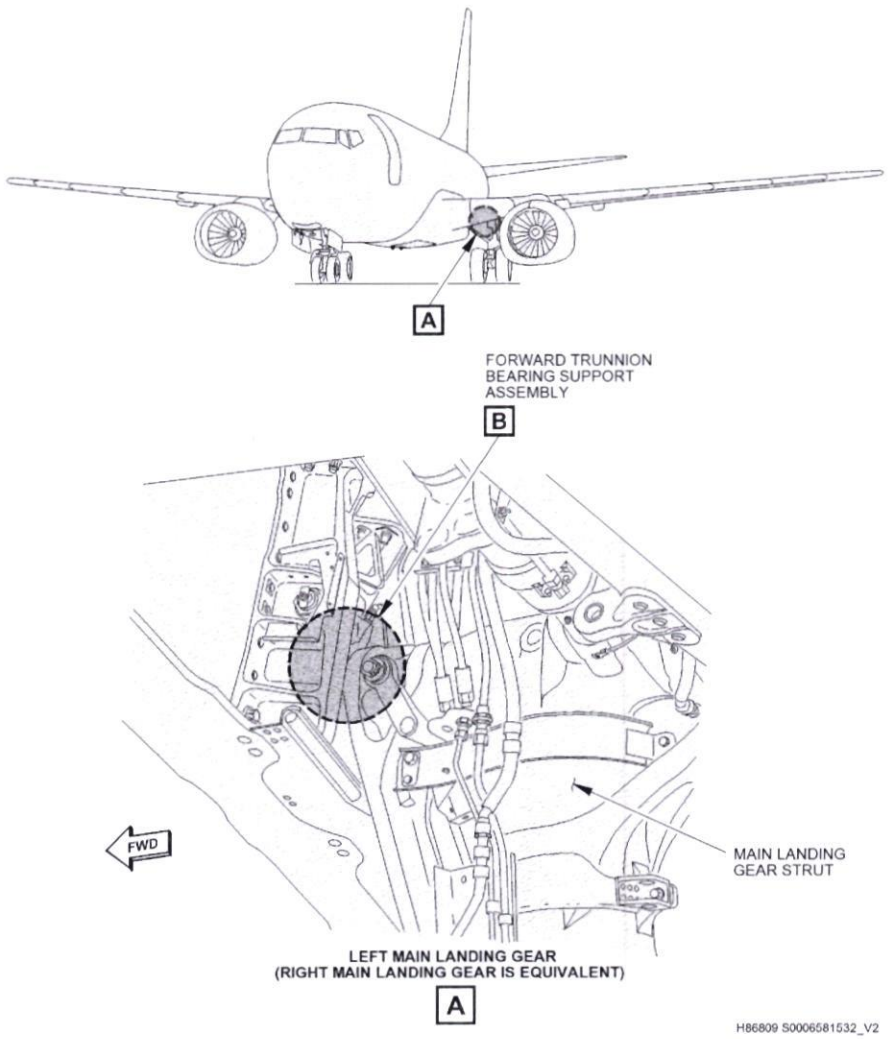


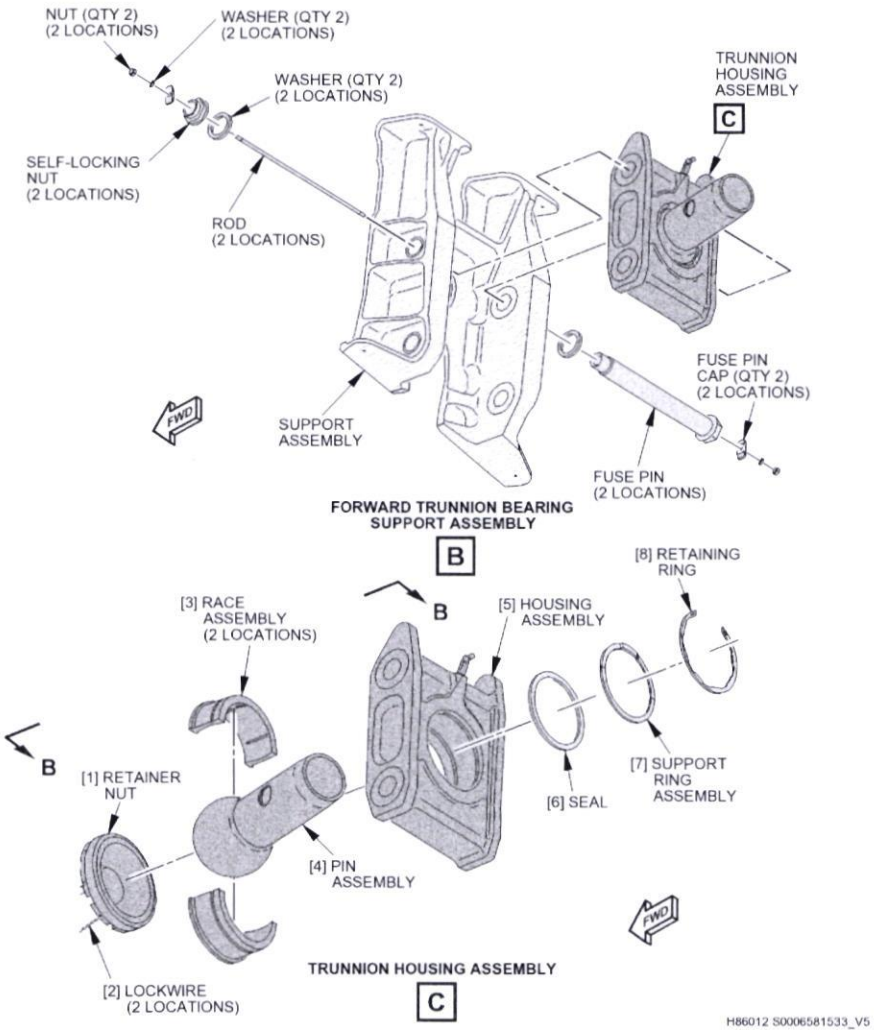
Figure 401. Forward Trunnion Housing Assembly Installation - Sheet 1
TASK 57-16-01-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 004, 005, 007-009, 014 PRE SB 737-32-1448



MR2008

21.3.25 GAT 492

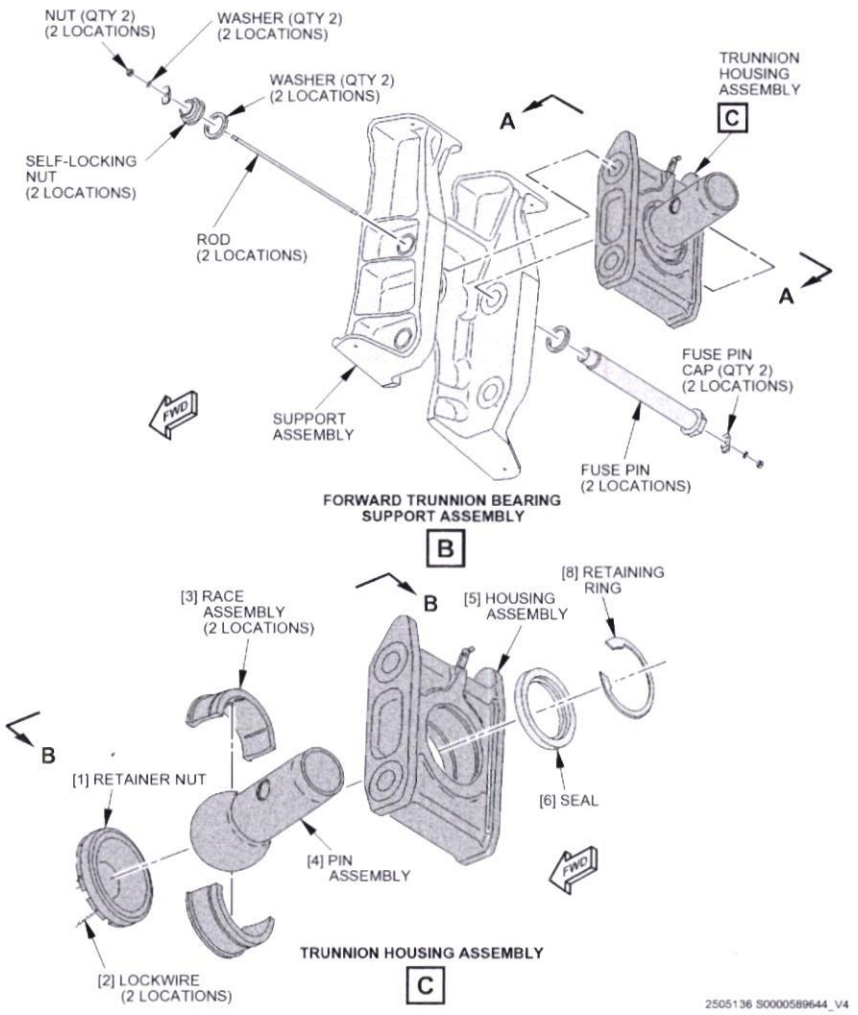
Figure 401. Forward Trunnion Housing Assembly Installation - Sheet 2
TASK 57-16-01-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448



20-8-25

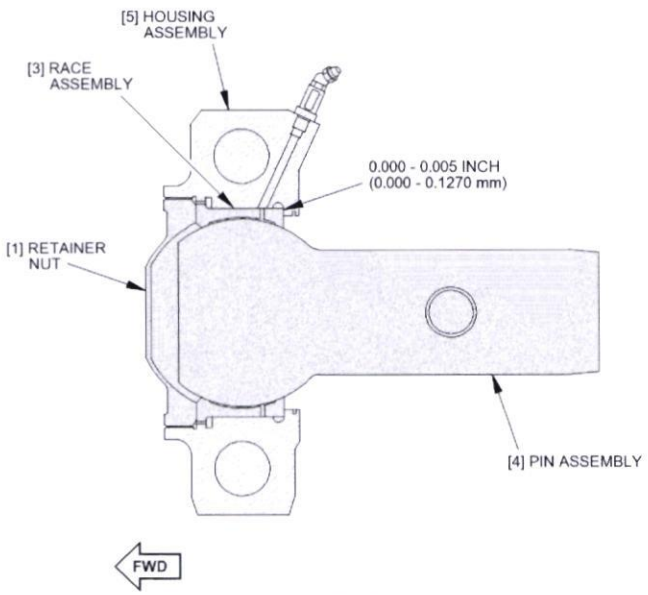
GAT 492

Figure 401. Forward Trunnion Housing Assembly Installation - Sheet 3
TASK 57-16-01-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



B-B

3013172 S0000786750_V1

M.3.25

GAT 492

MR2008

Figure 401. Forward Trunnion Housing Assembly Installation - Sheet 4
TASK 57-16-01-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 026-031, 033, 036, 037, 040-051, 053-999; JXB 004, 005, 007-009, 014 POST SB 737-32-1448

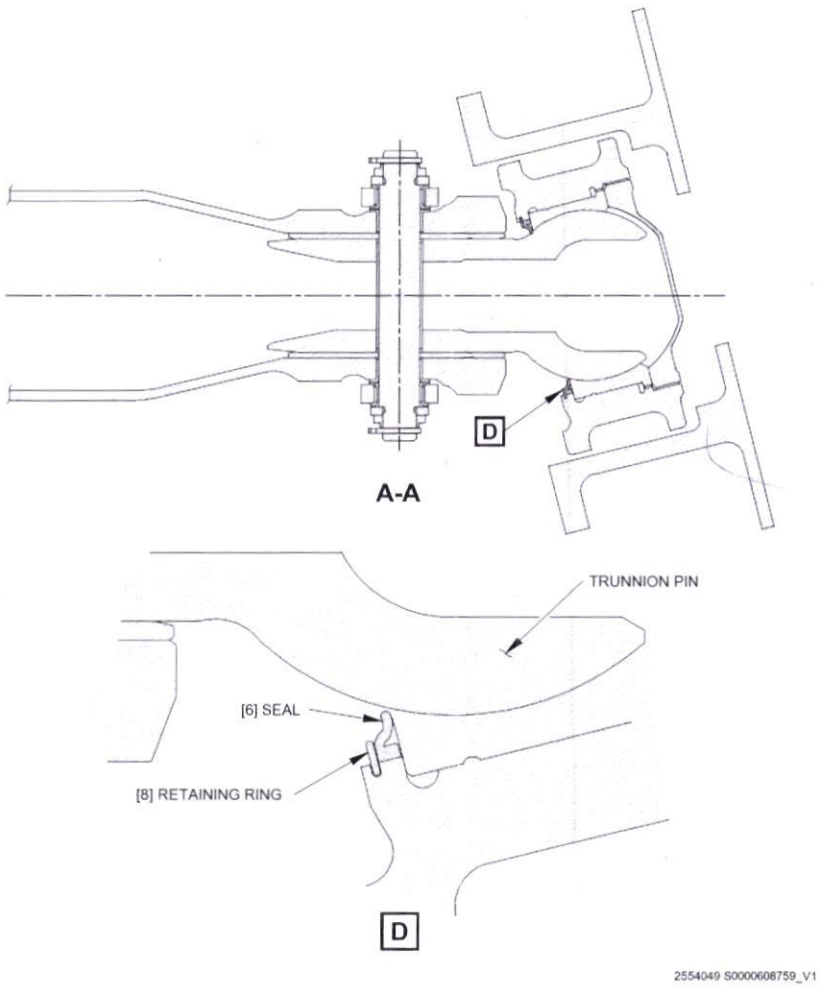
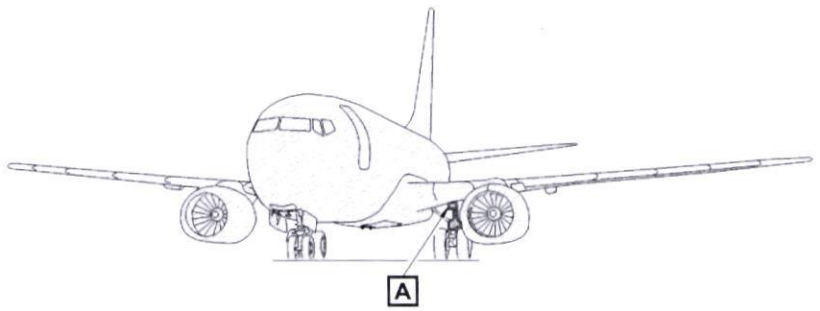


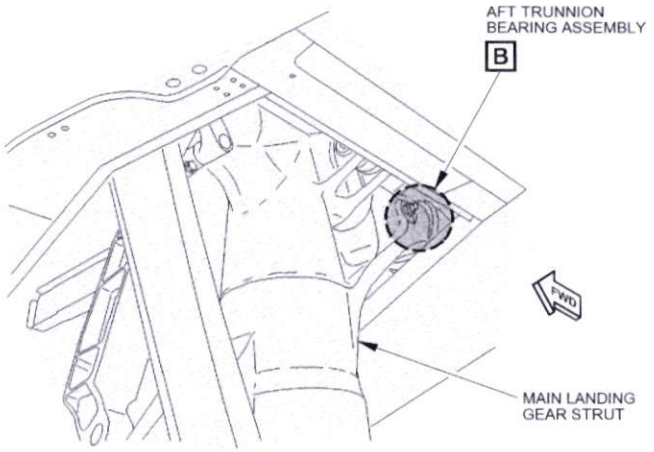
Figure 401. Forward Trunnion Housing Assembly Installation - Sheet 5
TASK 57-16-01-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____





LEFT MAIN LANDING GEAR
(RIGHT MAIN LANDING GEAR IS EQUIVALENT)
A

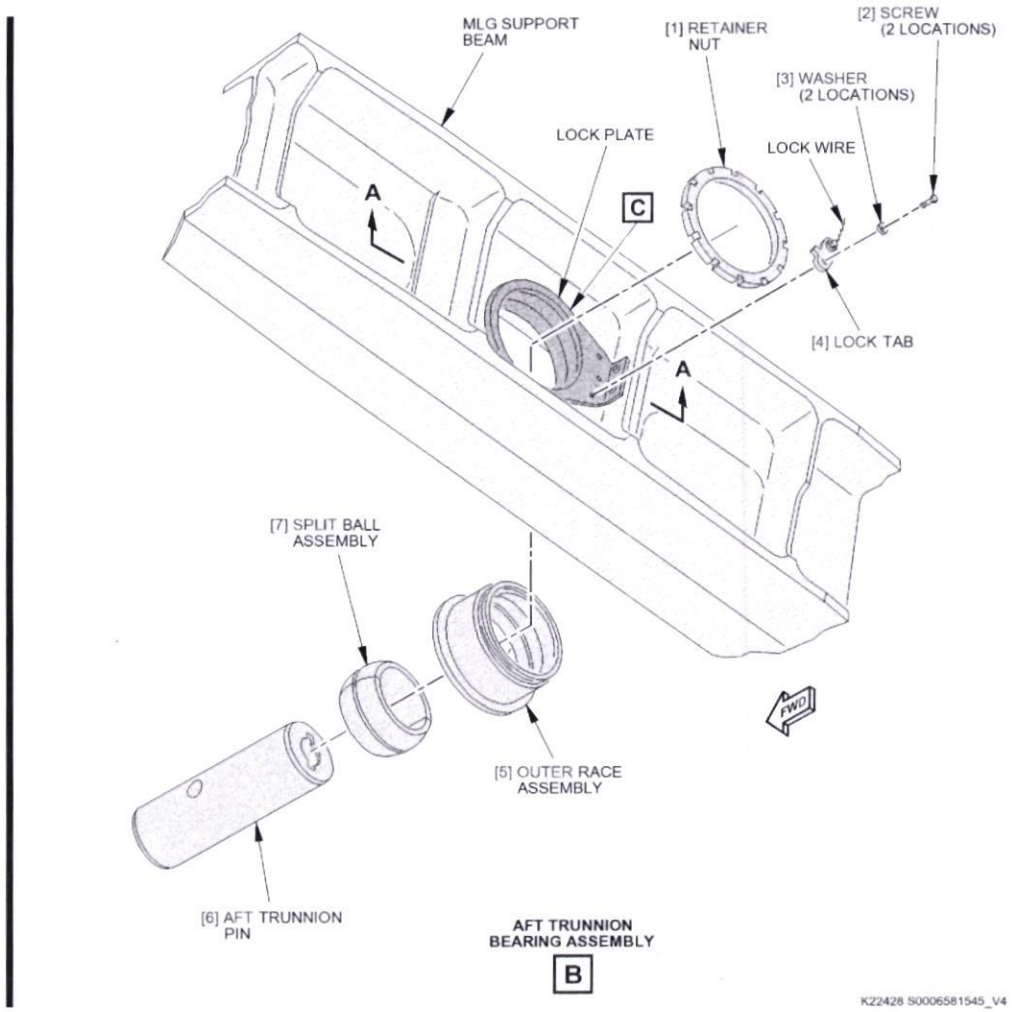
K22407 S0006581544_V2

Figure 401. MLG Aft Trunnion Bearing Assembly Installation - Sheet 1
TASK 57-16-02-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



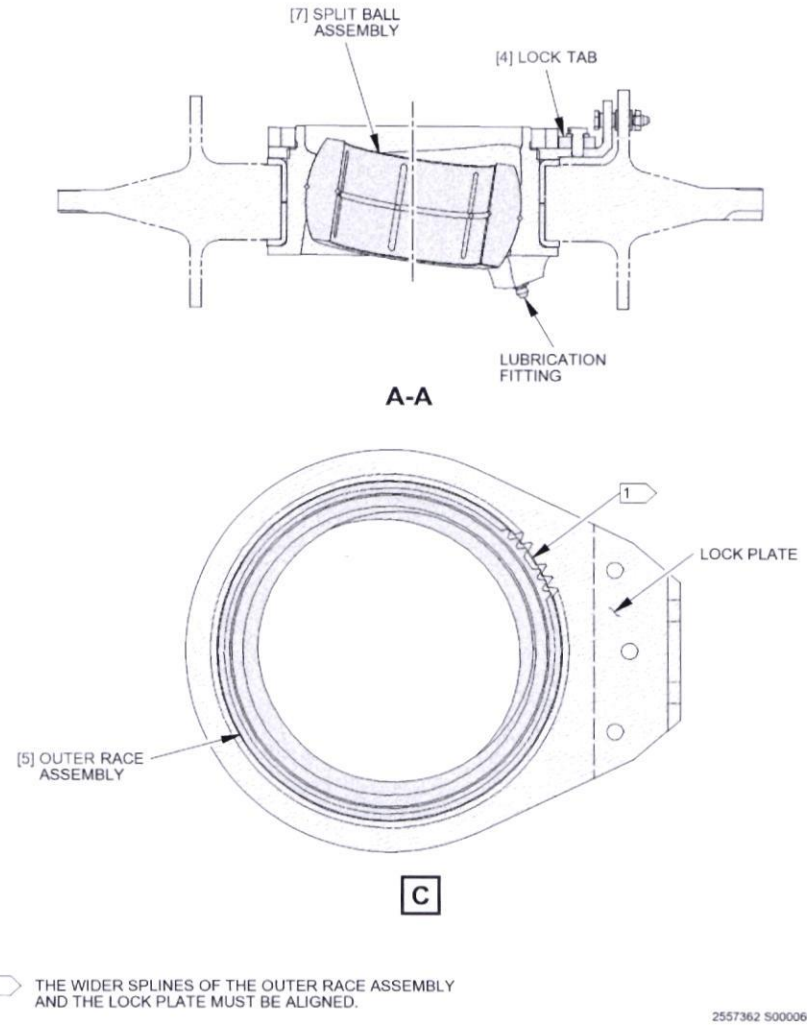
GAT 492

Figure 401. MLG Aft Trunnion Bearing Assembly Installation - Sheet 2
TASK 57-16-02-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



1 THE WIDER SPLINES OF THE OUTER RACE ASSEMBLY AND THE LOCK PLATE MUST BE ALIGNED.

2557362 S0000609931_V2

20.3.24

GAT 492

Figure 401. MLG Aft Trunnion Bearing Assembly Installation - Sheet 3
TASK 57-16-02-000-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

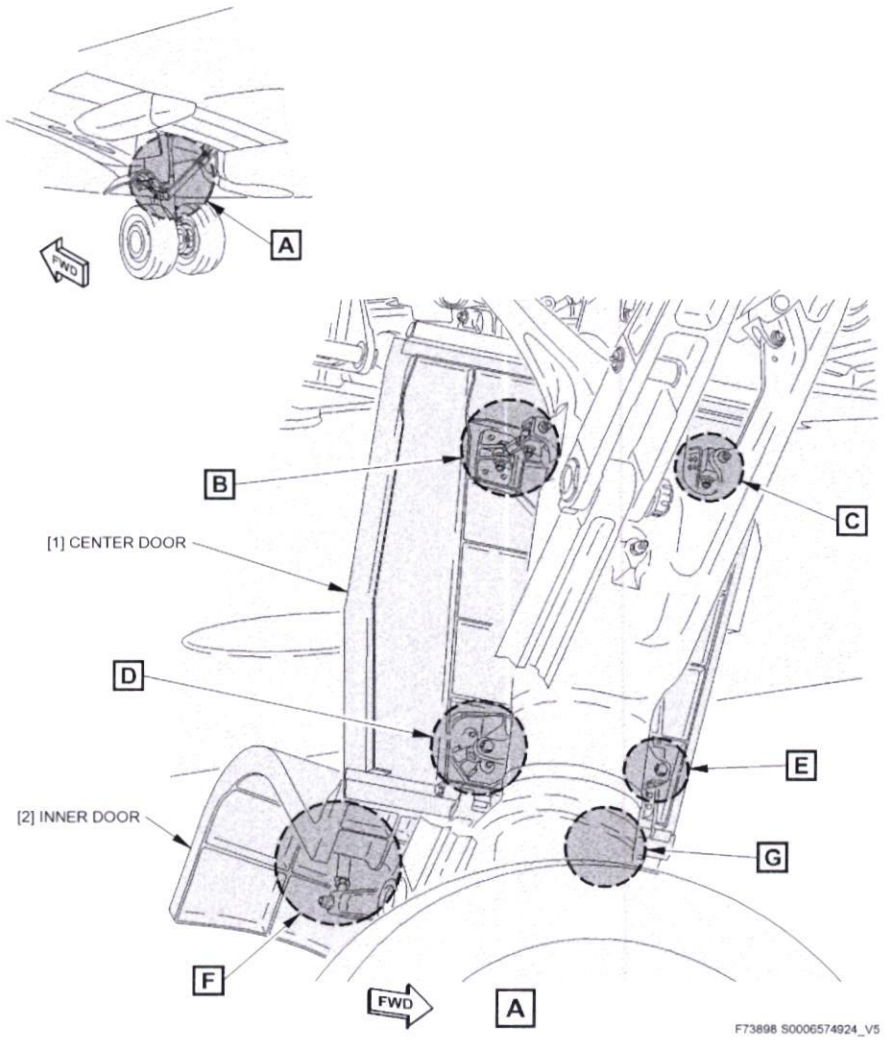


Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 1
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

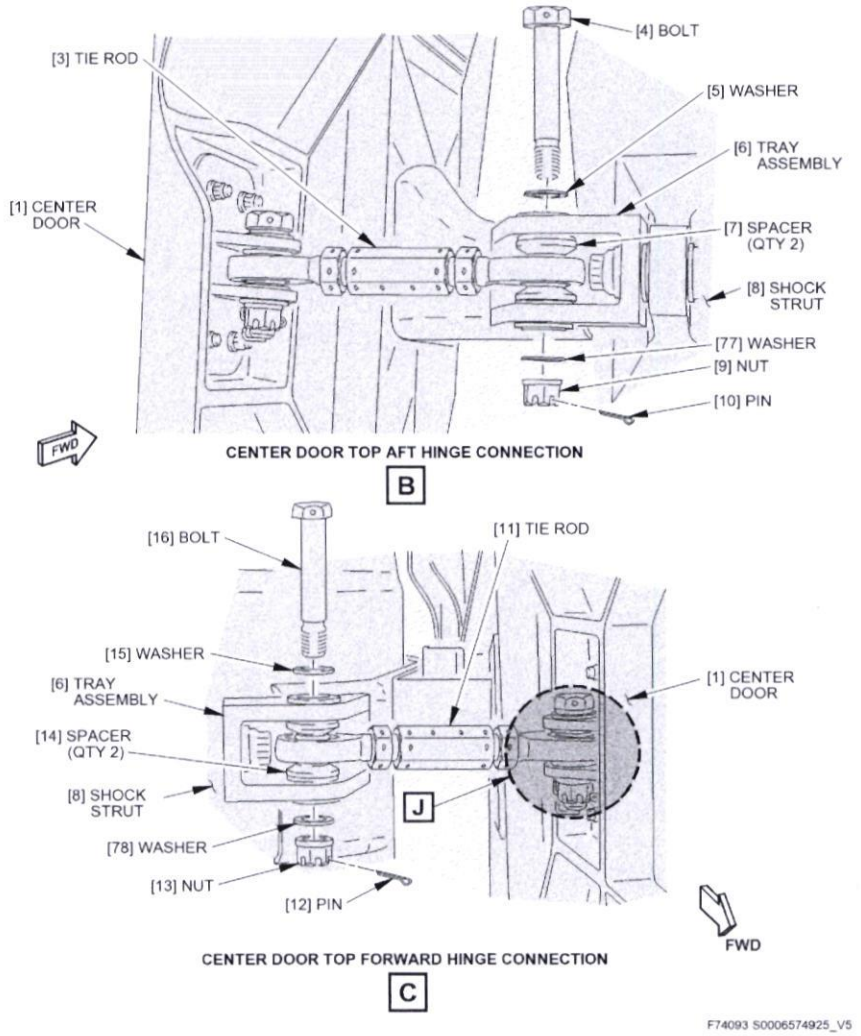


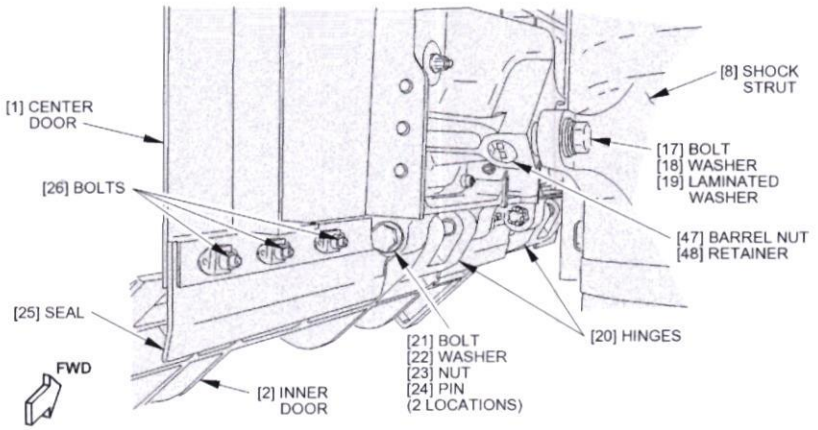
Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 2
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

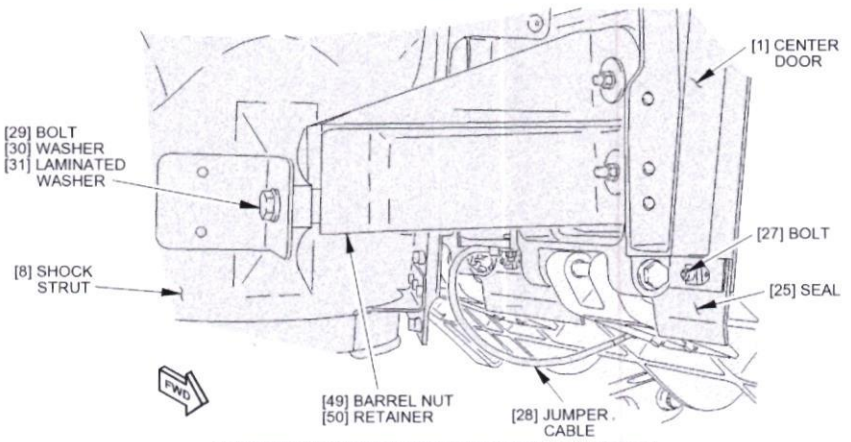
Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 036, 037, 040-051, 053-999 PRE SB 737-52-1170
MODIFICATION



CENTER DOOR BOTTOM AFT HINGE CONNECTION

D



CENTER DOOR BOTTOM FORWARD HINGE CONNECTION

E

F74190 S0006574926_V6

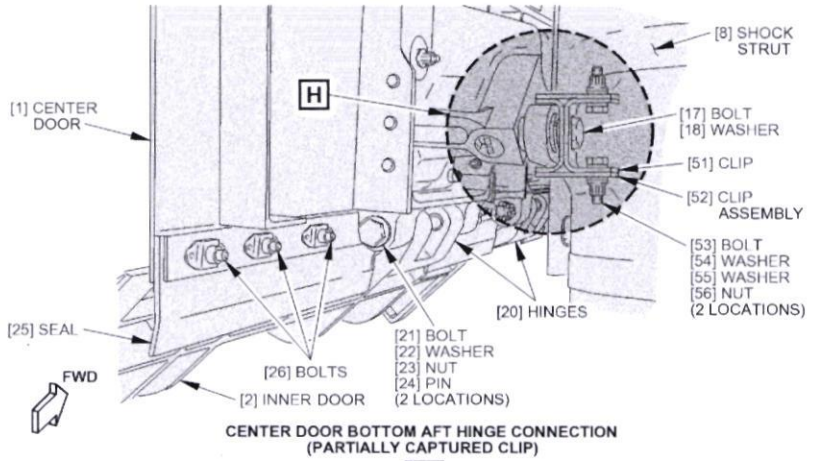
Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 3
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

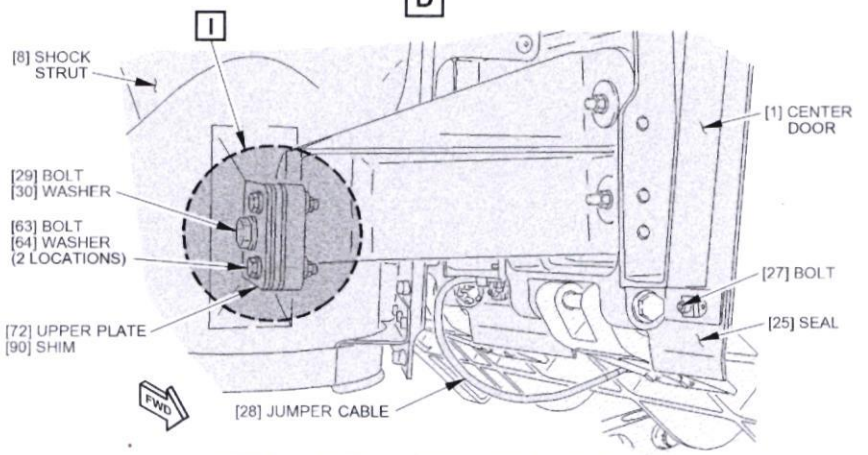
Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 004, 005, 007-009, 014, 028, 029 PRE SB
737-52-1170 MODIFICATION



CENTER DOOR BOTTOM AFT HINGE CONNECTION
(PARTIALLY CAPTURED CLIP)

D



CENTER DOOR BOTTOM FORWARD HINGE CONNECTION
(PARTIALLY CAPTURED CLIP)

E

K26037 S0006574927_V7

GAT 128

GAT 492

Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 4
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 004, 005, 007-009, 014, 028, 029 PRE SB
737-52-1170 MODIFICATION

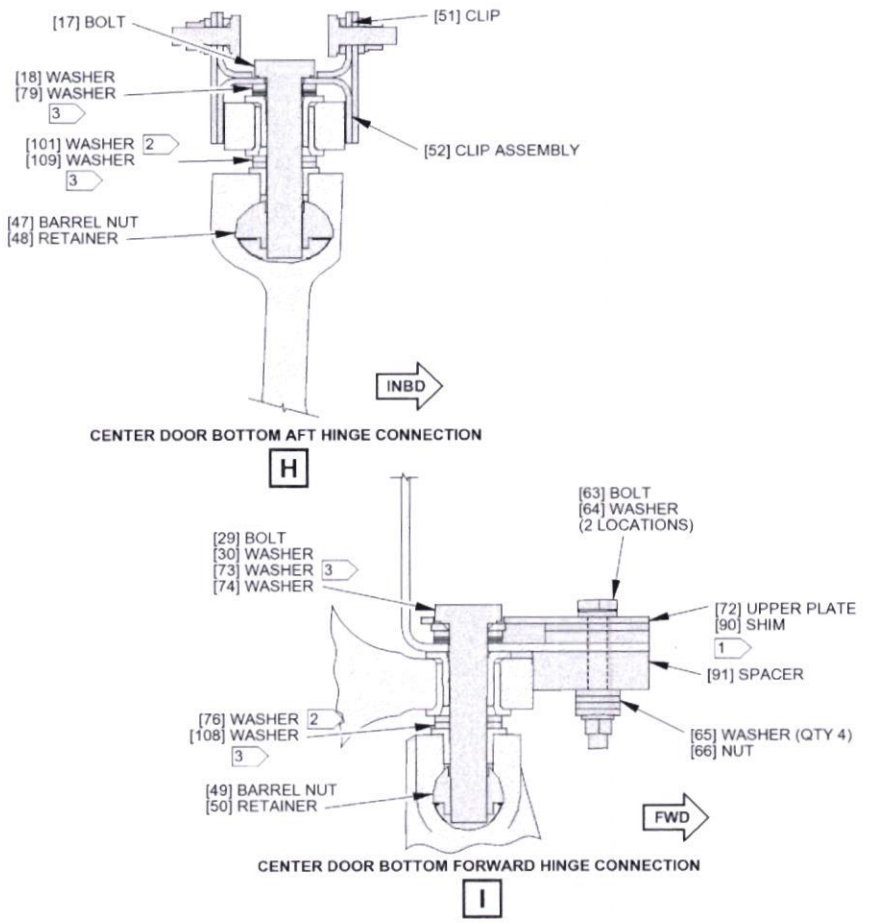


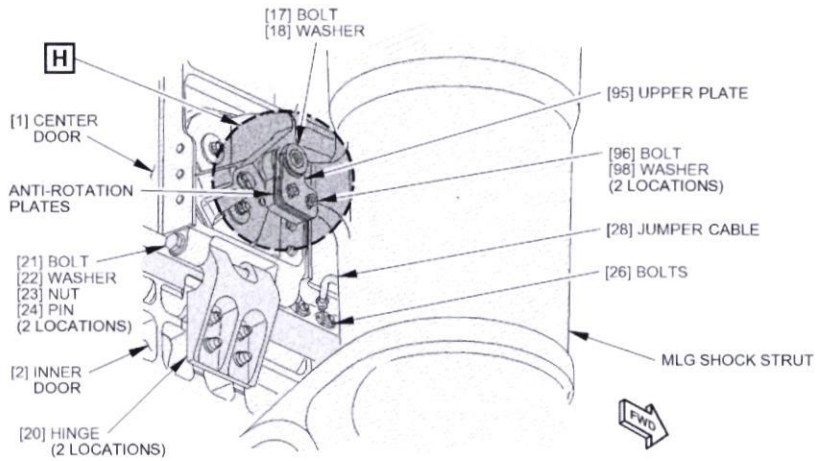
Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 5
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

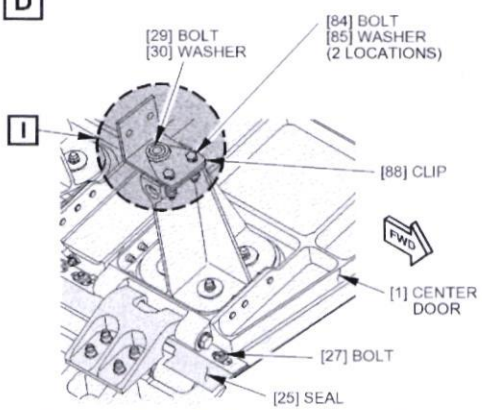
Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 030, 031, 033 PRE SB 737-52-1170 GRP 3 FULL CAPTURE



CENTER DOOR BOTTOM AFT HINGE CONNECTION (ANTI-ROTATION PLATES - TYPICAL)

D



CENTER DOOR BOTTOM FORWARD HINGE CONNECTION

E

2201364 S0000490249_V9

20.8.24 GAT 492

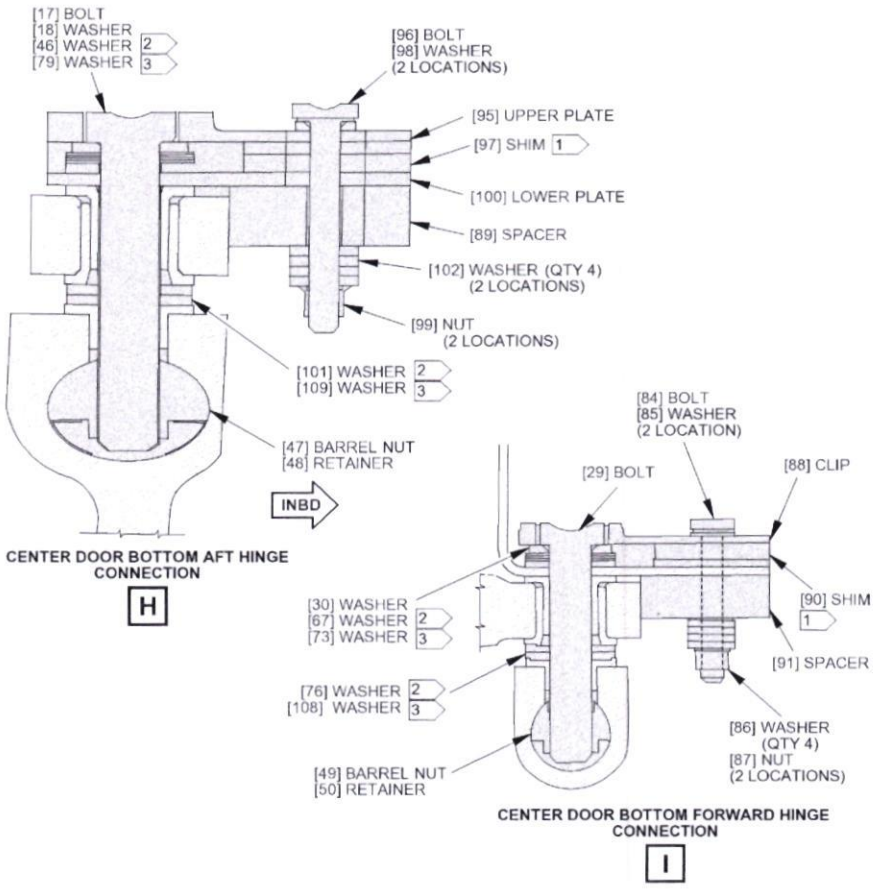
Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 6
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 030, 031, 033 PRE SB 737-52-1170 GRP 3 FULL CAPTURE



1 ADJUST SHIM THICKNESS TO MATCH WASHERS
2 USE 0 - 3 WASHERS TO GET CORRECT FIT/CLEARANCE
3 USE 0 - 4 WASHERS TO GET CORRECT FIT/CLEARANCE

2201987 S0000490305_V7

GAT 492

Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 7
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 026, 027; JXB 004, 005, 007-009, 014, 028-031, 033, 036, 037, 040-051, 053-999 POST SB 737-52-1170 MODIFICATION

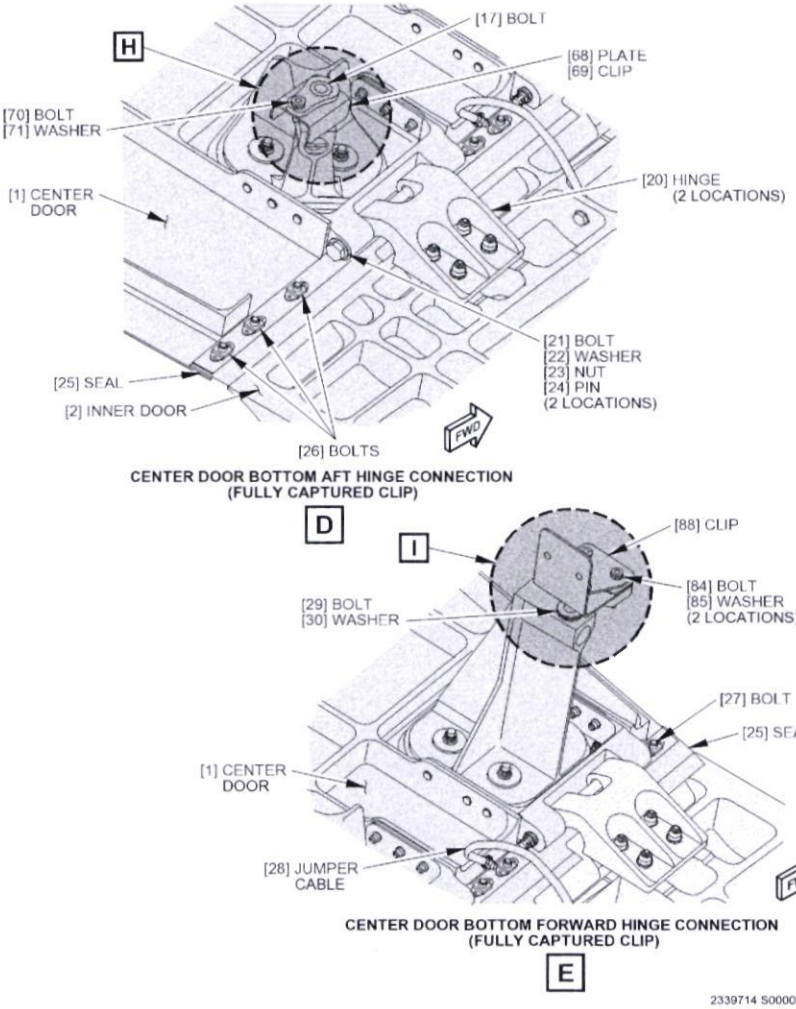


Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 8

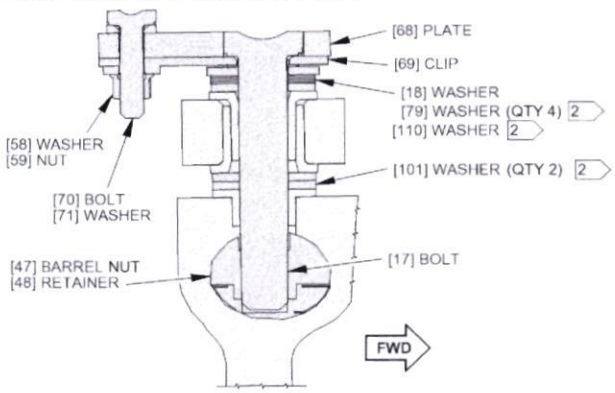
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

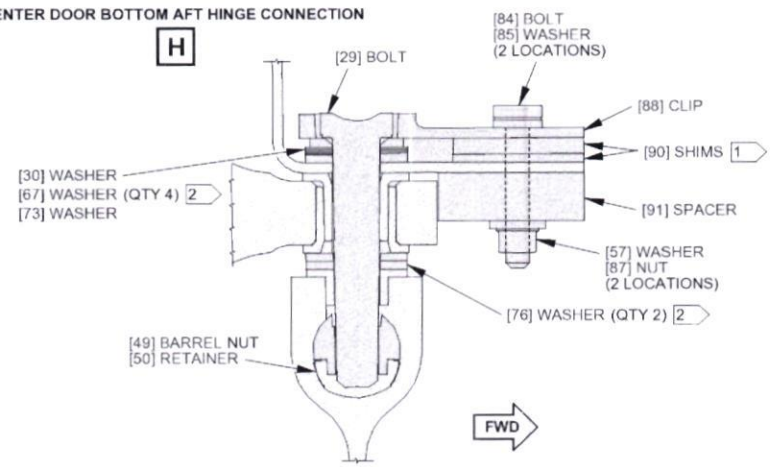
Item: _____ Completed through item: _____ Sign: _____

FIG. EFFECTIVITY: JXB AMM EFFECTIVITY 026, 027; JXB 004, 005, 007-009, 014, 028-031, 033, 036, 037, 040-051, 053-999 POST SB 737-52-1170 MODIFICATION



CENTER DOOR BOTTOM AFT HINGE CONNECTION

H



CENTER DOOR BOTTOM FORWARD HINGE CONNECTION

I

1 INSTALL A MAXIMUM OF THREE SHIMS. IF NECESSARY, REMOVE AND PEEL THE SHIM LAMINATION TO MAKE SURE THAT THE CLIP ENGAGES THE BOLT.

2 PRE-ADJUSTMENT WASHER STACK-UP IS SHOWN

2339746 S0000532809_V10

20.3.24

GAT 492

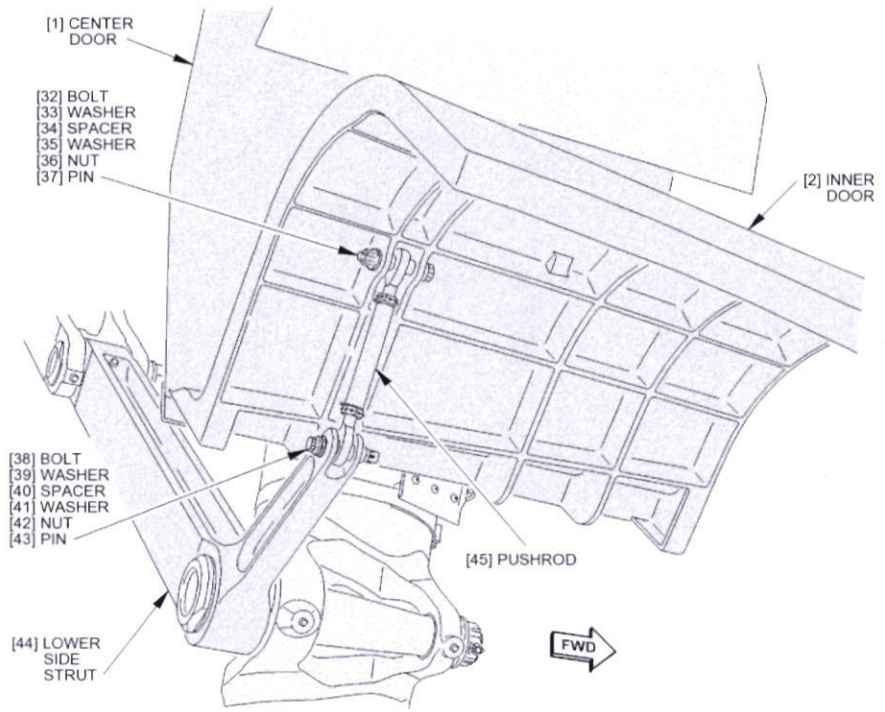
Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 9

TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



INNER DOOR PUSHROD CONNECTION
(RIGHT SIDE)
F

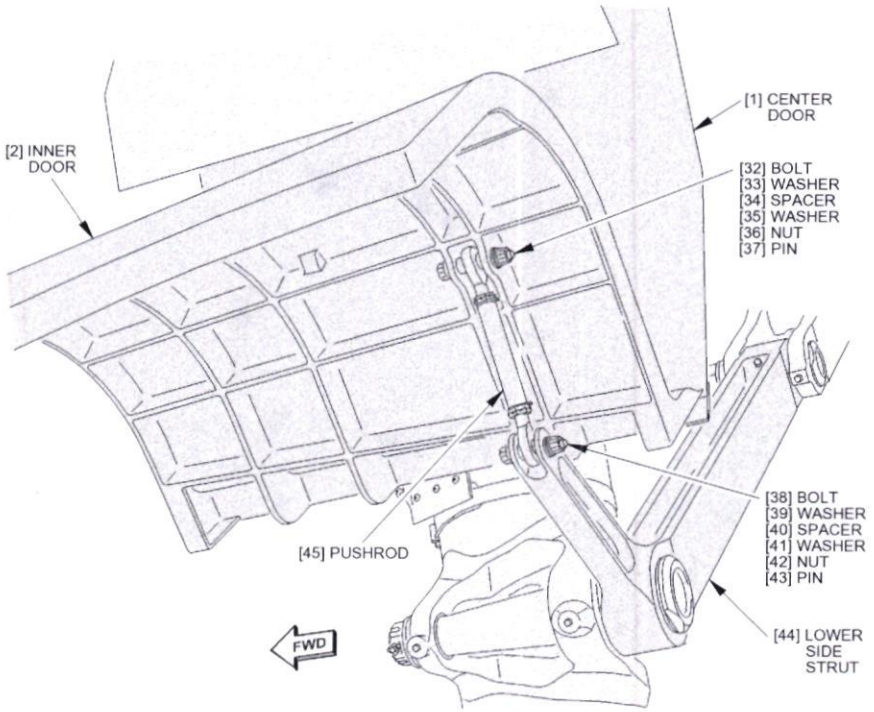
F74096 S0006574929_V2

Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 10
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____



INNER DOOR PUSHROD CONNECTION
(LEFT SIDE)

G

M57626 S0006574930_V2

20.8.24

GAT 492

Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 11
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

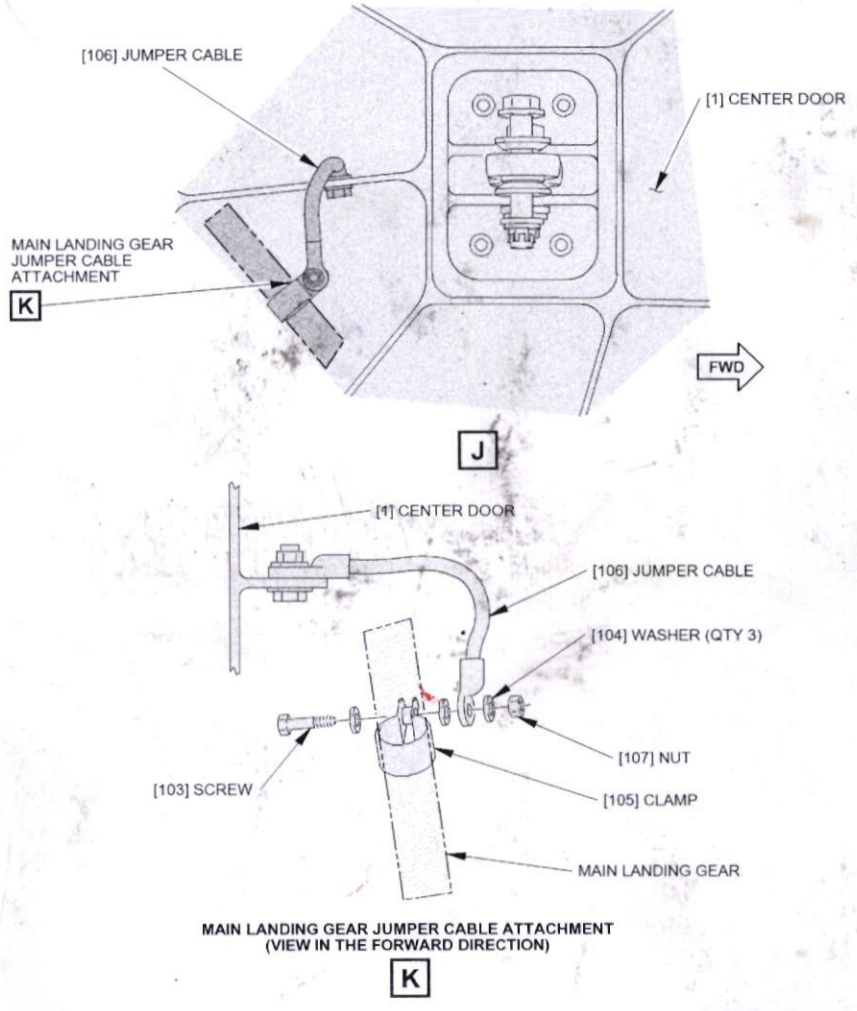


Figure 401. Main Landing Gear Shock Strut Doors Installation - Sheet 12
TASK 32-11-00-400-801

PARTIAL SIGN OFF STATUS:

Item: _____ Completed through item: _____ Sign: _____

Item: _____ Completed through item: _____ Sign: _____

flydubai	flydubai Dubai	W/O 7911047 Working Copy	Barcode WO7911047	Registration A6FEW 73NG 738

Type	Origin	ATA	Position	Zone	Area
S SCHED	V Replacement Event	32 LANDING GEAR	RH MLG RH MAIN LANDING GEAR		

Partnumber	Description	Serialnumber
161A1100-54	MLG COMPONENT INSTL	MAL05346S

Type	Reference	Description

Due Date	Due at TAH	Due at TAC
05.Jan.2029		31307

Description Step 1 — Udit Bansal (FZ04850), 02.Oct.2024

REPLACE PN:161A1100-54 SN:MAL05346S (MLG COMPONENT INSTL)

1. REPLACE PN:161A1100-54 SN:MAL05346S (MLG COMPONENT INSTL)
INSTALLED AT: RH MLG(RH MAIN LANDING GEAR)
INFO/TRIGGERED BY: 161A2330-2/E0130 RS(RESTORE) RESTORE / 161A2330-2 RESTORE

RECORD THE FOLLOWING:
OFF: P/N:_____ S/N:_____

ON: P/N:_____ S/N:_____

2. REMOVED MLG TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY

Action Step 1-1

1. REPLACED PN:161A1100-54 SN:MAL05346S (MLG COMPONENT
INSTL)
INSTALLED AT: RH MLG(RH MAIN LANDING GEAR)
INFO/TRIGGERED BY: 161A2330-2/E0130 RS(RESTORE) RESTORE /
161A2330-2 RESTORE REF MPD 32-040-02-01 REV 44
RECORDED THE FOLLOWING: DATED OCT17, 2024
OFF:

P/N:_161A1100-54 S/N:_MAL05346S

ON:

P/N:161A1100-68 S/N:MAL10792Y5241 22.3.25

2. REMOVED MLG PROVIDED WITH SERVICEABLE TAG AND
CREDITED TO STORES FOR ONWARD INSTALLATION ON A6- FEY
23.3.25

GAT 492

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate
161A1100-54	MAL05346S			161A1100-68	MAL10792Y5241		

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 2 — Munal Rao (FZ04849), 14.Jan.2025

REPLACEMENT OF SHIMMY DAMPER ASSY ALONG WITH RH MLG

1,CARRY OUT REPLACEMENT OF SHIMMY DAMPER ASSY P/N: 273A3610-4 REFER AMM 32-11-81.

RECORD THE FOLLOWING: OFF : P/N:_____ S/N:_____

ON: P/N:_____ S/N:_____

2.REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY

Action Step 2-1

1,CARRIED OUT REPLACEMENT OF SHIMMY DAMPER ASSY P/N:
273A3610-4 ALONG WITH RH MAIN LANDING GEAR ASSEMBLY REF MPD 32-040-02-01 REV 44 DATED OCT 17, 2024.
RECORDED THE FOLLOWING:
OFF :
P/N: 273A3610-4 S/N:5287

ON:

P/N:273A3610-6 S/N:0388 22.3.25

2.REMOVED UNIT PROVIDED WITH SERVICEABLE TAG AND CREDITED TO STORES FOR ONWARD INSTALLATION ON A6- FEY
23.3.25

GAT 492

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate
273A3610-4	5287			273A3610-6	0388		

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

flydubai	flydubai Dubai	W/O 7911047 Working Copy	Barcode WO7911047	Registration A6FEW 73NG 738

Description Step 3	Munal Rao (FZ04849), 14.Jan.2025	Action Step 3-1
REPLACEMENT OF MLG RETRACT ACTUATOR ALONG WITH RH MLG 1.CARRY OUT REPLACEMENT OF MLG RETRACT ACTUATOR P/N: 273A2101-101 REFER AMM 32-32-11. RECORD THE FOLLOWING: OFF : P/N: _____ S/N:_____ ON: P/N:_____ S/N:_____ 2.REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY.		1.CARRIED OUT REPLACEMENT OF MLG RETRACT ACTUATOR P/N: 273A2101-101 REFER AMM 32-32-11. AMM REV 86 RECORDED THE FOLLOWING: DATED 15 FEB 2025 OFF : P/N: 273A2101-2 S/N:2101/0493A ON: P/N:273A2101-101 S/N:2101/11250A 22.8.25 GAT 492 2.REMOVED UNIT PROVIDED WITH SERVICEABLE TAG AND CREDITED TO STORES FOR ONWARD INSTALLATION ON A6-FEY. 25.8.25 GAT 492

Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate
273A2101-2	2101/0493A			273A2101-101	2101/11250A		

Transfers									
Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 4	Munal Rao (FZ04849), 14.Jan.2025	Action Step 4-1
REPLACEMENT OF UPLOCK ACTUATOR ALONG WITH RH MLG 1.CARRY OUT REPLACEMENT OF UPLOCK ACTUATOR P/N: 273A2501-1. REFER AMM 32-32-41. RECORD THE FOLLOWING: OFF : P/N: _____ S/N:_____ ON: P/N:_____ S/N:_____ 2.REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY.		1.CARRIED OUT REPLACEMENT OF UPLOCK ACTUATOR P/N: 273A2501-1. REFER AMM 32-32-41.AMM REV 86 RECORDED THE FOLLOWING: DATED 15 FEB 2025 OFF : GAT 492 P/N: 273A2501-1 S/N:0015 ON: P/N:273A2501-1 S/N:10812 22.8.25 2.REMOVED UNIT PROVIDED WITH SERVICEABLE TAG AND CREDITED TO STORES FOR ONWARD INSTALLATION ON A6-FEY. 28.8.25 GAT 492

Component Changes							
PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate
273A2501-1	0015			273A2501-1	10812		

Transfers									
Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 5	Munal Rao (FZ04849), 14.Jan.2025	Action Step 5-1
REPLACEMENT OF MLG FWD TRUNNION HOUSING ASSY ALONG WITH RH MLG 1.CARRY OUT REPLACMENT OF MLG FWD TRUNNION HOUSING ASSY P/N: 115A5240-3. REFFR AMM 32-11-83. RECORD THE FOLLOWING: OFF : P/N: _____ S/N:_____		1.CARRIED OUT REPLACMENT OF MLG FWD TRUNNION HOUSING ASSY P/N: 115A5240-3. REFFR AMM 32-11-83. AMM REV 86 RECORDED THE FOLLOWING: DATED 15 FEB 2025 OFF : P/N: 115A5240-3 S/N:16503A-050254G-01

ON: P/N:_____ S/N:_____

2.REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY.

ON:

P/N:115A5240-3 S/N:P20135351

2.REMOVED UNIT PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY.

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate
115A5240-3	16503A-050254G-01			115A5240-3	P20135351		

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 6 — Munal Rao (FZ04849), 14.Jan.2025

REPLACEMENT OF MLG FWD TRUNNION BEARING ASSY ALONG WITH RH MLG.

1.CARRY OUT REPLACEMENT OF MLG FWD TRUNNION BEARING ASSY(REFERENCE P/N:115A5005-XXX).
REFER AMM 32-11-83.
RECORD THE FOLLOWING:
OFF : P/N: _____ S/N:_____

ON: P/N:_____ S/N:_____

2.REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY.

Action Step 6-1

1.CARRIED OUT REPLACEMENT OF MLG FWD TRUNNION BEARING ASSY(REFERENCE P/N:115A5005-XXX).
REFER AMM 32-11-83 AMM REV 86 DATED 15 FEB 2025
RECORDED THE FOLLOWING:

OFF :

P/N: 115A5005-102 S/N:DLH89185

ON:

P/N:115A5005-102 S/N:MAL10792Y5241

2.REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY.

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate
115A5005-102	DLH89185			115A5005-102	MAL10792Y5241		

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Description Step 7 — Munal Rao (FZ04849), 14.Jan.2025

REPLACEMENT OF MLG AFT TRUNNION BEARING ASSY ALONG WITH RH MLG

1.CARRY OUT REPLACEMENT OF MLG AFT TRUNNION BEARING ASSY(REFERENCE P/N:161A1500-X). REFER AMM 57-16-02.

RECORD THE FOLLOWING:

OFF : P/N: _____ S/N:_____

ON: P/N:_____ S/N:_____

2.REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY.

Action Step 7-1

1.CARRIED OUT REPLACEMENT OF MLG AFT TRUNNION BEARING
ASSY(REFERENCE P/N:161A1500-X). REFER AMM 57-16-02.
RECORD THE FOLLOWING: AMM REV 86
DATED 15 FEB 2025

OFF :

P/N: 161A500-6 S/N:DLH89186

ON:

P/N:161A1500-6 S/N:MAL10792Y5241

2.REMOVED UNIT TO BE PROVIDED WITH SERVICEABLE TAG AND
CREDIT TO STORES FOR ONWARD INSTALLATION ON A6- FEY.

flydubai	flydubai Dubai	W/O **7911047** Working Copy	Barcode WO7911047	Registration **A6FEW** 73NG 738

Component Changes

PN Off	SN Off	Label	Position	PN On	SN On	Description	Certificate
161A500-6	DLH89186			161A500-6	MAL10792Y5241		

Transfers

Station	Date	TAH	TAC	Days / Due Date	Hours / Due at TAH	Cycles / Due at TAC	Approval No.	Release Sign	Reason

Workorder Transfer

Flight Hours	Flight Cycles	Days	Work deferred in accordance with	Limitation / Downgrades	Date
Due at TAH	Duet at TAC **31307**	Due Date **05.Jan.2029**			Stamp / Sign

Released To Service

Certifies that the work specified except as otherwise specified was carried out in accordance with CAR 145 and in respect to that work the aircraft /aircraft component is considered ready for release to service

Date: 23.3.24

Stamp / Sign: 23.3.24 GAT 492

	P.NO	DESCRIPTION	LOT
1)	D5026NS	→ CORROSION PREVENTION	→ LHG-166322-1122
2)	MS28778-4	→ O-RING	→ LHG-195875-0723
3)	PS870B2	→ SEALANT	→ LHG-341281-0325
4)	CORBAN 27L	→ CORROSION INHIBITANT	→ LHG-329790-0225
5)	MSBT-8H	→ NEVER SEEZ	→ LHG-167751-1122
6)	GREASE 33	→ AIRFRAME GREASE	→ LHG-251169-0624
7)	ROYCO LGF	→ ROYCO LGF	→ LHG-248723 → 0524
8)	SHC 100	→ MOBIL GREASE	→ LHG-178573-0223

Source Task/Discrep	Part #	Required Qt	Qty. Availal	UOM	Part Description
32-040-01-01	5649-0615-369	1.00	0.00	EA	SEAL-TURN
32-040-01-01	AEROSHELL GREASE	5.00	36.25	KG	SYNTHETIC UNIVEF
32-040-01-01	BACN10JC4CM	4.00	24.00	EA	NUT
32-040-01-01	BACP18BC02A06P	12.00	1002.00	EA	PIN
32-040-01-01	BACP18BC02A10P	6.00	1284.00	EA	PIN
32-040-01-01	BACP18BC03A06P	1.00	610.00	EA	PIN- SPLIT
32-040-01-01	BACP18BC03A08P	1.00	262.00	EA	PIN
32-040-01-01	BACP18BC03A10P	3.00	208.00	EA	PIN
32-040-01-01	BACP18BC04A10P	1.00	522.00	EA	PIN
32-040-01-01	BACP18BC04A14P	1.00	173.00	EA	PIN
32-040-01-01	BMS10-11TY1	1.00	0.00	KIT	PRIMER
32-040-01-01	CORBAN27L	1.00	0.00	CAN	CORROSION INHIBI
32-040-01-01	D5026NS	1.00	0.00	EA	CORROSION PREVE
32-040-01-01	LD4	10.00	90.50	QT	HYDRAULIC FLUID
32-040-01-01	MS20995NC32	5.00	0.00	LB	LOCKWIRE
32-040-01-01	MS28778-4	1.00	20.00	EA	O RING
32-040-01-01	NAS1612-4	2.00	283.00	EA	PACKING
32-040-01-01	NASM20995N32	1.00	0.00	EA	LOCKING WIRE
32-040-01-01	PS870B2	1.00	122.00	KIT	Sealant
32-040-01-01	ROYCO LGF	2.50	0.00	GAL	ROYCO LGF
32-040-01-01	SHC100	1.00	0.00	KG	AVIATION SYNTHET
32-040-01-01	161A1100-53	0.00	0.00	EA	MAIN LANDING GE
32-040-01-01	161A1100-67	1.00	0.00	EA	LH MLG
32-040-01-01	BACP18BC03A06P	1.00	610.00	EA	PIN- SPLIT
32-040-01-01	BACP18BC03A08P	1.00	262.00	EA	PIN

27.8.25

GAT 492

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
D5026NS		CORROSION PREVENTION	Consumable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-166322-1122 / 21337A/22180	4.00 EA	4P0018961122	GR22/004696/1122	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
DIVYANSHI AVIATION SERVICES PVT LTD	46586361	07/07/2022	07/07/2025

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 09/0404 03X

STORAGE REMARKS

5°C TO 25°C

INSPECTED BY		INSPECTED DATE
MR1144 /Manmohan Dash		22/11/2022
SIGNATURE		**STAMP**
		GAT 351

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

ZIP-CHEM PRODUCTS
A DIVISION OF ANDPAK, INC
400 JARVIS DRIVE
MORGAN HILL, CA. 95037
(408) 782-2335

MATERIAL CERTIFICATION

We hereby certify that the goods supplied on this purchase order were produced in compliance with the requirements of the Fair Labor Standards Act, as ammended, and of regulations and orders of the United States Department of Labor issued thereunder. Materials furnished have been manufactured and or packaged in accordance with all applicable instructions and specifications.

The user shall determine the suitability of this product for its intended use and user assumes all risk and liability in connection therewith. Certification, test reports or any applicable documents pertaining to this purchase order are kept on file and available for review at Zip-Chem Products.

Date:	JULY 7, 2022	Batch#:	21337A/22180
To:	BOEING DISTRIBUTION INC.	Date of manufacture:	06/2022
Cust. P.O.#:	46586361	Date of shipment:	07/07/2022
Product:	D-5026NS W/CO2 (DPM 5707-1)	Date of expiration:	07/07/2025
Net contents:	12 Ounces	Quantity shipped:	92 CASES

MIL-PRF-81309H, TYPE II & III, CLASS 2, GRADE B: CO2
TEST DATA REQUIREMENTS
JOB# 24079J

TEST PROPERTY	REQUIREMENT	TEST METHOD	RESULTS
Acid salt fog protection	No visible corrosion.	4.6.1	Pass
Application	Sprayable after 20 hours at 40° F.	4.6.2	Pass
Fill weight	11.0 Ounces minimum	4.6.7	Pass
Film thickness	0.2 mil maximum	4.6.9	0.2
Flash point	60° C (140° F) Minimum.	4.6.21	67°C
Friction coefficient	0.20 maximum	4.6.10	0.17
Leakage / Distortion of cans	None	4.6.12	None
Net content of container	As specified in contract or PO.	4.6.19	12 Oz.
Non-volatile content	Report	4.6.14	42.4
Water displacement	No visible corrosion.	4.6.18	Pass

Note: D-5026NS is on the QPL for MIL-PRF-81309 which meets the QPL requirement for SS8536 Type I.

Shelf life: Warranted 3 years from the date of shipment, when properly stored at 40° - 100° F (4.4° - 37.8° C).
DAC shelf life is unlimited when properly stored at ambient temperature.

Matt Bradley

ZIP-CHEM PRODUCTS
MATT BRADLEY,
QUALITY CONTROL

GMR Aero Technic

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
MS28778-4		O RING	Expendable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-195875-0723 / 0021528875	60.00 EA	4P0034080623	GR23/002079/0723	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
BOEING DISTRIBUTION SERVICES INC.	N/A	03/06/2023	01/07/2038	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 01/030605B

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1144 /Manmohan Dash		11/07/2023
SIGNATURE		**STAMP**
		GAT 351

Freudenberg-NOK
Sealing Technologies
Phone(714) 8340602
Fax (714) 8340590

2041 East Wilshire Ave.
Santa Ana, CA 92705 USA

International Seal Company, Inc. Certification

-----Certification of Conformance------

PROPONENT MAIN WHSE
3120 E. ENTERPRISE STREET
BREA, US, 92821

Delivery Note # : 0078462755
Material : 62AS568-904 N459
Specification : AMS-P-5510B

Customer PO : 677288-00 Line : 10

Customer Part : MS28778-4

Customer Info : REV C

ISC PartNumber	Quantity	Batch No	Cure Date	Cure Quarter
0072207068	4130	0021528875	04/27/2023	2Q23

Test	Description	Test Results	Min	Max
Specific Gravity	ASTM D297	1.34	1.25	1.45
Tensile Strength, p.s.i.	ASTM D1414	1659	1450	N/A
Modulus @ 50% Elongation, p.s.i.	ASTM D412	1379	500	N/A
Ultimate Elongation, %	ASTM D1414	95	80	N/A
Hardness, Durometer A	ASTM D2240 and ASTM D1414	88	85	95

This is to certify that the material used to fabricate the above product meets the specification listed. This report shall not be reproduced, except in full, without written approval of ISC-FNGP laboratory. Furnished material is Mercury free. Strorage conditions, age control and maximum storage life (shelf and expiration date) per AS5316.*** Country of Origin is USA***

Signed :

Name: Gyorgy Magyar
Title: Quality Assurance Representative
Date : Jun 3, 2023

FREUDENBERG-NOK
INNOVATING TOGETHER

ICN: 848934, Doc Date: 6/7/2023 CERTIFIED TRUE COPY

GAR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
PS870B2		Sealant	Consumable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-341281-0325 / AC6477DB	300.00 KIT	4P0032090225	GR24/011151/0325	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
PPG Industries	204674905	05/03/2025	31/12/2025

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 09/01 FLOOR.

STORAGE REMARKS

05°C TO 27°C

INSPECTED BY		INSPECTED DATE
MR1144 /Manmohan Dash		15/03/2025
SIGNATURE		STAMP
		GAT 351

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

Certificate of Conformance

PACK LIST #: 2046749

PACK TICKET

05/03/25 12:49:32

PRC-DeSoto International, Inc.
A PPG Industries Company

PPG INDUSTRIES MIDDLE EAST FZE
PLOT NUMBER MO 0738
P.O.BOX 17725 , Jebel Ali Free Zone
DUBAI
Phone: +971 4 883 9666
Fax: +971 4 883 9665
UNITED ARAB EMIRATES

Ship From: 3817EU
Ship Via: AIR
FOB Point: Exworks-Jafz-Dxb
Mode Of Transp:
Carrier: 2046749
Vehicle ID:
Total Pallets: 0

Sold-To:

GMR AIRCARGO AND AEROSPACE
ENGINEERING LTD
PLOT#1,GMR HYDERABAD AVIATION SEZ
RAJIV GANDHI INTERNATIONAL AIRPORT
SHAMSHABAD,HYDERABAD 500108
Phone: +914067251000
Fax: +914067251010
INDIA

Ultimate Consignee AS100340

GMR AIR CARGO AND AEROSPACE
ENGINEERING LTD PLOT#1 GMR HYDERABAD
AVIATION SEZ AREA, RAJIV GANDHI
INTERNATIONAL AIRPORT, SHAMSHABAD
HYDERABAD
INDIA

Fwd Agent FF001099

FedEx International Priority# 2863935

UNITED ARAB EMIRATES.

Sales Order	Order Date	Cust. No.	Terms	Cust. P.O.	CSR	Transaction Type	Shipment Terms	DS-PO#
DUB24014	24/02/25	AS000340	15FM-30D	4P0032090225	MN			

SO Ln#	Qty Ordered	UM	Description	Req Date	Perform Date	Qty Shipped
1	20	EA	0870C012CA654SK PRC STANDARD PS 870 C 12 654SK SPEC: PRC STANDARD REV NOT CONTROLLED Shelf Life: 9 MONTHS FROM DATE OF PACKAGING WHEN STORED BELOW 80°F % Shelf Life Remaining : 80 Primary Specification : SP: PRC STD POLYSULFIDE Date of Manufacture : 05/03/25 Lot No: AC6479DB *Location: A11 61402330 Expiration Date: 12/25	06/03/25	24/02/25	20
2	300	EA	1828B1/2CA654SK PRC STANDARD PR 1828 B 1/2 654 SK SPEC: PRC STANDARD REV NOT CONTROLLED Shelf Life: 9 MONTHS FROM DATE OF PACKAGING WHEN STORED BELOW 80°F % Shelf Life Remaining : 80 Primary Specification : SP: PRC STD P3 Date of Manufacture : 05/03/25 Lot No: AC6486DB *Location: A11 60059490 Expiration Date: 12/25	06/03/25	24/02/25	300
3	300	EA	0870B002CA654SK PPG STANDARD PS 870 B 2 654 SK SPEC: PRC STANDARD REV NOT CONTROLLED Shelf Life: 9 MONTHS FROM DATE OF PACKAGING WHEN STORED BELOW 80°F	06/03/25	24/02/25	300

GAR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
CORBAN 27L		CORROSION INHIBITANT	Consumable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-329790-0225 / 24031A/24051	5.00 CAN	4P0028580125	GR24/009377/0225	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
TENTACLE AEROLOGISTIX PRIVATED LIMITED	ZIPCHEMCOC/258510	28/05/2024	28/05/2027

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 09/040502X

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

STORE B/W 4° - 37°C

INSPECTED BY		INSPECTED DATE
MR1591 /Sunil kumar Puli		04/02/2025

SIGNATURE		STAMP
		GAT 456

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

ZIP-CHEM PRODUCTS
A DIVISION OF ANDPAK, INC
400 JARVIS DRIVE
MORGAN HILL, CA. 95037
(408) 782-2335

Certification of Conformance

We hereby certify that the goods supplied on this purchase order were produced in compliance with the requirements of the Fair Labor Standards Act, as ammended, and of regulations and orders of the United States Department of Labor issued thereunder. Materials furnished have been manufactured and/or packaged in accordance with all applicable instructions and specifications.
The user shall determine the suitability of this product for its intended use and user assumes all risk and liability in connection therewith. Certification, test reports or any applicable documents pertaining to this purchase order are kept on file and available for review at Zip-Chem Products.

Date shipped:	MAY 28, 2024	Batch#:	24031A/24051
To:	BOEING DISTRIBUTION INC.	Date of manufacture:	01/2024
Cust. P.O.#:	46881979	Job Number:	258510
Product:	Cor-Ban 27L CIC (009404)	Date of expiration:	05/28/2027
Net contents:	PINT	Quantity shipped:	25 CASES

ACCEPTANCE TEST DATA

TEST PROPERTY	REQUIREMENT	TEST METHOD	RESULTS
Consistency, 77° F (25 C), ± 5° F and 50± 10% relative humidity. Consistency, 130° F (54 C)	Smooth in texture, free of entrapped air when examined visually. No flow at 130 F (54 C).	Visual inspection of texture and high temperature flow test	Pass
Appearance	Shall spread smoothly to result in an even film and have a uniform composition, with the same physical properties throughout. No lumps or separated material shall be present. Color shall be Amber / Tan.	4.8.7 (GMS 4109)	Pass
Odor	No rancid, perfume or alcohol odor.	ASTM D 4339	Pass
Application Properties	A minimum of 15 grams per minute at 40° F (4 C).	SLP/CORBAN27L/APP	149
Nonvolatile Content	92.77% ± 5% (88.13 – 97.41) (per BMS 3-38)	ASTM D 1644 Test Method A	91.76%
Preload Test	The average readings must be between 4,940 and 5,460 pounds. All readings must be between 4,800 and 5,800 pounds. (Avg)	SLP/CORBAN27L/LOAD	5210
Nonvolatile Content	90% minimum (per MEP 09-075)	SLP-CORBAN27L-NVE	92

This product is qualified to BMS 3-38B, HMS 20-1267/2372, MEP 09-075F, LGQP 5010E Ty V, CSNP038H Ty X, CML 05-133, GMS 4109, TY. NC1, Rev. C and is referenced under DPM 6498. The specific batch stated above, fully meets the requirements of these specifications.
Shelf life: 3 years from the DOS unless otherwise indicated above, when properly stored.
Storage: Material should be stored at 40 to 100 degrees Fahrenheit in the original unopened containers.

QA
1
MEPIX

ZIP-CHEM PRODUCTS
MIKE SMITH,
QUALITY MANAGER

CERTIFIED TRUE COPY

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
NEVER-SEEZ NSBT-8N		NICKEL NEVER-SEEZ	Consumable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-167751-1122 / 2734576	5.00 CAN	4P0017541022	GR22/004881/1122	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING DISTRIBUTION PTE LTD (FORMALLY AVIALL PTE LTD)	118768	04/10/2022	08/05/2027

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 09/040406X

STORAGE REMARKS

5°C TO 25°C

INSPECTED BY		INSPECTED DATE
MR1144 /Manmohan Dash		30/11/2022

SIGNATURE		STAMP
		GAT 351

GMR Aero Technic

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

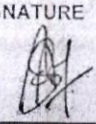
PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
AEROSHELL GREASE 33		SYNTHETIC UNIVERSAL AIRFRAME GREASE (CAN=3	Consumable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-251169-0624 / 12119150	90.00 KG	4P0003930524	GR24/001883/0624	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
TEAM AVIATION	SHELL COA	27/04/2023	04/04/2029	

WAREHOUSE / ZONE / BIN / RACK DETAILS

~~LM-HYD-SER~~

STORAGE REMARKS

0°C TO 50°C

INSPECTED BY		INSPECTED DATE
MR1427 /Sharath Kumar Ayitha		03/06/2024
SIGNATURE		**STAMP**
		GAT 458

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

Z042-CH4500 Bern LubProd
Steigerhubelstrasse 8
Bern
3008
Switzerland

Certificate of Analysis

Material	550043673	Date Manufactured	05 Apr 2023
Material Description	AeroShell Grease 33 4*3kg	Date Tested	17 Apr 2023
Batch Number	12119150		

Test Description	Result	Unit	Method
Aspect/Consistency/Texture	Smooth	-	VISUAL
Bluegreen Colour	Bluegreen equal to standard	-	VISUAL
CU Corrosion 24h/100°C / 1b	1b	-	ASTM D4048
Dirt Count 25-74 µm (Part/ml)	0	******	FED-STD-791 3005
Dirt Count 75-124 µm (Part/ml)	0	******	FED-STD-791 3005
Dropping point	229,0	CEL	IP 396
Evaporation Loss 22h 100°C	0,98	%(m)	ASTM D2595
Four Ball Weld Load	500,0	kg	ASTM D2596
Low Temp Torque -73°C - Run	0,055	N/m	ASTM D1478
Low Temp Torque -73°C - Start	0,515	N/m	ASTM D1478
Load Wear Index	78,2	kg	ASTM D2596
Odour Aeroshell Grease	Correspond to Reference	-	NONE
Oil Separation 30h 100°C	1,9	%(m)	ASTM D6184
Oxidation Stability 100h 99°C	16,0	kPa	ASTM D942
Penetration Unworked 25°C	281	dmm	ASTM D217
PEN_WORKED_25C	290	dmm	ASTM D217
Pen Worked 100000 25°C	301	dmm	ASTM D217
Pen Worked 100000 25°C FTM	303	dmm	FED-STD-791 313
Work Stability 10% H2O (DEF) 05-50/P63	25	dmm	DEF. STAN.
Rust Test	Pass	-	ASTM D1743
Water Washout 1h 38°C	3,3	%(m)	ASTM D1264

Date	27 Apr 2023
Certified by	Andreas Sigrist
Preferred contact	Foranyqueries,pleasecontactShellCustomerServiceCenterinyourregion.

Shell Lubricants Switzerland maintains Control Systems to ensure product supplied meets required specifications. Certified to ISO 9001 / 14001 / 45001. Approved to AMS 3052, AIMS 09-06-002, IPS 09-06-002-03, MIL-PRF-23827C Type 1 and BMS 3-33C Type 1. Document is valid without a signature.
Shelf life of AeroShell Grease is 6 years from date of manufacture. The validity of shelf life is based on the container remaining unopened, undamaged and stored under shelter at constant ambient temperature condition.

For TEAM AVIATION INDIA PVT. LTD.

RAJEEV RANJAN SHARMA
MANAGER QUALITY CONTROL

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
ROYCO LGF		ROYCO LGF	Consumable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-248723-0524 / 0000120684	25.00 GAL	4P0003910524	GR24/001377/0524	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
TENTACLE AEROLOGISTIX PRIVATED LIMITED	5000532346/000010	02/02/2024	29/11/2027

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/s /09 /06 FLOOR 06 22 01 X

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1427 /Sharath Kumar Ayitha		18/05/2024
SIGNATURE		**STAMP**
		GAT 458

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

Certificate of Analysis

LANXESS
Energizing Chemistry

Boeing Distribution Inc.
Hermann Warehouse Corporation
100 South Royal Lane
Coppell TX 75019-3851
Texas

E-mail: lanxess@hermanntds.com

Company
LANXESS Corporation
111 RIDC Park West Drive
15275-1112 PITTSBURGH
Pennsylvania
USA

Date: 01/31/2024

Material No.	Material Description
58323794	ROYCO LANDING GEAR FLUID YELLOW 5 GALLON STEEL PAIL

Customer Order Data

Order No.	Your Order No.	Ship-to Party
4000302298/000010	46836768	4000056682 Boeing Distribution Inc.

Delivery Data

Delivery No.	Delivered Quantity	Planned Delivery Date
5000532346/000010	160.000 PL	02/02/2024

Batch	Delivered Quantity	Date Of Manufacture	Best Before
0000120684	160.000 PL	11/30/2023	11/29/2027

Certificate of Analysis

LANXESS
Energizing Chemistry

Batch	Material Description	Delivery No.	Planned Delivery Date
0000120684	ROYCO LANDING GEAR FLUID YELLOW 5 GALLON STEEL PAIL	5000532346/000010	02/02/2024

Inspection Method / Characteristic	Result	Specification	Unit
1) ASTM D445 Viscosity (40°C)	13.97	>= 13.20	cSt
2) ASTM D4052 API Gravity	30.31		DegAPI
3) ASTM D4052 Pounds per Gallon	7.284		lb/gal
4) ASTM D974 Total Acid Number	2.44	1.50 - 5.00	mgKOHg
5) Materials	Conforms		
6) ICP-AES Zn Concentration	1538	1400 - 2000	ppm
7) Lubrizol 1395 Used in Batch	1.5	1.4 - 1.6	%
8) EMERST Used in Batch	1.0	0.9 - 1.1	%
9) ASTM D1500 Color, ASTM: 1.0 to 2.0	L2.0		

This Batch has been Tested, Inspected and is in Compliance with the Specification Requirements of AIRCRAFT LANDING GEAR FLUID, BMS3-32C TYPE 2.

CERTIFICATE OF COMPLIANCE:

Data appearing on this quality report are derived from process data and the testing of samples taken during the manufacturing process. Applicable International standard or internal test methods are used to control the process and characterize the product.

This report has been processed electronically and is therefore not signed.

CERTIFIED TRUE COPY

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ..... (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
SHC 100		MOBIL GREASE SHC 100 (2KG/CAN)	Consumable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-178573-0223 / 70422587T	2.00 KG	4P0025190223	GR22/006342/0223	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
DIVYANSHI AVIATION SERVICES PVT LTD	10122G26A	26/07/2022	26/07/2028	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S

STORAGE REMARKS

5°C TO 27°C

INSPECTED BY		INSPECTED DATE
MR1144 /Manmohan Dash		14/02/2023
SIGNATURE		**STAMP**
		GAT

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

Certificate of Analysis

ExxonMobil

Beaumont Lube Blending Plant
2805 Sycamore St. Beaumont, Texas 77701
Date: 08/10/2022

Product: MOBIL AVIATION GREASE,SHC 100

Batch Number	70422587T	Quantity / Package	TIN-START
Order Key		Manufacture Date	07/26/2022
Export# / P.O.#		Destination	
Yardmark	10122G26A	Reference #	
CMCS Code/ Prod#	201550402010	T/C or T/T	

Test Description	Method	Test Result
Odor	OLFACTORY	PASS
Texture	VISUAL	SMOOTH
Color	VISUAL	RED
Dropping Point, C	ASTM D2265	295
Four-Ball Wear Test, Scar Diameter, mm	ASTM D2266	0.6
Penetration, 60X, 0.1 mm	ASTM D217	274
Dirt, # particles 125u or larger	FTM 3005	0
Low Temperature Torque, Starting @ -54 C, Nm	ASTM D1478	0.6
Low Temp Torque, Running@-54 C, 60 min, Nm	ASTM D1478	0.1
Pen Worked X 100,000, 1/16" holes, 0.1 mm	FTM 313	301
Dirt, # Particles 25 - 124u	FTM 3005	0
Copper Strip Corrosion, 24h at 100C	ASTM D4048	1A
Oil Separation, 30 h @ 177 C, mass%	ASTM D6184	3.7

This material meets the ExxonMobil Sales specification established for this product and has been produced in a facility complying with the requirements of the ISO 9001 certified Global Product Integrity Management System (GPIMS). Test results on this certificate represent the most recent inspections done on this product for the stated characteristics and may be based on tank certification, manufacturing data, periodic testing and / or most recent product restock.

Direct Inquiries to:
Lorynda Clopton, Laboratory Supervisor
Phone: 1-409-240-4388
Fax: 1-409-240-8479

This document is electronically generated and does not require a physical signature to be valid.

ABSC, DUNLOP, GOODRICH, HONEYWELL, SAFRAN LANDING SYSTEMS, PARKER/CLEVELAND Wheel Bearing Approved.

Shelf life is 6 years from manufacturing date providing the container has remained unopened, undamaged, and was stored indoors.

Tests conducted according to ASTM Standard Test Methods are routinely verified to be in compliance with the latest published versions. Minor changes may be made where they have no material impact on test schedules and are necessitated by reasons such as safety, environmental standards, and method effectiveness.

BMT4464620_CofA (BMT4462306_CofA obsoleted)

* Duplicate Tag *

GMR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC03A08P		PIN	Expendable	Serviceable

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-006731-0619 / CSMS102808A	700.00 EA		GR08/000403/0518	OWNED

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING	N/A	26/04/2018	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S /02 /040308B

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1591/ Sunil Kumar Puli		15/11/2024
SIGNATURE		**STAMP**
		GAT 456

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

P.O. Box 025263, Miami, FL 33102-5263 • Tel: 305.925.2600 • Fax: 305.507.7191
Plant Location: 10000 N.W. 15th Terrace, Miami, FL 33172 • SITA: MIAMMCR
www.KLXaerospace.com

Shipped From: 10000 N W 15TH TERRACE, MIAMI, FL 33172

Material Certification

The items set forth on the purchase order referred to below have been visually inspected and the dimensions thereof have been measured by us, and based on the aforesaid, as well as the representation made to us by the manufacturers of the items subject of such purchase order, we hereby certify that such items are in conformity with all current governmental and manufacturer's requirements, specifications, drawings, and conform to the purchase order requirements. Said items are in new condition and have not been obtained from any U.S. Government or Military source and are traceable to KLX Aerospace Solutions.

FIRM: GMR AERO TECHNIC LIMITED

PURCHASE ORDER#: 8P0001420418

LINE#	QUANTITY	U/M	PART NUMBER	COST REF#	LOT NUMBER	MANUFACTURER	CCODE REV	EFF Date	EXP DATE
5	700	EA	BACP10PC03A06P		CSN0102008A	WESTERN WIRE/FENTON	55039 D		

04/26/15

Jason Lewis
Senior Director, Global Quality

Inv # 04/26/15

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [ ] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC03A10P		PIN	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-249037-0524 / 03817337	300.00 EA	4P0003790524	GR24/001463/0524	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING	03817337	13/05/2024	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 02 / 030809B

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1538 /Ramesh Manchu		20/05/2024
SIGNATURE		STAMP

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

1. Approving Civil Aviation Authority/Country: FAA/UNITED STATES	2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	3. Form Tracking Number: 03817337
4. Organization Name and Address: THE BOEING COMPANY,737 LOGAN AVE N, RENTON,WA 98057-0000 - PC #700		5. Work Order/Contract/Invoice Number: 3609669

6. Item:	7. Description:		8. Part Number:	9. Quantity	10. Serial Number:	11. Status/Work:
000001	PIN, COTTER (SPLIT)	8008388199	BACP18BC03A10P	300 EA	N/A	NEW
000002	PIN, COTTER (SPLIT)	8008388200	BACP18BC02A06P	1000 EA	N/A	NEW

12. Remarks: 5009 1GM

Page 1 of 1

SHIPMENT:3609669

SUPPLEMENTAL SHIPMENT - AIRWORTHINESS DOCUMENT

13a. Certifies the item identified above were manufactured in conformity to:

☒ Approved design data and are in a condition for safe operation.
☐ Non-approved design data specified in Block 12.

14a. ☐ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No: PC 700	14b. Authorized Signature:	14c. Approval/Certificate No:
13d. Name (Typed or Printed): LARRY W. GAC	13e. Date (dd/mmm/yyyy): 13/MAY/2024	14d. Name (Typed or Printed):	14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

X37968 NEW (30 JAN 2017)

GMR Aero Technic

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC04A14P		PIN	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-081665-0521 / CDMS030320B	200.00 EA	3P0000890421	GR21/001170/0521	Owned

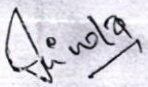
SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING DISTRIBUTION SERVICES INC.	MFG COC	20/04/2020	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S /02 /030809B

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR0742 /Jayanta N		18/05/2021
SIGNATURE		STAMP
		GAT 206

Form No GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

Boeing Distribution Services Inc.
P.O. Box 025263, Miami, FL 33102-5263 • Tel: 305.925.2600 • Fax: 305.507.7191
Plant Location: 3760 W. 108th Street, Miami, FL 33018 • SITA: MIAMMCR
www.BoeingDistribution.com

Shipped From: 3760 W. 108TH ST DOOR B, MIAMI, FL 33018

Material Certification

The items set forth on the purchase order referred to below have been visually inspected and the dimensions thereof have been measured by us, and based on the aforesaid, as well as the representation made to us by the manufacturers of the items subject of such purchase order, we hereby certify that such items are in conformity with all current governmental and manufacturer's requirements, specifications, drawings, and conform to the purchase order requirements. Said items are in new condition and have not been obtained from any U.S. Government or Military source and are traceable to Boeing Distribution Services.

FIRM: GMR AIR CARGO AND AEROSPACE ENG.LTD

PURCHASE ORDER#: 3P0000890421

LN#	QUANTITY	U/M	PART-NUMBER	CUST REF#	LOT-NUMBER	MANUFACTURER	CCODE	Eff Date	EXP DATE
1	200	EA	BACP18BC04A14P		CDMS030320B	WESTERN WIRE/FENTON	65029		

BDSI CTRL/LOT # :2020DM869A

04/12/21

Jason Lewis
Senior Director, Global Quality

Inv # 04/12/21

For

EMASGAR Aero Technic

STORE ACCEPTANCE TAG

19/0603050A
02/040805B

Description: Cotter pin GRN: 445-1111

Part no. BACP18BC04A10P Cond: NEW / REPAIRED 19/0603050A

Serial No.: — Location: 02020305C

Batch/ Lot No. CAM5022111A C of C CASE No.: 02/0204080C

Release Note No. CofC 784096 Date: 24-10-2011

Supplier: Aviall Asia Pte Ltd

P.O. No.: PETT-000093-0911 Date: 28/9/2011

Shelf Life: — Quantity: 167 UOM: each

Signature of Inspector: [signature] Date: 1/12/2011

Regulations

- [ ] CAR 145
- [x] EASA Part 145
- [ ] FAR Part 145
- [ ] DCA, Malaysia

INSPECTION STAMP: MGAT 024

RECEIVED

Tag no.: MGAT/MT/005

AVIO-DIEPEN B.V.
P.O. BOX 566
2400 AN ALPHEN AAN DEN RIJN
NETHERLANDS
FAX : +31-(0)172-449789
SITA: HAGME7X

PACKING LIST Page 3 of

Number : 784096
Date : 2011-10-24
Contact: Fred van der Jagt
Phone : +31172449769

-Consignee- (1)
MAS GMR AEROTECHNIC LTD
PLOT NO:1 C/O GMR HYDERABAD AVI SEZ
SY 97P: 98/P 99/P 133/P 134/P
MAMIDIPALLI VILLAGE HYDERABAD500409
INDIA

-Sold to-
MAS GMR AEROTECHNIC LTD
PLOT NO 1 C/O GMT HYDERABAD AVI SE;
133/P 134/P MAMIDIPALLI VILLAGE
HYDERABAD 500409
INDIA

It.	Partnumber / Description	Serialno	Quantity	Unit	RcptNo Line	Locatn TchCnd	Certno Certificate
006	AS3208-06 PACKING YOUR ITEM: 111 CUREDATE: JUL 07 BATCHNUMBER: 723505 ECCN CODE: 9A991.D HARMONISED SYSTEM: 40169300		4.0	EACH	387466 4	E07D04A MAN	378153 CMA CAD
007	BACN10JC4CM NUT, SELF-LOCK YOUR ITEM: 115 BATCHNUMBER: N/A HARMONISED SYSTEM: 73181630 QUANTITY ADJUSTED TO MINIMUM		15.0	EACH	534594 8	B14C05J MAN	TX6612 CMA CAD
008	BACP18BC02C06P COTTER PUN YOUR ITEM: 118 BATCHNUMBER: BA040811A HARMONISED SYSTEM: 73182400		1000.0	EACH	534126 13	E16E04D MAN	369899-00 CMA CAD
009	BACP18BC04A10P PIN, COT YOUR ITEM: 122 BATCHNUMBER: CDMS022111A HARMONISED SYSTEM: 73182400 QUANTITY ADJUSTED TO MINIMUM		167.0	EACH	534126 15	B17H01B MAN	368470-00 CMA CAD
010	NAS1352-08-8P SCREW YOUR ITEM: 251 BATCHNUMBER: N/A HARMONISED SYSTEM: 73181590		50.0	EACH	530766 3	B04E03J MAN	229400 CMA CAD

GMR Aero Technic

STORE ACCEPTANCE TAG

Acceptance

Regulations	
☑ CAR 145	
☑ EASA Part 145	
☑ FAA Part 145	
☐ Others	
______ (Specify)	

Description:	PIN- SPLIT		
Part no :	BACP18BC03A06P	GRN :	GR08/001912/0918
Serial No :		Cond :	~~NEW/REPAIRED~~
Batch/ Lot No :	CDMS090616A-5 / LNG-149973-0918	Location :	~~[illegible]~~
Release Note No:	MAT CERT	C of C CASE No:	02/d/02050
Supplier:	KLX AEROSPACE SOLUTIONS	Date:	27-08-2018
P.O. No.:	8P0007530818		
Quantity:	500	Date:	22-08-2018
		UOM:	EA
Shelf Life :	Not Available	Expire On :	Not Available
Signature of Inspector:	Qi'nola	Date:	13/09/2018

INSPECTOR STAMP GAT 206

Tag No. GAT/MS/005,Issue Initial,Rev , Dated 18/08/2016

Western Wire Products Company

770 Sun Park Drive - Fenton, MO 63026
Phone: 800.325.3770 or 636.305.1100 - Fax: 636.305.1119
www.WesternWireProd.com - Sales@WesternWireProd.com

February 27, 2017

Made in USA

KLX INC
9835 N.W 14TH STREET
MIAMI, FL 33172

Re: Purchase Order # OO53016
Western Wire Order # 504842-00

Dear Madam or Sir:

This is to certify that the referenced shipment contained the items and quantities below.

Quantity	Part Description	Lot	Material
25,000	MS24665-300 AND BACP18BC03A06P 3/32 X 3/4 STAINLESS EXT. PRONG - CHISEL PT. COTTER PINS	CDMS090616A-5(20,000 PCS) CDMS090616A-4(5,000 PCS)	13

are made of the following material:

13 18/8 TYPE 302/304 CHROME-NICKEL, CORROSION RESISTANT STAINLESS STEEL, CONFORMING TO ASME B18.8.1 - 2014 IN ACCORDANCE WITH NATIONAL AEROSPACE STANDARD NASM24665 REV 2 AND BOEING SPECIFICATIONS REV. D AND PASSIVATED IN ACCORDANCE WITH AMS2700, METHOD 2. DFAR 252.225-7009

The material as shipped is free from mercury contamination.

L. Braddum , Chief Clerk

* Duplicate Tag *

GMR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC03A08P		PIN	Expendable	Serviceable

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-006731-0619 / CSMS102808A	700.00 EA		GR08/000403/0518	OWNED

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING	N/A	26/04/2018	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S /02 /040308B

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1591/ Sunil Kumar Puli		15/11/2024
SIGNATURE		**STAMP**
		

KLX

P.O. Box 025263, Miami, FL 33102-5263 • Tel: 305.925.2600 • Fax: 305.507.7191
Plant Location: 10000 N.W. 15th Terrace, Miami, FL 33172 • SITA: MIAMMCR
www.KLXaerospace.com

Shipped From: 10000 N W 15TH TERRACE, MIAMI, FL 33172

Material Certification

The items set forth on the purchase order referred to below have been visually inspected and the dimensions thereof have been measured by us, and based on the aforesaid, as well as the representation made to us by the manufacturers of the items subject of such purchase order, we hereby certify that such items are in conformity with all current governmental and manufacturer's requirements, specifications, drawings, and conform to the purchase order requirements. Said items are in new condition and have not been obtained from any U.S. Government or Military source and are traceable to KLX Aerospace Solutions.

FIRM: GMR AERO TECHNIC LIMITED

PURCHASE ORDER#: 8P0001420418

LINE#	QUANTITY	U/M	PART-NUMBER	CUST REF#	LOT-NUMBER	MANUFACTURER	CCODE	REV	Eff Date	MAF DATE
5	700	EA	BACP18BC03A09P		CSM8102908A	WESTERN WIRE/FENTON	65039	D		

04/26/18
Jason Lewis
Senior Director, Global Quality

Inv # 04/26/18

Form 67 Rev 1.

* Duplicate Tag *

GAR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC03A08P		PIN	Expendable	Serviceable

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-006731-0619 / CSMS102808A	700.00 EA		GR08/000403/0518	OWNED

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING	N/A	26/04/2018	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S /02 /040308B

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1591/ Sunil Kumar Puli		15/11/2024
SIGNATURE		**STAMP**
		GAT 456

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

KLX

P.O. Box 025263, Miami, FL 33102-5263 • Tel: 305.925.2600 • Fax: 305.507.7191
Plant Location: 10000 N.W. 15th Terrace, Miami, FL 33172 • SITA: MIAMMCR
www.KLXaerospace.com

Shipped From: 10000 N W 15TH TERRACE, MIAMI, FL 33172

Material Certification

The items set forth on the purchase order referred to below have been visually inspected and the dimensions thereof have been measured by us, and based on the aforesaid, as well as the representation made to us by the manufacturers of the items subject of such purchase order, we hereby certify that such items are in conformity with all current governmental and manufacturer's requirements, specifications, drawings, and conform to the purchase order requirements. Said items are in new condition and have not been obtained from any U.S. Government or Military source and are traceable to KLX Aerospace Solutions.

FIRM: GMR AERO TECHNIC LIMITED

PURCHASE ORDER#: 8P0001420418

LNE#	QUANTITY	U/M	PART-NUMBER	CUST REF#	LOT-NUMBER	MANUFACTURER	CCODE	REV	Eff Date	EXP DATE
5	700	EA	BACP18BC03A05P		CSMS102908A	WESTERN WIRE/FENTON	55019	D		

04/26/18

Jason Lewis
Senior Director, Global Quality

Inv # 04/26/18

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [ ] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC02A06P		PIN	Expendable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-249036-0524 / 03817337	1000.00 EA	4P0003790524	GR24/001463/0524	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
BOEING	03817337	13/05/2024	NOT APPLICABLE	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 02/030307A

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1538 /Ramesh Manchu		20/05/2024
SIGNATURE		**STAMP**
		GAT 408

Authority/Country:
FAA/UNITED STATES

AUTHORIZED RELEASE CERTIFICATE

FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG

3. Form Tracking Number:
03817337

4. Organization Name and Address:
THE BOEING COMPANY,737 LOGAN AVE N, RENTON,WA 98057-0000 PC #700

5. Work Order/Contract/Invoice Number:
3609669

6. Item:	7. Description:	8. Part Number:	9. Quantity	10. Serial Number:	11. Status/Work:
000001	PIN, COTTER (SPLIT) 8008388199	BACP18BC03A10P	300 EA	N/A	NEW
000002	PIN, COTTER (SPLIT) 8008388200	BACP18BC02A06P	1000 EA	N/A	NEW

12. Remarks: 5009 1GM
SUPPLEMENTAL SHIPMENT - AIRWORTHINESS DOCUMENT

SHIPMENT:3609669

13a. Certifies the item identified above were manufactured in conformity to:

☒ Approved design data and are in a condition for safe operation.
☐ Non-approved design data specified in Block 12.

14a. ☐ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No: PC 700	14b. Authorized Signature:	14c. Approval/Certificate No:
13d. Name (Typed or Printed): LARRY W. GAC	13e. Date (dd/mmm/yyyy): 13/MAY/2024	14d. Name (Typed or Printed):	14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the installer before the aircraft may be flown.

BOEING

See back: Terms & Conditions

X37968 NEW (30JAN2017)

GAR Aero Technic

Regulations

- [x] DGCA CAR 145
- [ ] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
BACP18BC02A06P		PIN	Expendable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-249036-0524 / 03817337	1000.00 EA	4P0003790524	GR24/001463/0524	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
BOEING	03817337	13/05/2024	NOT APPLICABLE	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 02/030307A

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1538 /Ramesh Manchu		20/05/2024
SIGNATURE		**STAMP**
		GAT 408

1. Authority/Country:
FAA/UNITED STATES

AUTHORIZED RELEASE CERTIFICATE

FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG

3. Form Tracking Number:
03817337

4. Organization Name and Address:
THE BOEING COMPANY,737 LOGAN AVE N, RENTON,WA 98057-0000 PC #700

5. Work Order/Contract/Invoice Number:
3609669

6. Item:	7. Description:	8. Part Number:	9. Quantity	10. Serial Number:	11. Status/Work:
000001	PIN, COTTER (SPLIT) 8008388199	BACP18BC03A10P	300 EA	N/A	NEW
000002	PIN, COTTER (SPLIT) 8008388200	BACP18BC02A06P	1000 EA	N/A	NEW

12. Remarks: 5003 1GM
SUPPLEMENTAL SHIPMENT - AIRWORTHINESS DOCUMENT

SHIPMENT:3609669

13a. Certifies the item identified above were manufactured in conformity to:

☒ Approved design data and are in a condition for safe operation.
☐ Non-approved design data specified in Block 12.

14a. ☐ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No: PC 700	14b. Authorized Signature:	14c. Approval/Certificate No:
13d. Name (Typed or Printed): LARRY W. GAC	13e. Date (dd/mmm/yyyy): 13/MAY/2024	14d. Name (Typed or Printed):	14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

BOEING

See back: Terms & Conditions

X37968 NEW (30JAN2017)

GAR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
MS24665-157		PIN COTTER	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-300563-1024 / DSS071922A/136939	700.00 EA	4P0018081024	GR24/006258/1024	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING DISTRIBUTION SERVICES INC.	00RS441	28/12/2022	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 09/03081C

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1427 /Sharath Kumar Ayitha		25/10/2024
SIGNATURE		STAMP
		GAT 458

Form No: GAT/MS/005,Issue 2,Rev.0,Dated 06-11-2020

Boeing Distribution Services Inc.
P.O. Box 025263, Miami, FL 33102-5263 • Tel: 305.925.2600 • Fax: 305.507.7191
Plant Location: 3760 W. 108th Street, Miami, FL 33018 • SITA: MIAMMCR
www.BoeingDistribution.com

Shipped From: 3760 W. 108TH ST DOOR B, MIAMI, FL 33018

Material Certification

The items set forth on the purchase order referred to below have been visually inspected and the dimensions thereof have been measured by us, and based on the aforesaid, as well as the representation made to us by the manufacturers of the items subject of such purchase order, we hereby certify that such items are in conformity with all current governmental and manufacturer's requirements, specifications, drawings, and conform to the purchase order requirements. Said items are in new condition and have not been obtained from any U.S. Government or Military source and are traceable to Boeing Distribution Services.

FIRM: GMR AIR CARGO AND AEROSPACE ENG.LTD

PURCHASE ORDER#: 4P0018081024

LNE#	QUANTITY	U/M	PART-NUMBER	CUST REF#	LOT-NUMBER	MANUFACTURER	CCODE	Eff Date	EXP DATE
3	700	EA	MS24665-157		DSS071922A/136939	WESTERN WIRE/FENTON	65029		

BDSI CTRL/LOT # :2023AMS0KW

10/17/24

Valentin Usov
Supply Chain Quality Leader

Inv #KYRHS3 10/17/24

GAR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
MS24665-157		PIN COTTER	Expendable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-300563-1024 / DSS071922A/136939	700.00 EA	4P0018081024	GR24/006258/1024	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
BOEING DISTRIBUTION SERVICES INC.	00RS441	28/12/2022	NOT APPLICABLE	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 02/03081C

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1427 /Sharath Kumar Ayitha		25/10/2024
SIGNATURE		**STAMP**
		GAT 458

Boeing Distribution Services Inc.
P.O. Box 025263, Miami, FL 33102-5263 • Tel: 305.925.2600 • Fax: 305.507.7191
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FIRM: GMR AIR CARGO AND AEROSPACE ENG.LTD

PURCHASE ORDER#: 4P0018081024

LNE#	QUANTITY	U/M	PART-NUMBER	CUST REF#	LOT-NUMBER	MANUFACTURER	CCODE	Eff Date	EXP DATE
3	700	EA	MS24665-157		DSS071922A/136939	WESTERN WIRE/FENTON	65029		

BDSI CTRL/LOT # :2023AMS0KW

10/17/24

Valentin Usov
Supply Chain Quality Leader

Inv #KYRHS3 10/17/24

GAR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
10P4-2NF/EC-117S		EPOXY PRIMER GREEN BAC452 2GL/KT	Consumable	NEW
LOT # / MFR. LOT #	**QTY / UOM**	**REF. DOC. #**	**RECEIPT NO. #**	**STOCK STATUS**
LNG-306110-1124 / 8474116021/164081	3.00 KIT	4P0020631124	GR24/006976/1124	Owned
SUPPLIER NAME	**CERTIFICATE NO.**	**CERTIFICATE DATE**	**EXPIRY/ CAL. DUE DATE**	
TENTACLE AEROLOGISTIX PRIVATED LIMITED	8474116021	16/07/2024	30/06/2026	

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 09/061203X

STORAGE REMARKS

4.4°C TO 37.8°C

INSPECTED BY		INSPECTED DATE
MR1144 /Manmohan Dash		20/11/2024
SIGNATURE		**STAMP**
2		GAT 351

AkzoNobel Aerospace Coatings, a division of International Paint LLC
1 East Water Street
Waukegan, IL 60085
USA
T +1-847-623-4200
F +1-847-625-3200
www.akzonobel.com/aerospace

AkzoNobel

Certificate of Conformance

We certify that the following batch meets the standard quality control requirements of:
AkzoNobel Aerospace Coatings

Product Code:	Description:	Batch #:	Date of Mfg.
	BAC 452		
10P4-2NF	Fluid Resistant Epoxy Primer Base, Component A, Green	8474116021	06/24

Shelf Life: Twenty-four months from date of manufacture.

Scott D. W. Kuebler

Date: 07/16/24 LD

Scott Kuebler
Quality Control Manager

CERTIFIED TRUE COPY

GMR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others _____ (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
MS20995C20		WIRE	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-324425-0125 / 43958	4.00 EA	4P0026110125	GR24/008749/0125	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
BOEING DISTRIBUTION SERVICES INC.	00Y0A07	19/06/2024	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S 02/03 10 08A

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1427 /Sharath Kumar Ayitha		16/01/2025
SIGNATURE		**STAMP**
		GAT 458

Boeing Distribution Services Inc.
P.O. Box 025263, Miami, FL 33102-5263 • Tel: 305.925.2600 • Fax: 305.507.7191
Plant Location: 3760 W. 108th Street, Miami, FL 33018 • SITA: MIAMMCR
www.BoeingDistribution.com

Shipped From: 3760 W. 108TH ST DOOR A, Miami,FL 33018

Material Certification

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FIRM: GMR AIR CARGO AND AEROSPACE ENG.LTD

PURCHASE ORDER#: 4P0026110125

LNE#	QUANTITY	U/M	PART-NUMBER	CUST REF#	LOT-NUMBER	MANUFACTURER	CCODE MFR DTE	EXP DATE
3	4	EA	MS20995C20		43958	AMERICAN WIRE	84127 06/07/24	

BDSI CTRL/LOT # :2024FM1AJL

01/07/25

Valentin Usov
Supply Chain Quality Leader

Inv #KZZ1L6 01/07/25

Form 217 Re

GAR AERO TECHNIC

Regulations

- [x] DGCA CAR 145
- [x] EASA Part 145
- [x] FAA Part 145
- [ ] Others ---------- (Specify)

STORE ACCEPTANCE TAG

PART	MFR. SERIAL #	PART DESCRIPTION	PART TYPE	PART CONDITION
MS20995NC20		WIRE	Expendable	NEW

LOT # / MFR. LOT #	QTY / UOM	REF. DOC. #	RECEIPT NO. #	STOCK STATUS
LNG-324022-0125 / 43791	2.00 ROLL	4P0021991124	GR24/008718/0125	Owned

SUPPLIER NAME	CERTIFICATE NO.	CERTIFICATE DATE	EXPIRY/ CAL. DUE DATE
DIVYANSHI AVIATION SERVICES PVT LTD	15920-MAL	02/02/2024	NOT APPLICABLE

WAREHOUSE / ZONE / BIN / RACK DETAILS

H/S

STORAGE REMARKS

FOLLOW GENERAL STORAGE REQUIREMENTS

INSPECTED BY		INSPECTED DATE
MR1427 /Sharath Kumar Ayitha		15/01/2025

SIGNATURE		STAMP
		GAT 458

MALIN CO.
Wire Specialists Since 1884.

5400 Smith Road · Brook Park, OH 44142 · Phone: 216-267-9080 · Fax: 216-267-9077 · Toll Free: 800-967-9697

CERTIFICATE OF COMPLIANCE

To: BOEING DISTRIBUTION, INC
PO#4679229000 1
PN#MS20995NC20MON-05-0201-1bkc
PACKING SLIP#15920-MAL
.0201 MONEL #24 1LB. RED CAN MS20995NC20

Lot: 43791
Mill Order: L0264030
Type: .0201 MONEL #24
ASTM B164

THIS IS TO CERTIFY THAT THE MATERIAL SHIPPED ON THE ABOVE PURCHASE ORDER COMPLIES WITH THE SPECIFICATIONS NOTED HEREON:

PHYSICALS:

AVERAGE SIZE	TENSILE	YIELD	ELONG % IN 10"	REDUCTION IN AREA	WRAP/BEND TEST
.02	77,762 (PSI)	35,169	34.41	79.25	PASS

CHEMISTRY: (43791)

C	Mn	Si	S	P	Cr	Ni	Mo	Cu
0.0600	1.0100	0.1800	0.0036	0.0010		64.500	0.0050	32.340

N	Co	Ti	Al	Fe	Bal
	0.0100	0.0970	0.2030	1.3900	

OTHER:

Country of Origin: FR (FRANCE)

HEAT NUMBER
77698

NOTES:

Printed: 02/02/2024

Contact: Tracy Byrne x204

Authorized Signature

J Hauck S Schall CSR

www.malinco.com

DA-True Certified Copy